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HIGH-TEMPERATURE, LOW-CYCLE FATIGUE
OF COPPER-BASE ALLOYS FOR ROCKET
NOZZLES; PART I - DATA SUMMARY FOR
MATERIALS TESTED IN PRIOR PROGRAMS.

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MAR-TEST INC.
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16. Abstract A more detailed analysis of the results obtained in 188 previously reported low-cycle fatigue tests of various candidate materials for regeneratively-cooled, reusable rocket nozzle liners is reported. Plots of load range versus cycles are reported for each test along with a stress-strain hysteresis loop near half-life. In addition, a summary table is provided to compare N_5 (cycles to a five percent load range drop) and N_f (cycles to complete specimen separation) values for each test.					
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I - SUMMARY

Five recent NASA Contractor Reports were written and published by Mar-Test Inc. to describe short-term tensile and low-cycle fatigue evaluations of eighteen (18) candidate materials (17 copper-base alloys along with pure silver) for regeneratively-cooled, re-usable rocket nozzle liners. Test temperatures ranged from room temperature to 593°C with all the elevated temperature tests being performed in high purity argon. All low-cycle fatigue tests were performed in axial strain control with strain ranges being selected to define the fatigue life to about 3000 cycles. A limited evaluation of strain-rate and hold-time effects was also included for selected alloys.

Following the formal publication of the results of these tests it was decided that a more detailed analysis should be made of each test to furnish a more comprehensive description of the low-cycle behavior of each material. This re-analysis focused on the 188 previously reported low-cycle fatigue tests and led to a plot of load range versus cycles for each test. It also yielded a typical hysteresis loop for each test chosen near the region of half-life. And, finally, this re-analysis led to a summary table to allow the values of N_5 (cycles to a five percent load range drop) and N_f (cycles to complete specimen separation) to be compared for each test.

In many of these tests a cyclic softening behavior was exhibited and it was not possible to identify a stabilized load range value for use in calculating N_5 . For these instances, the interpretation adopted was based on a five percent reduction from the load range value obtained at the point where a decided change in curvature was noted in the semi-logarithmic plot of load range versus cycles in the regime near fracture.

II - INTRODUCTION

In recent NASA programs, NAS3-16753 and NAS3-17777, a detailed evaluation was made of the short-term tensile and low-cycle fatigue behavior of numerous candidate materials for regeneratively-cooled, reusable rocket nozzle liners such as found in the engines of the Space Shuttle, Orbit-to-Orbit Shuttle, Space Tug, etc. The results obtained in these studies have been described in the following series of NASA Contractor Reports:

- NASA CR-121259 -- High Temperature, Low-cycle Fatigue of Copper-base Alloys in Argon; Part I - Preliminary Results for 12 Alloys at 1000°F (538°C).
- NASA CR-121260 -- High Temperature, Low-cycle Fatigue of Copper-base Alloys in Argon; Part II - Zirconium-copper at 482°C, 538°C and 593°C.
- NASA CR-121261 -- High Temperature, Low-cycle Fatigue of Copper-base Alloys in Argon; Part III - Zirconium-copper; Thermal-mechanical Strain Cycling, Hold-time and Notch Fatigue Results.
- NASA CR-134627 -- High Temperature, Low-cycle Fatigue of Advanced Copper-base Alloys for Rocket Nozzles; Part I - Narloy Z.
- NASA CR-134628 -- High Temperature, Low-cycle Fatigue of Advanced Copper-base Alloys for Rocket Nozzles; Part II - NASA 1.1, Glidcop, and Sputtered Copper Alloys.

Subsequent assessments by NASA personnel of the material property data presented in the above reports indicated that from a design point of view a more detailed evaluation of the stress-strain information recorded in the low-cycle fatigue tests was desirable. It was decided, therefore, that in Task I of this program a more comprehensive analysis would be made of each low-cycle fatigue test

performed during the last two NASA-sponsored programs performed by Mar-Test Inc. The strip chart records for each fatigue test were subjected to additional analysis so that a plot of load versus cycles could be prepared. In addition, it was decided that a typical hysteresis loop, selected near half-life, would be presented for each test. And finally, this Task I effort was to present a tabular summary for all the fatigue tests to compare the values of N_5 (cycles corresponding to a five percent load range drop) and N_f (cycles to complete specimen separation) for each test condition and material.

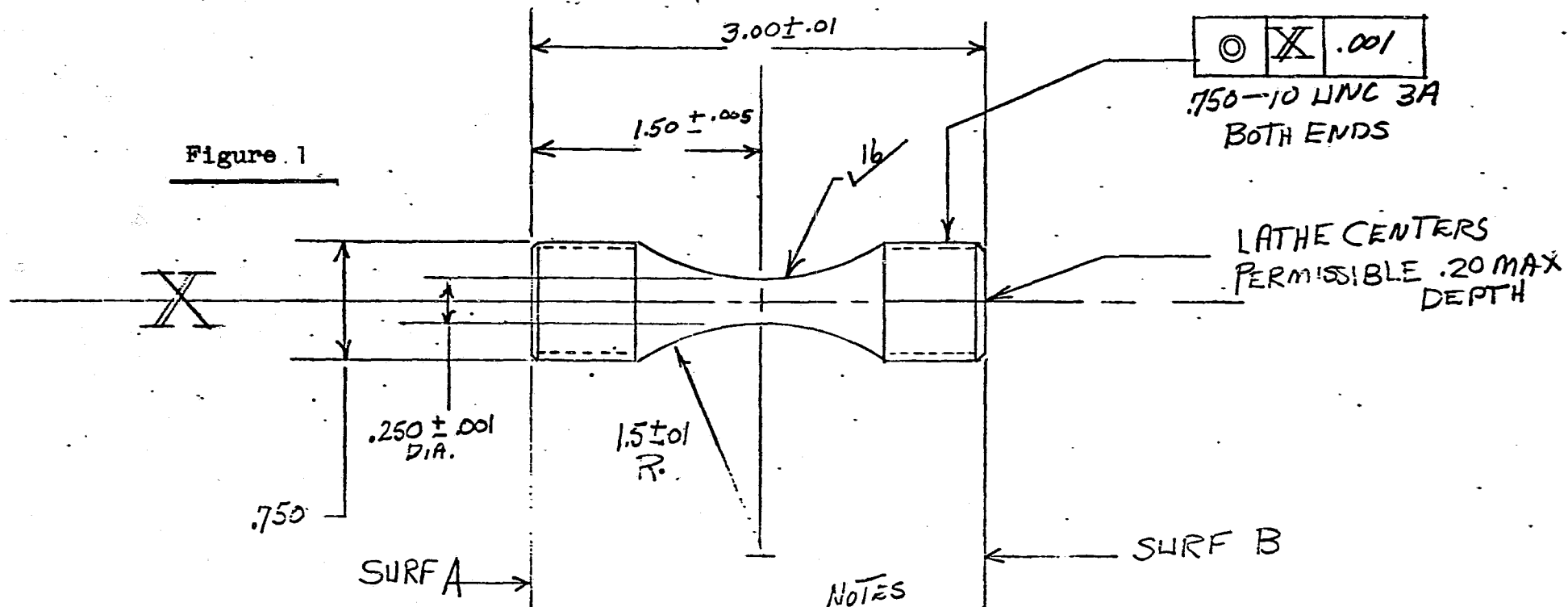
III - MATERIALS AND SPECIMENS

A total of nineteen (19) alloy designations was involved throughout the course of the material evaluations performed by Mar-Test Inc. on NASA Contracts NAS3-16753 and NAS3-17777. These designations together with a brief alloy definition are as follows:

R-0	Zirconium-copper, annealed
R-1	Zirconium-copper, $\frac{1}{4}$ hard
R-2	Zirconium-copper, $\frac{1}{2}$ hard
R-3	Tellurium-copper, $\frac{1}{2}$ hard
R-4	Chromium-copper, solution annealed and aged
R-5	OFHC copper, hard
R-6	OFHC copper, $\frac{1}{4}$ hard
R-7	OFHC copper, annealed
R-8	Silver, as-drawn
R-9	Zr-Cr-Mg copper, solution annealed, cold worked and aged
R-10	Electroformed copper, 30-35 ksi
R-13	Co-Be-Zr copper, solution annealed and aged
R-20	A second lot of zirconium-copper, $\frac{1}{2}$ hard
R-21	NASA 1-1A copper alloy, aged
R-22	NASA 1-1B copper alloy, as-received
R-23	Glidcop AL-10
R-24	Narloy Z, centrifugally cast, hot-rolled, solution annealed and aged
R-25	Sputtered zirconium-copper, annealed
R-26	Sputtered zirconium-copper, as-sputtered

All these materials were supplied by NASA-Lewis Research Center to Mar-Test Inc. and test specimens were then fabricated in accordance with the design shown in Figure 1. Special storage procedures were observed for the as-fabricated and post-test specimens and these are described in the contractor reports covering the test programs mentioned above.

Figure 1



5- SCREW THREADS TO BE
AS LISTED IN NBS
HAND BOOK H 28

NOTES

- 1- SURFACES A, & B TO BE PARALLEL WITHIN .001
- 2- SURFACES A, & B TO BE PERPENDICULAR TO
CENTER LINE OF SPECIMEN WITHIN .0005 TIR
- 3- CONTOURED PORTION OF SPECIMEN TO HAVE
A $\sqrt{16}$ FINISH OR BETTER. FINISHING SHOULD
BE IN THE AXIAL DIRECTION USING LOW
STRESS LAPPING OR POLISHING OPERATION
- 4- ALL DIA'S TO BE CONCENTRIC WITHIN .001 TIR

UNLESS OTHERWISE SPECIFIED DIMENSIONS ARE IN INCHES TOLERANCES ON FRACTIONS DECIMALS ANGLES \pm \pm \pm ALL SURFACES $\sqrt{16}$ MATERIAL GOVT. OR COML. To BE SPECIFIED	DRAWN		SPECIMEN Low CYCLE FATIGUE		Mar-Test inc.	
	DATE				CINCINNATI, OHIO	
	APPD					
	ISSUED					
APPROVED		DATE				
ENGR <i>StC</i>		1-12-71			SIZE	MTI-1002
MFG						
MATL			SCALE 1/1	WT CALC ACTUAL	CONT ON SHEET	SH NO.

IV - DISCUSSION OF DATA

A review of all the low-cycle fatigue tests performed on the two previous NASA-sponsored programs at Mar-Test Inc. led to the summary presented in Table 1. Each task of each program is treated individually to indicate the number of low-cycle fatigue tests performed for each material. This sequencing was also used in the presentation of the data plots and summary tables for this report. In other words, the data for the R-0 tests are presented first, followed by the data for the R-1 tests, etc. It will be noted that there are 188 tests involved in this data reporting activity.

For each test the load traces were reviewed first and a plot was made of the load range as a function of the number of cycles. In general, the first and second cycle information was read from the x-y traces (stress-strain hysteresis loops) and these points were plotted. Then this plotting was continued for subsequent cycles using load range information as read from the strip-chart recordings. A sufficient number of data points was selected throughout the fatigue life to define a fairly smooth pattern for the plot of load range versus cycles. Such plots for the 188 tests involved in this evaluation are presented in Figures 2 through 189. Each figure identifies alloy code number (see Table 1), specimen number, test temperature, strain range, strain rate and cycles to failure.

A typical hysteresis loop for each test is presented in Figures 190 through 288. It was the intent here to show the cyclic stress-strain behavior near half-life but in many cases no hysteresis loops were recorded in the immediate vicinity of $N_f/2$. In these instances the hysteresis loop selected for reporting was the one closest to half-life and usually in the region prior to $N_f/2$. The particular cycle used in such plots is recorded in the figure along with the temperature, strain range and cycles to failure.

A final data summary prepared for this report is presented in Table 2. For each test, values of N_5 , the cycles to a five percent load range drop, and N_f , cycles to complete specimen separation, are compared. Some interpretation of N_5 had to be made in those instances where cyclic softening was exhibited. In these situations the plot of load range versus cycles on semi-logarithmic coordinates indicated a gradual decrease in the load range as cycling progressed and it was not possible to identify a stabilized load range for calculating a five percent reduction. The special interpretation adopted for these cases involved the selection of that point on the load range plot where a change in curvature first began to appear as the load range began to decrease rapidly as failure approached. A five percent drop from this load range value was then used to establish the value for N_5 . In a few cases no value of N_5 is reported (in Table 2) due either to the fact that the specimen failed before a five percent load range reduction was reached or, because of the orientation of the crack with respect to the extensometer tips, the load range actually increased slightly near the failure point.

Table 1 - Summary of Low-Cycle Fatigue Tests Performed by Mar-Test Inc.
in Two Previous Programs

CONTRACT NAS3-16753; initiation date: June 29, 1972

<u>Alloy Designation</u>	<u>Material</u>	<u>Number of Tests</u>
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Task I

R-0	Zirconium-copper, annealed	6
R-1	Zirconium-copper, $\frac{1}{4}$ hard	6
R-2	Zirconium-copper, $\frac{1}{2}$ hard	8
R-3	Tellurium-copper, $\frac{1}{2}$ hard	6
R-4	Chromium-copper, SA and aged	6
R-5	OFHC copper, hard	6
R-6	OFHC copper, $\frac{1}{4}$ hard	7
R-7	OFHC copper, annealed	6
R-8	Silver, as-drawn	6
R-9	Zr-Cr-Mg copper, SA, CW and aged	7
R-10	Electroformed copper, 30-35 ksi	6
R-13	Co-Be-Zr copper, SA and aged	6

Task II

R-2	Zirconium-copper, $\frac{1}{2}$ hard	25
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Task III

R-2	Zirconium-copper, $\frac{1}{2}$ hard	17
R-20	Zirconium-copper, $\frac{1}{2}$ hard	6

8

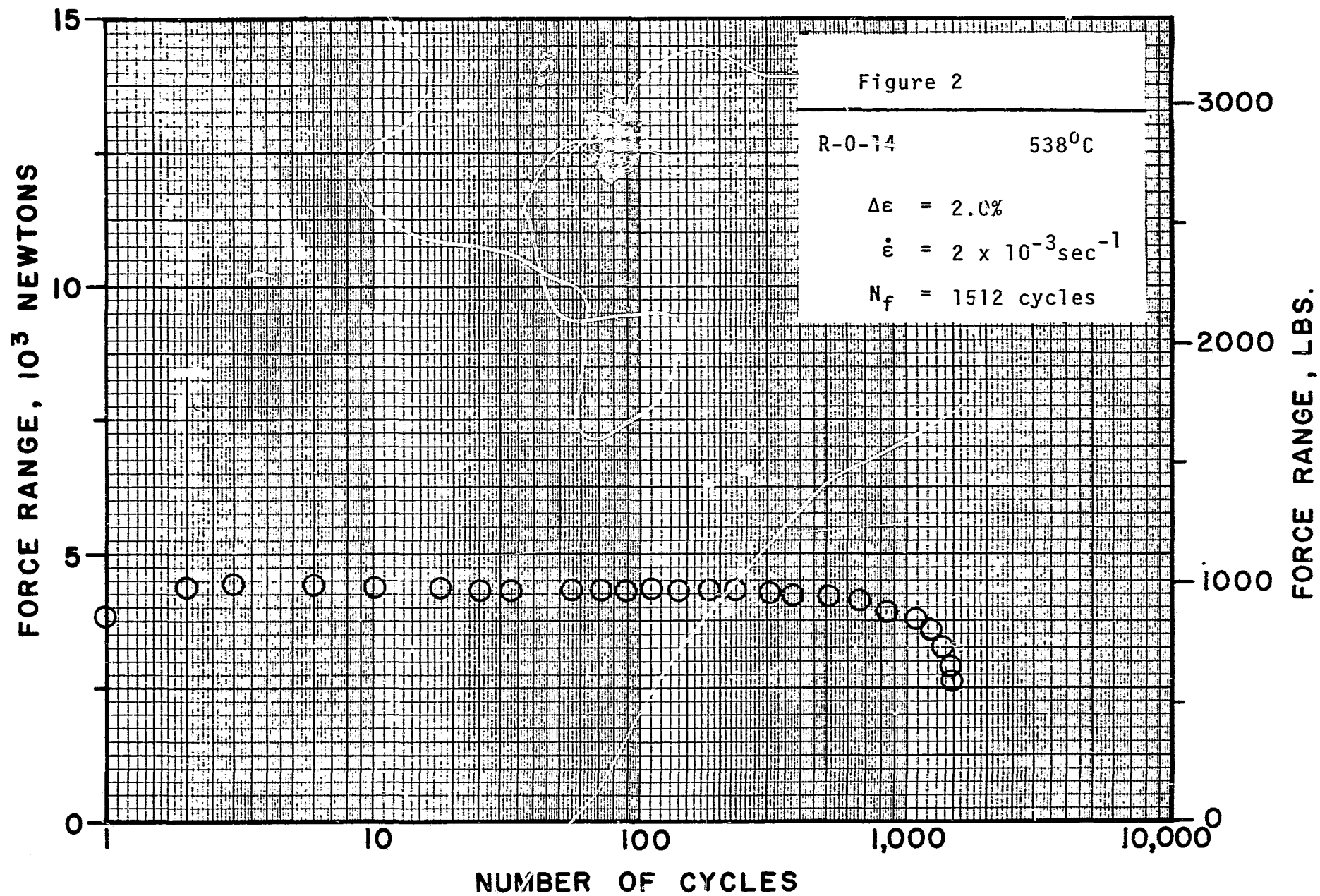
Table 1 continued - Summary of Low-Cycle Fatigue Tests Performed by Mar-Test Inc.
in Two Previous Programs

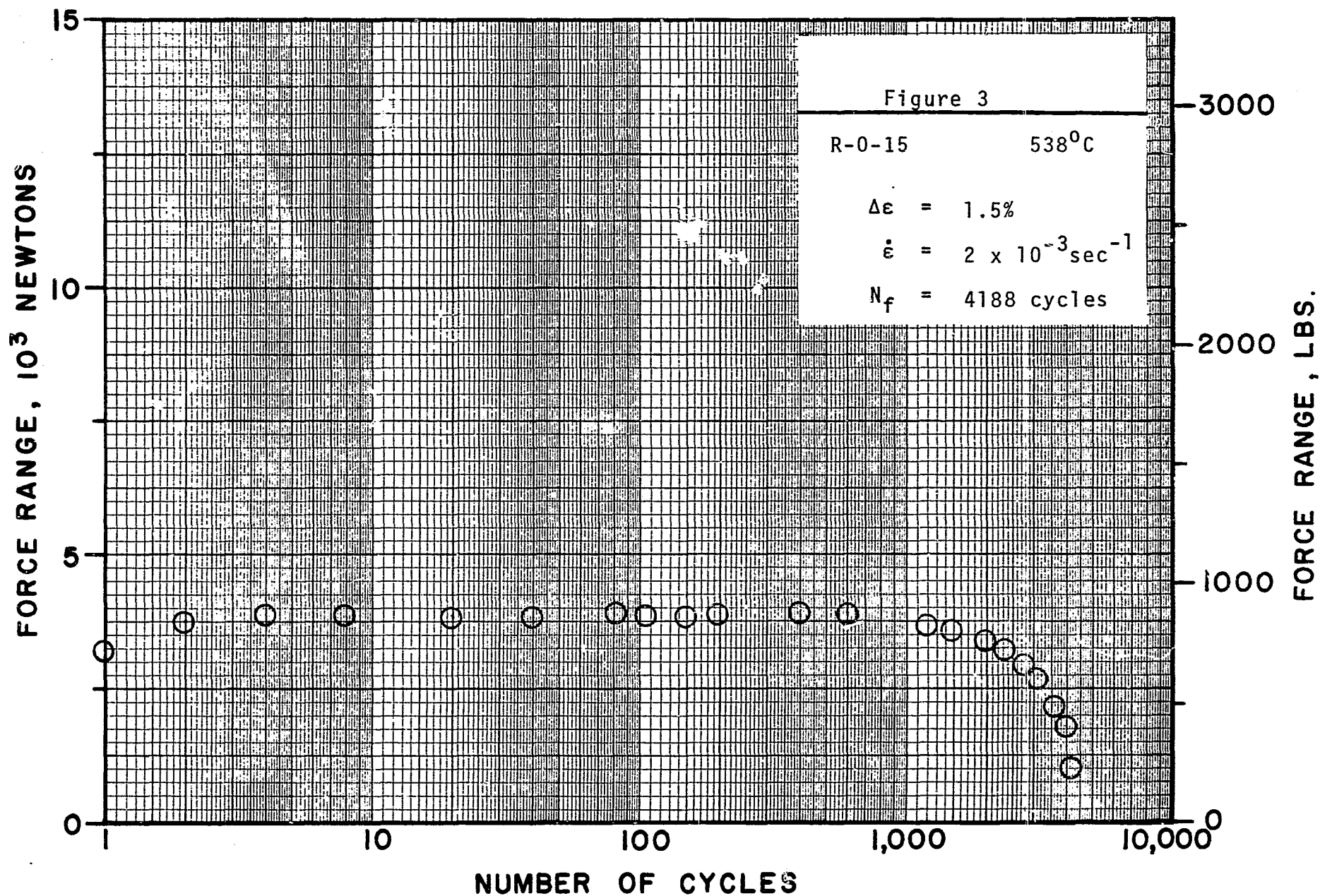
CONTRACT NAS3-17777; initiation date: June 28, 1973

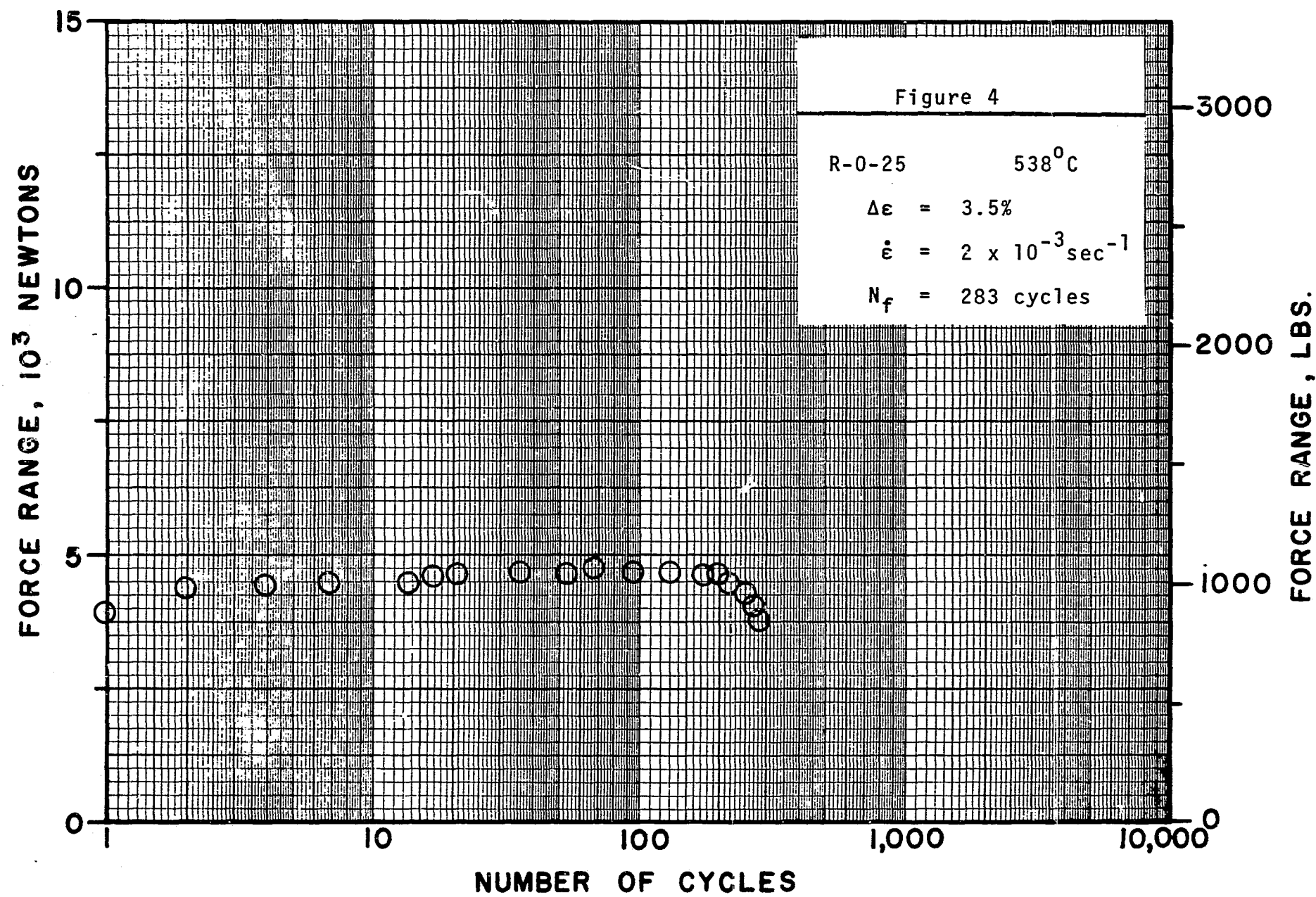
<u>Alloy Designation</u>	<u>Material</u>	<u>Number of Tests</u>
Task I *****		
R-24	Narloy Z, HR, SA and aged	29
R-2	Zirconium-copper, ½ hard	2
Task II *****		
R-21	NASA 1-1A copper alloy, aged	4
R-22	NASA 1-1B copper alloy, as-received	20
R-23	Glidcop AL-10	4
R-25	Sputtered zirconium-copper, annealed	4
R-26	Sputtered zirconium-copper, as-sputtered	1
Overall Total:		188 tests

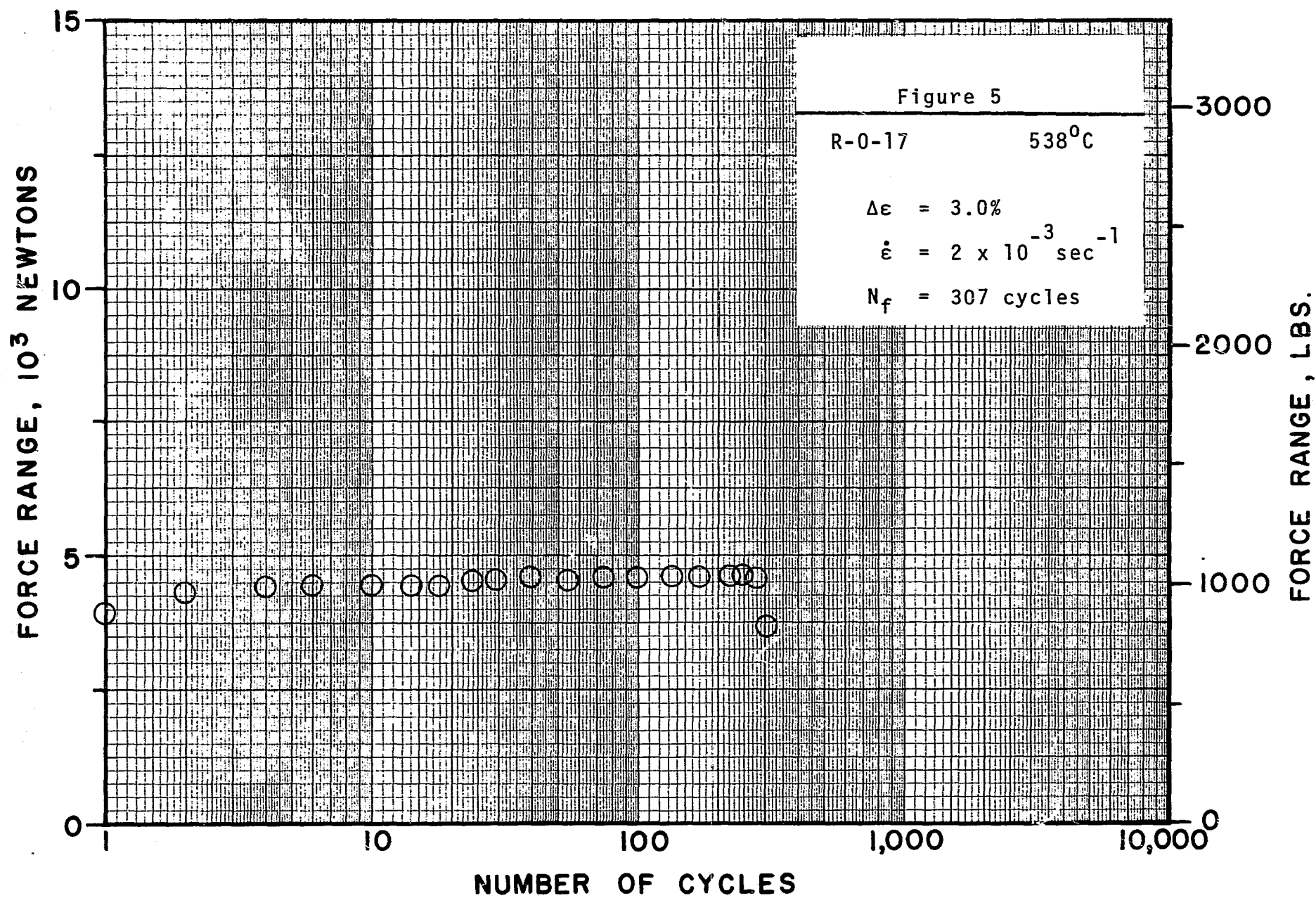
a) LOAD RANGE PLOTS

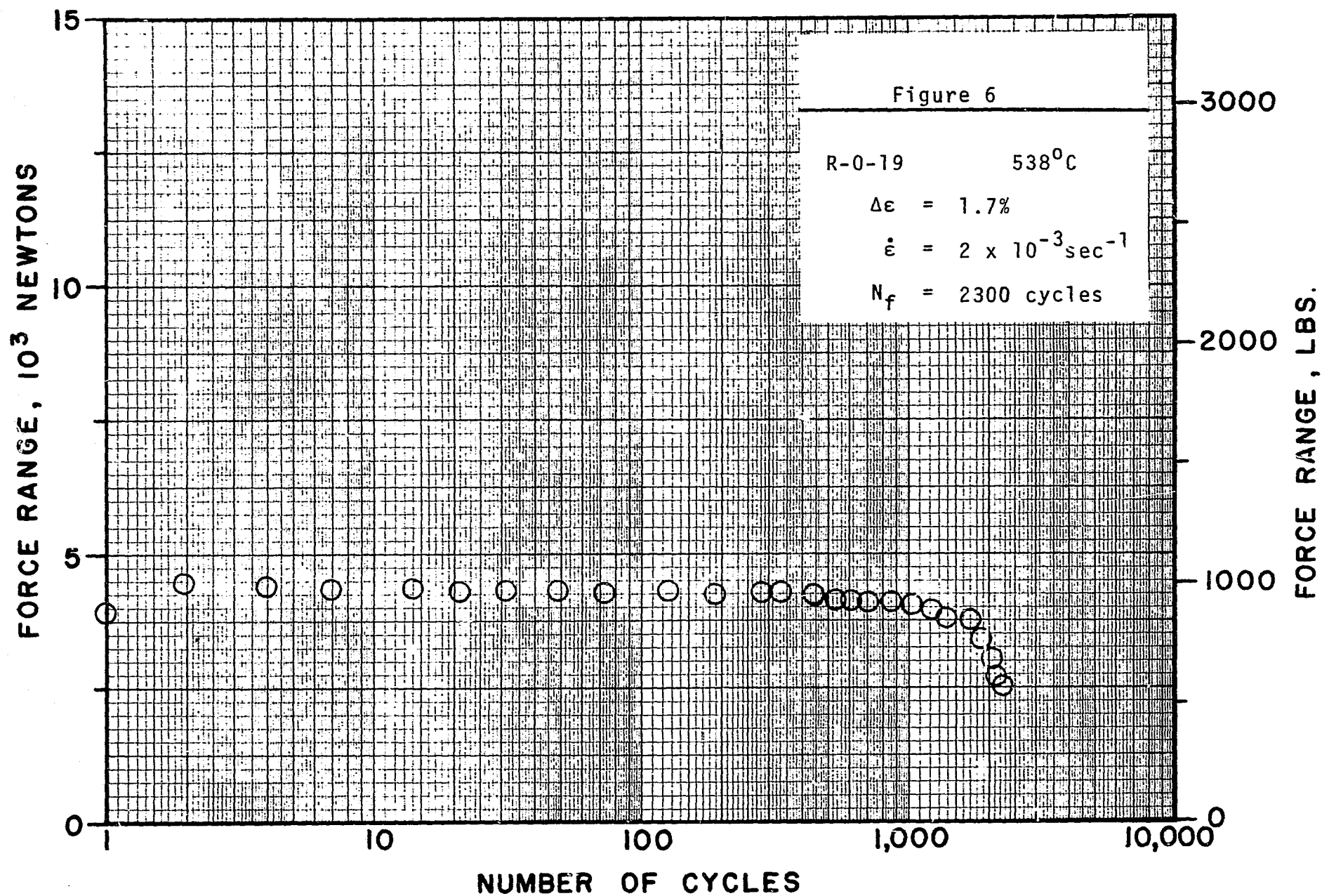
Figures 2 through 189

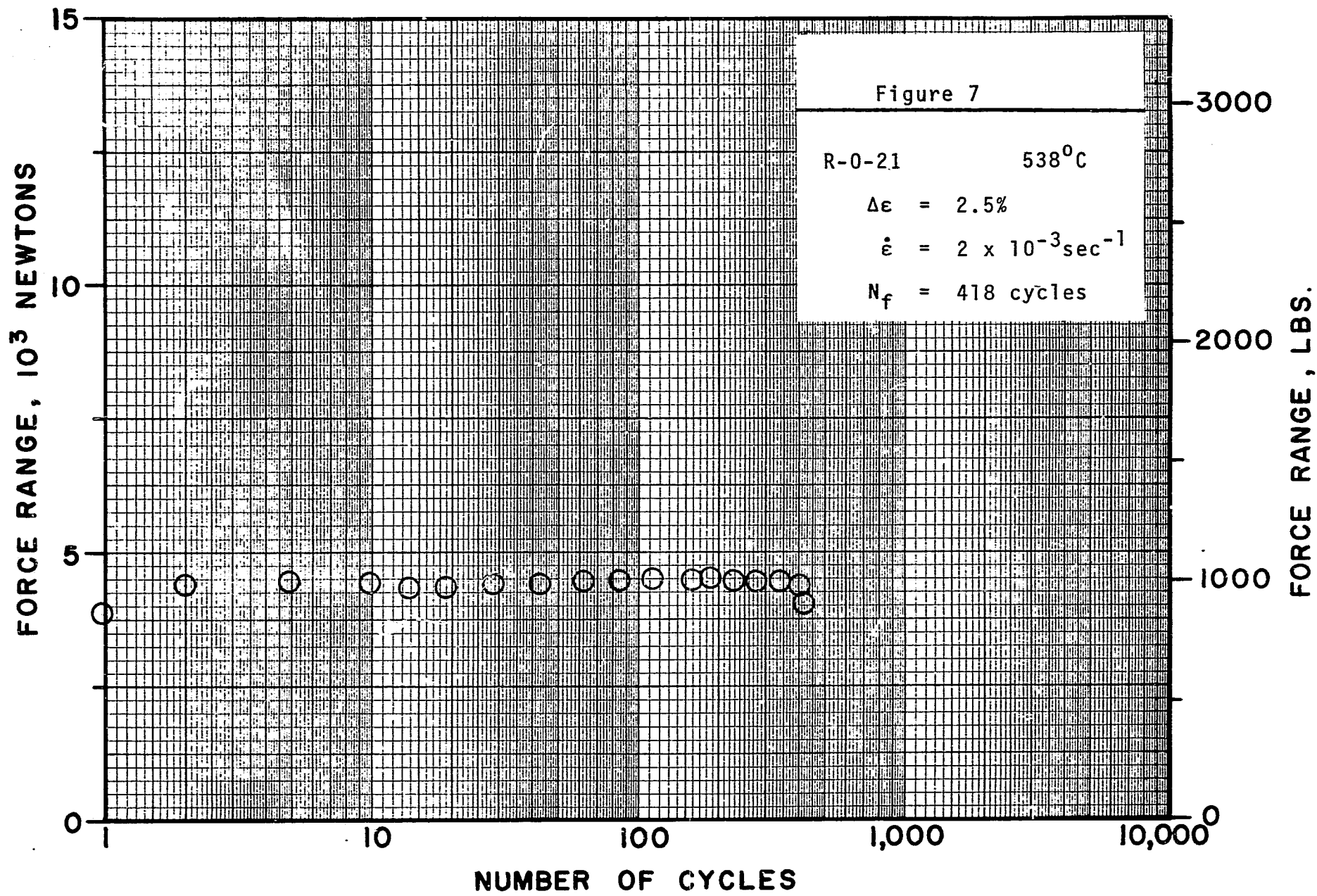


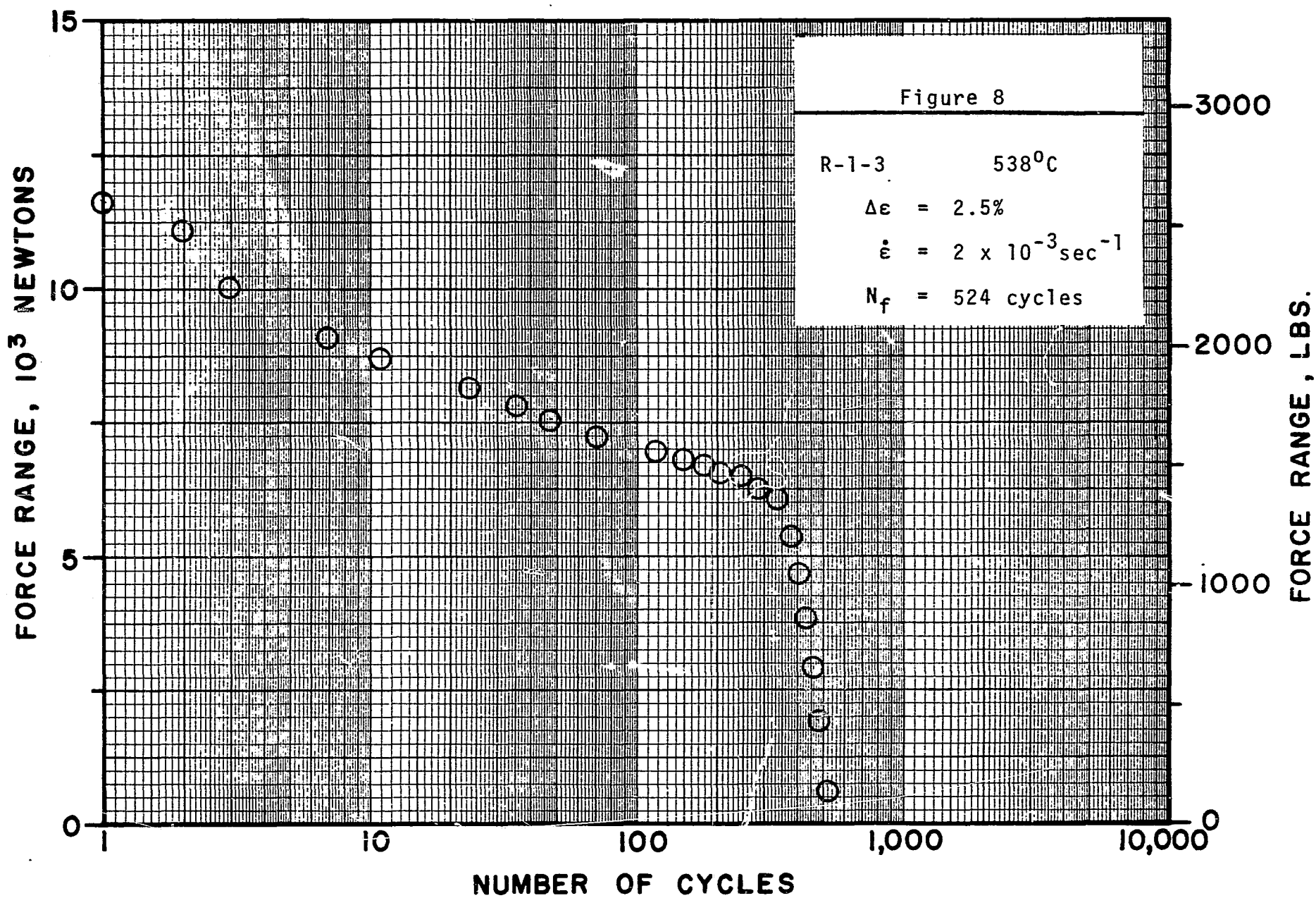


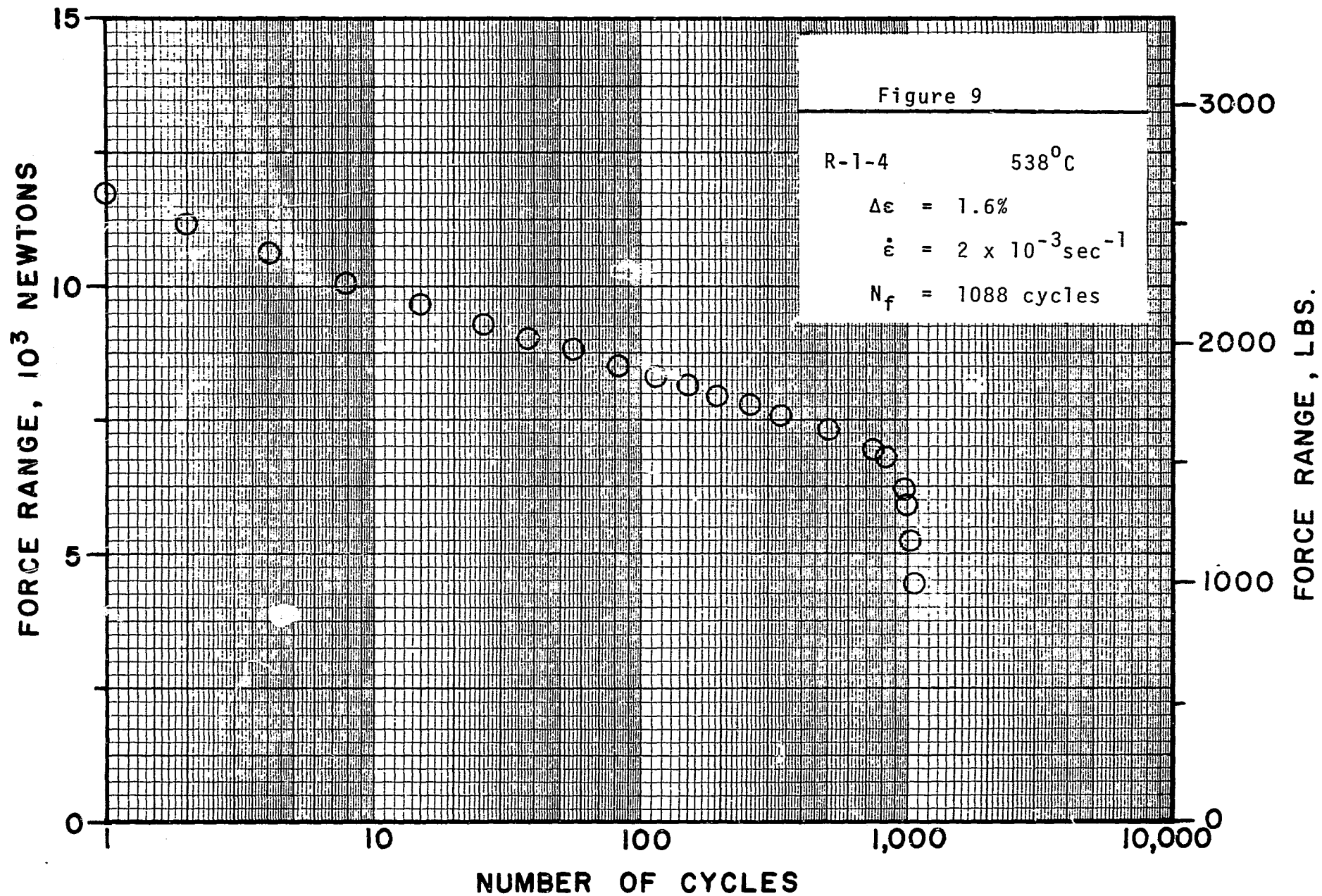


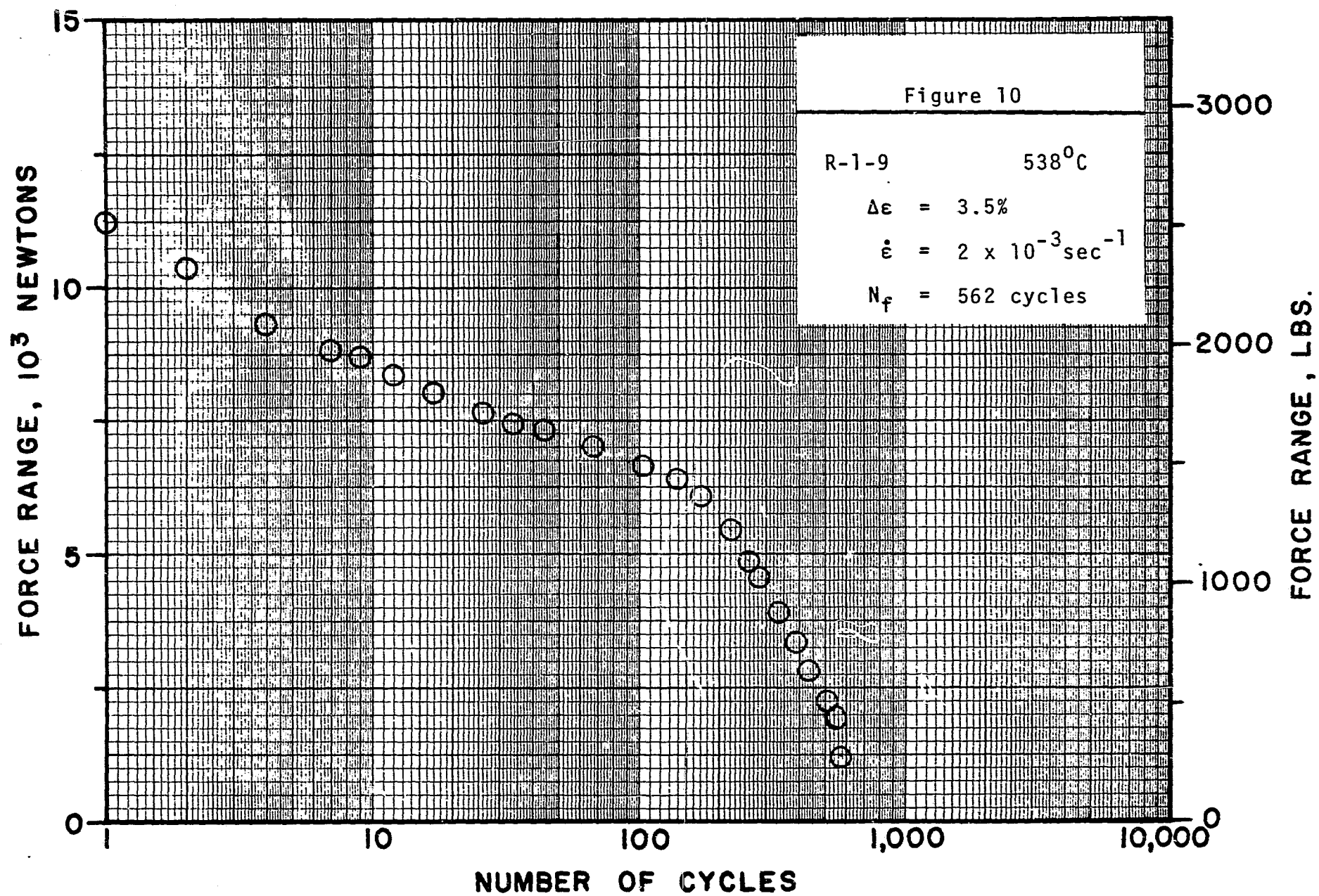


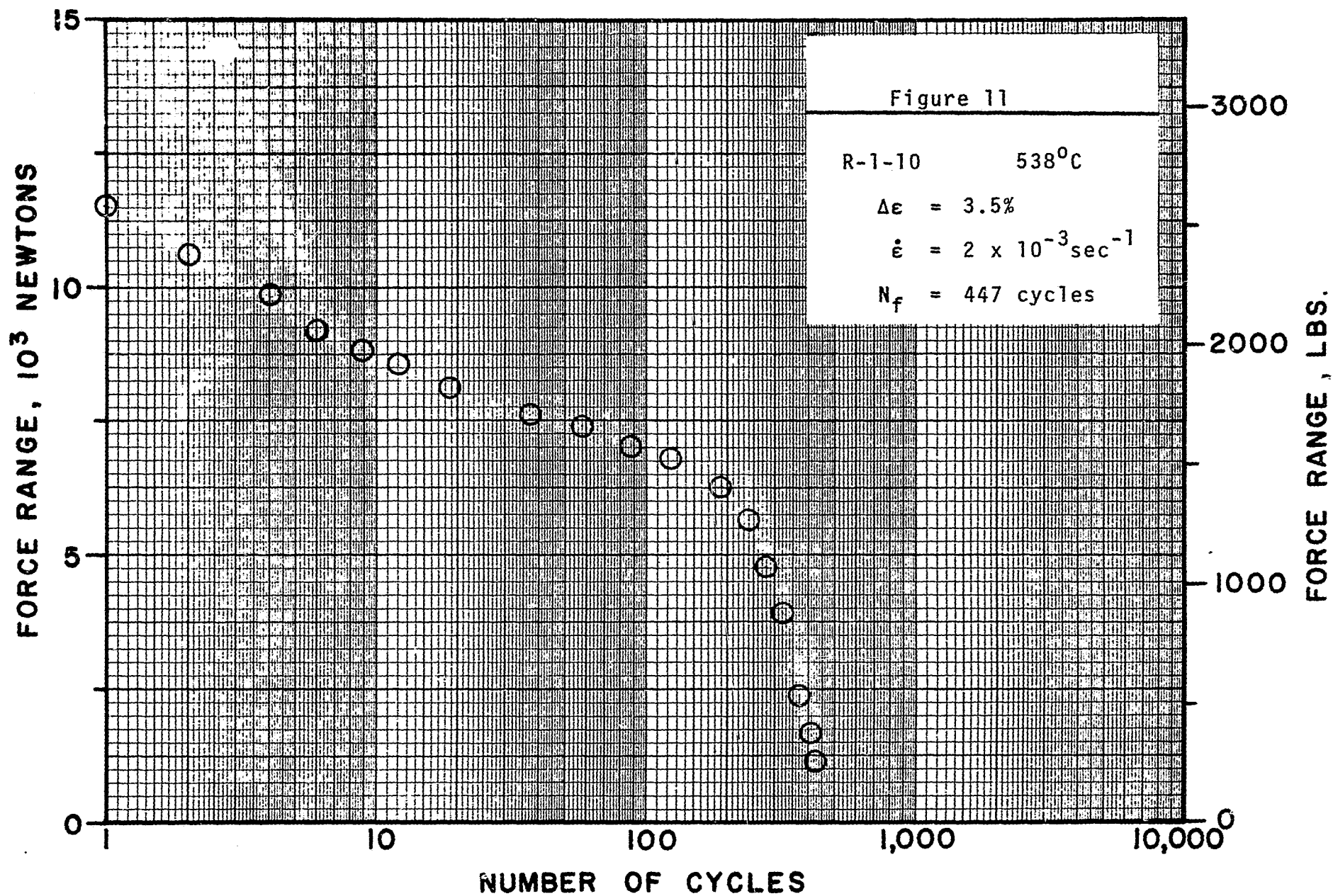


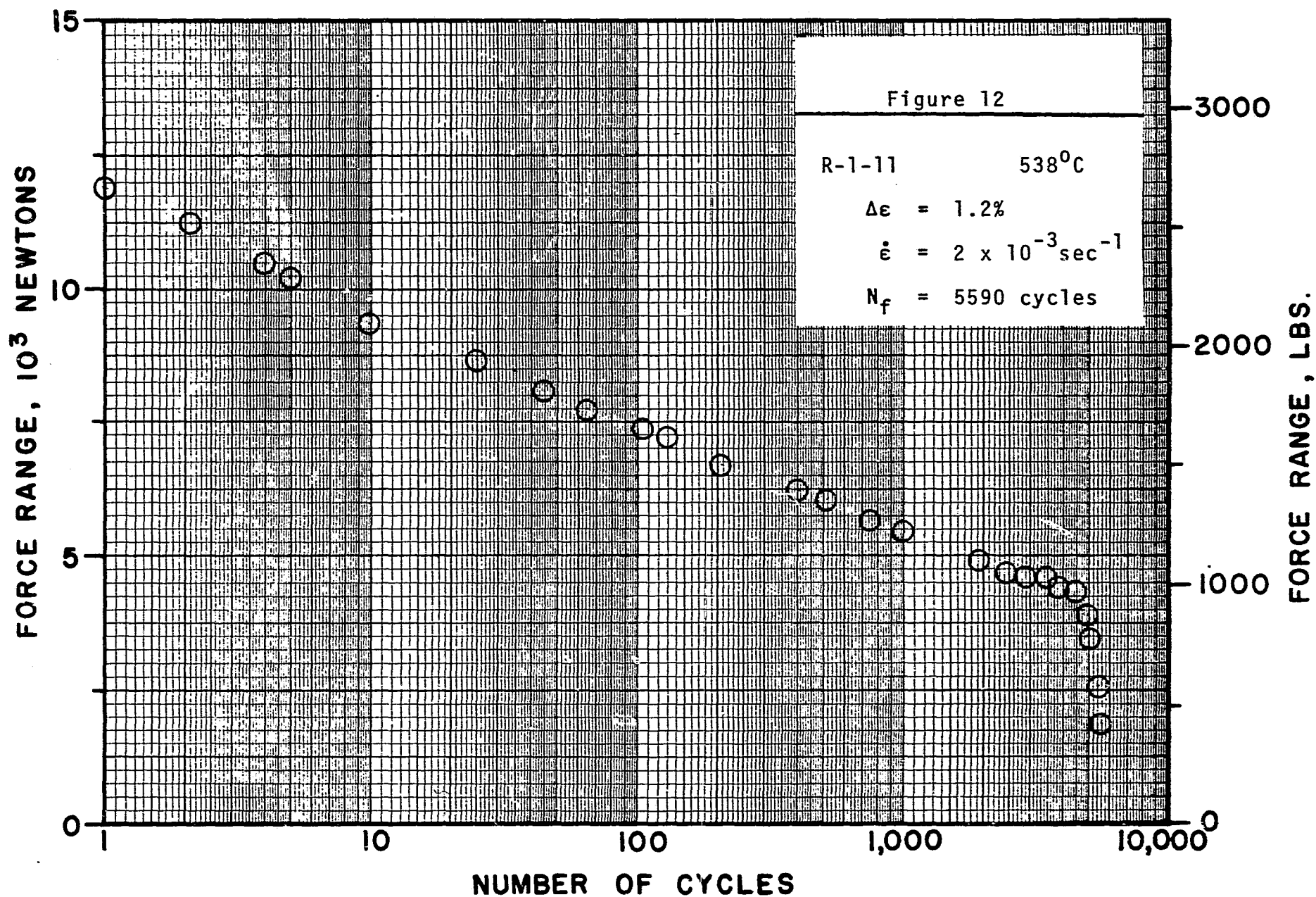


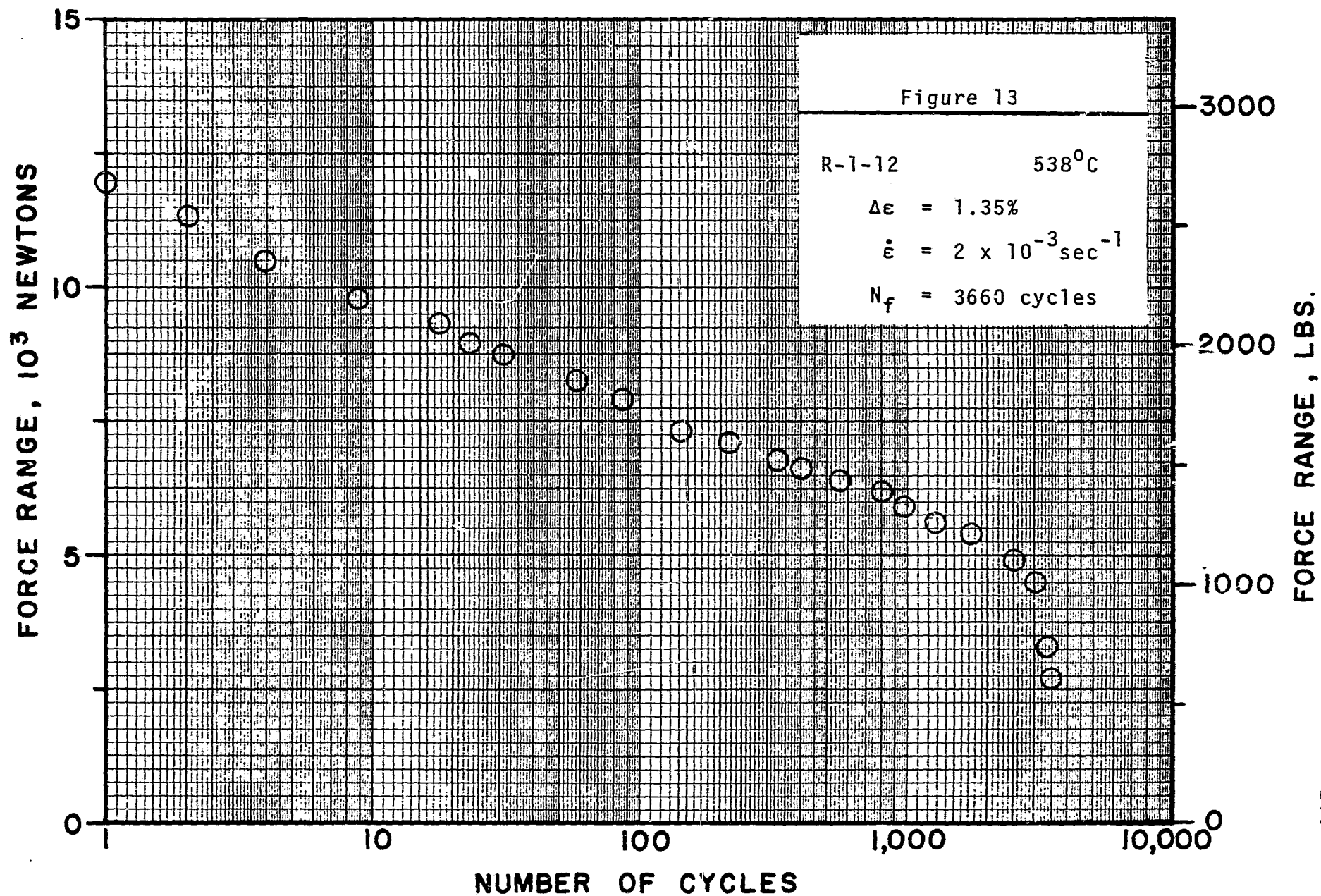


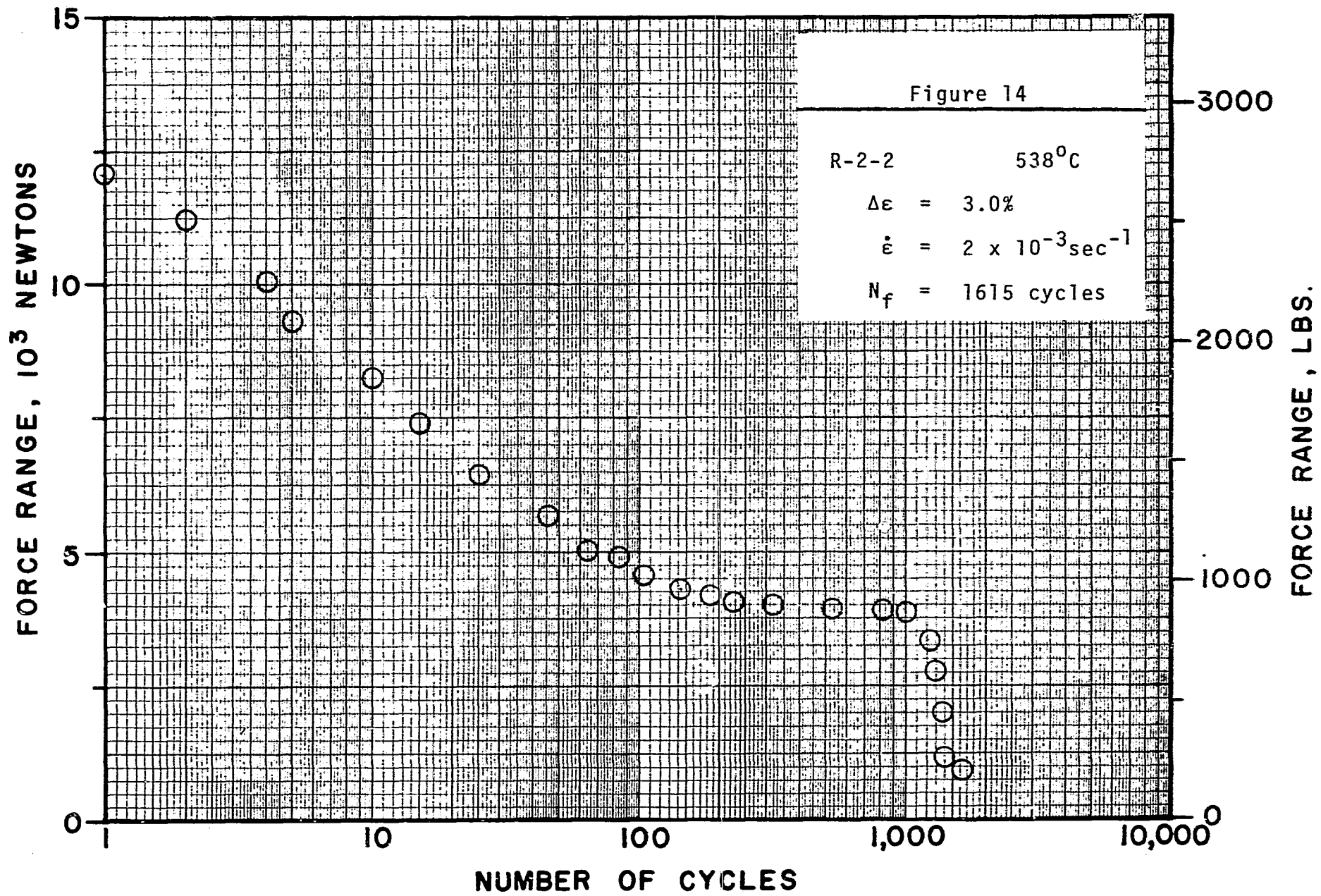


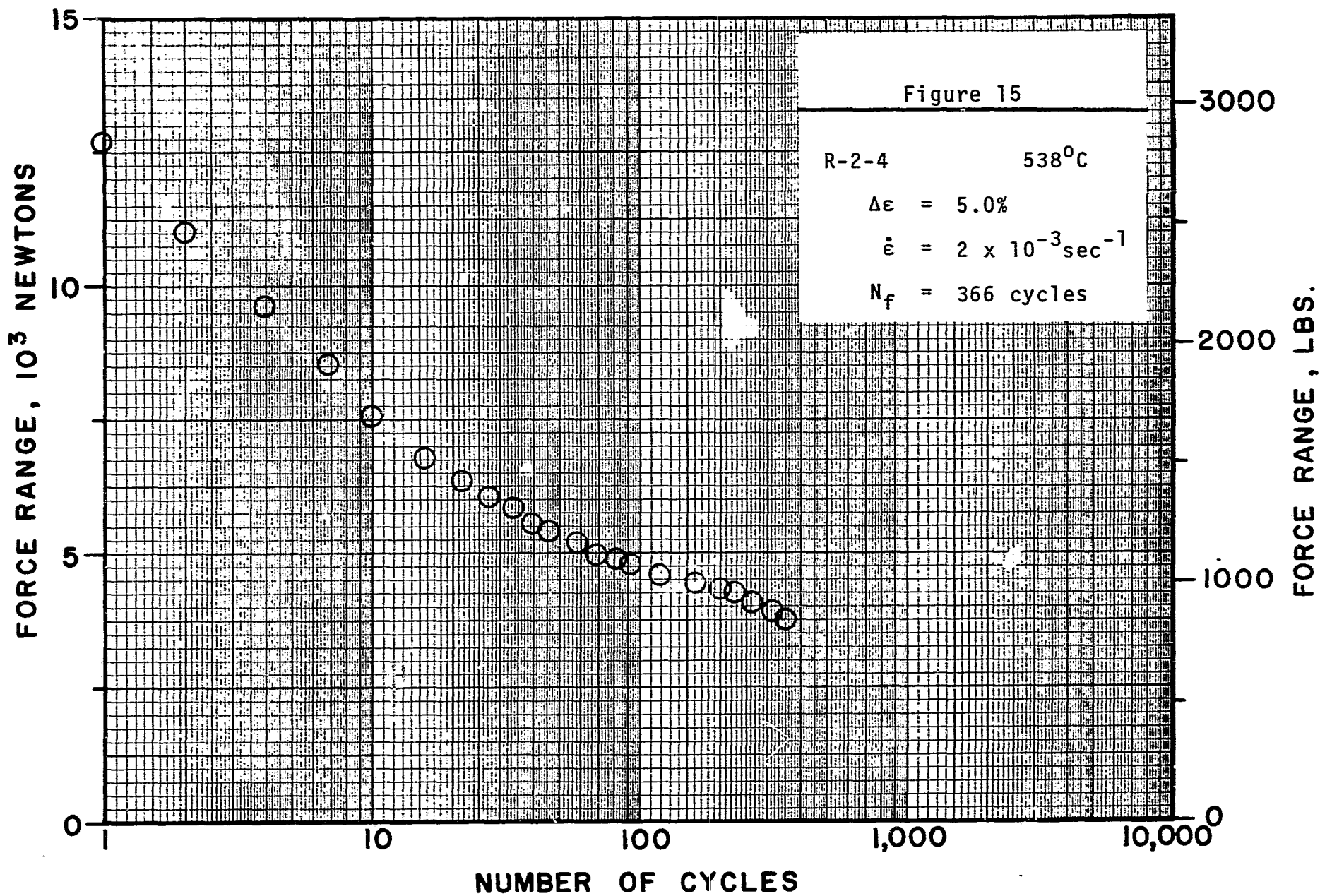


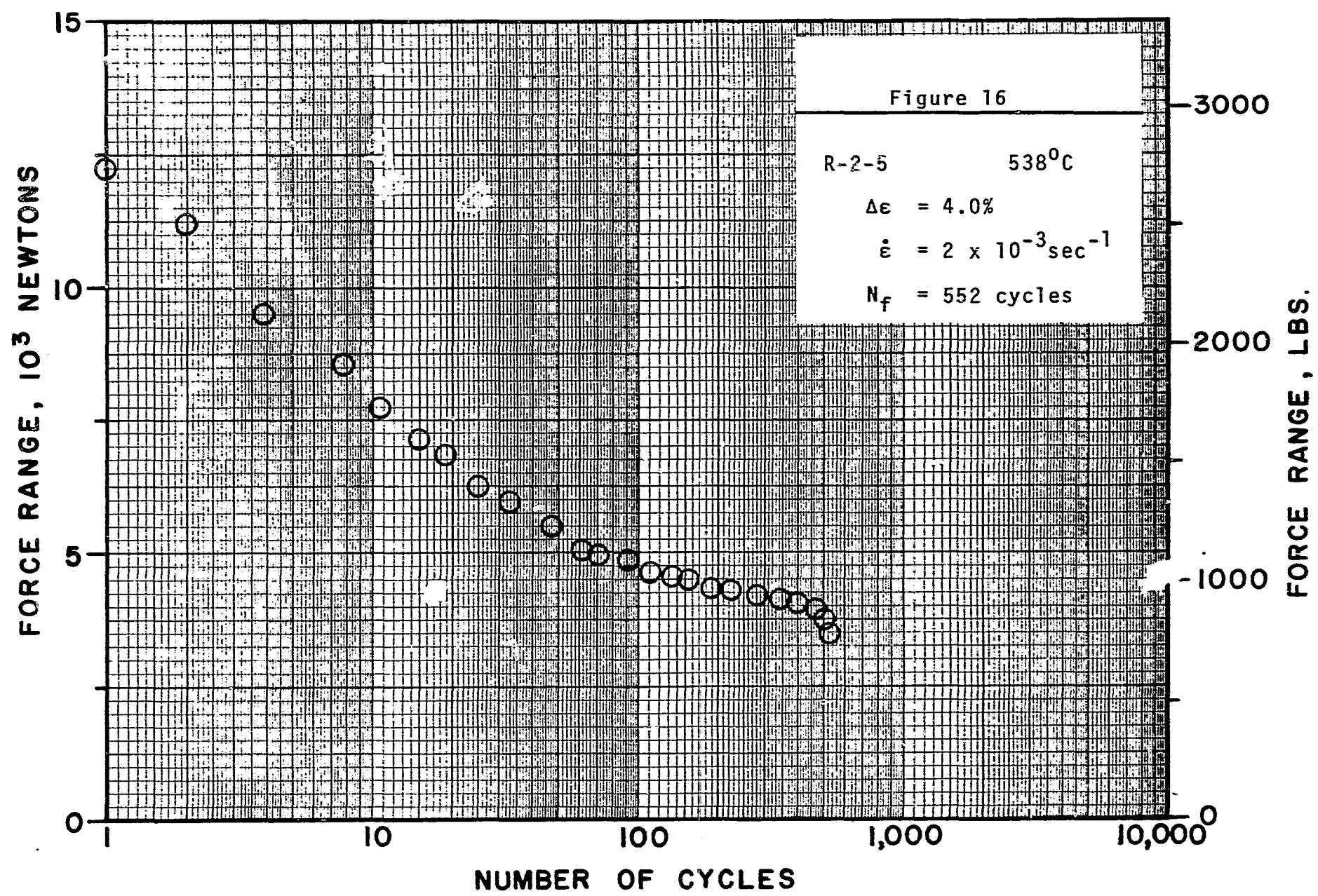


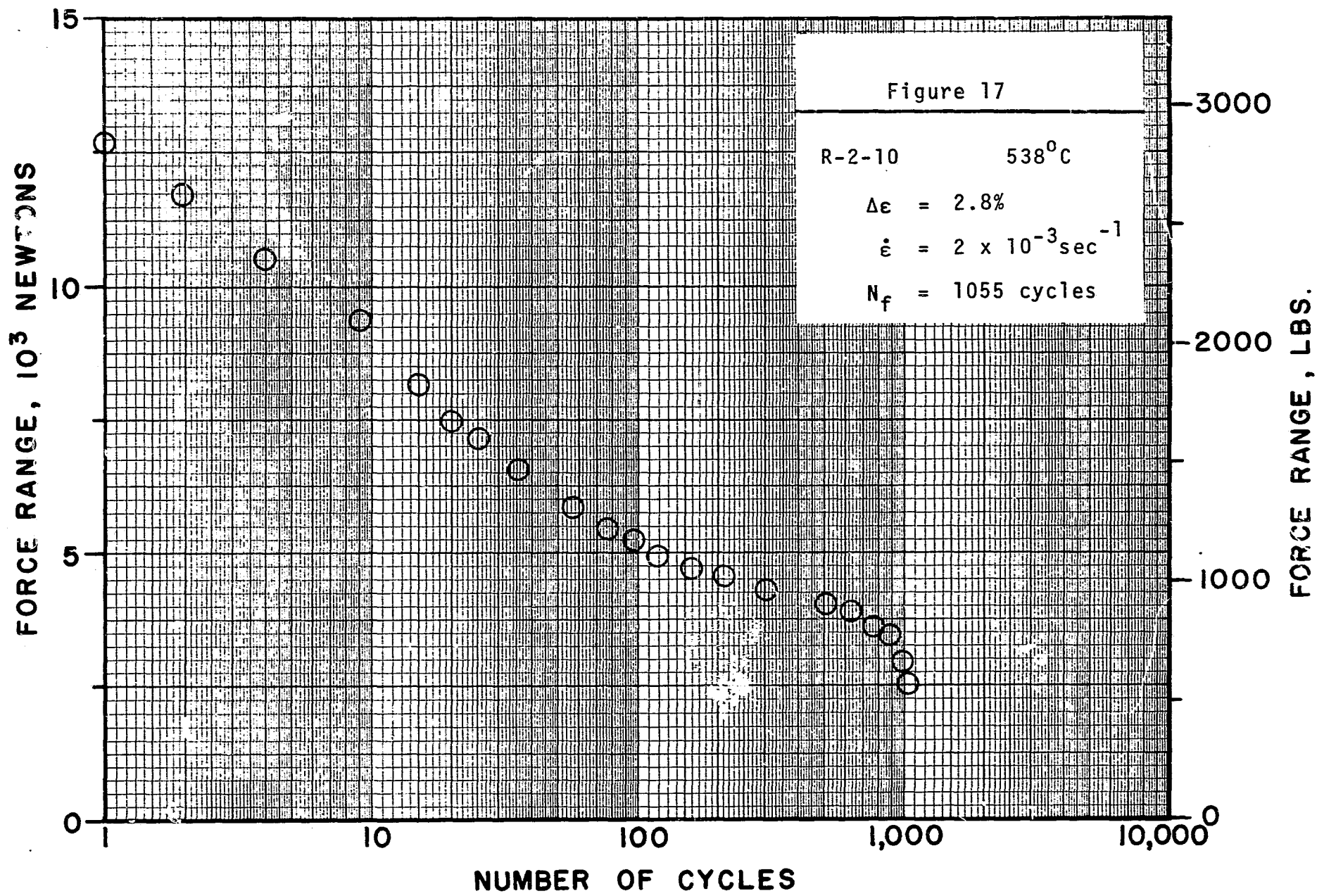


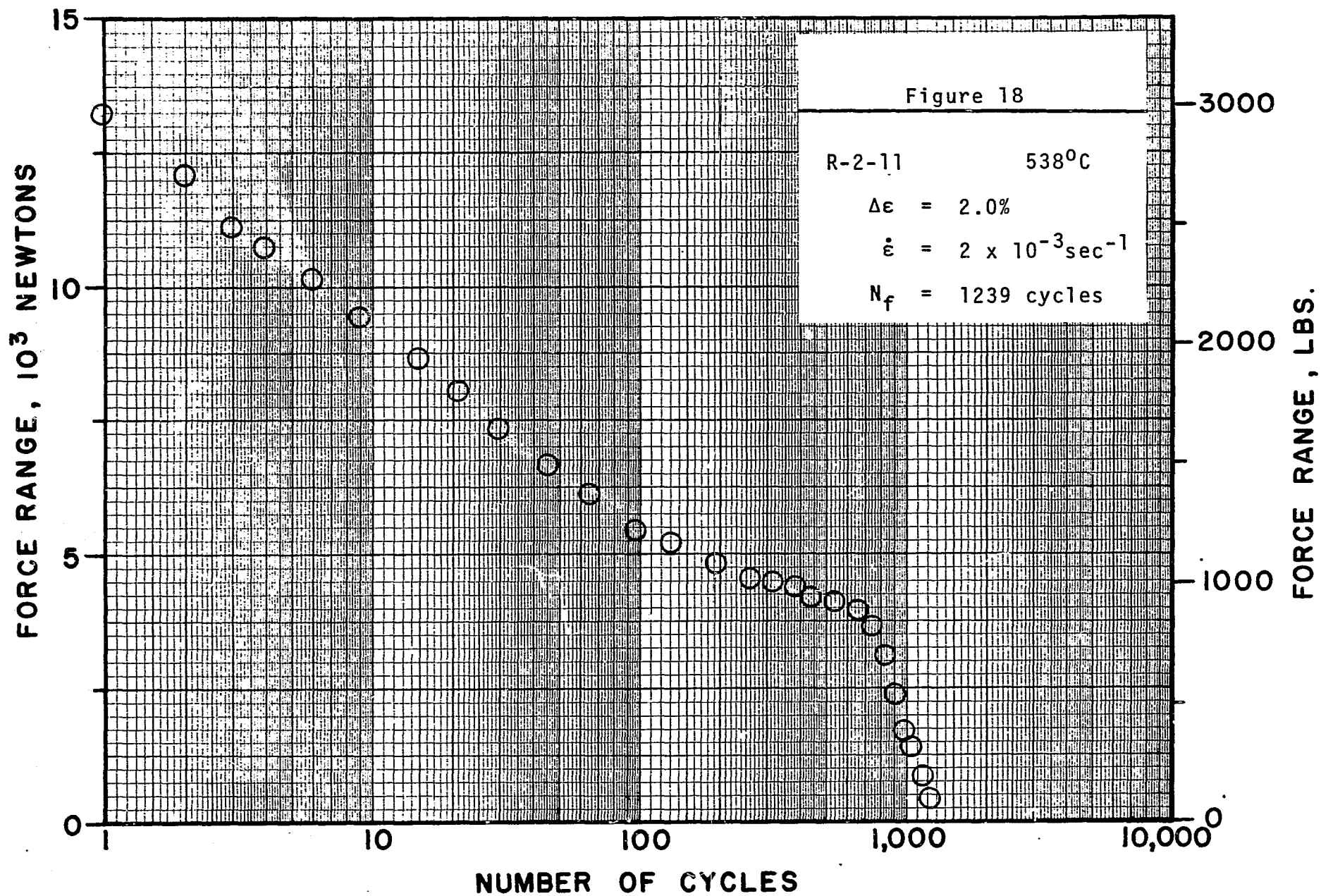


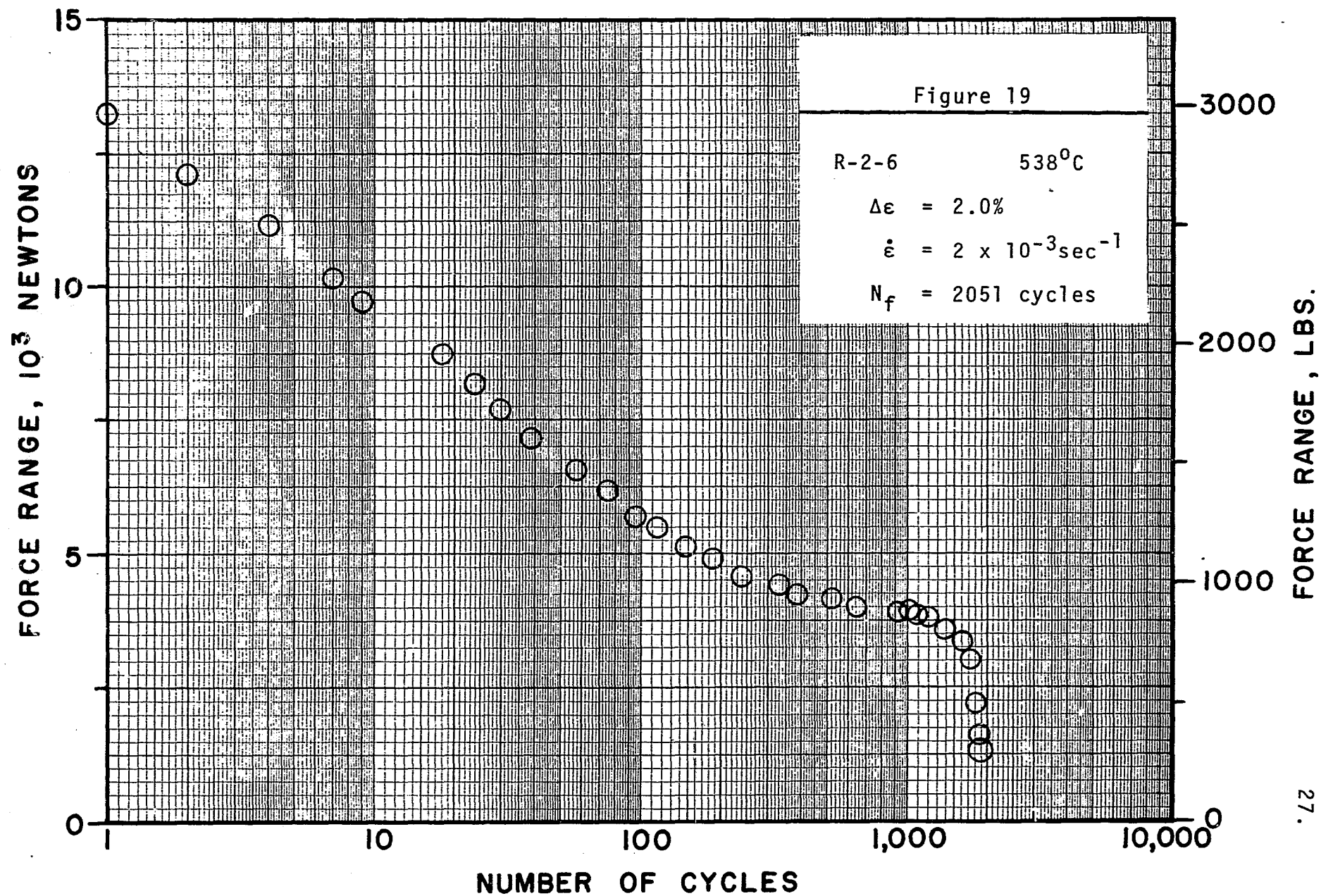


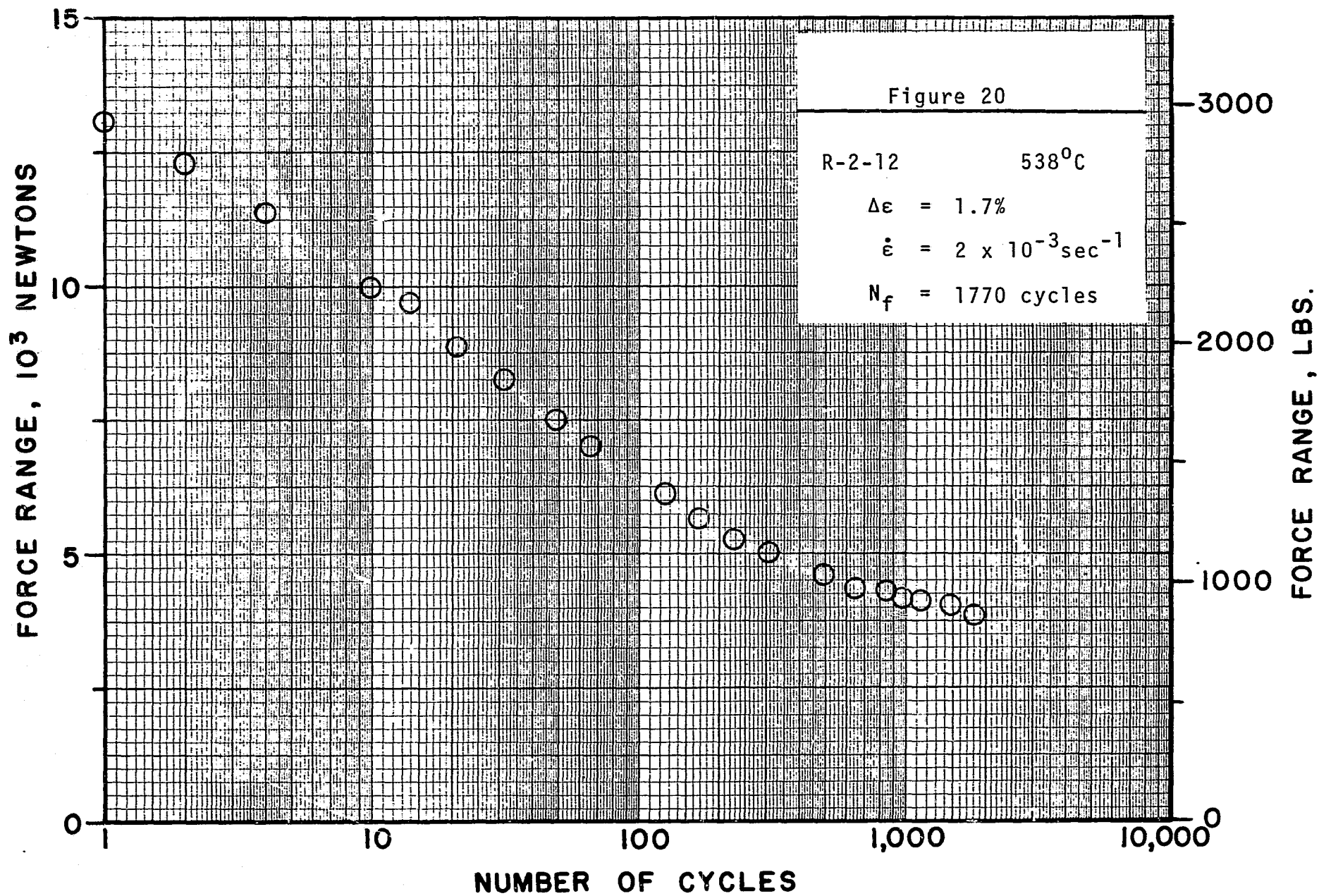


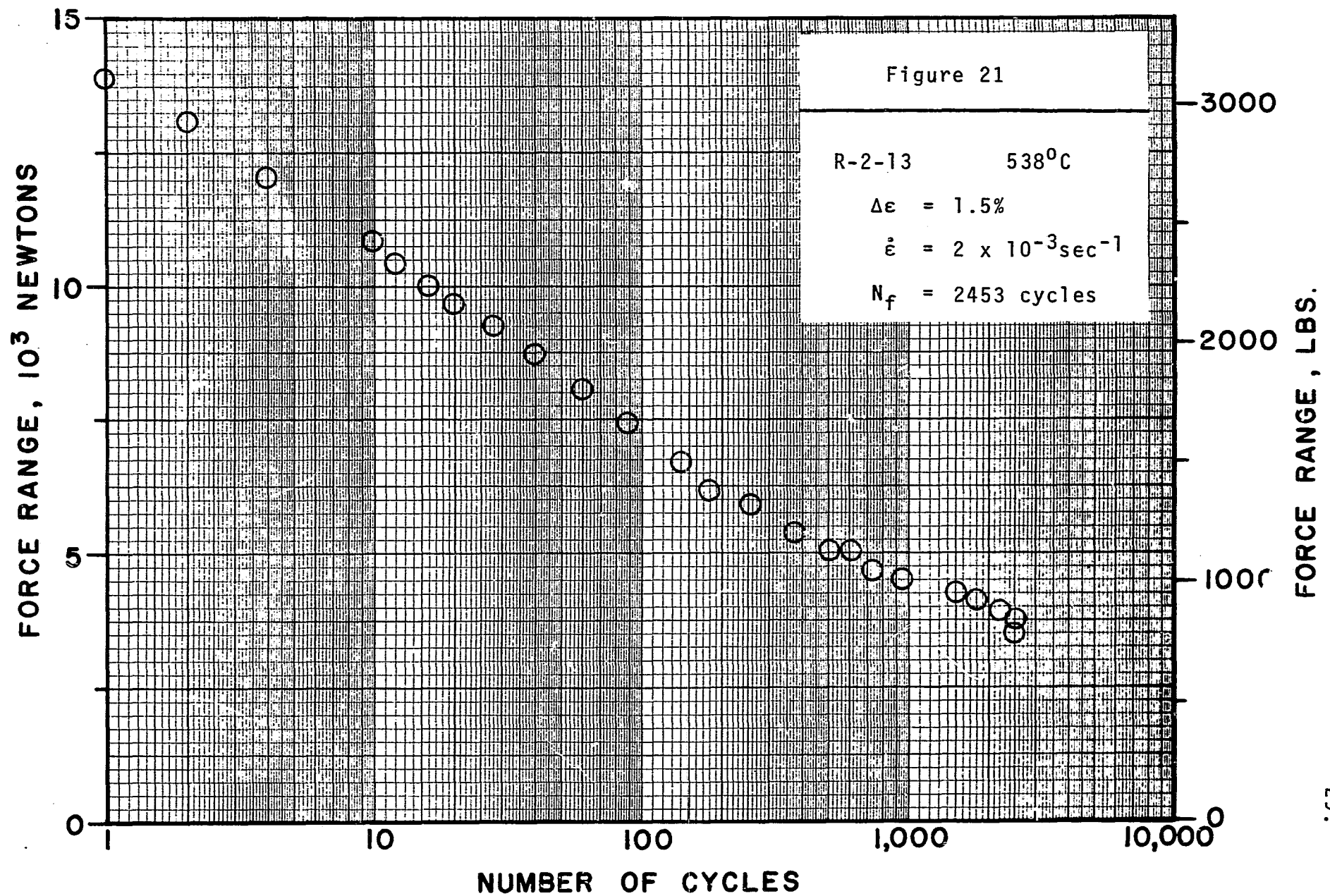


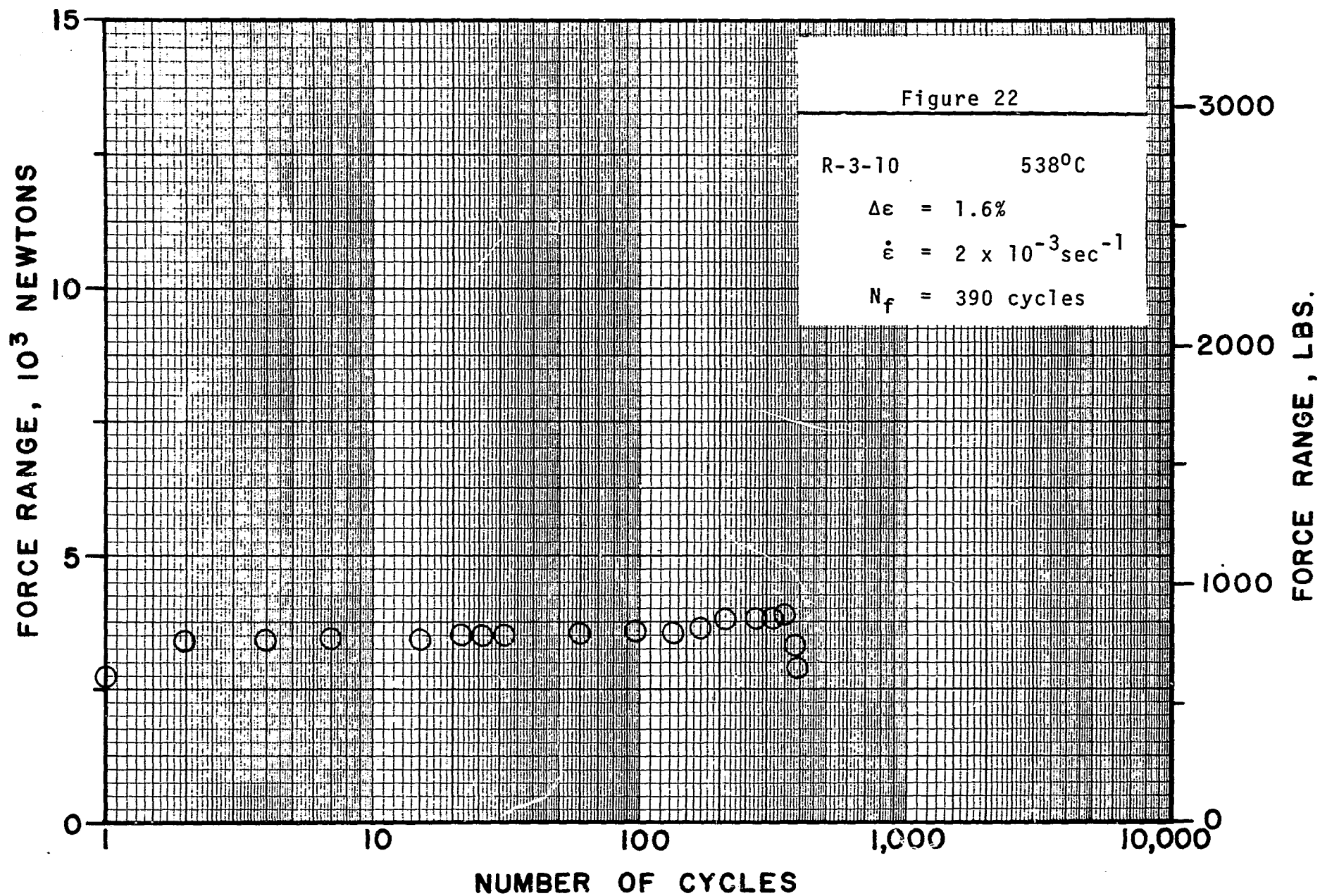


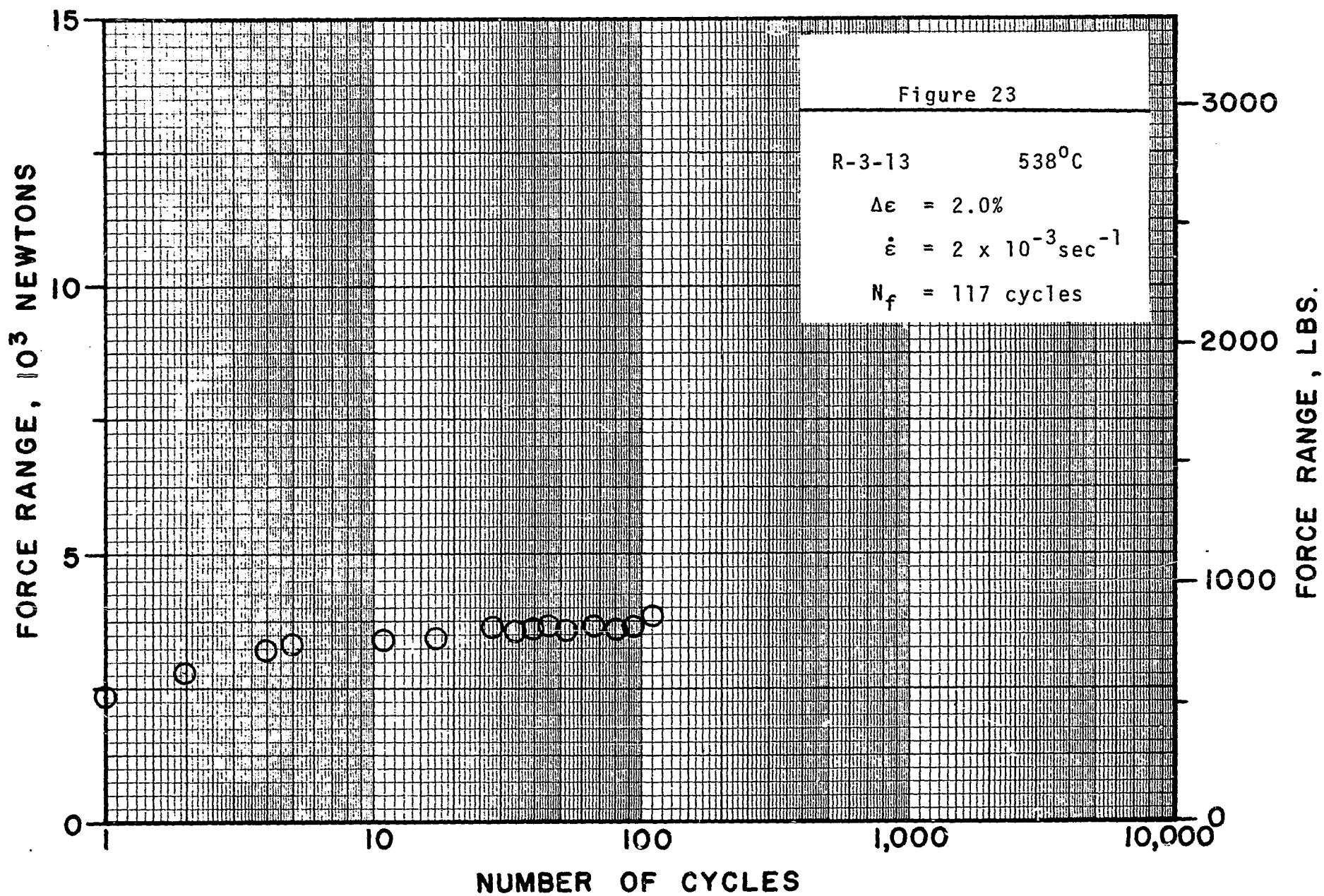


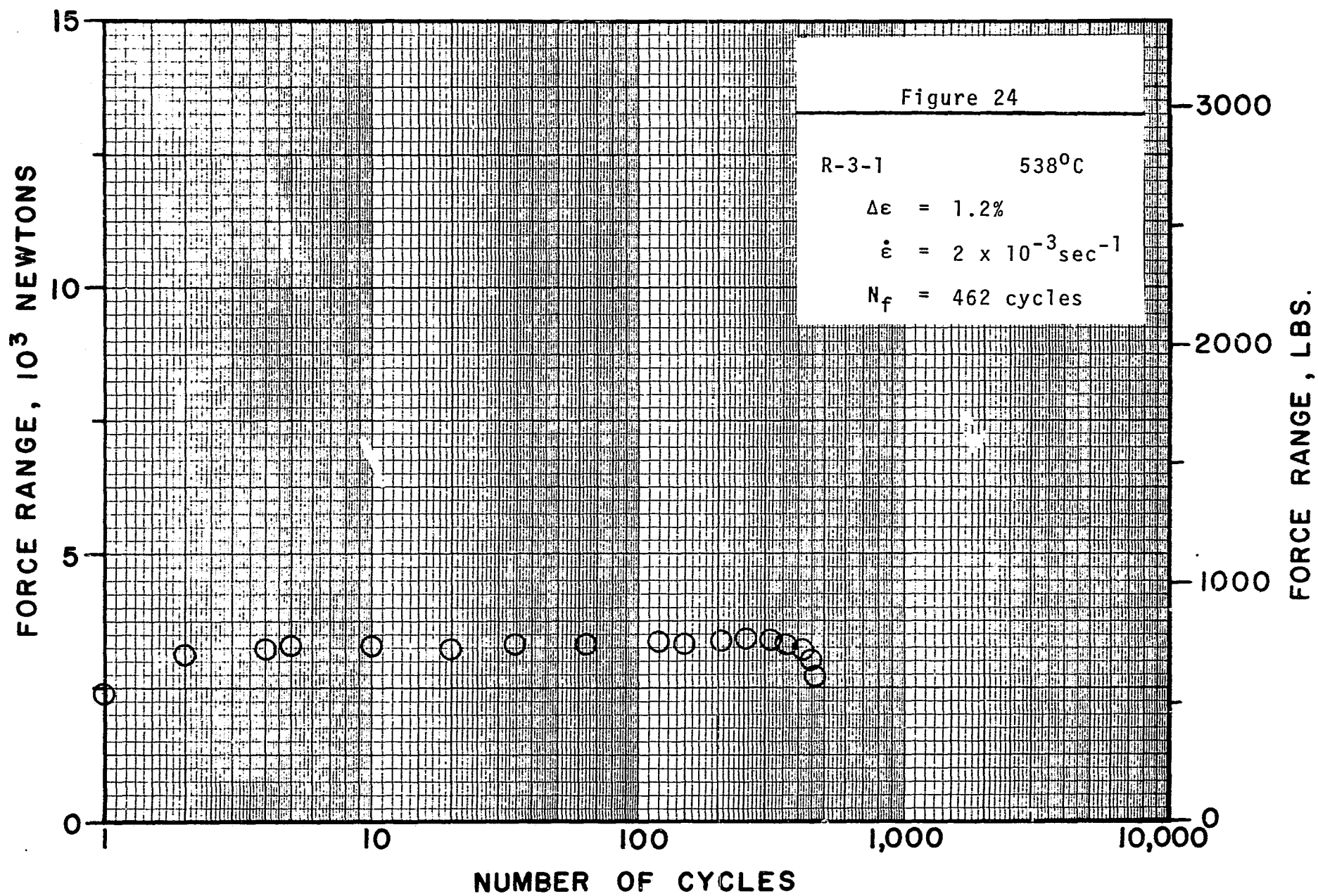


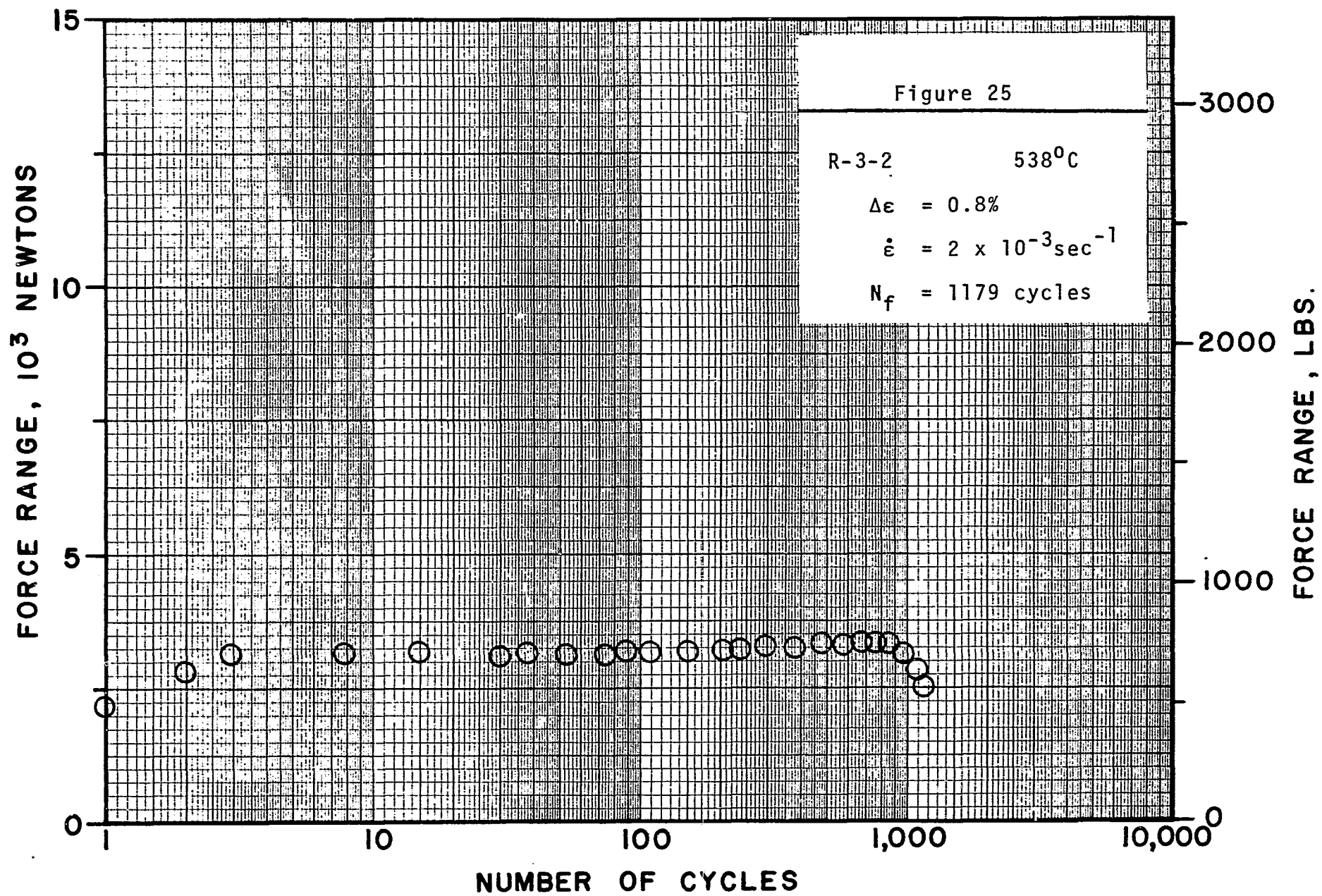


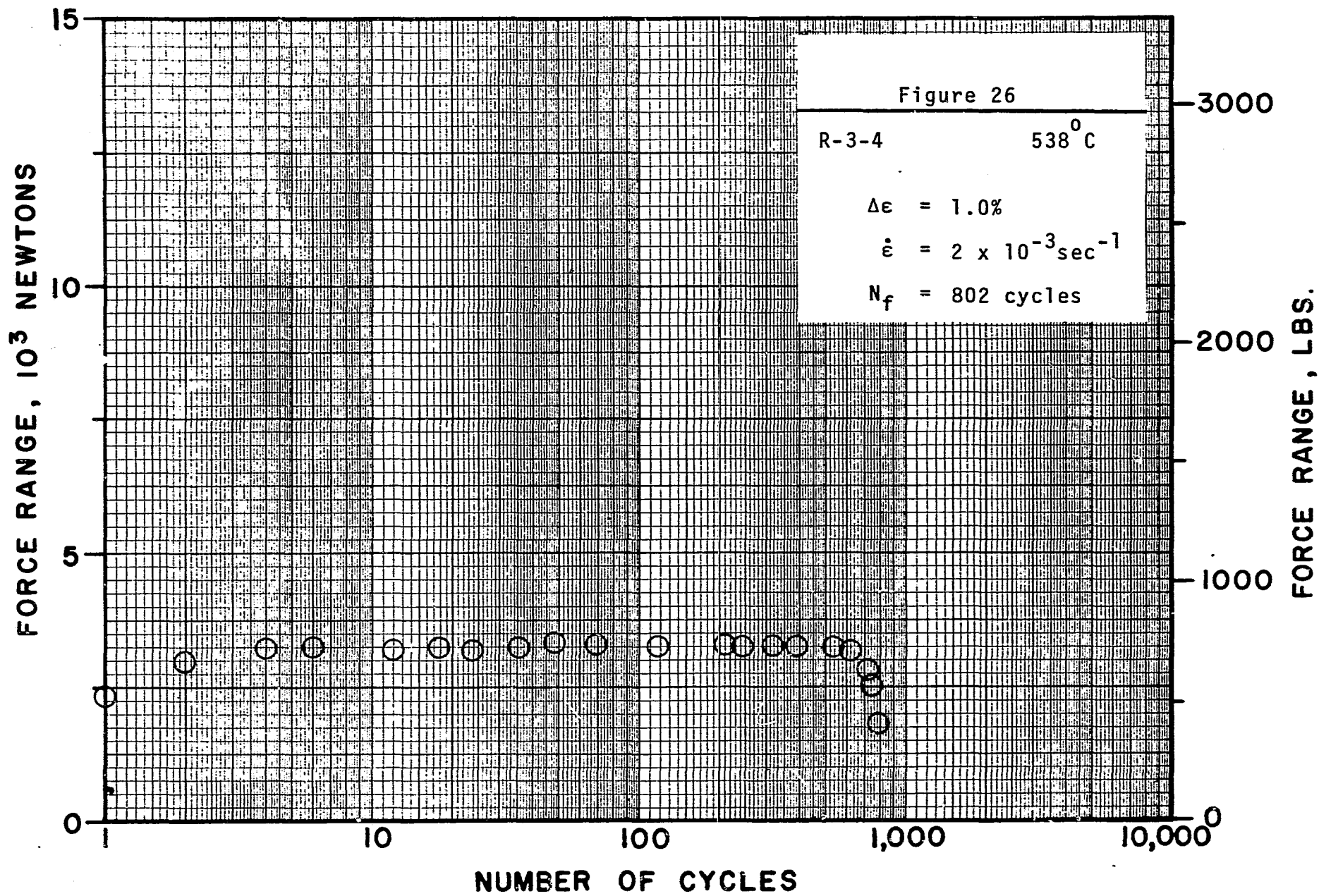


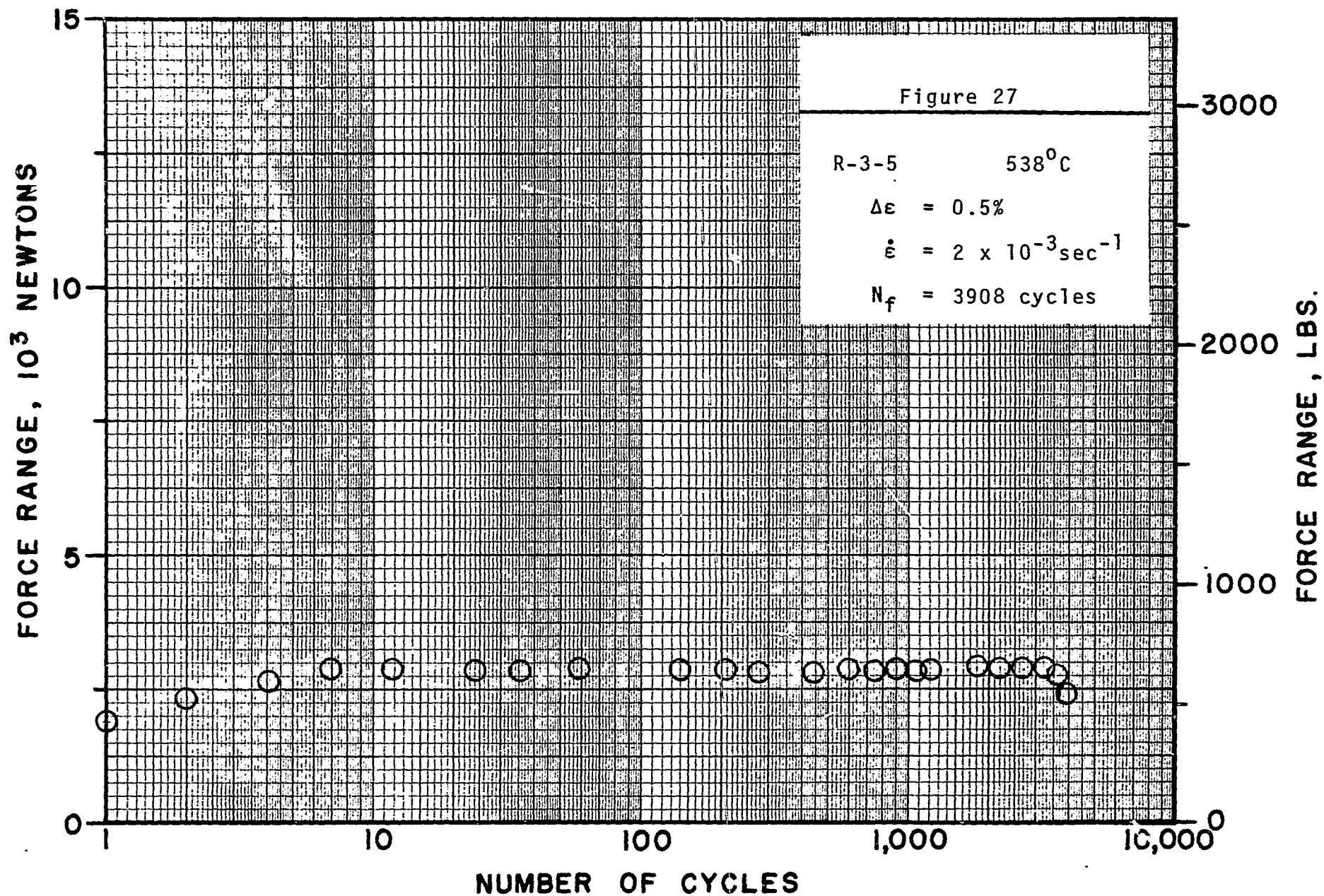


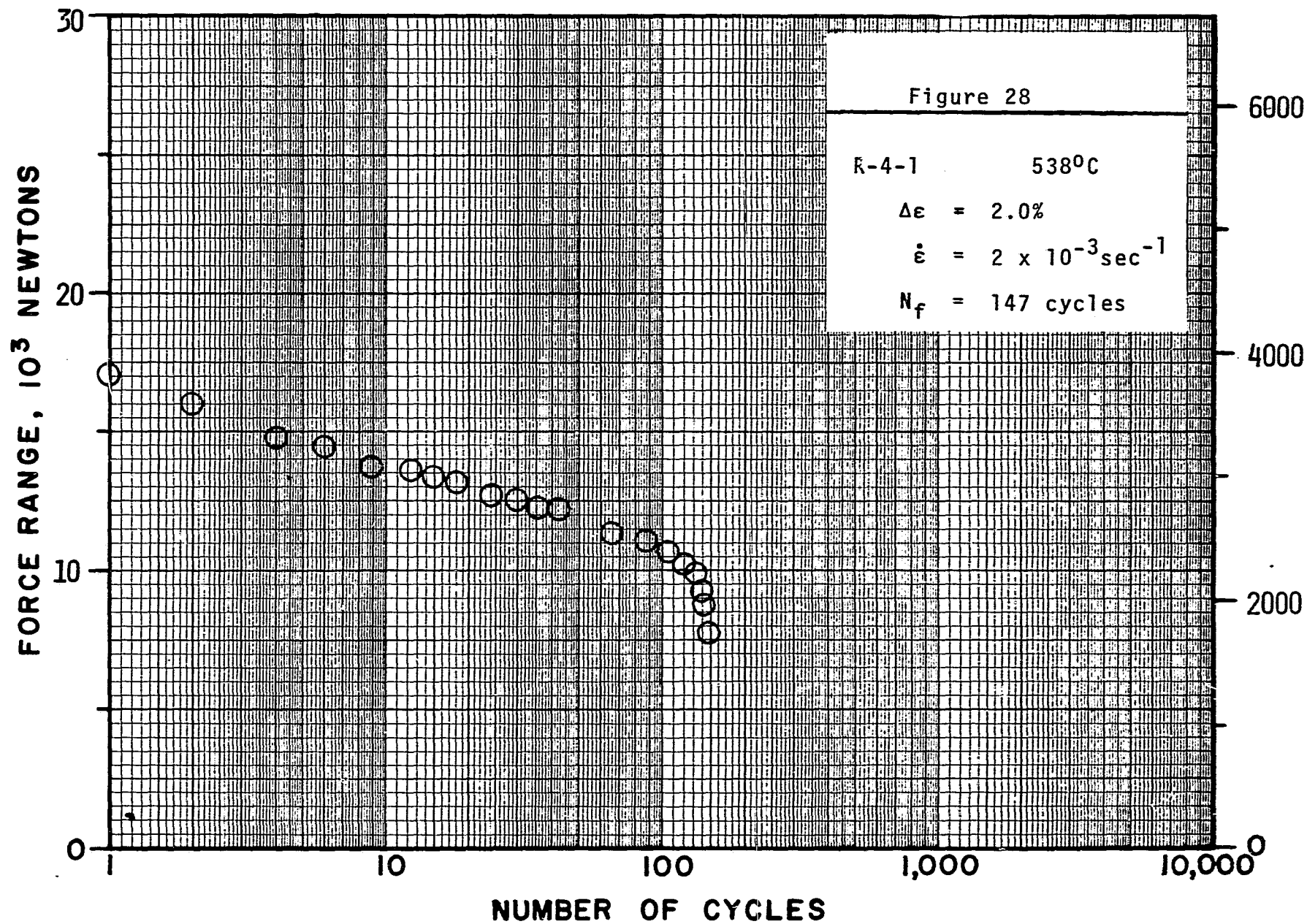


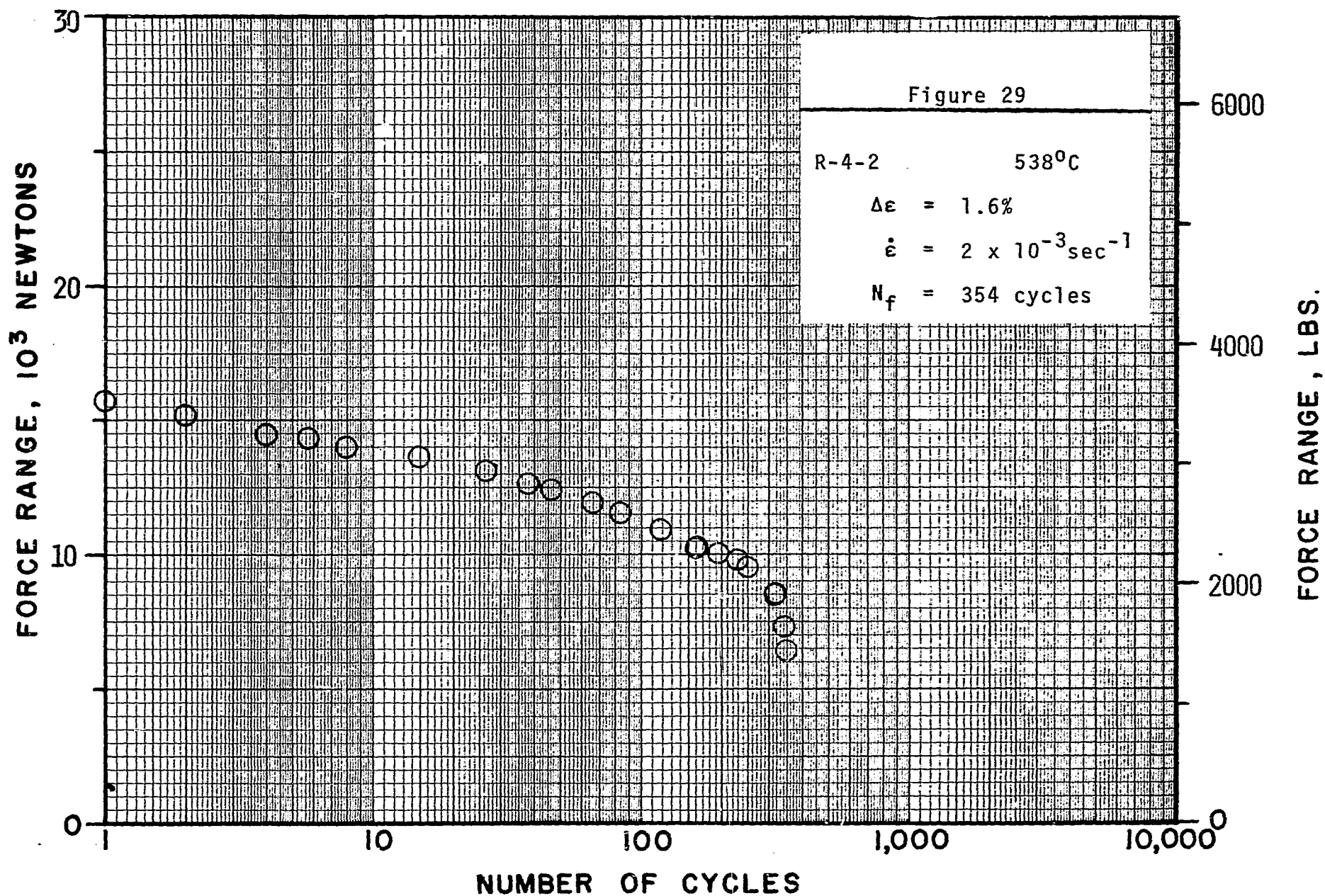


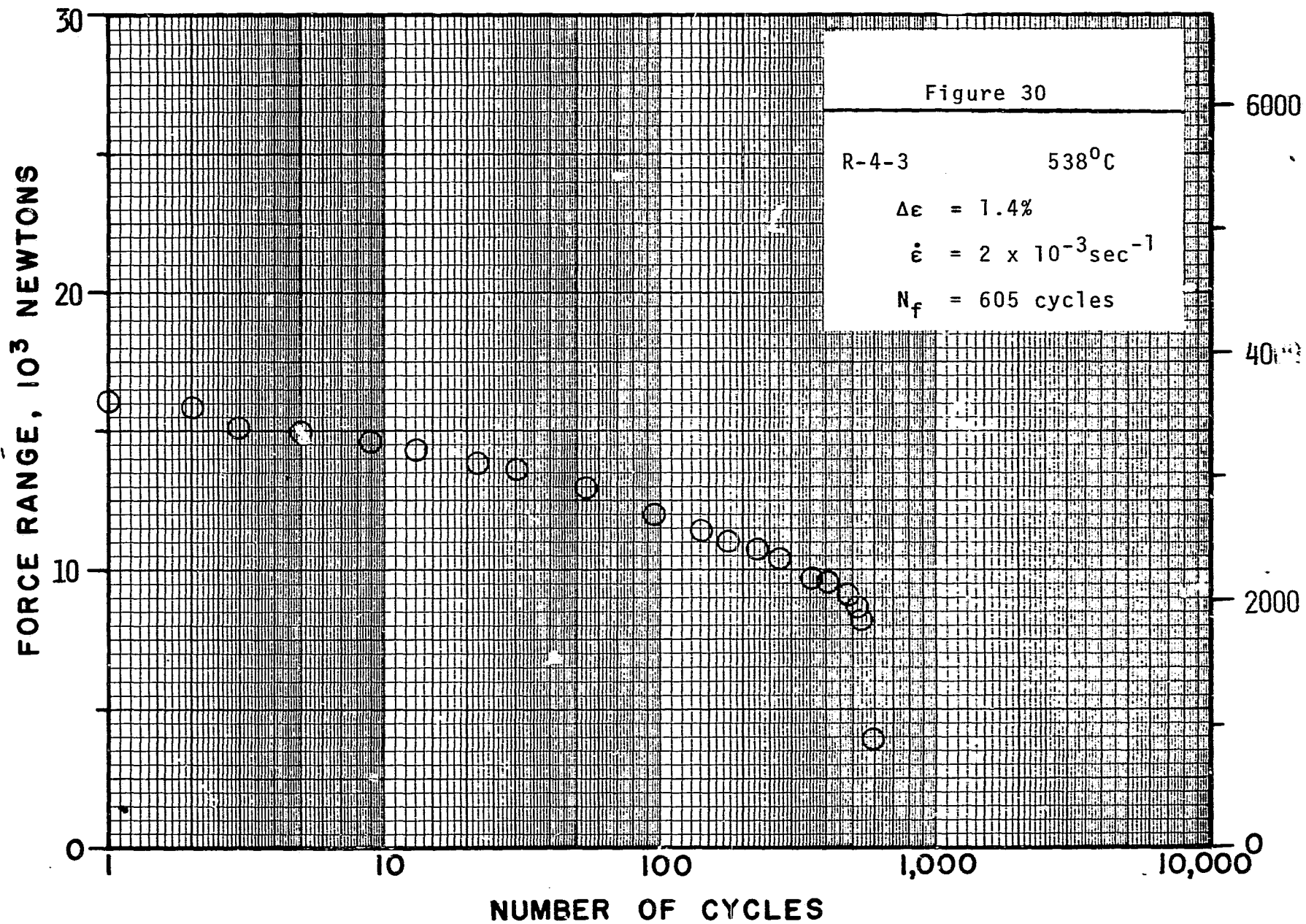


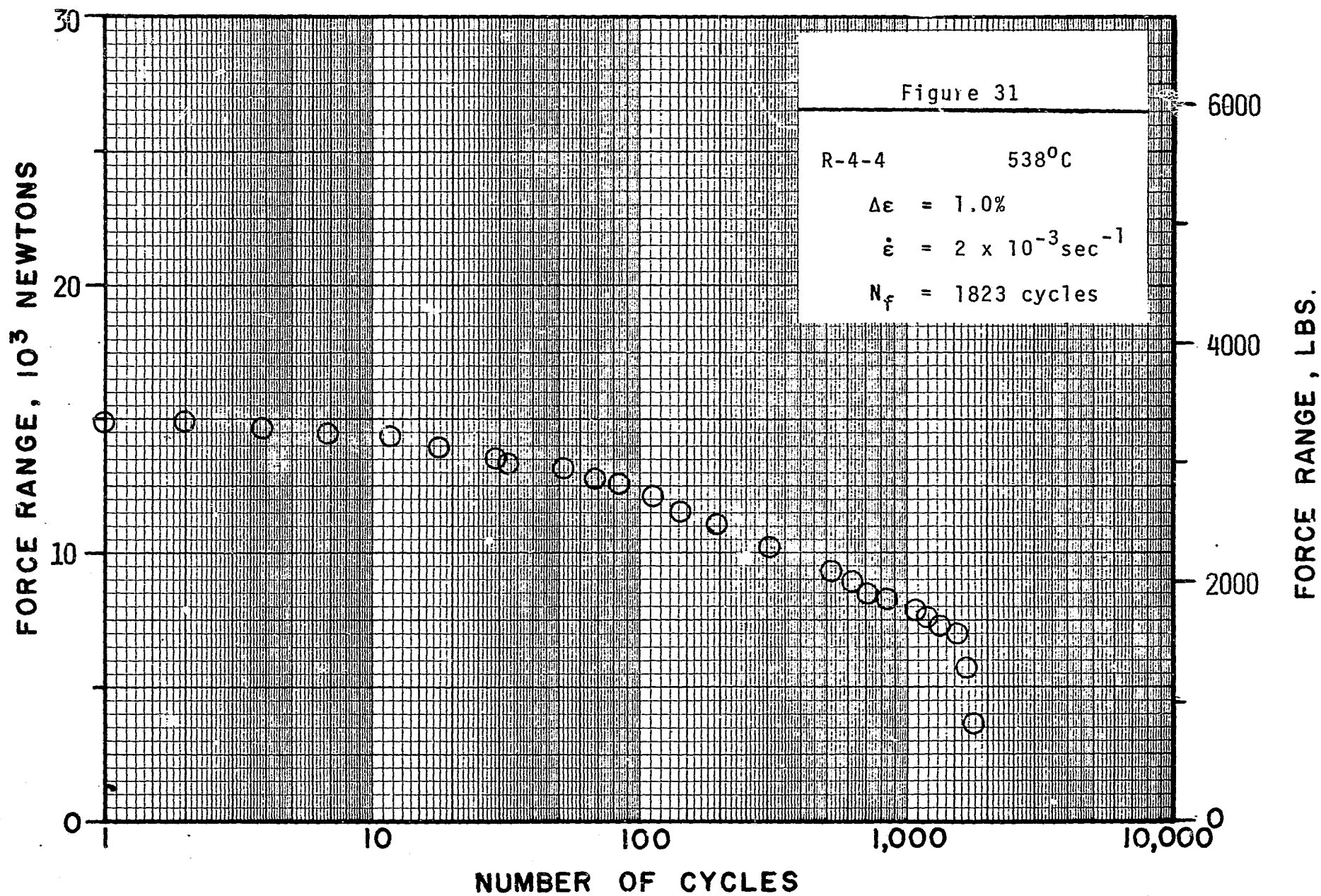


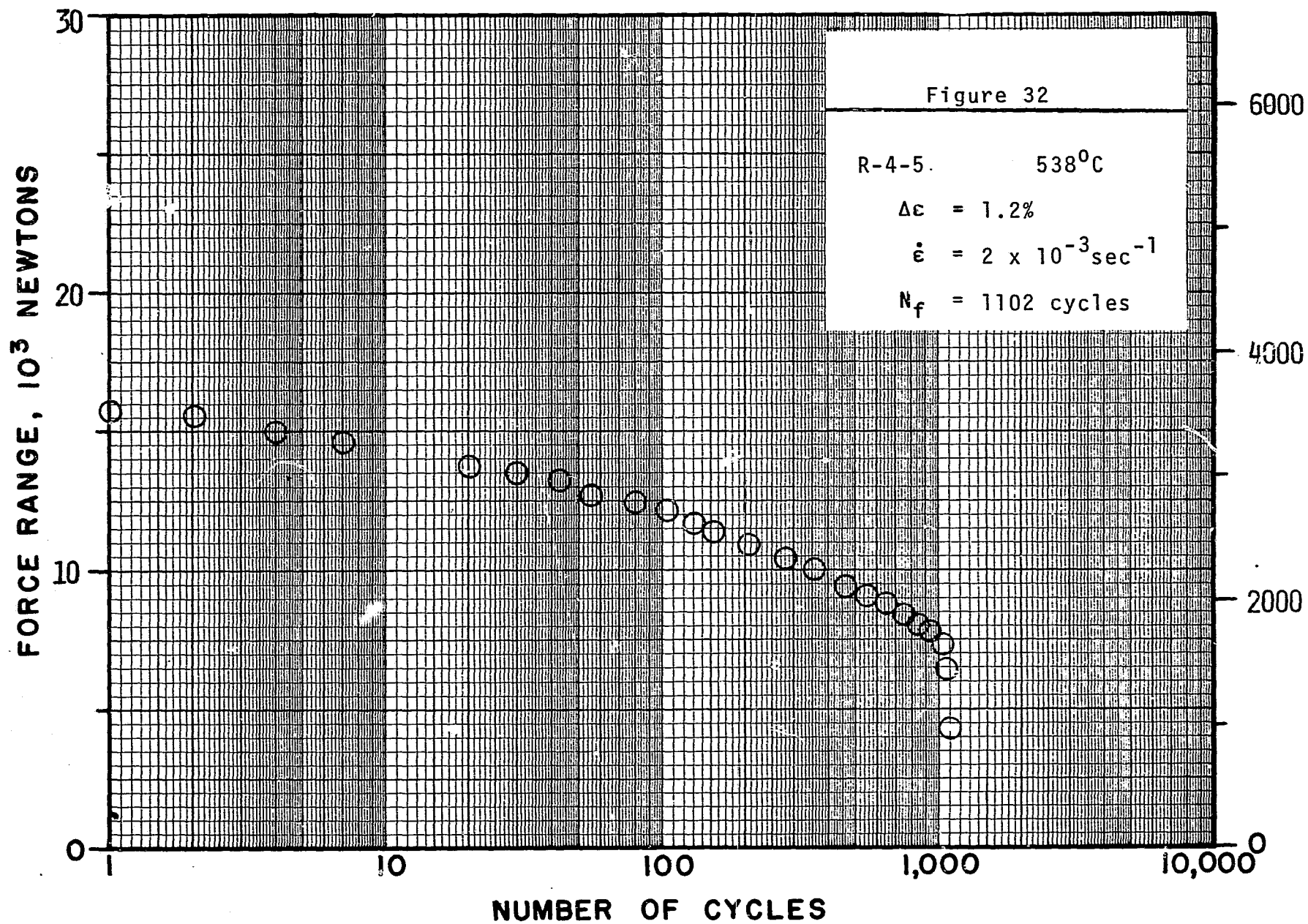


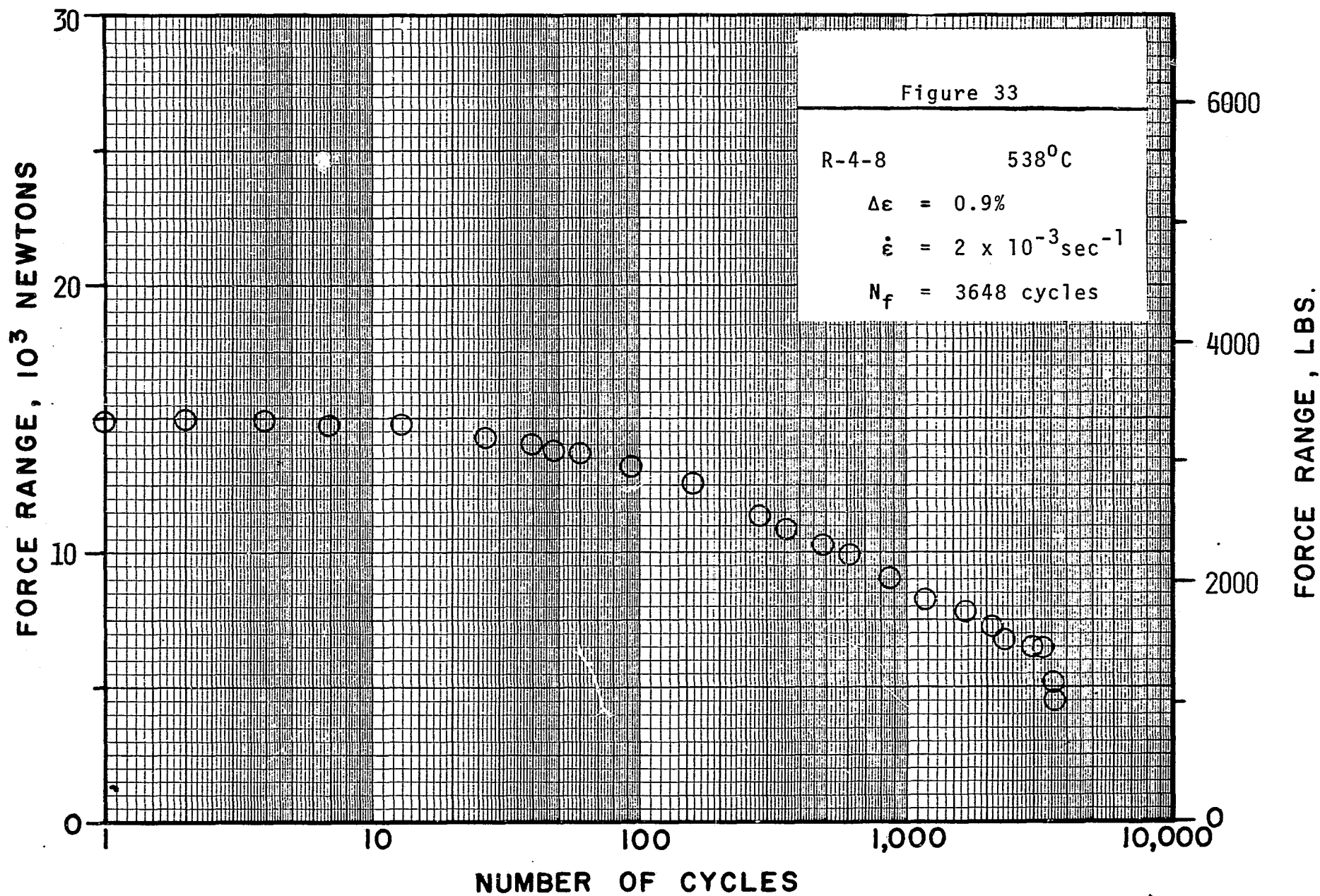


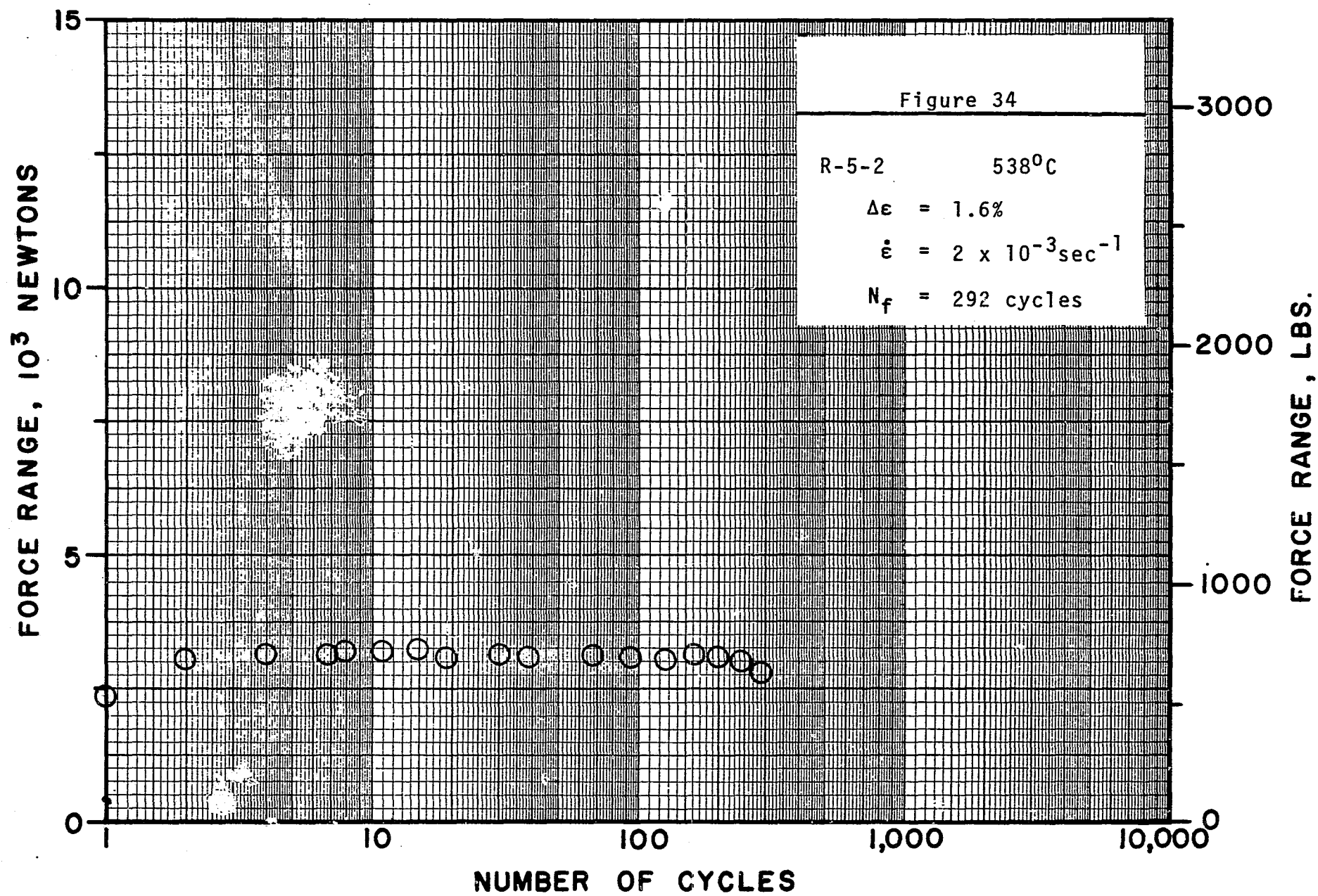


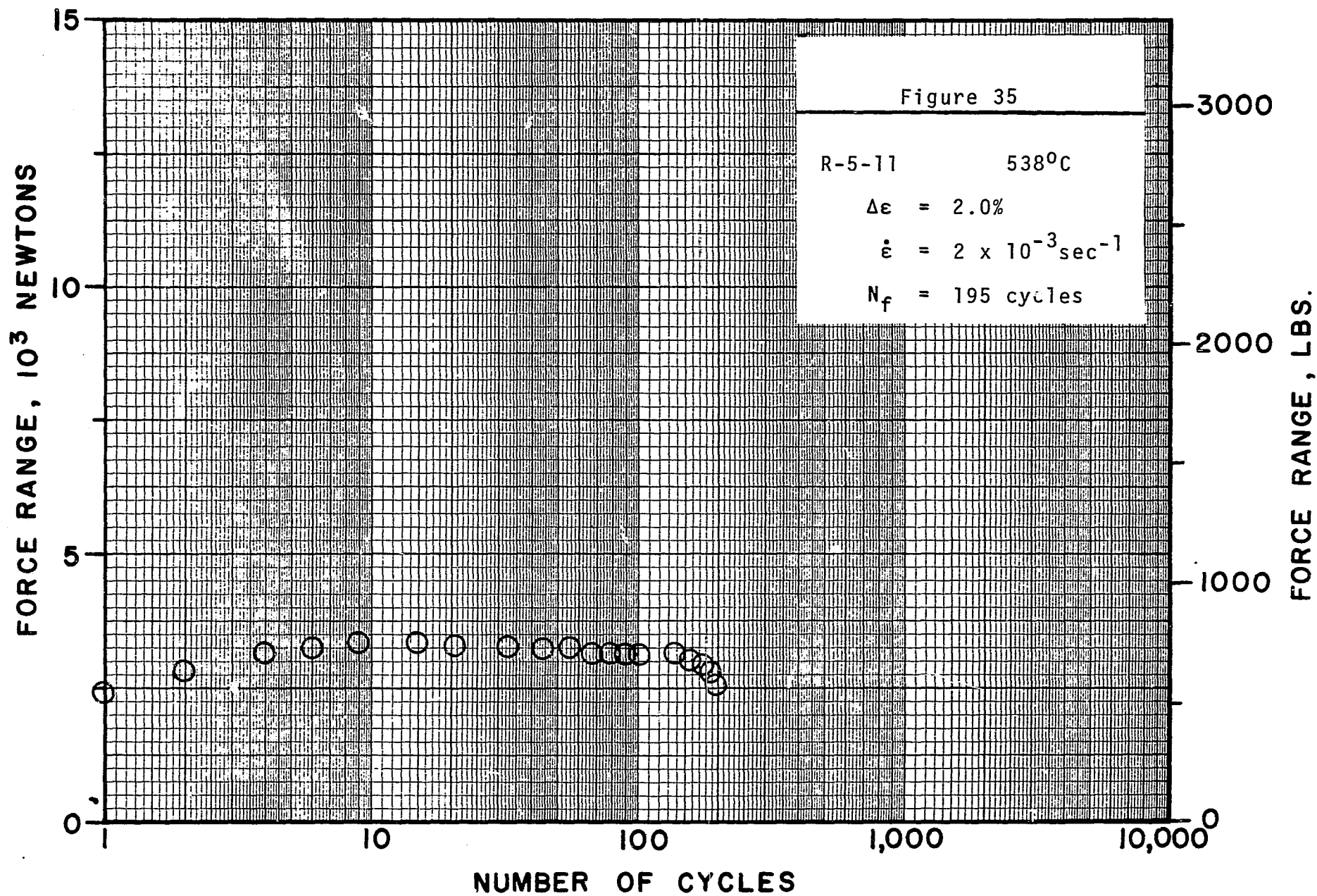


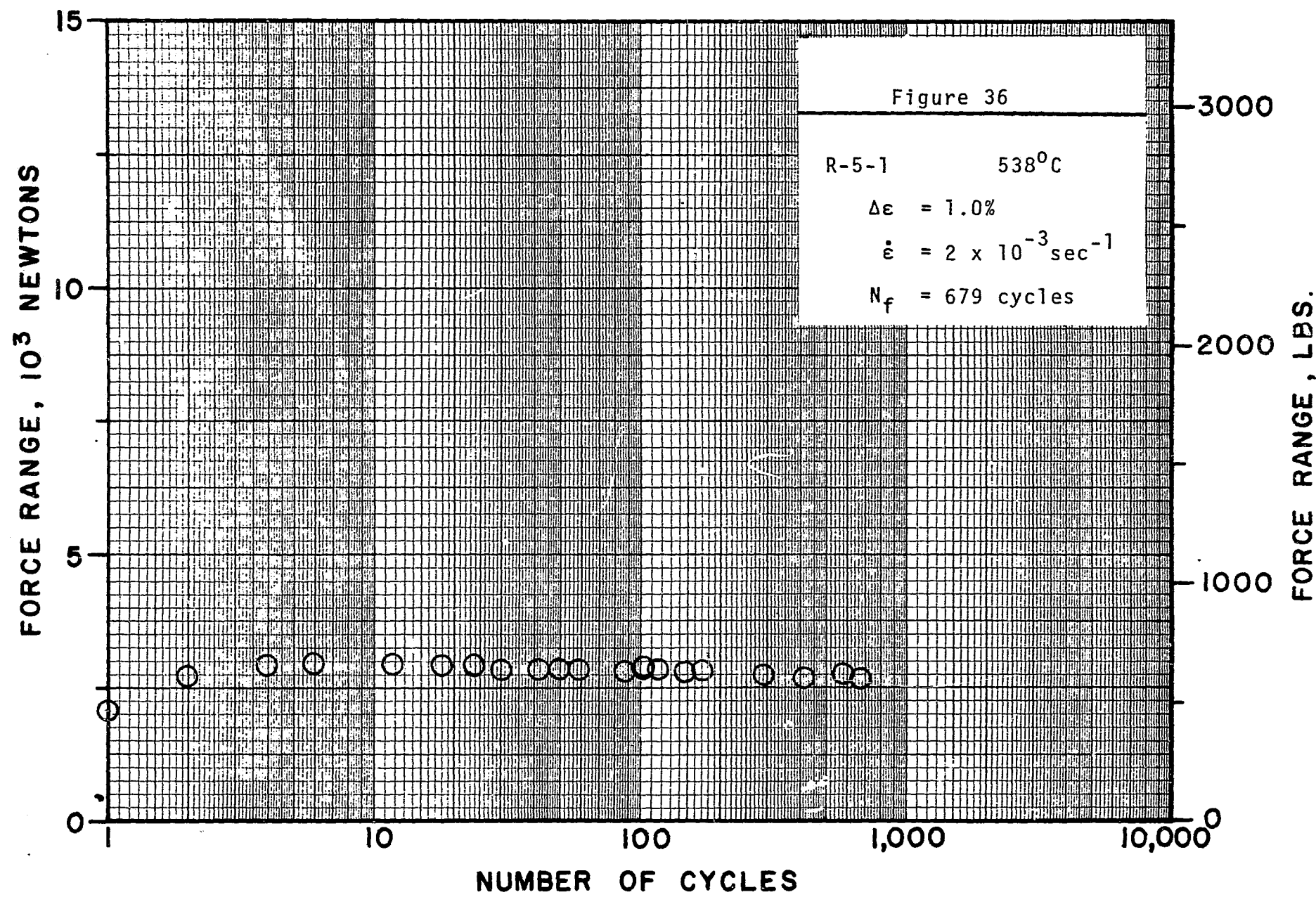


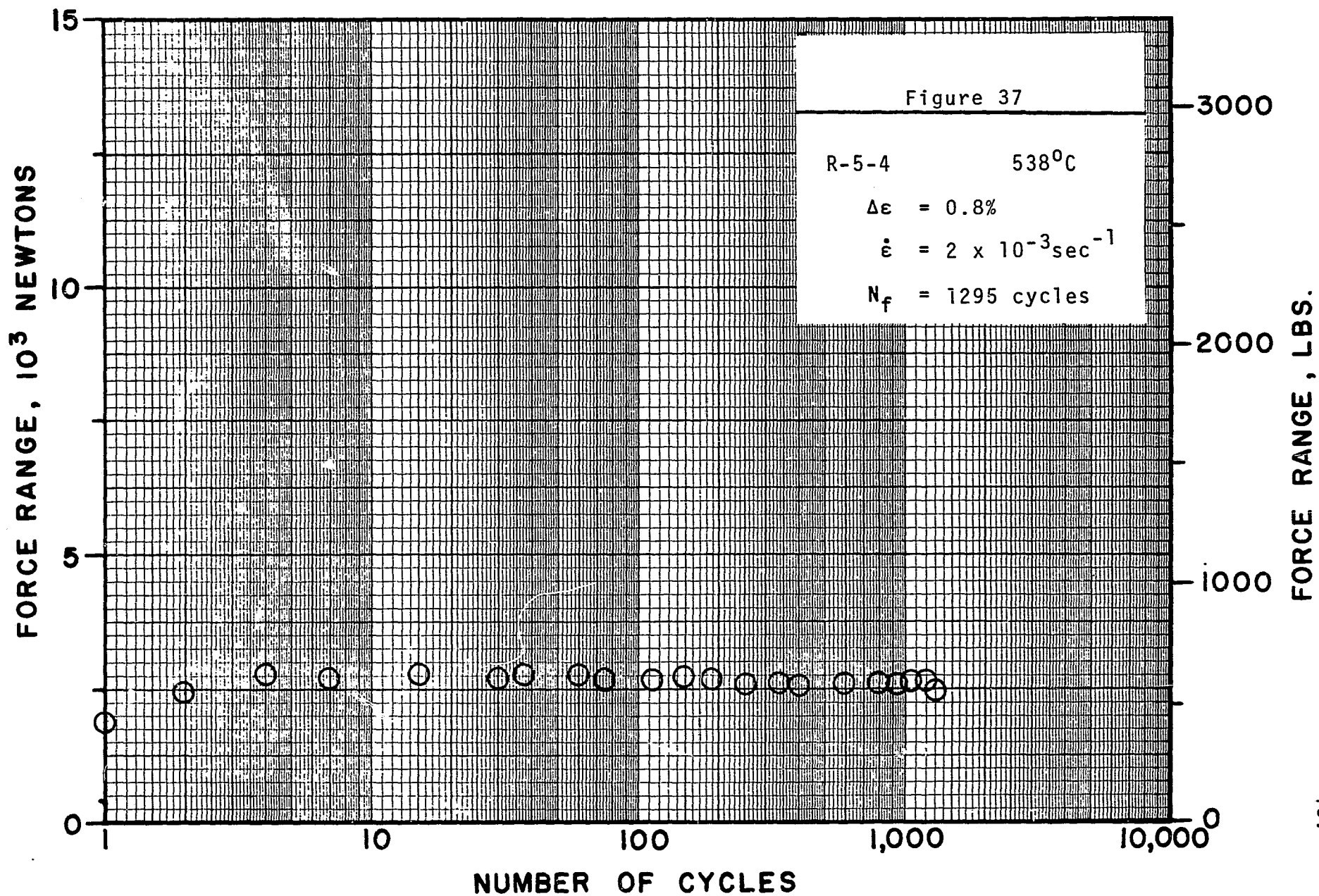


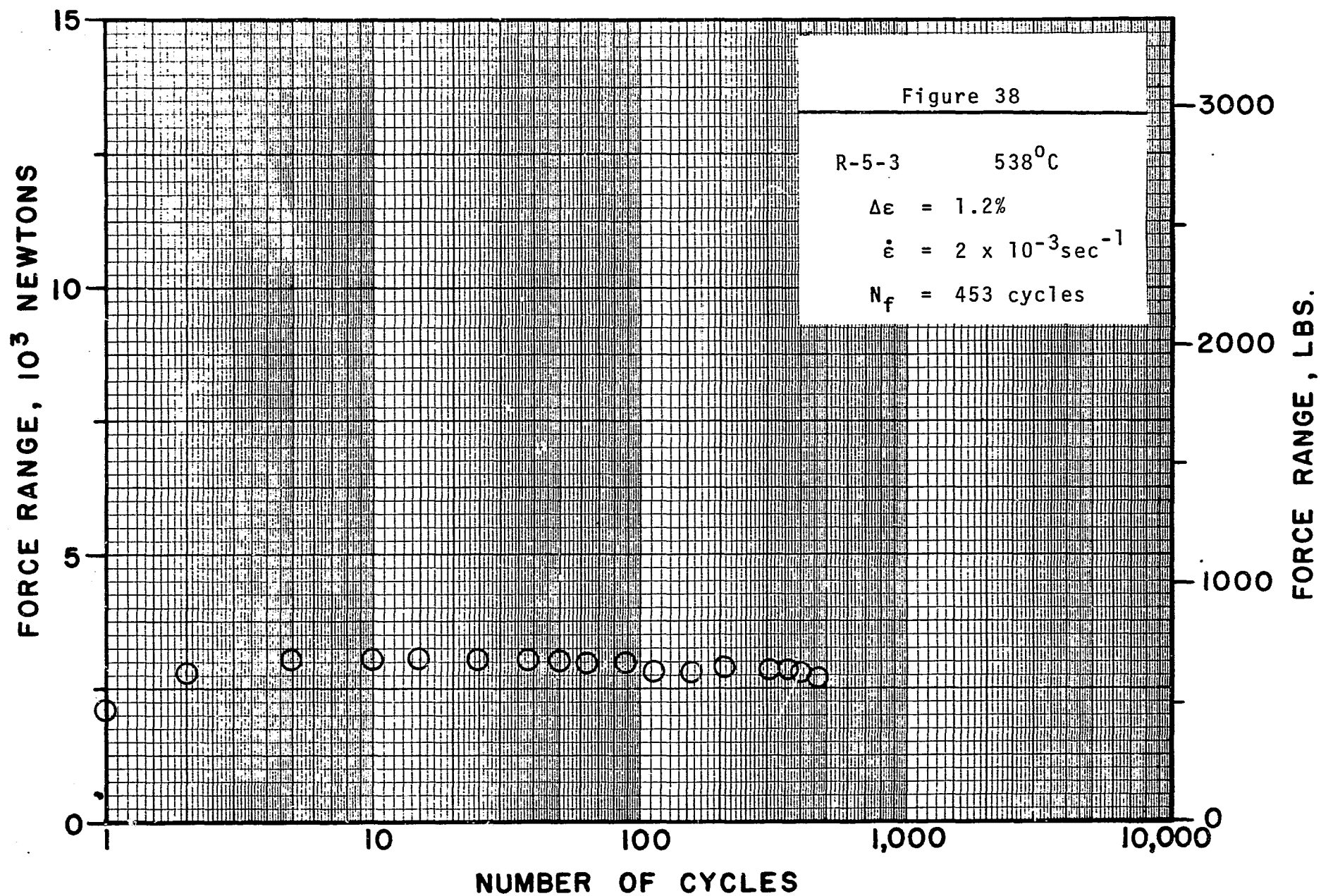


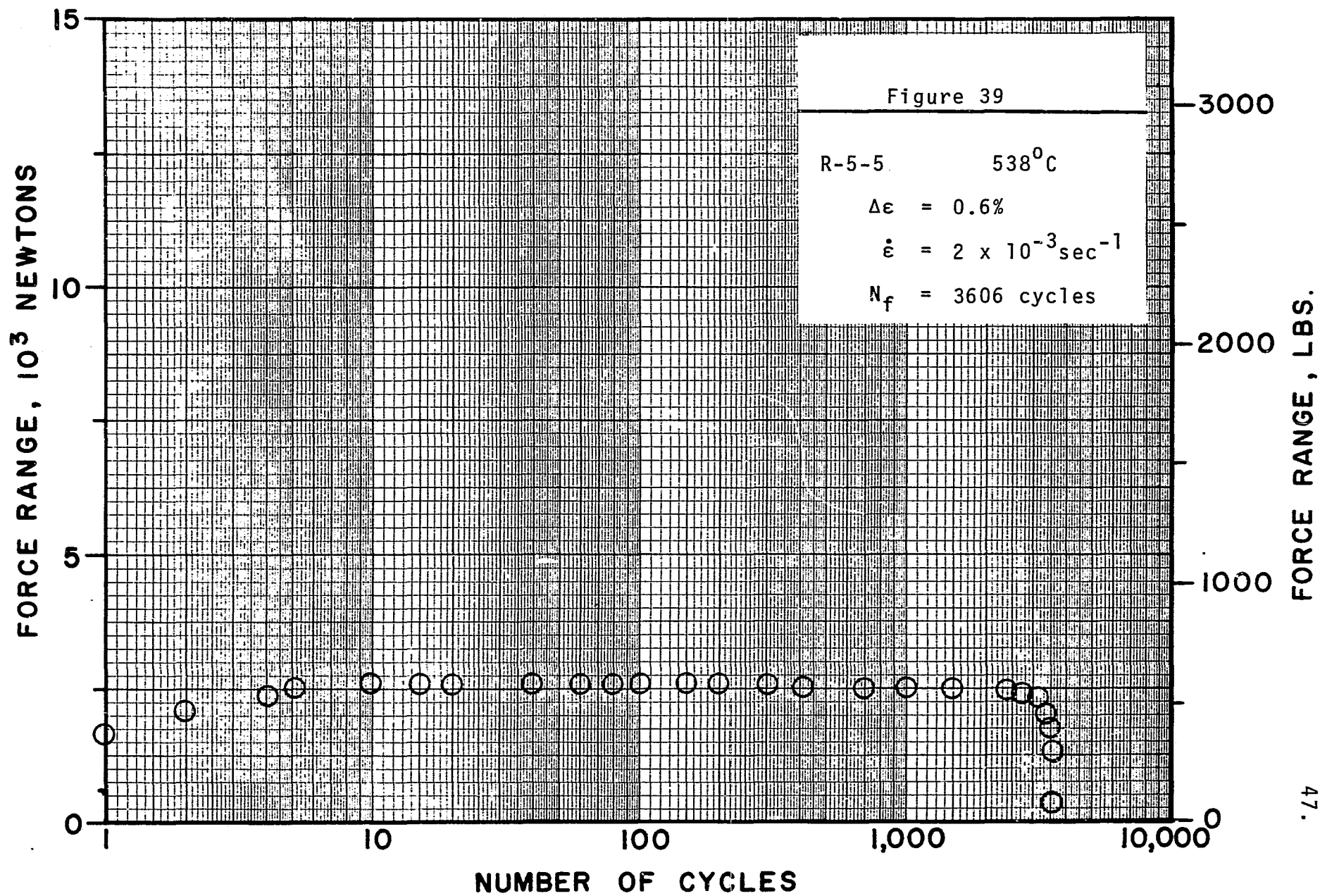


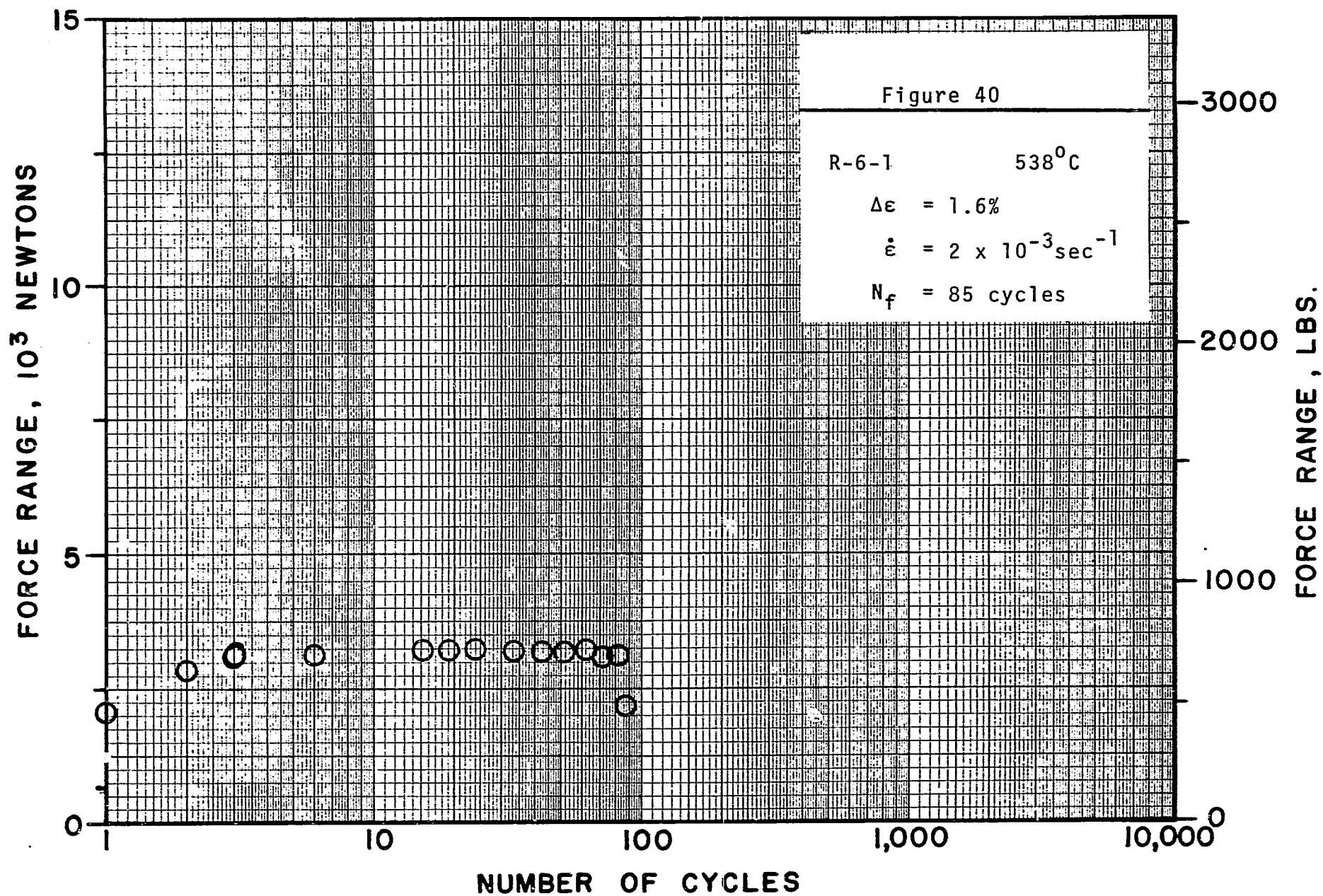


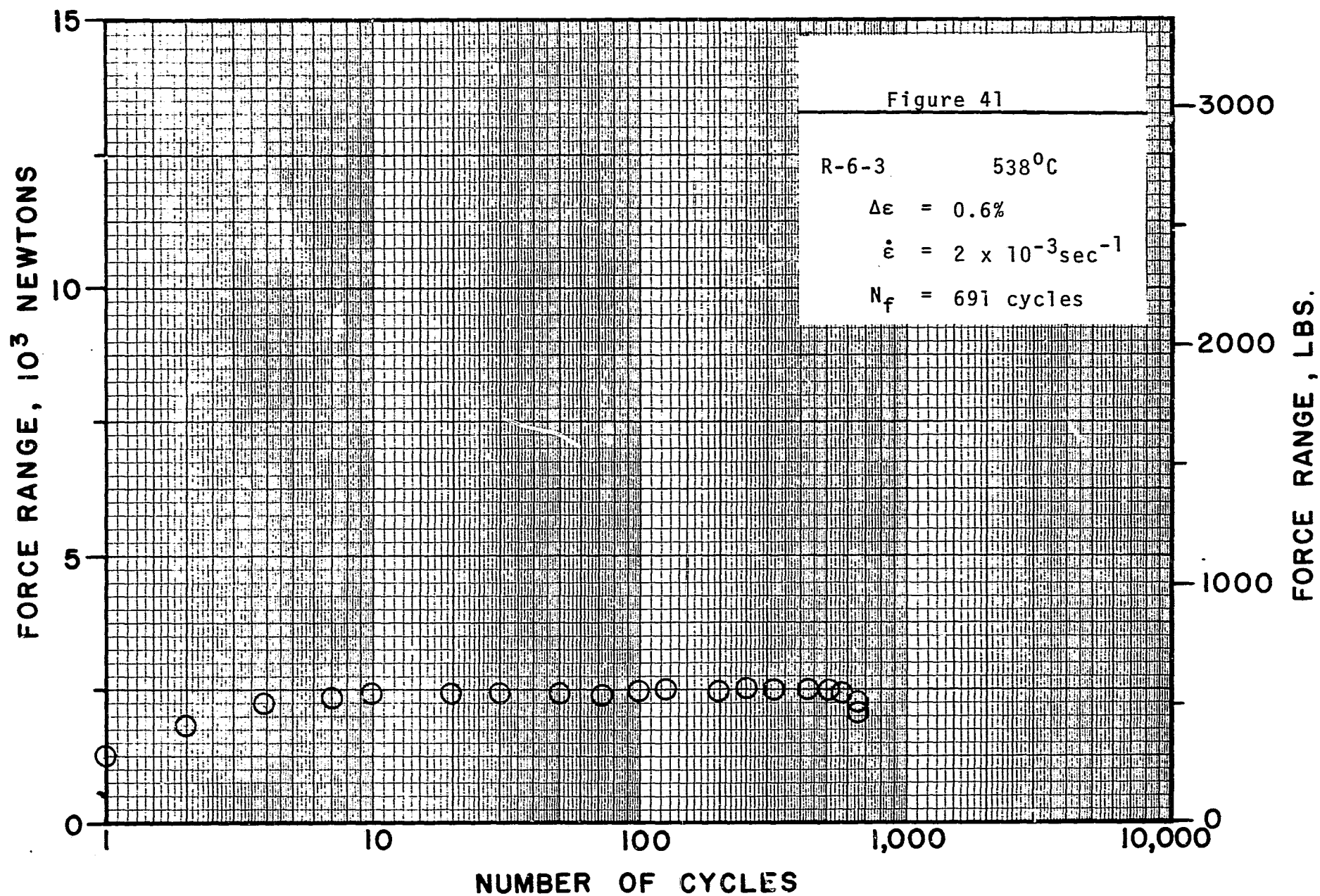


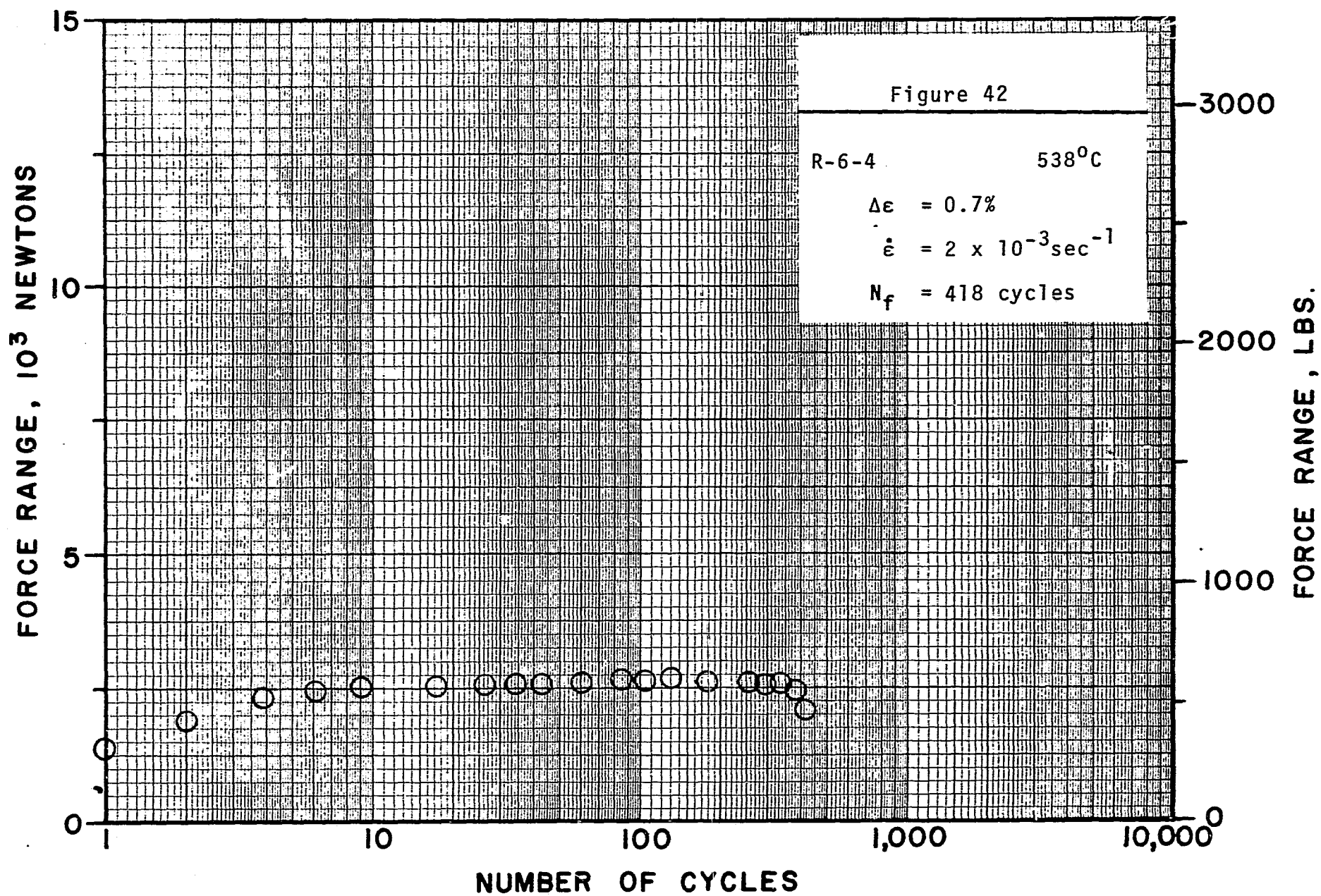


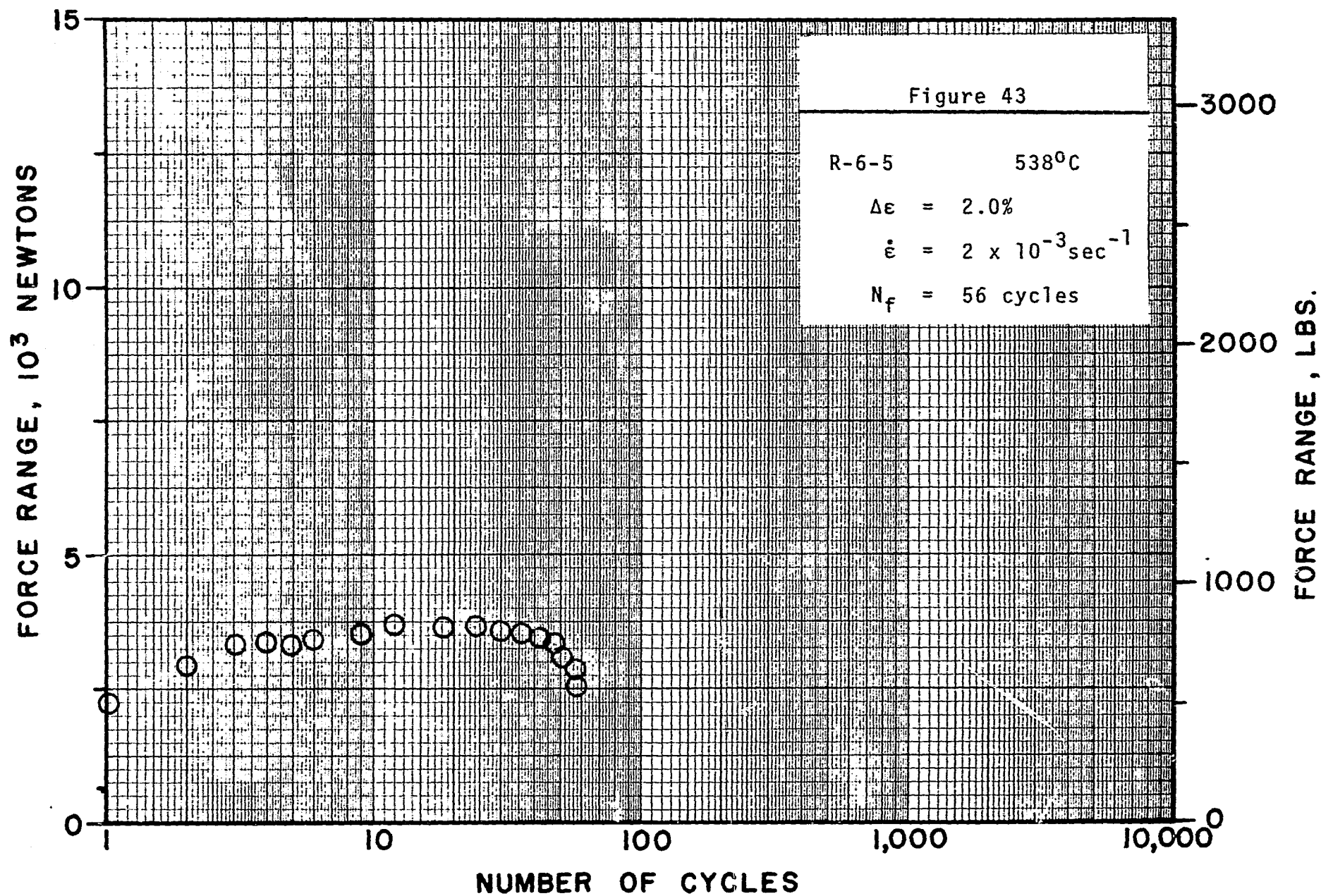


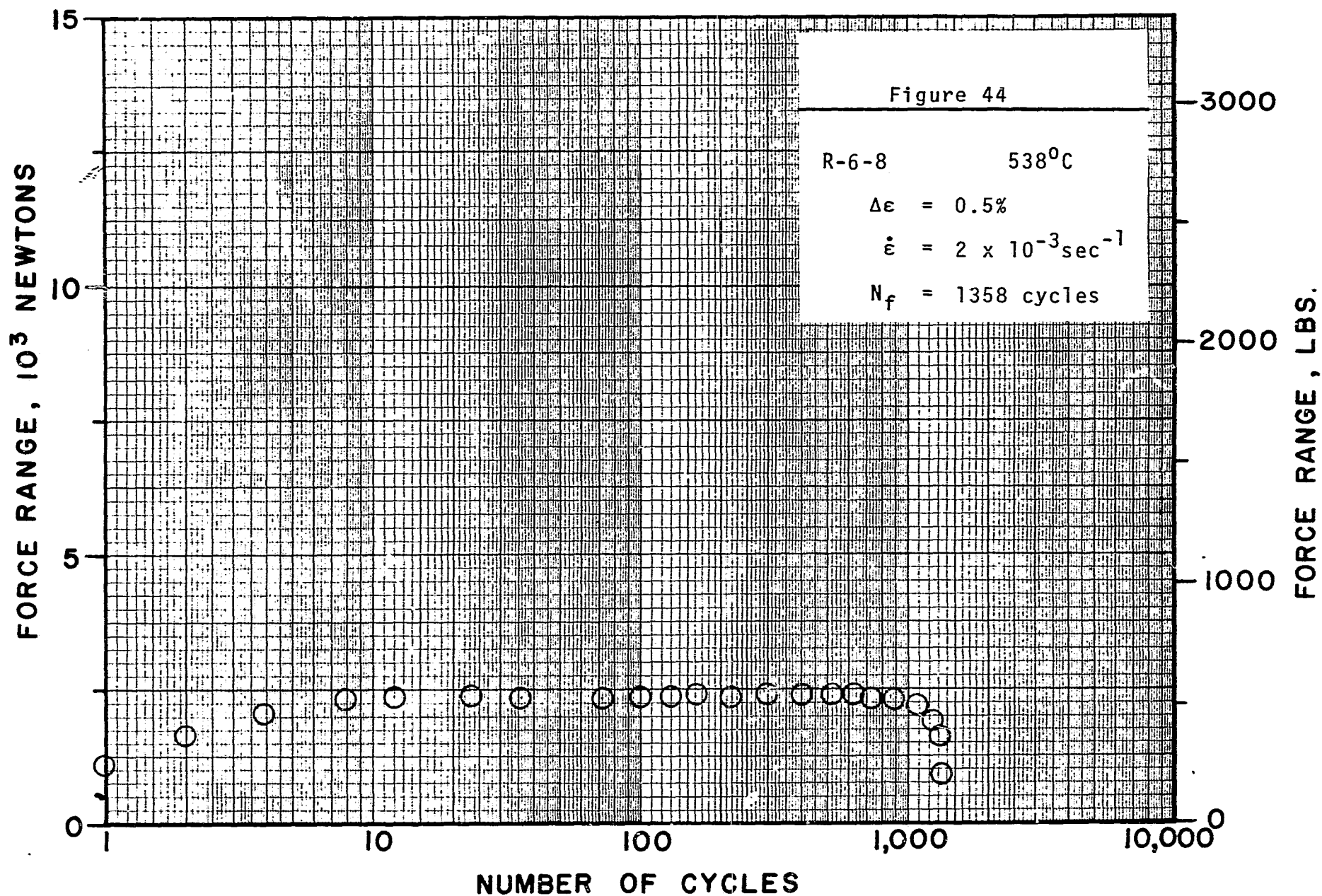


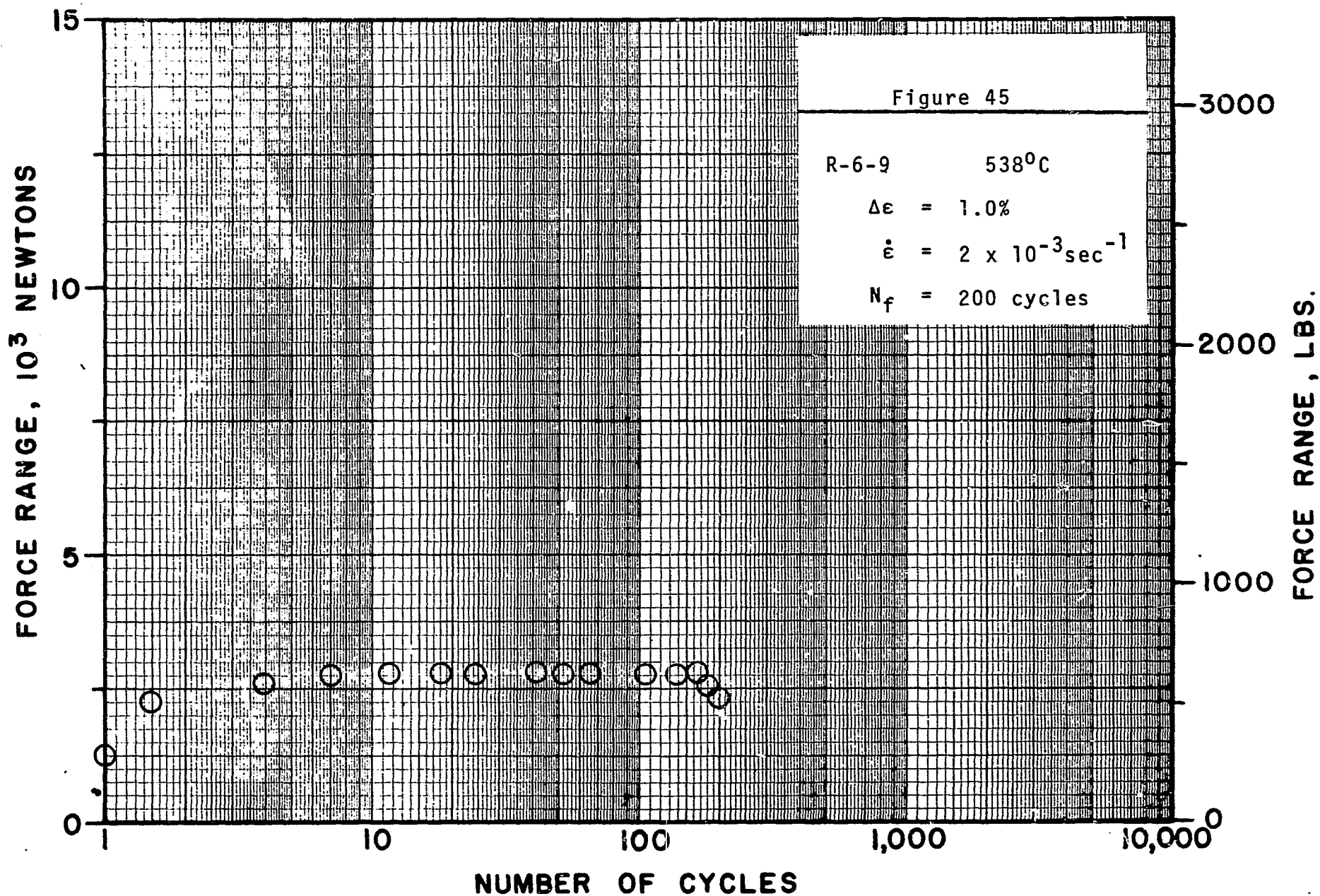


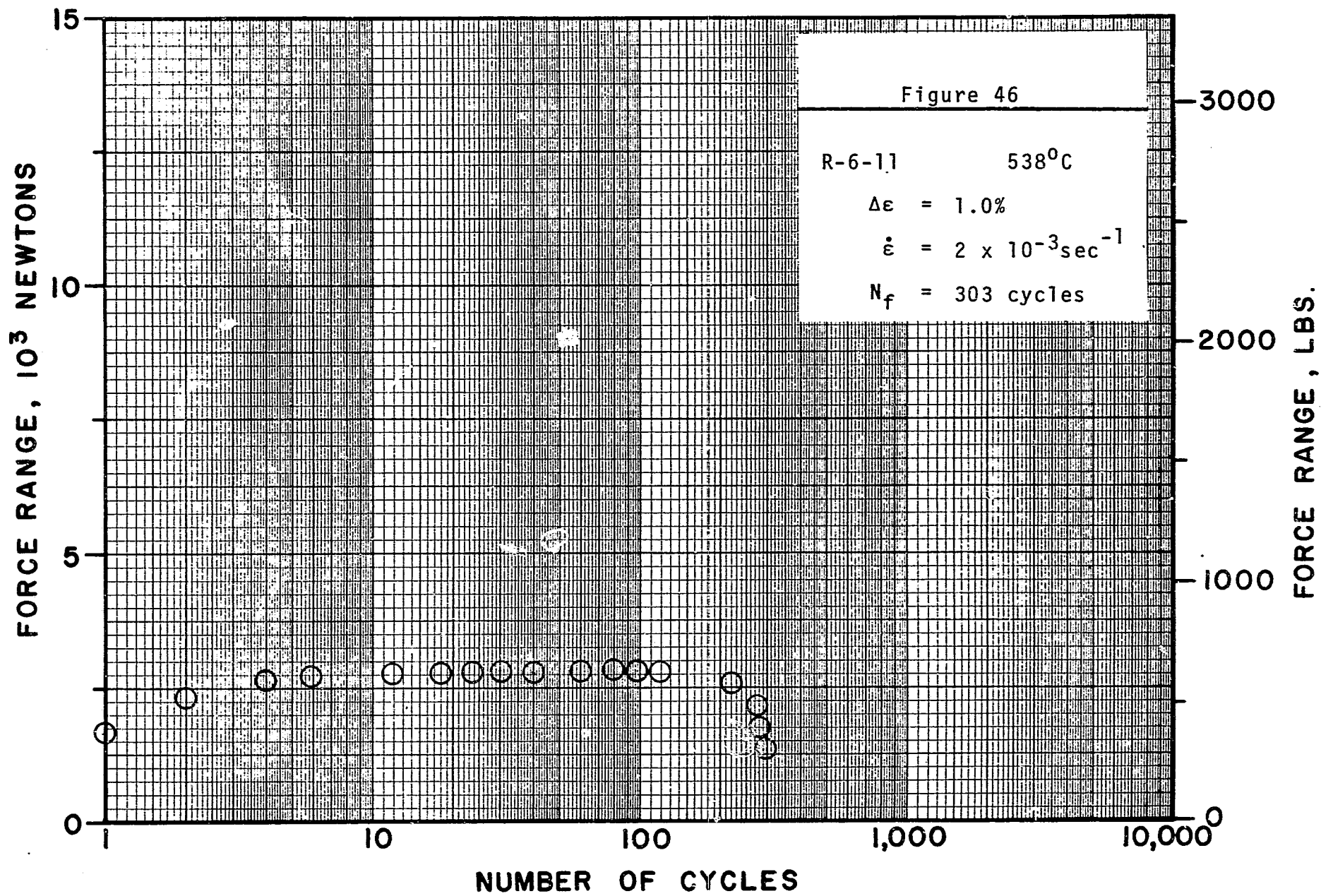


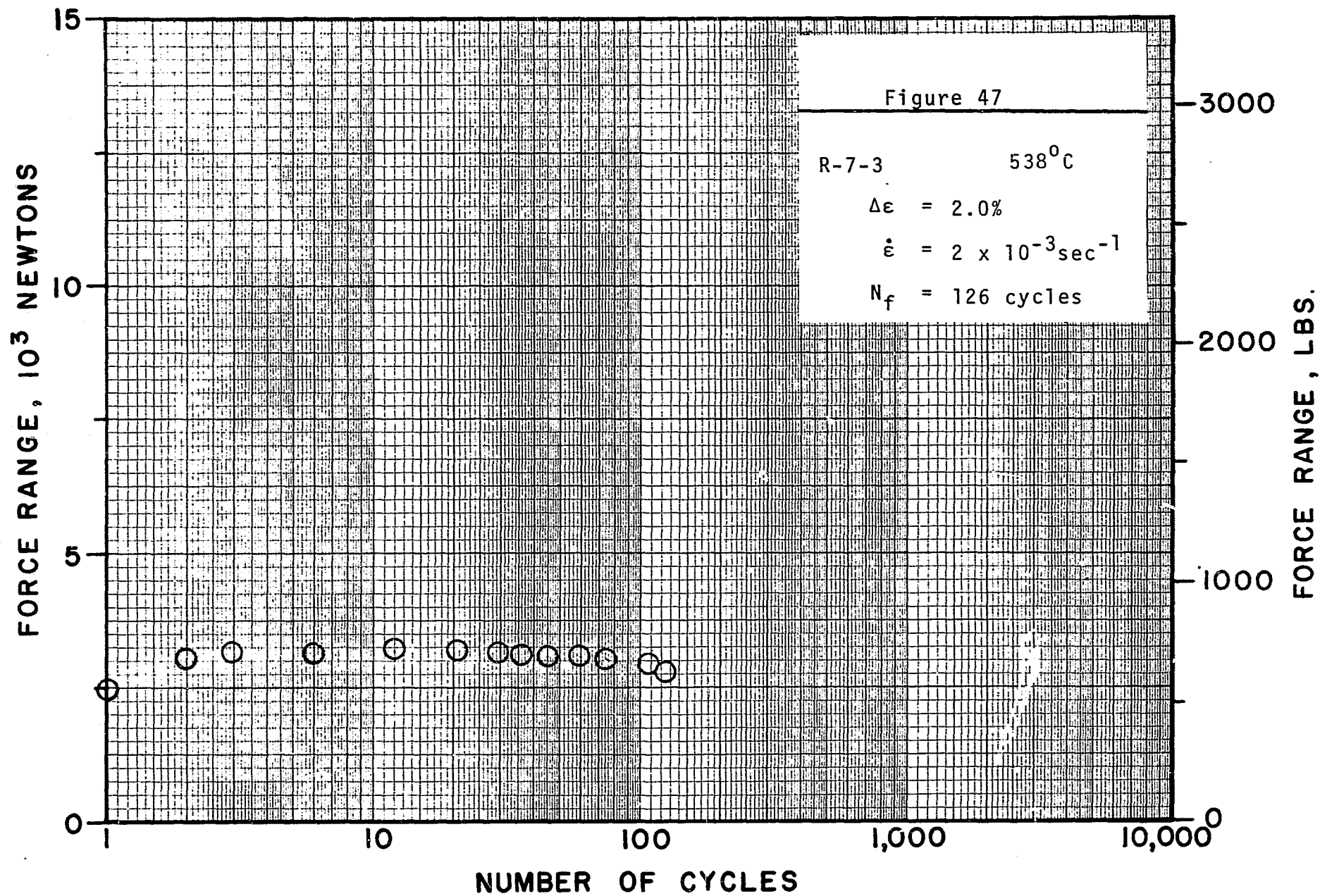


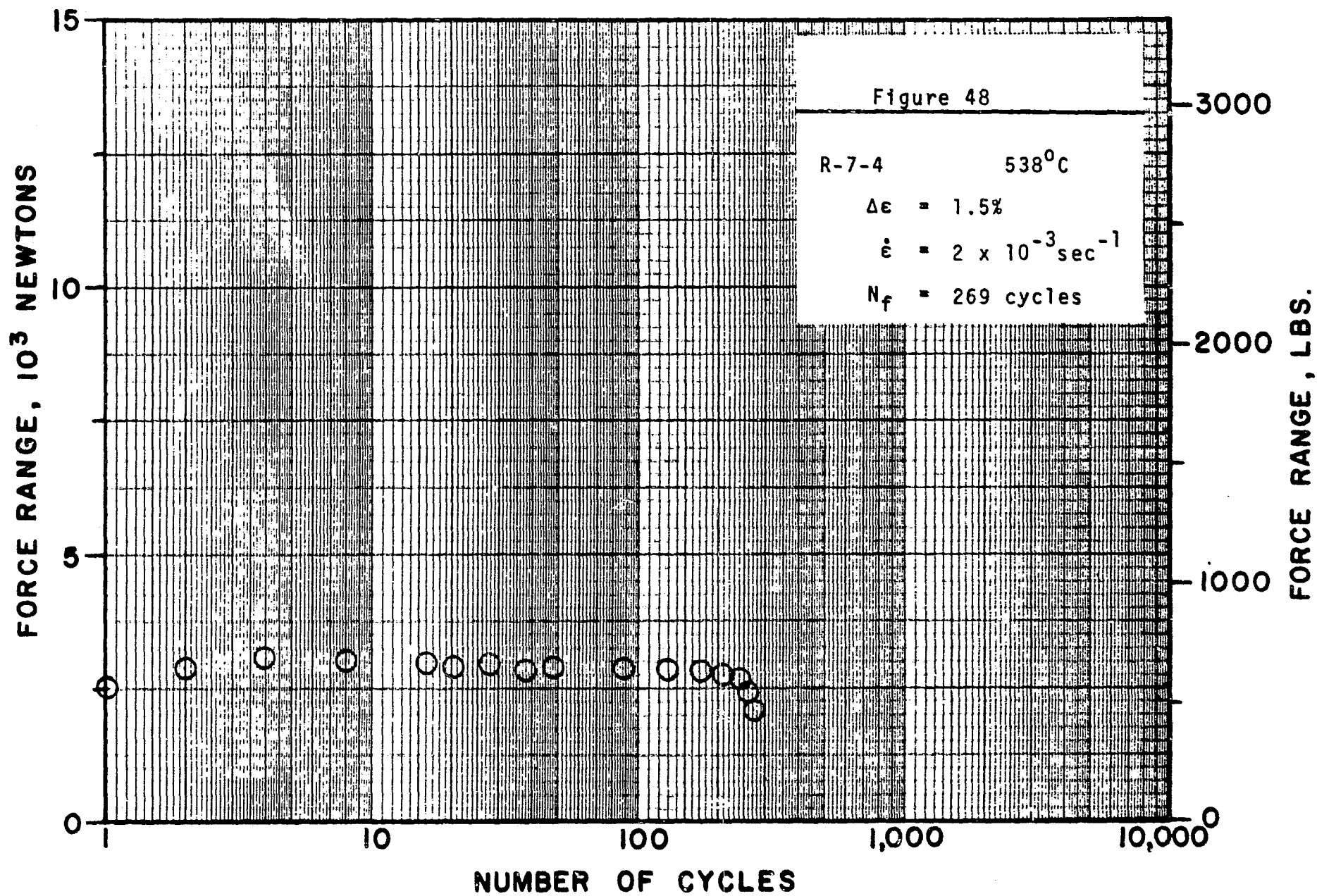


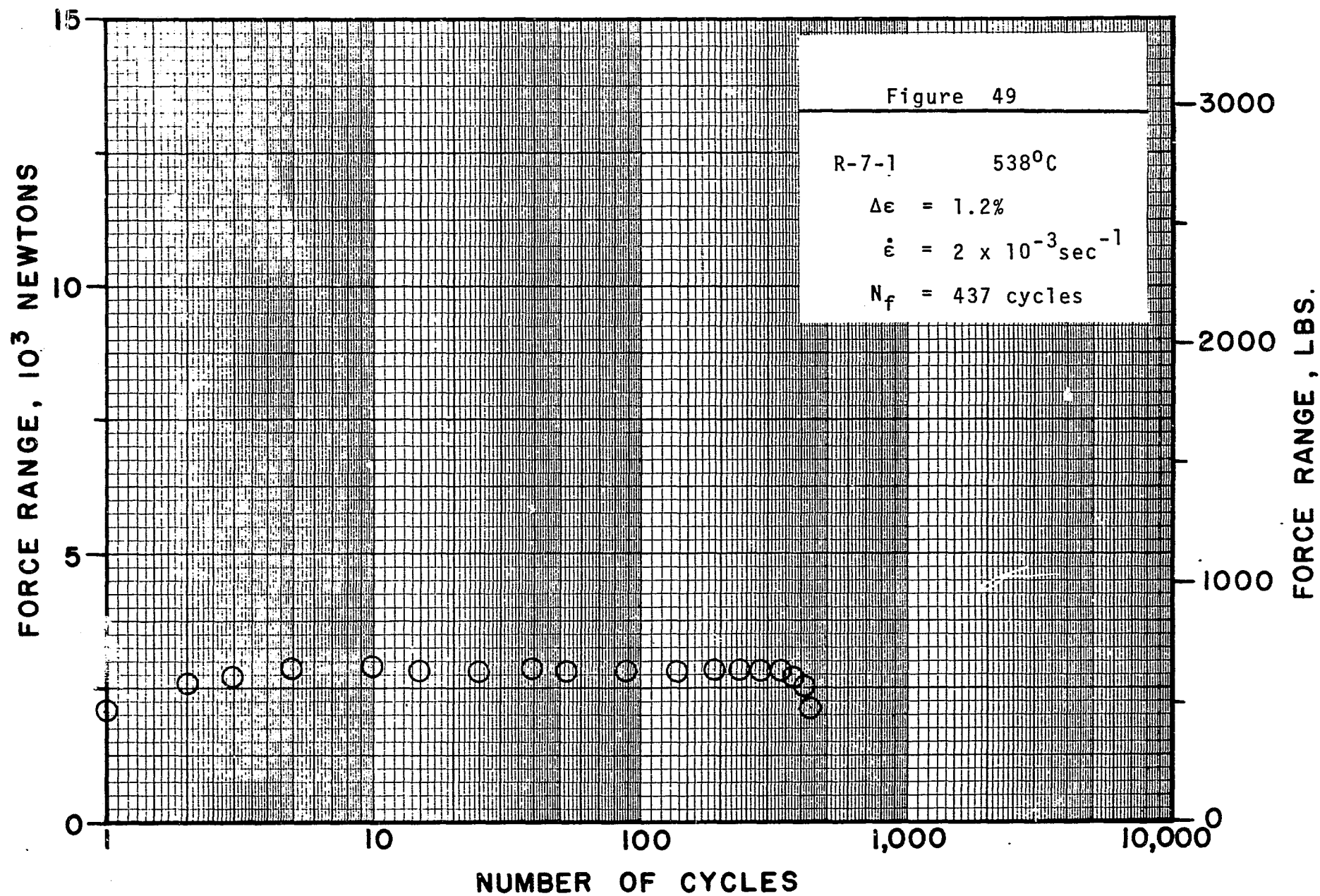


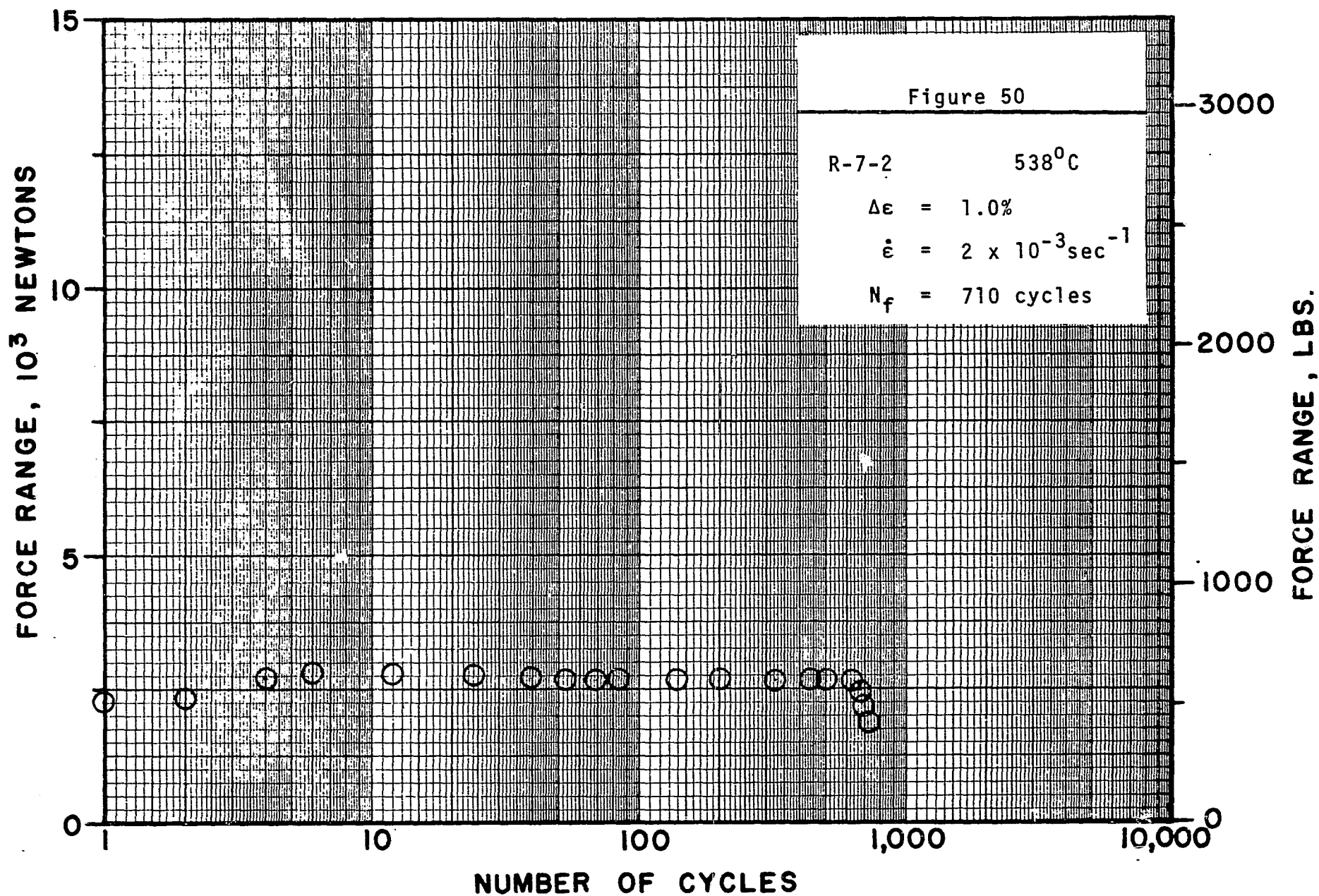


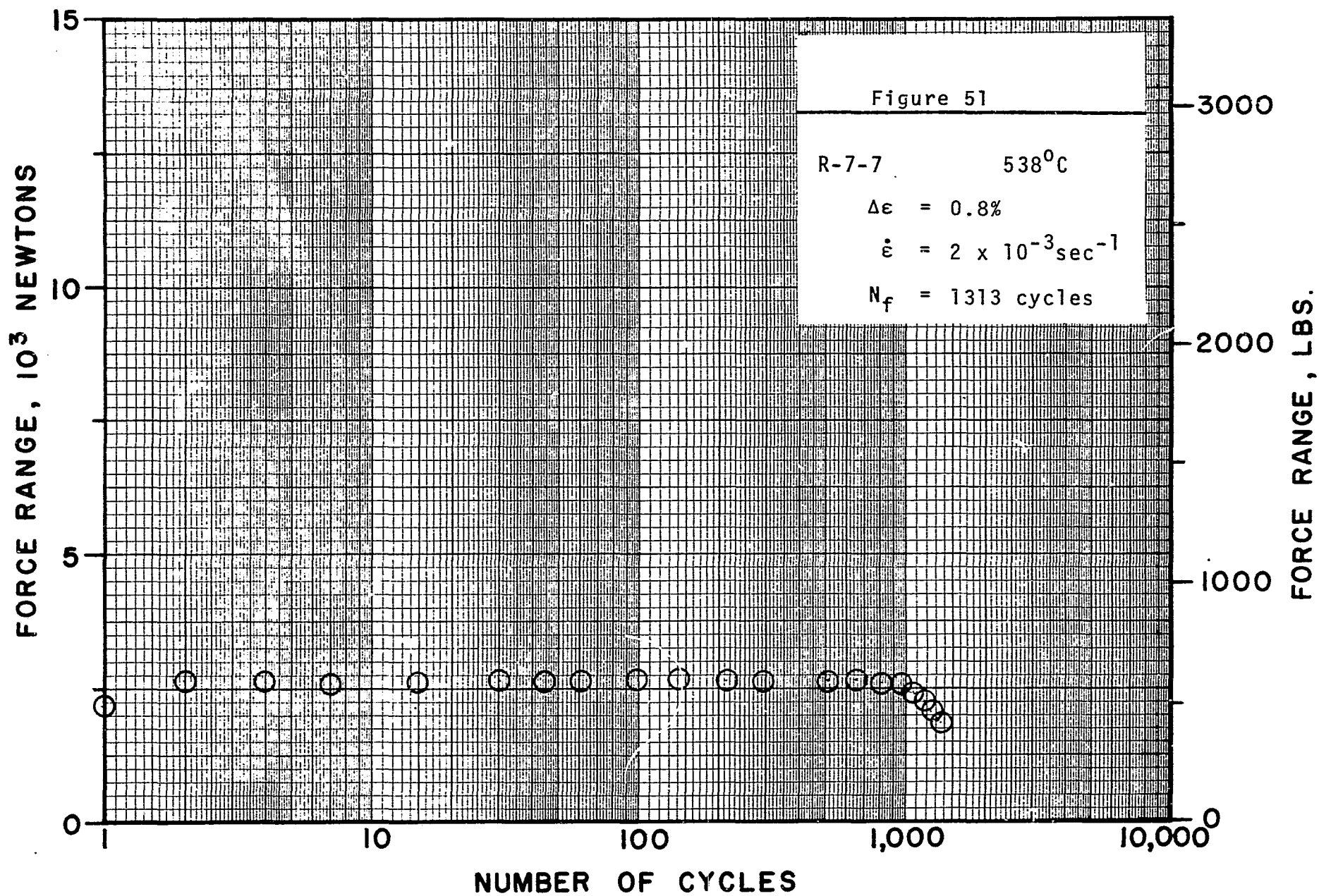


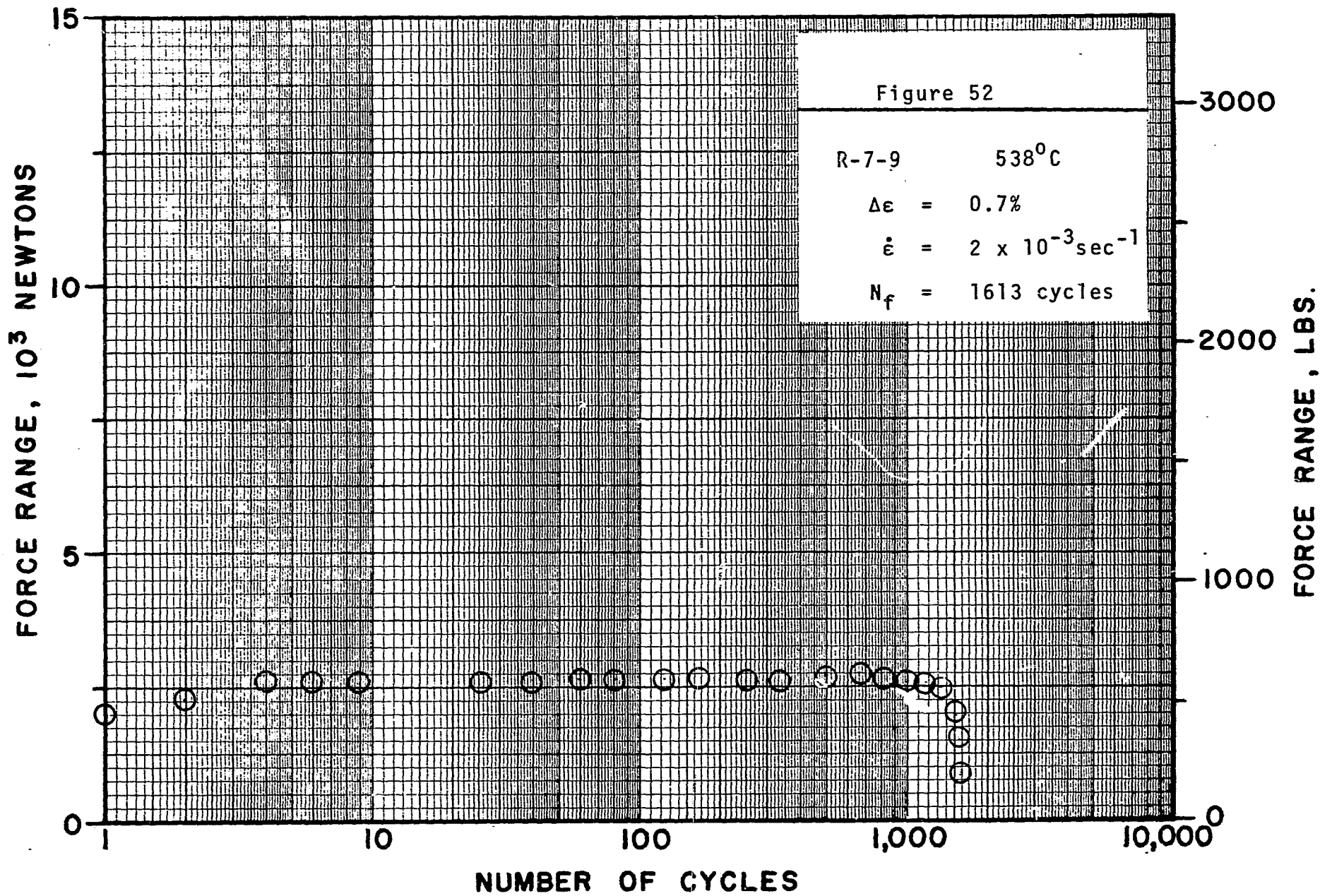


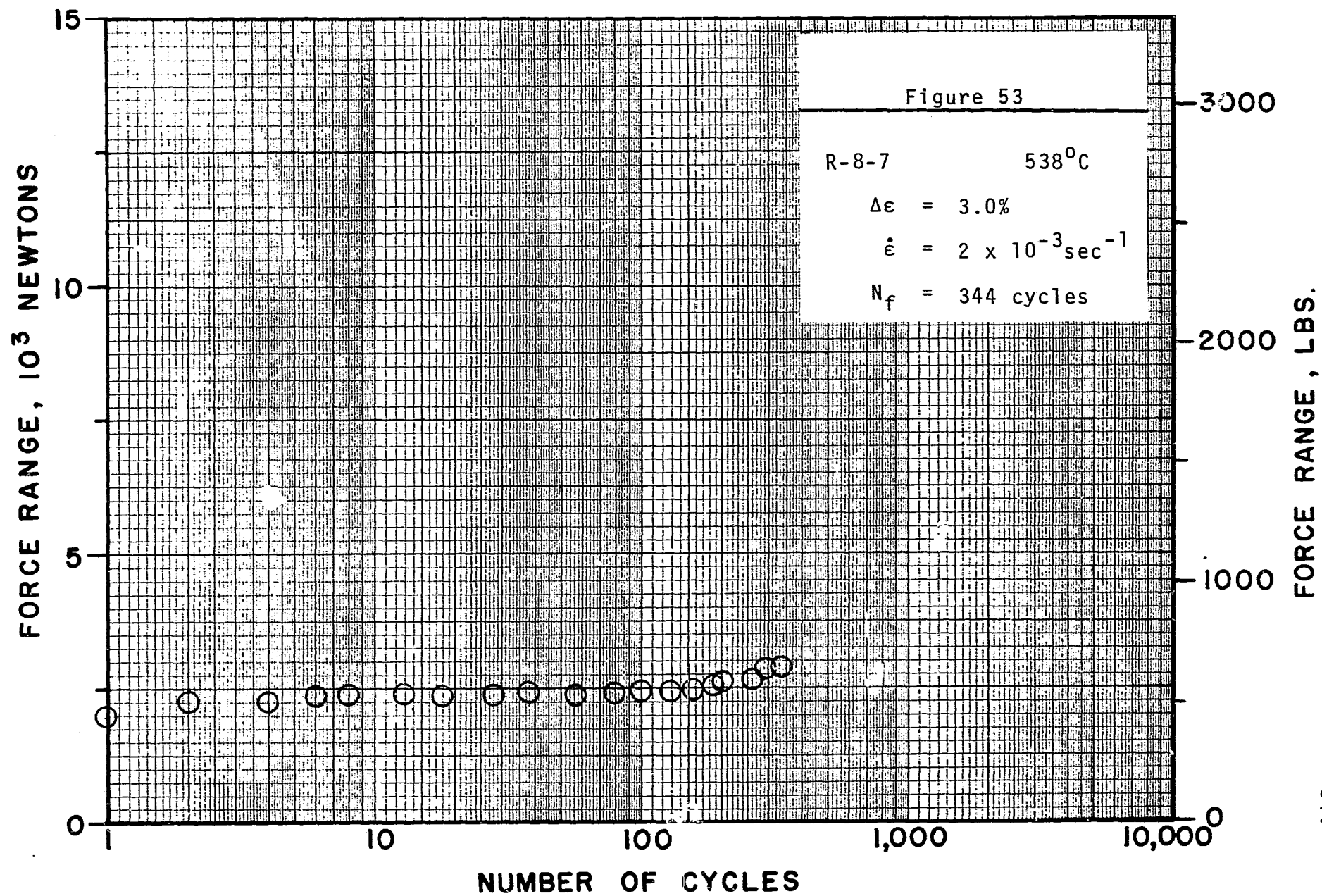


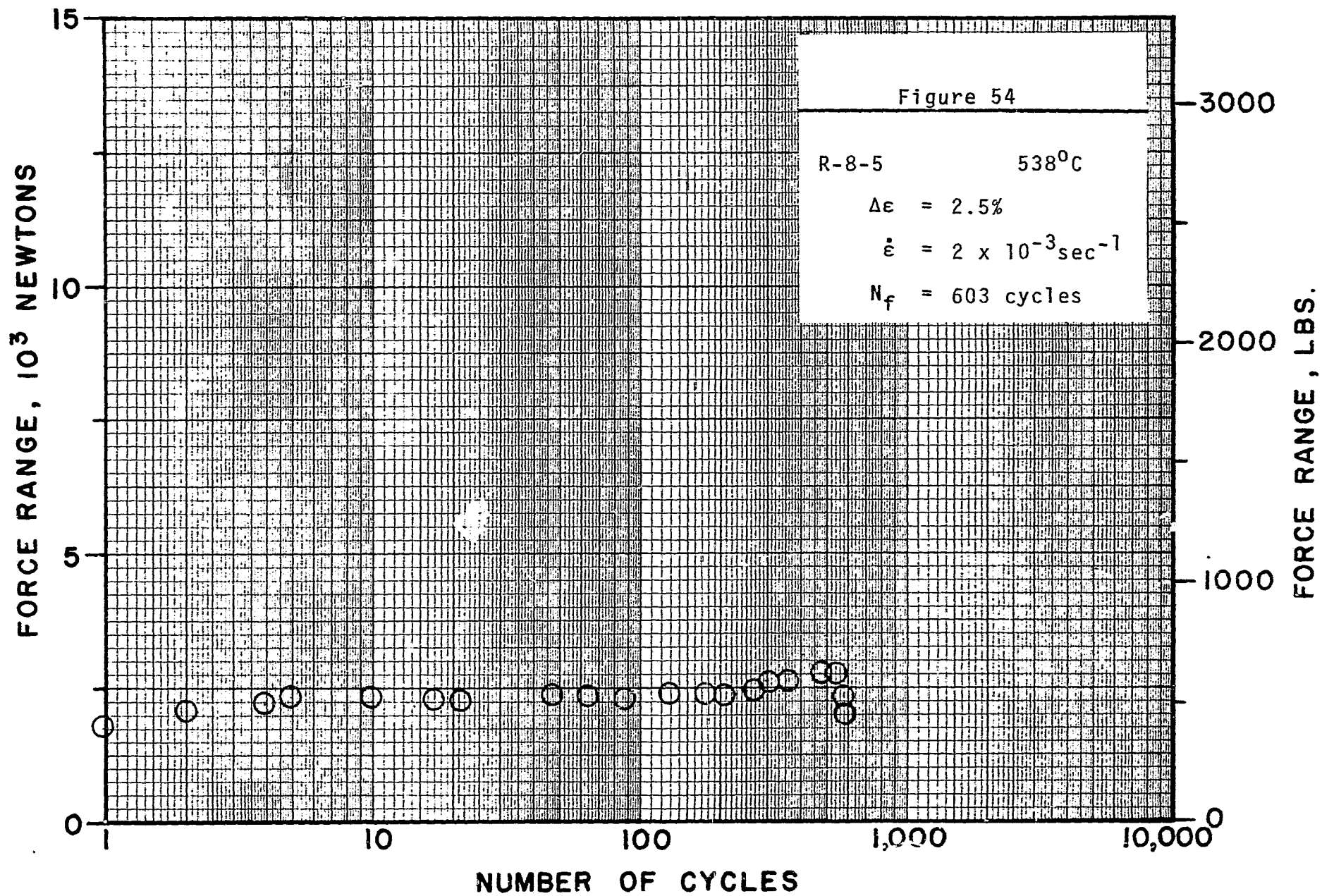


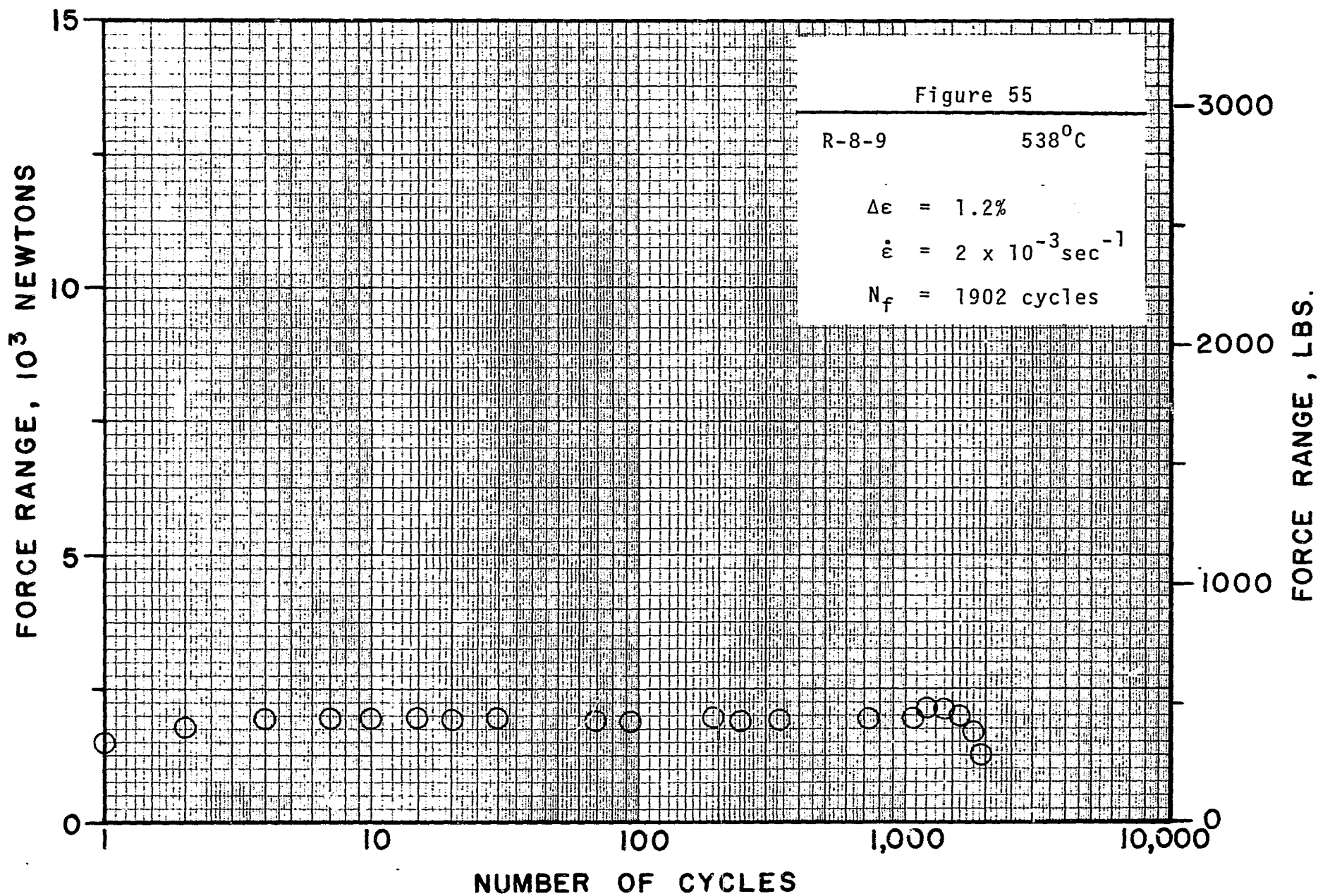


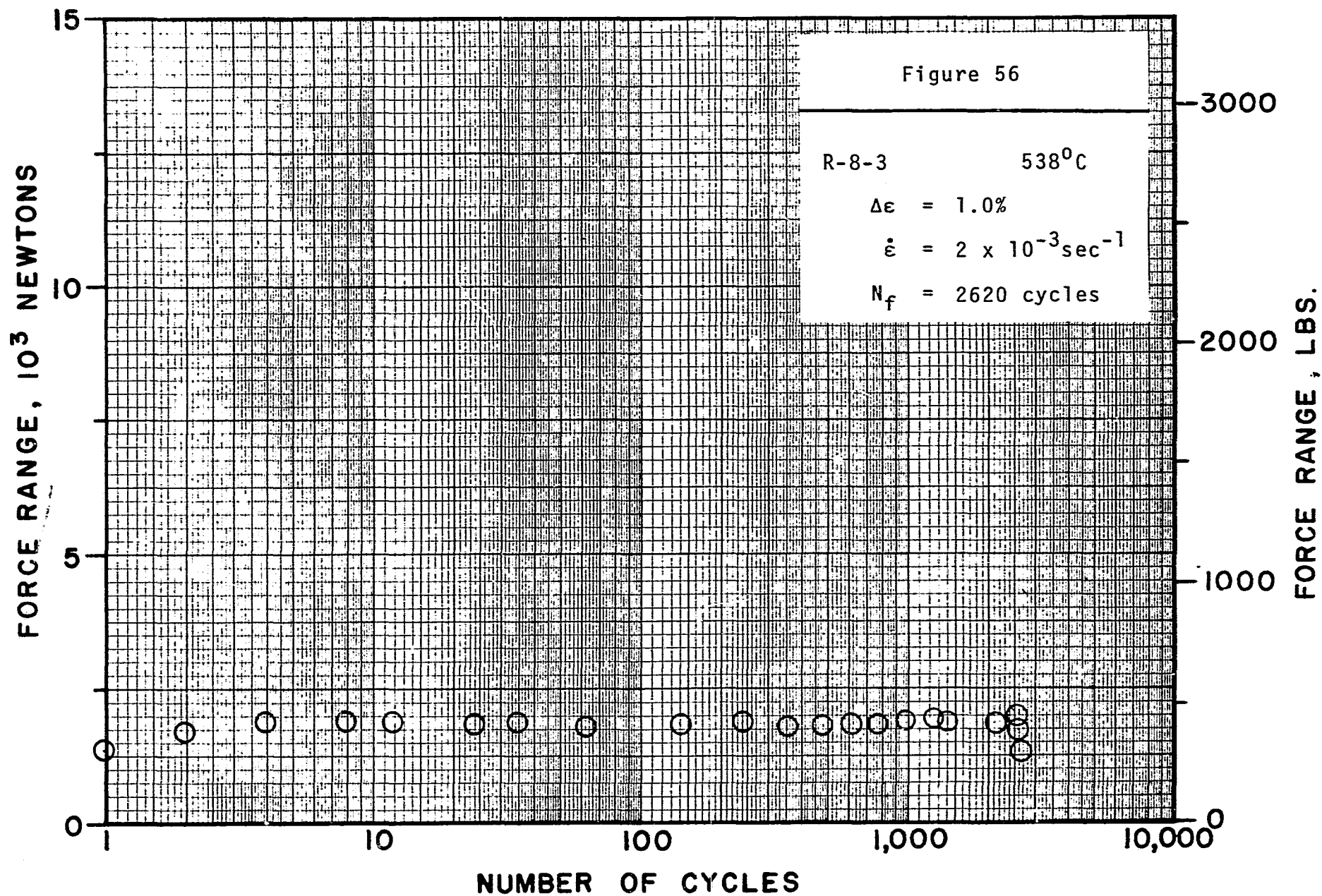


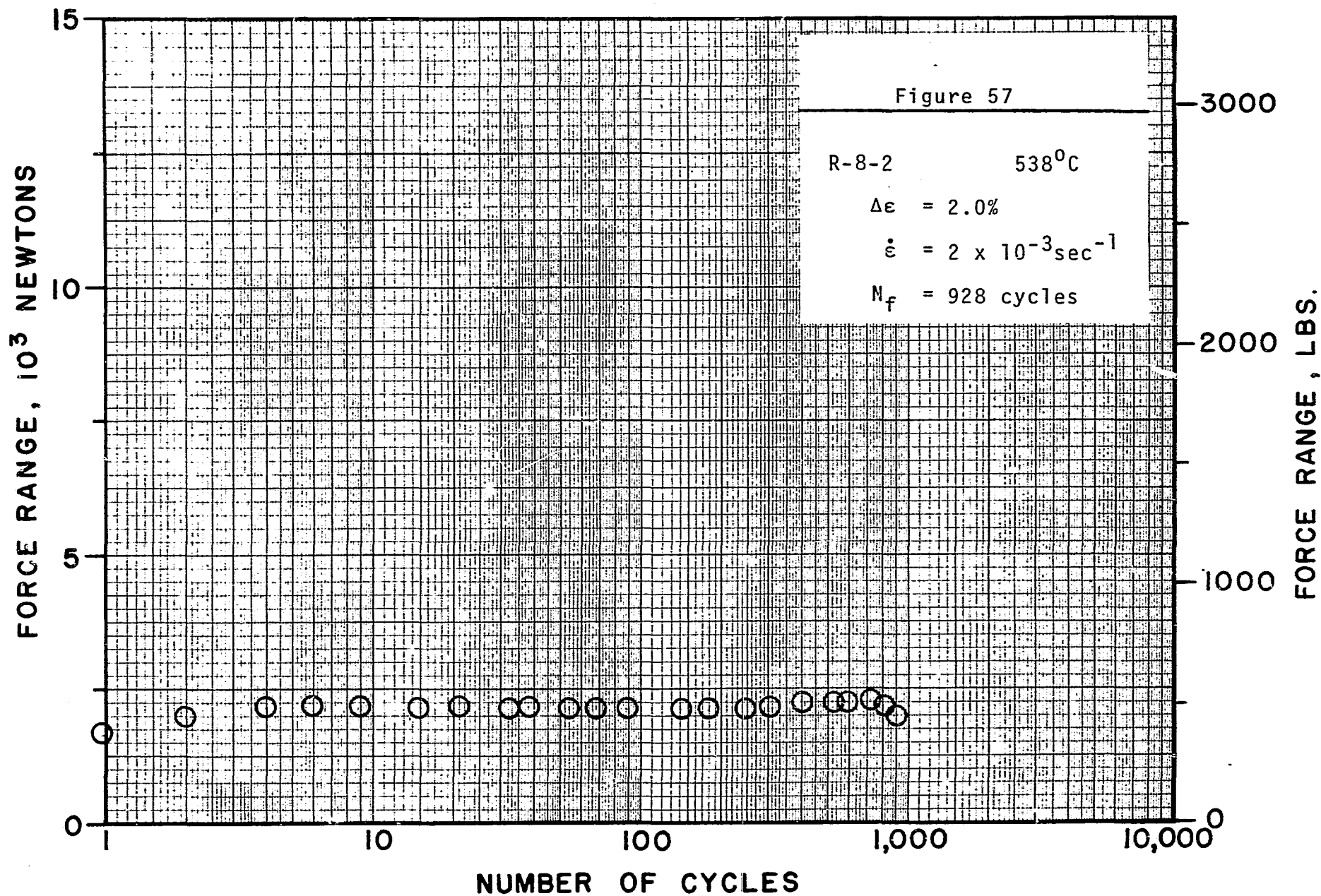


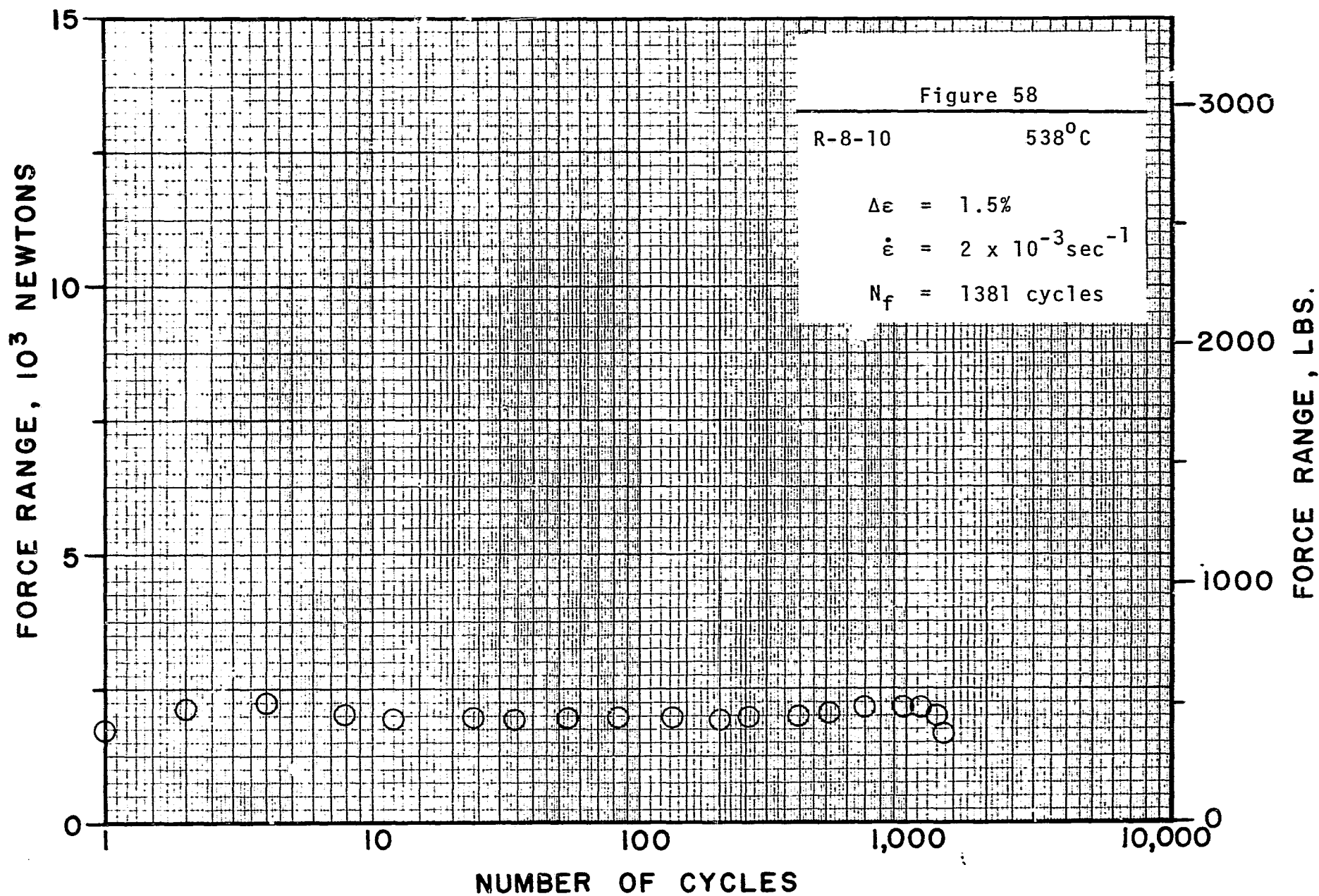


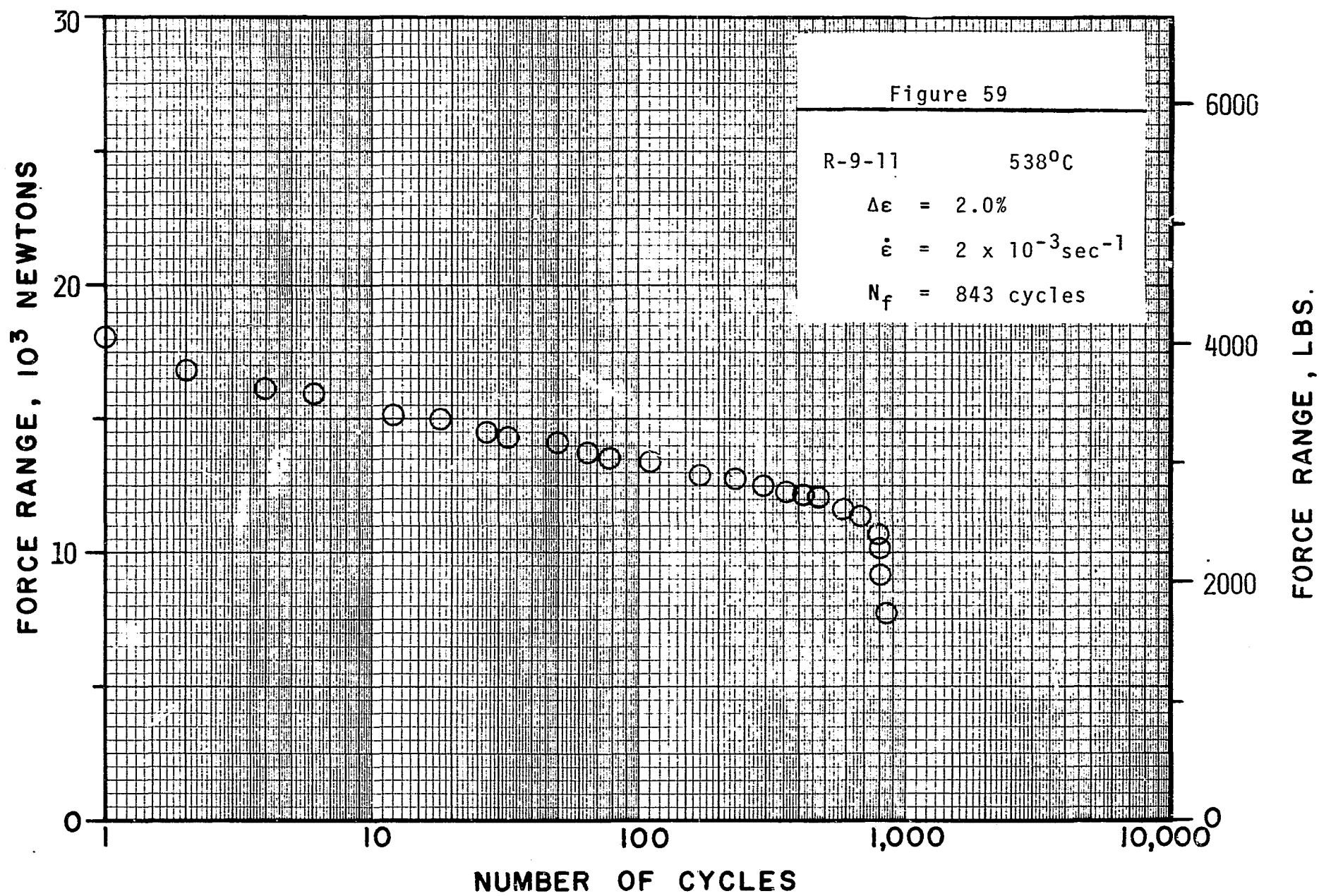


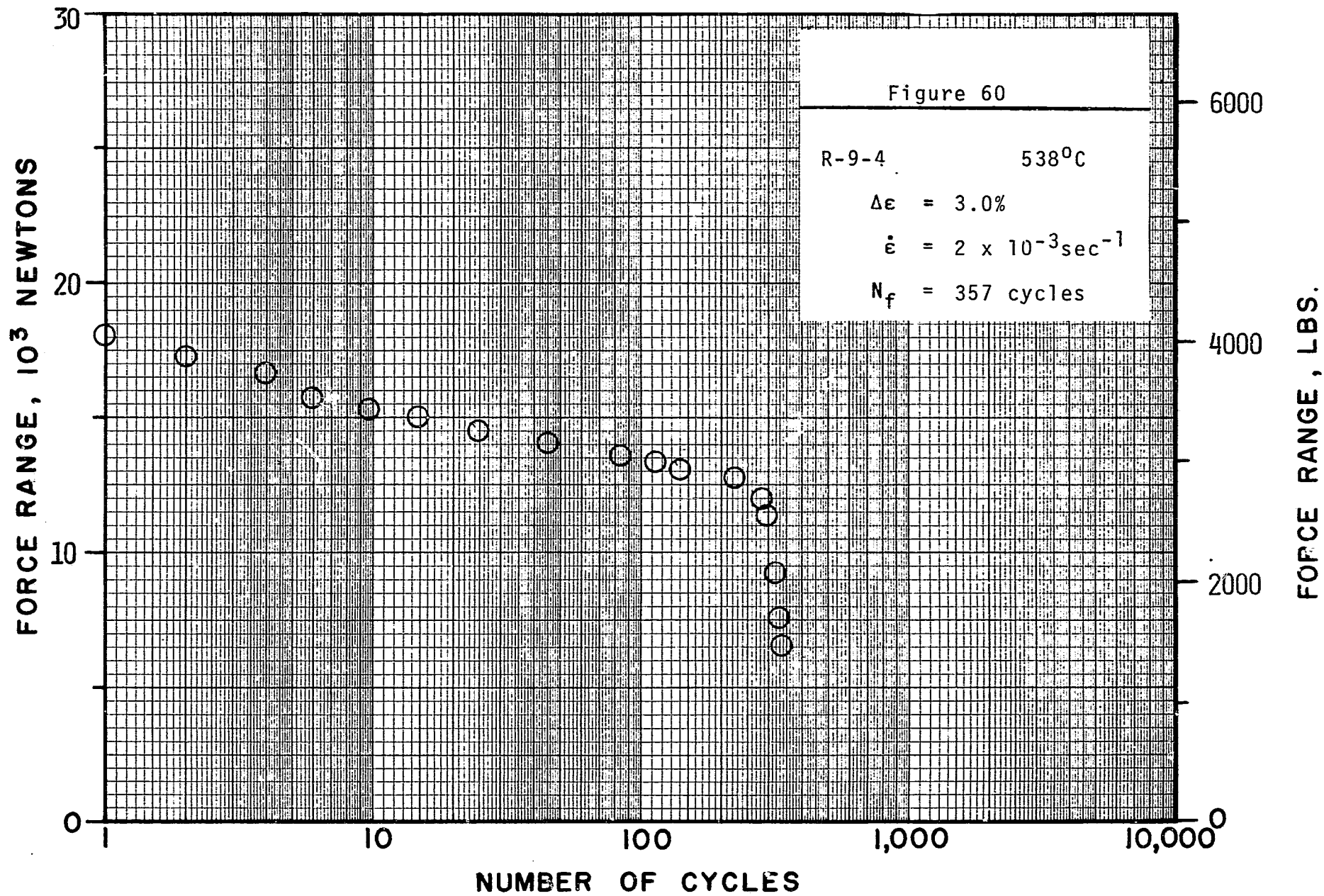


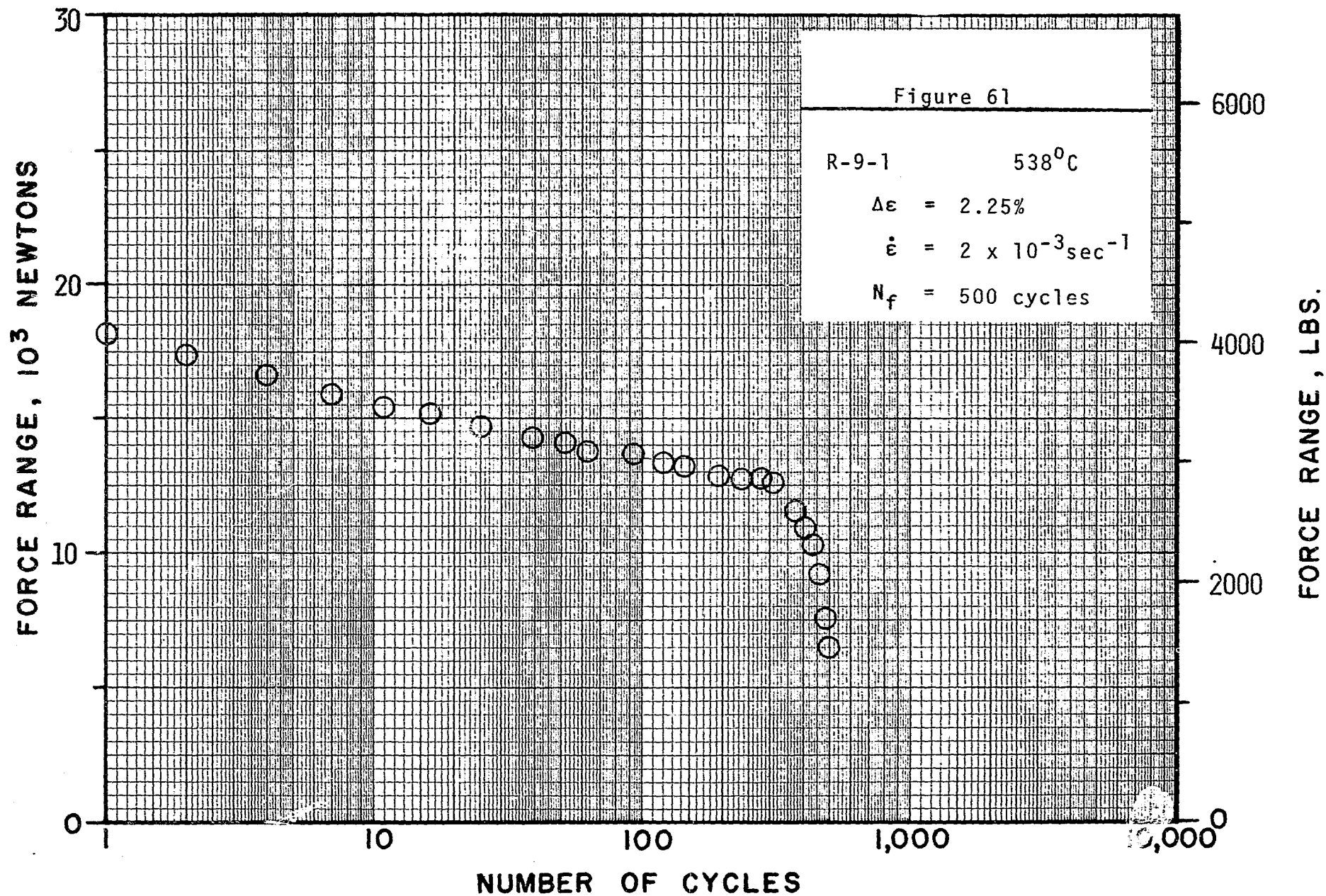


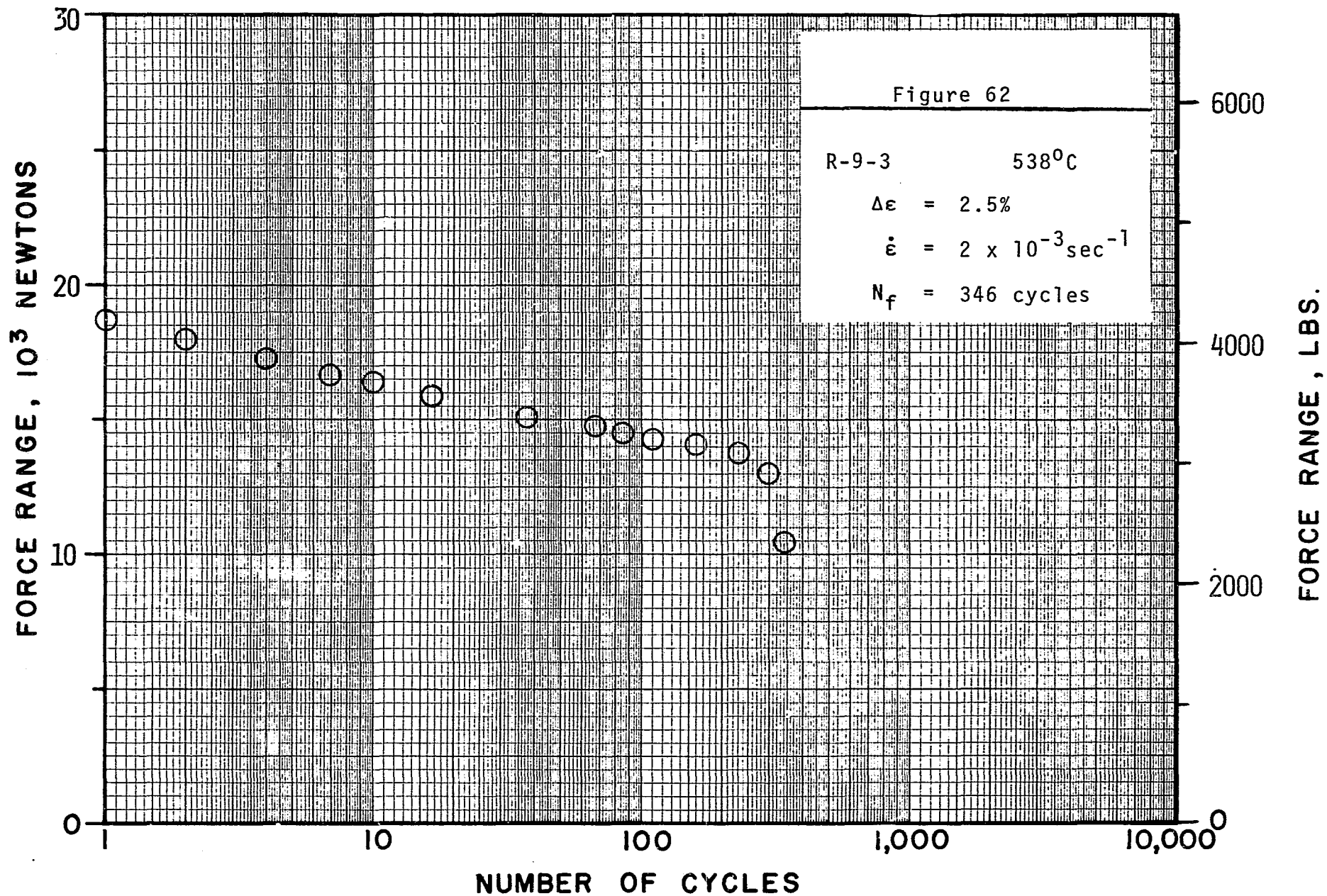


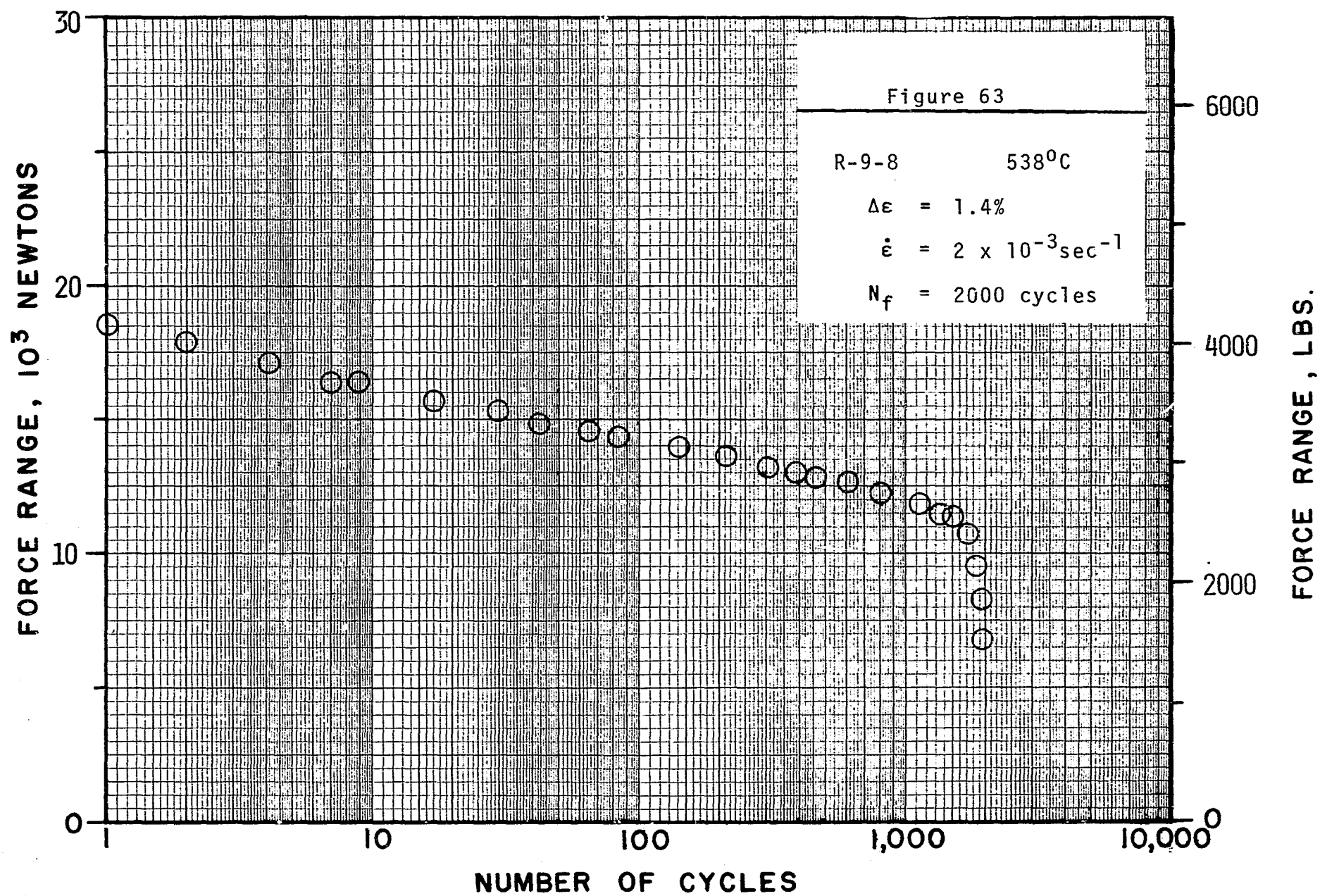


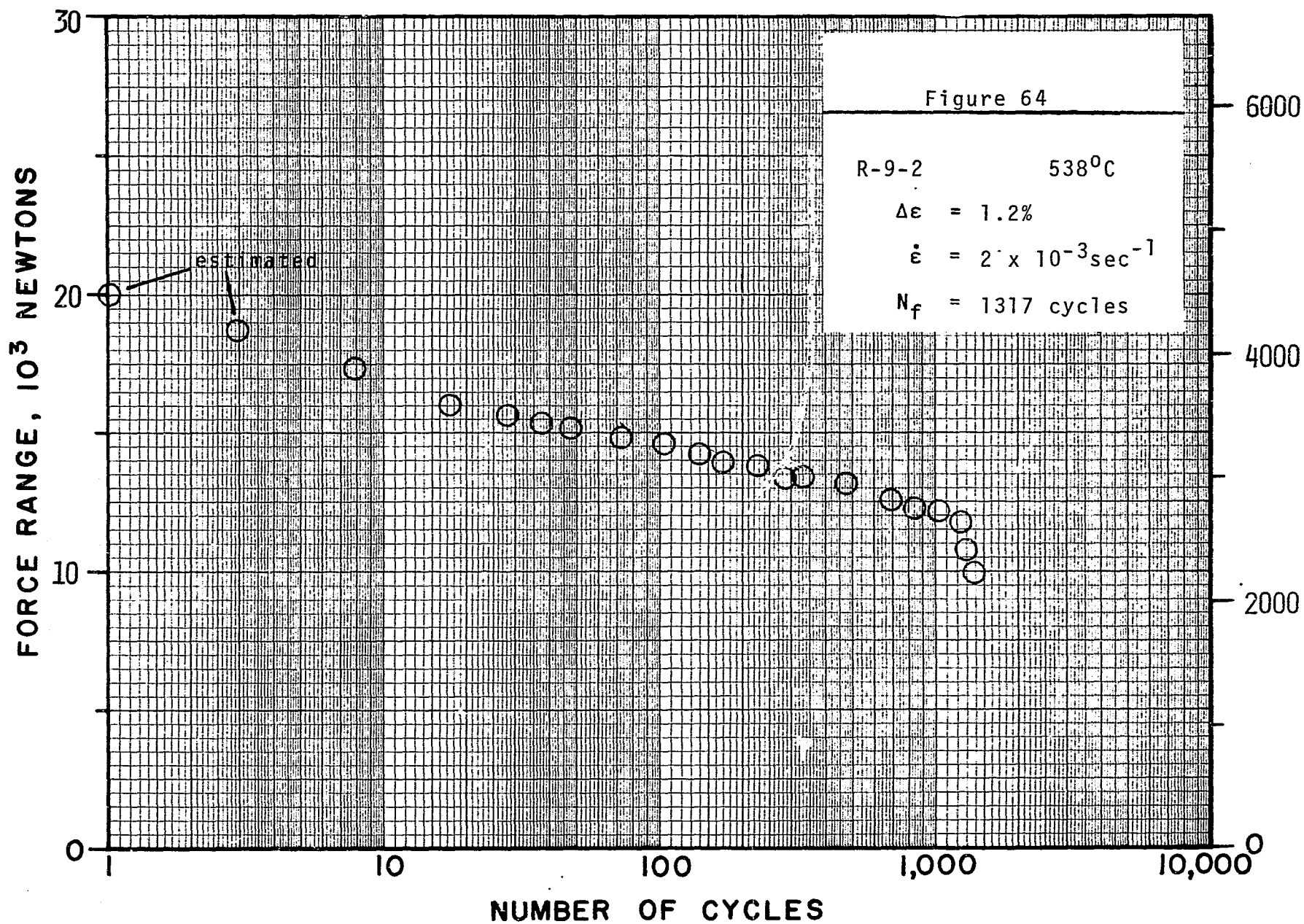


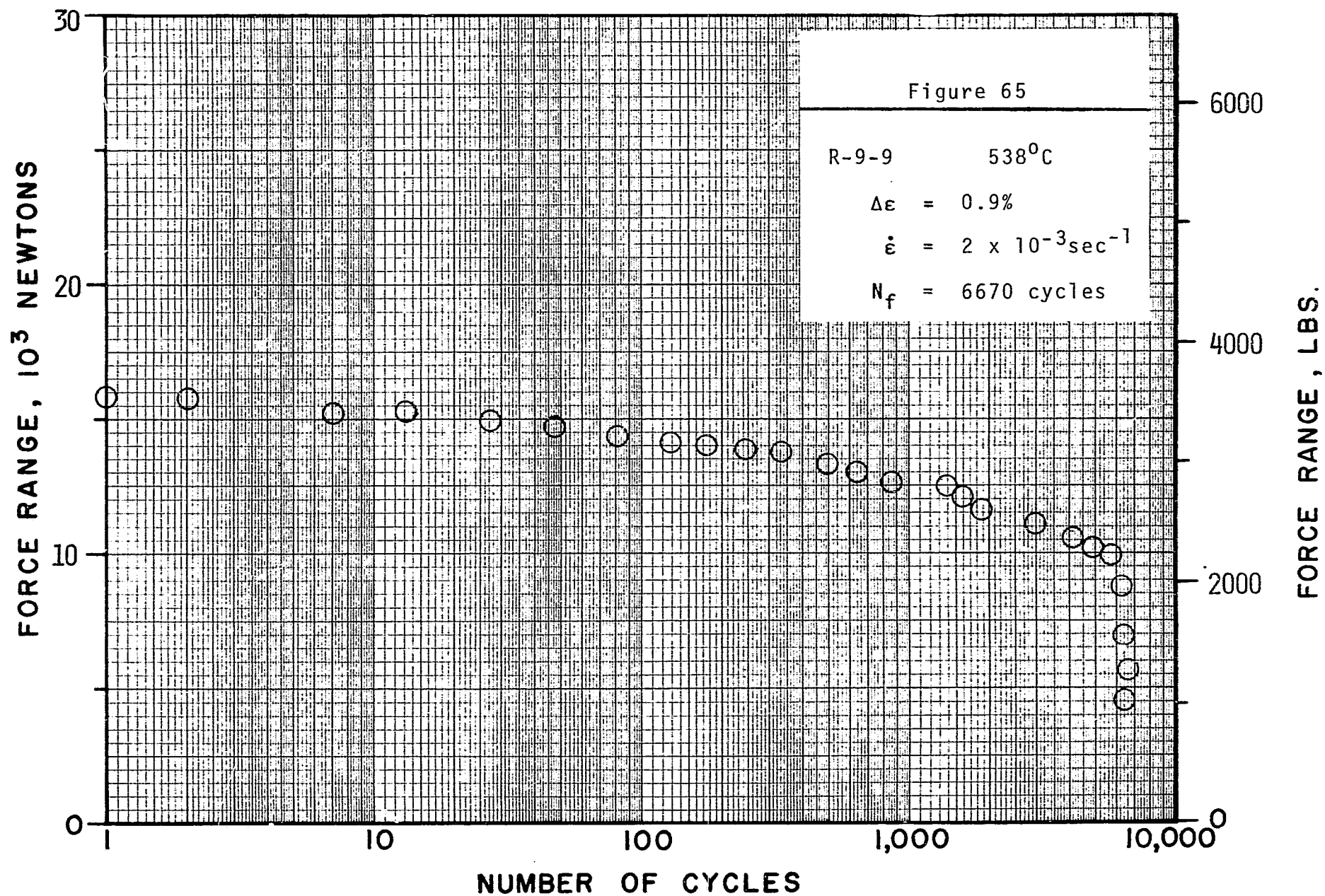


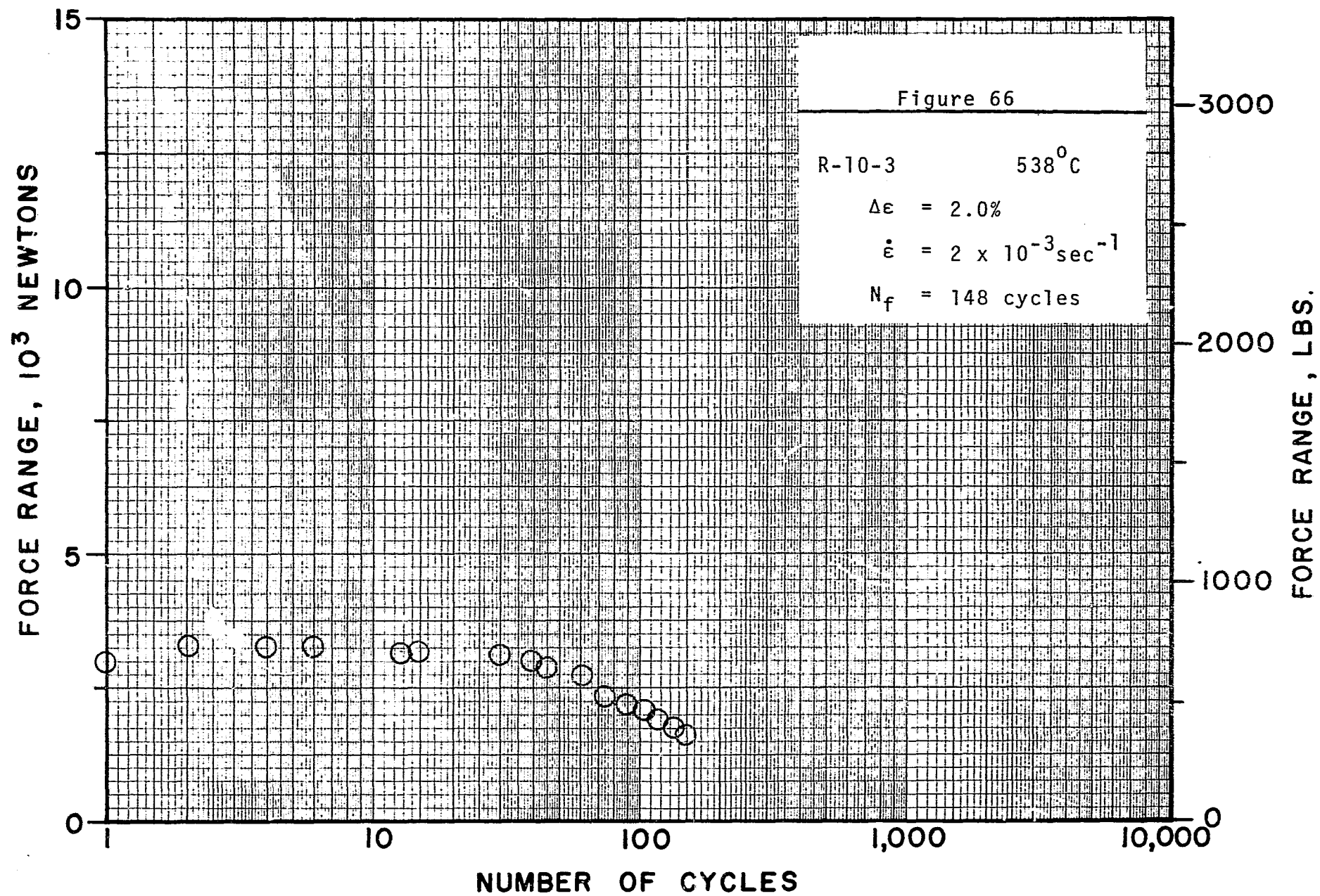


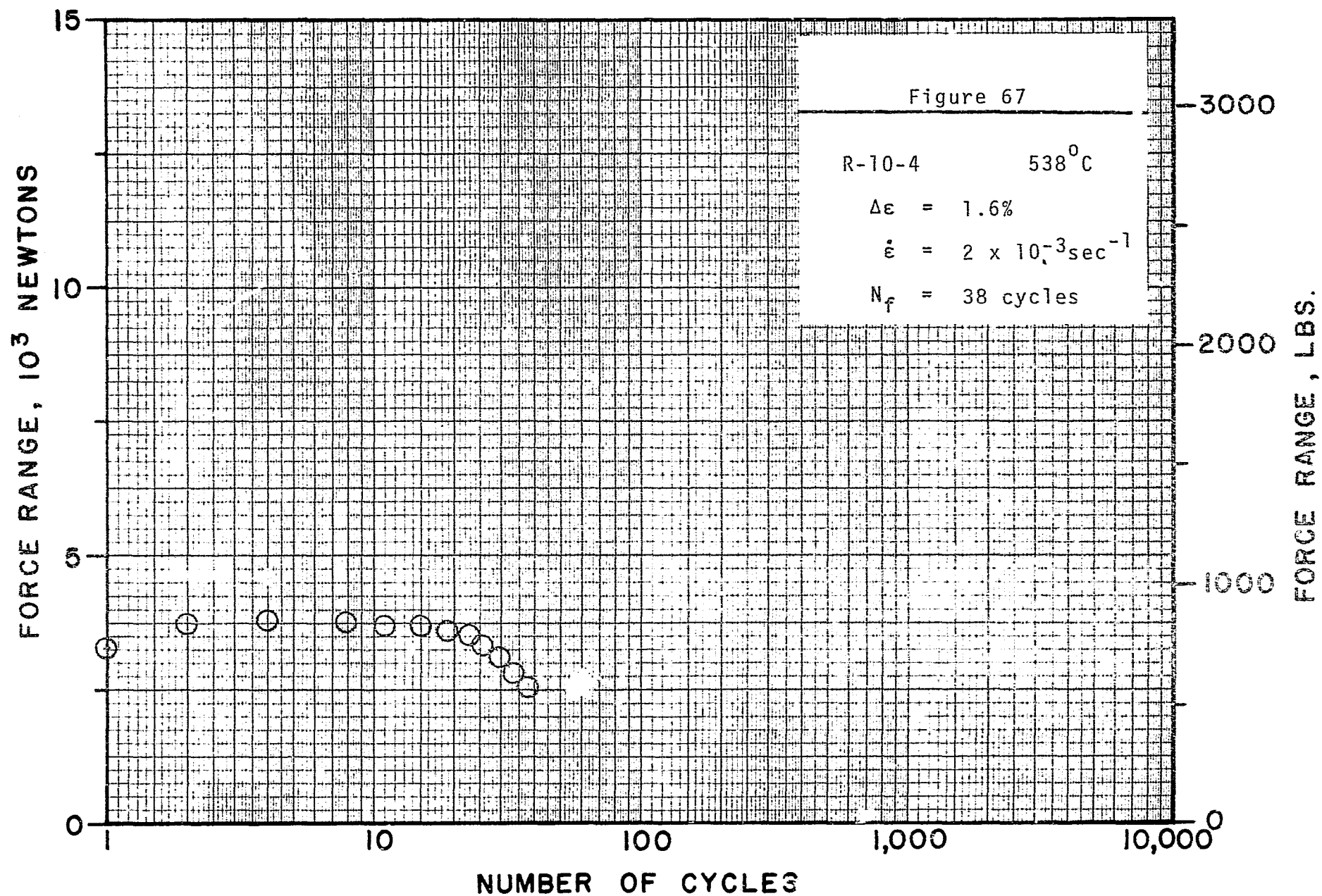


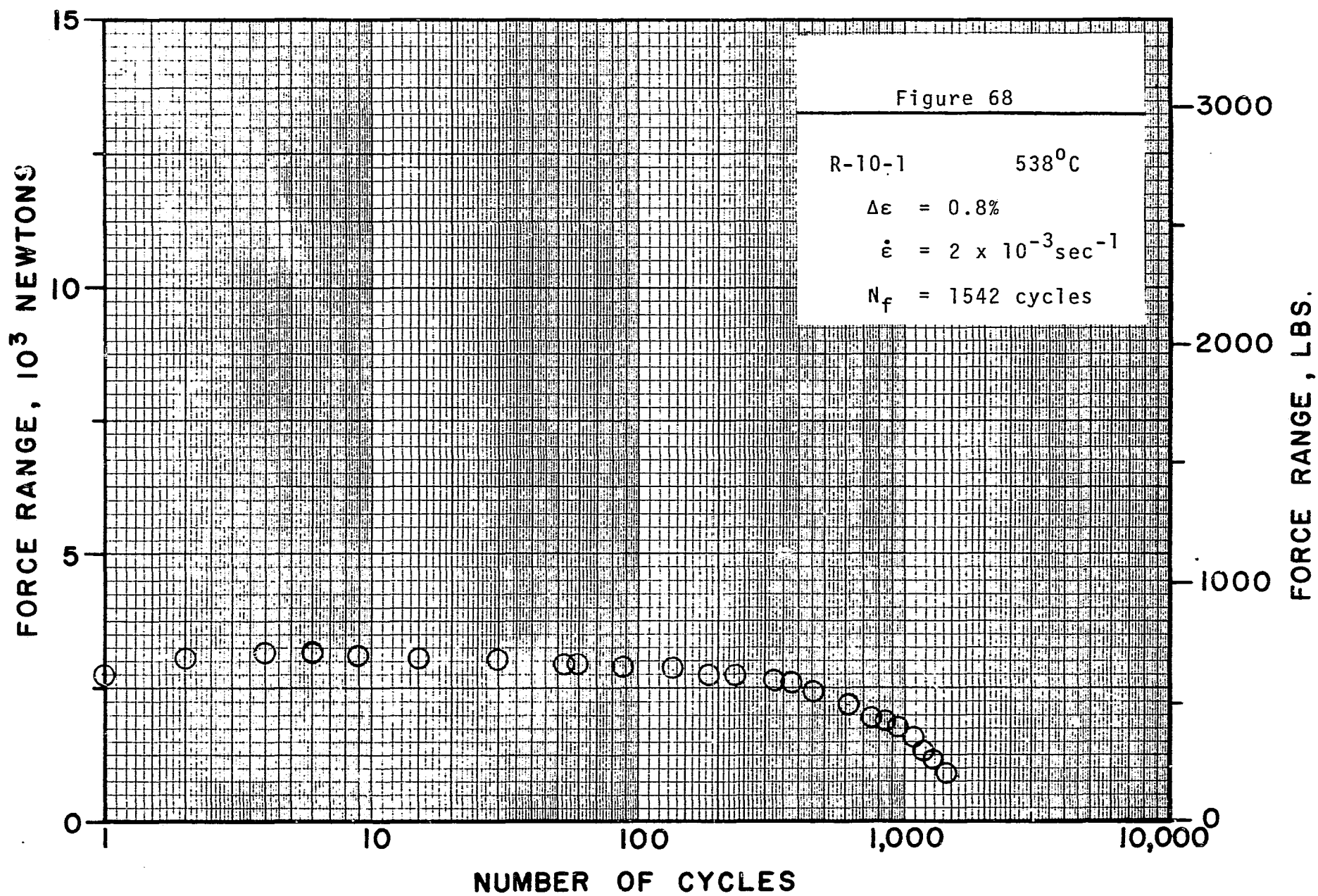


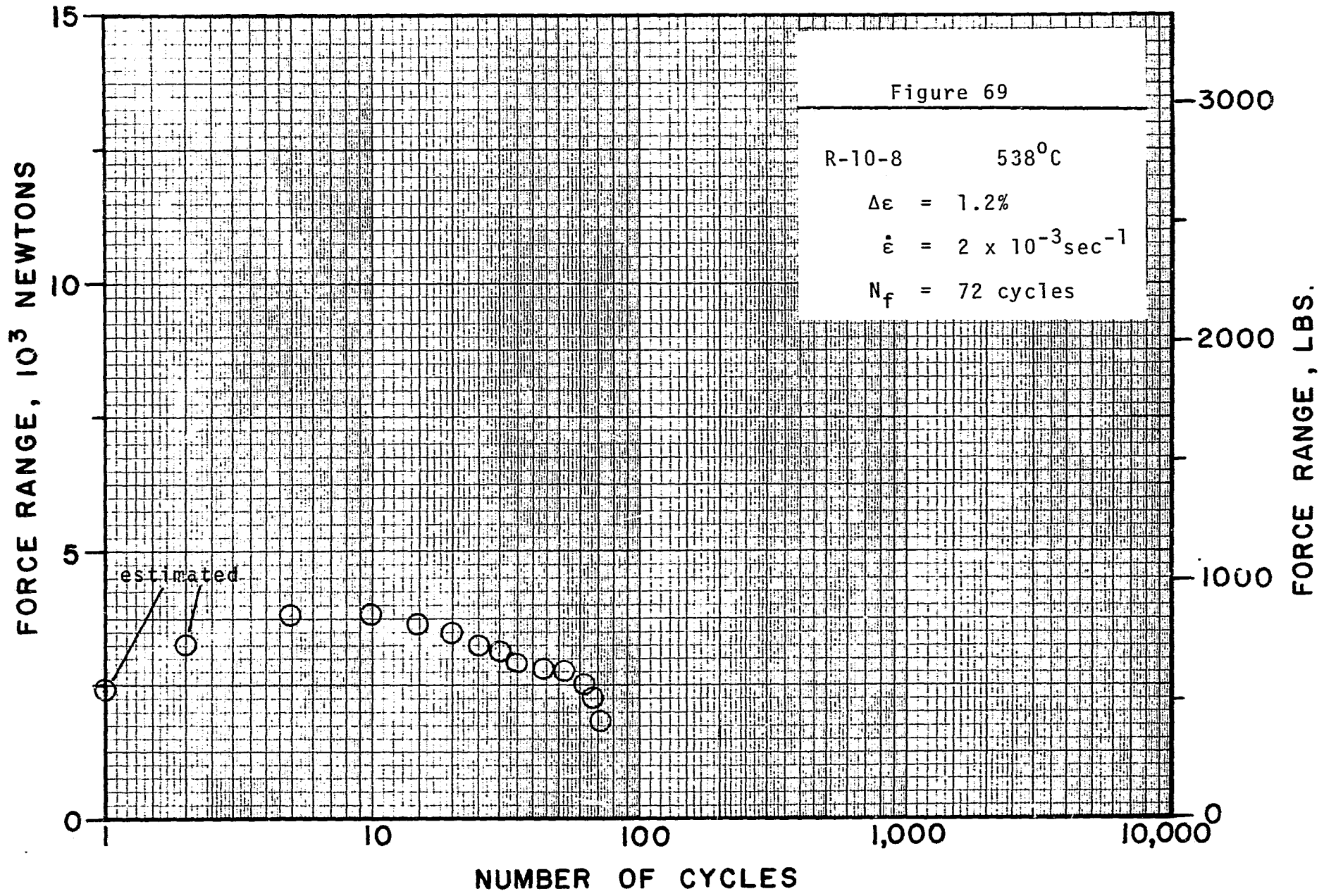


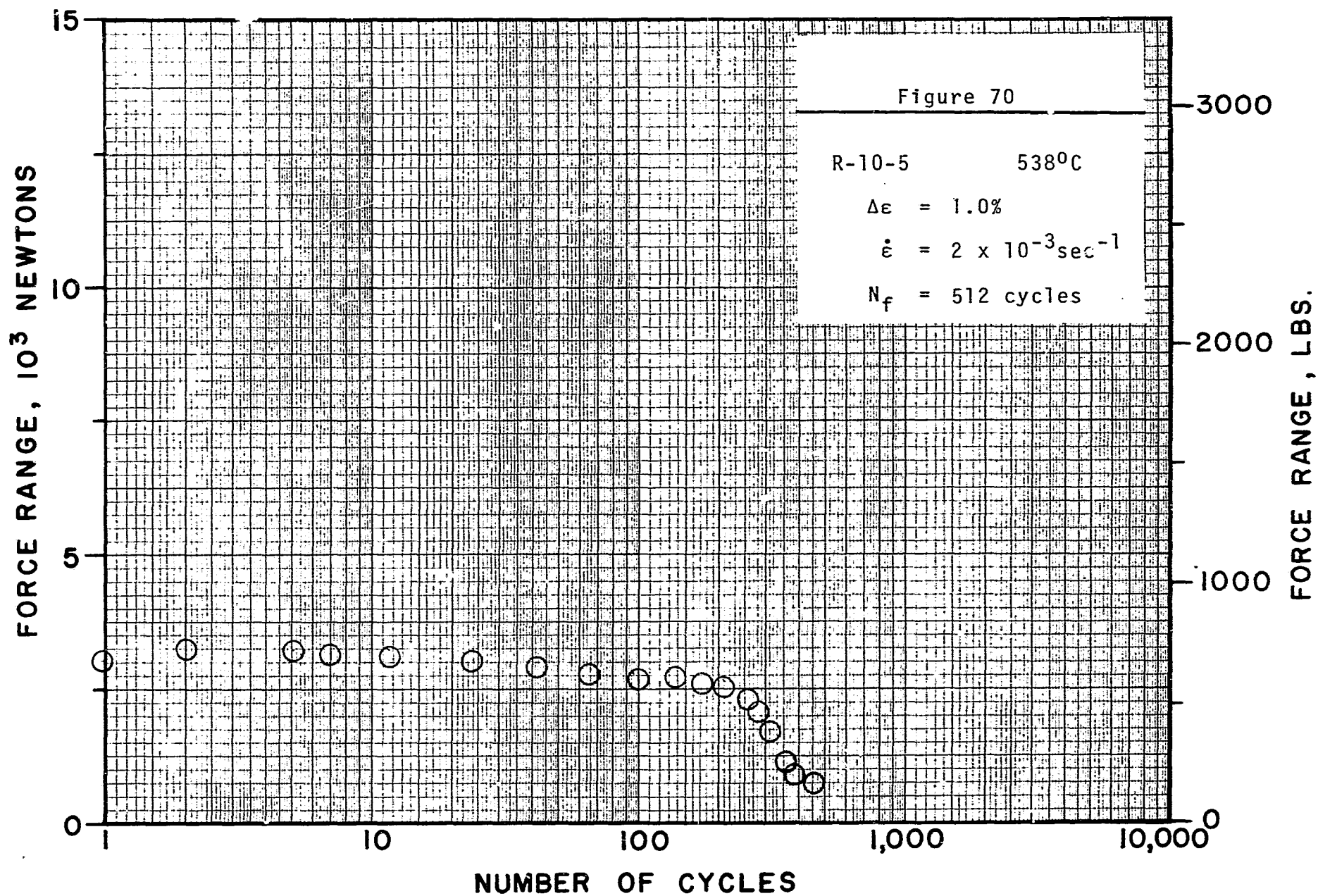


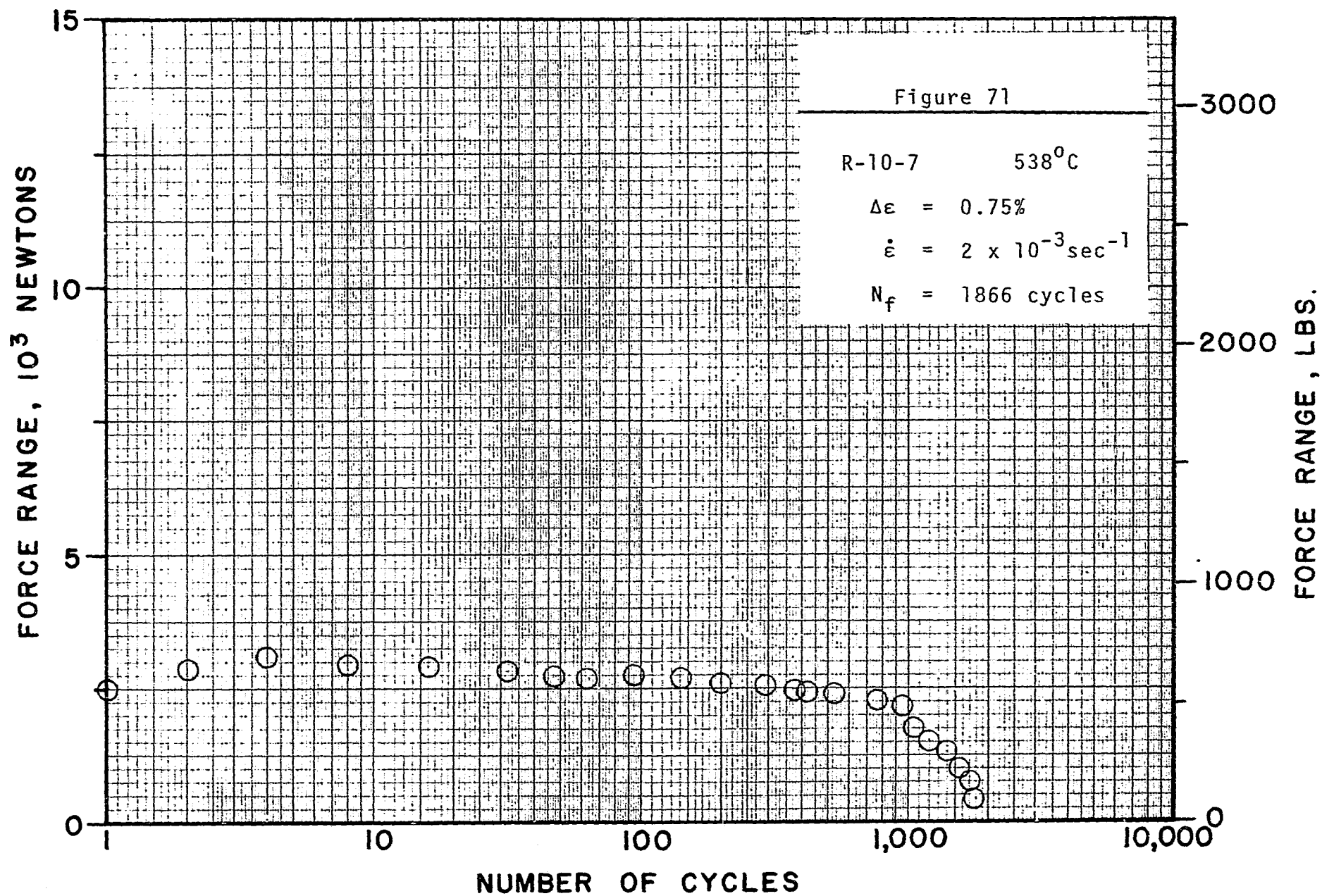


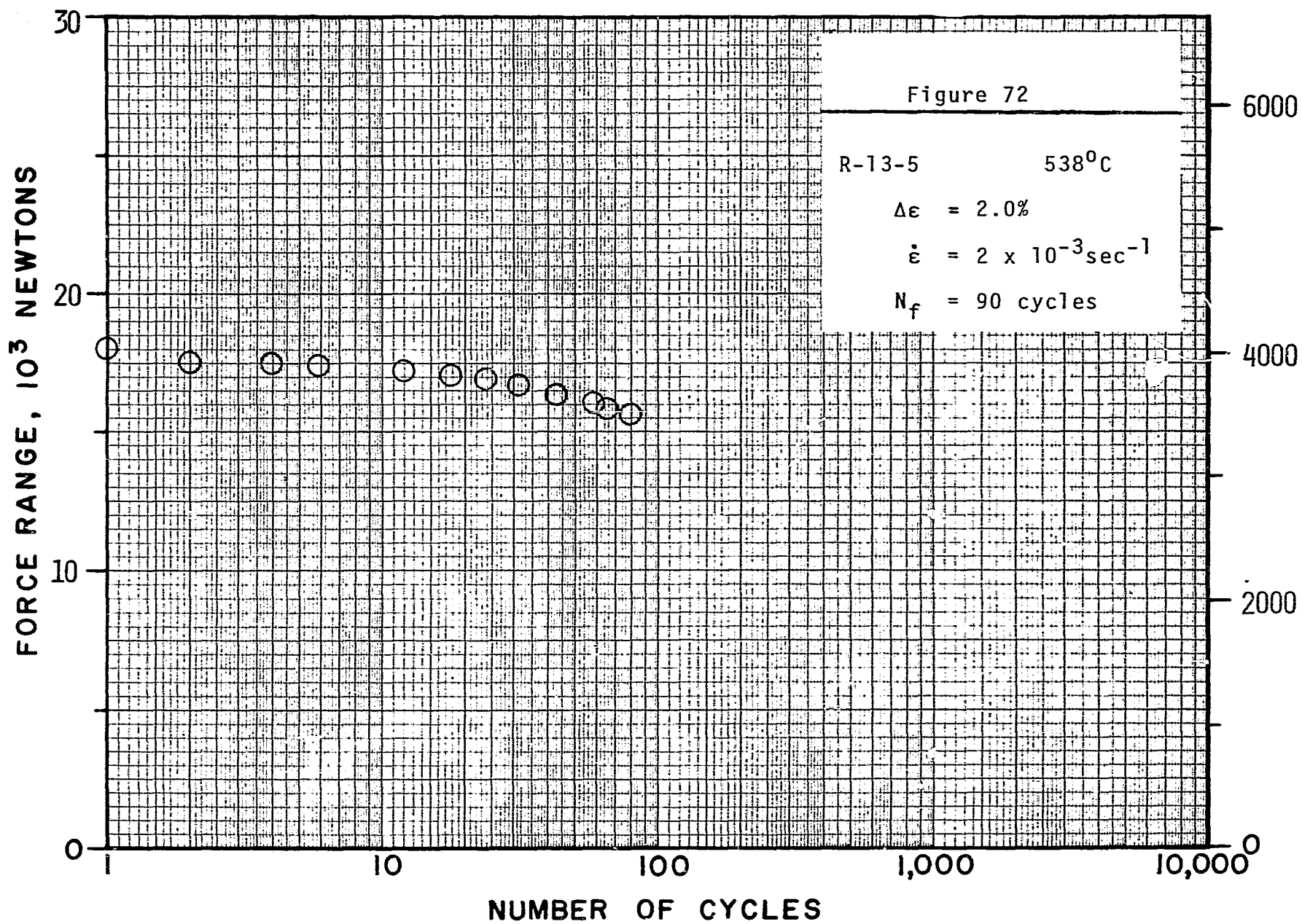


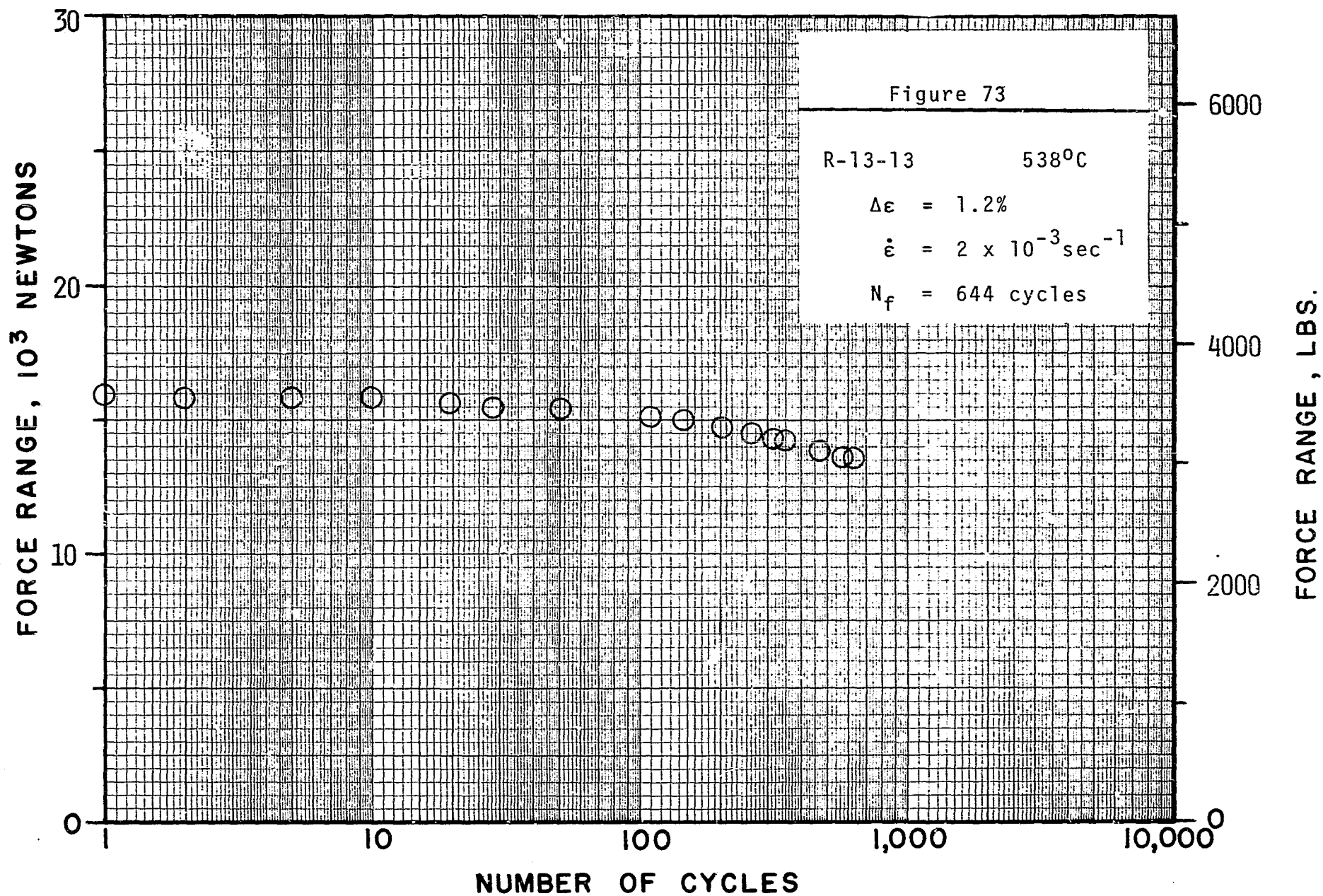


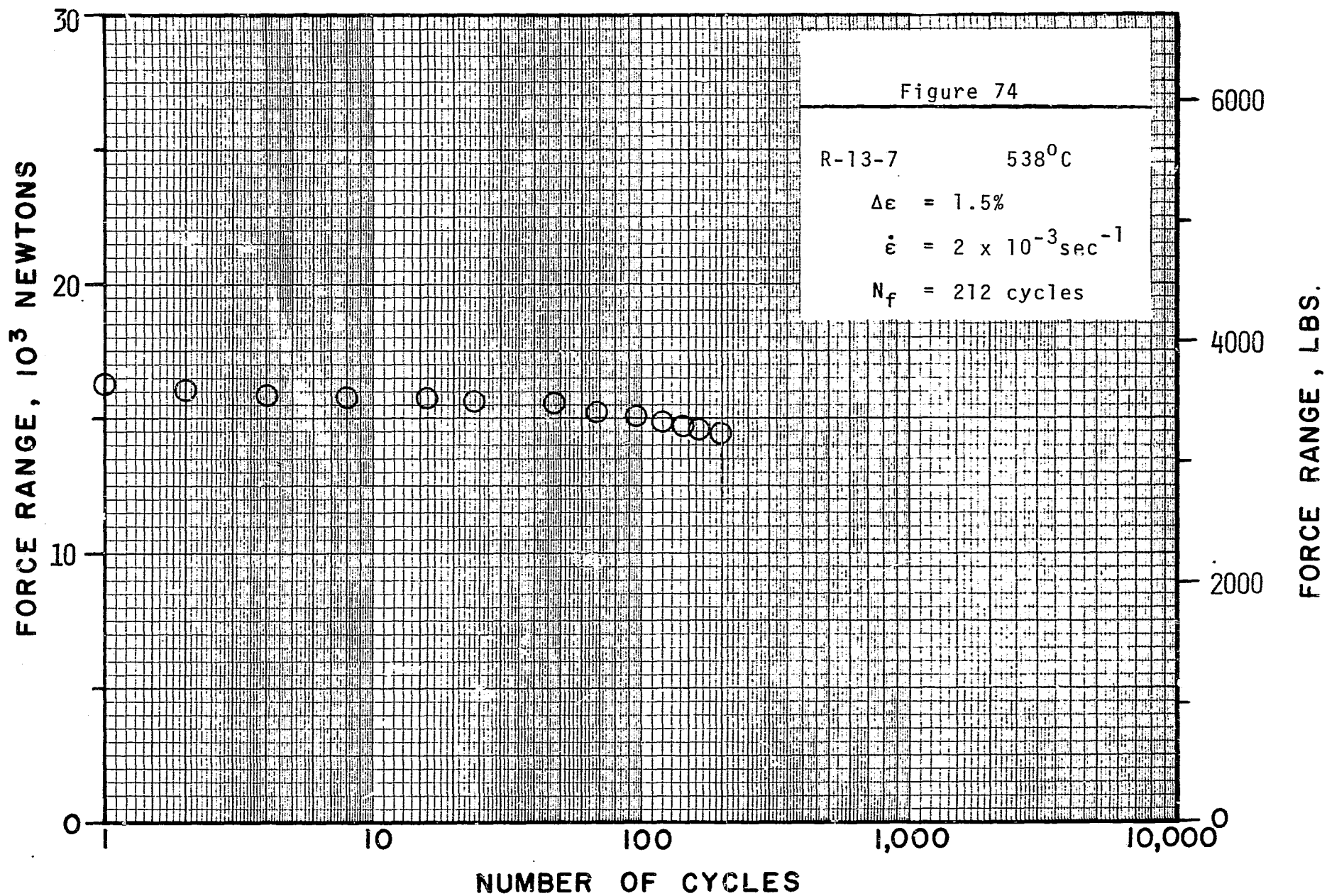


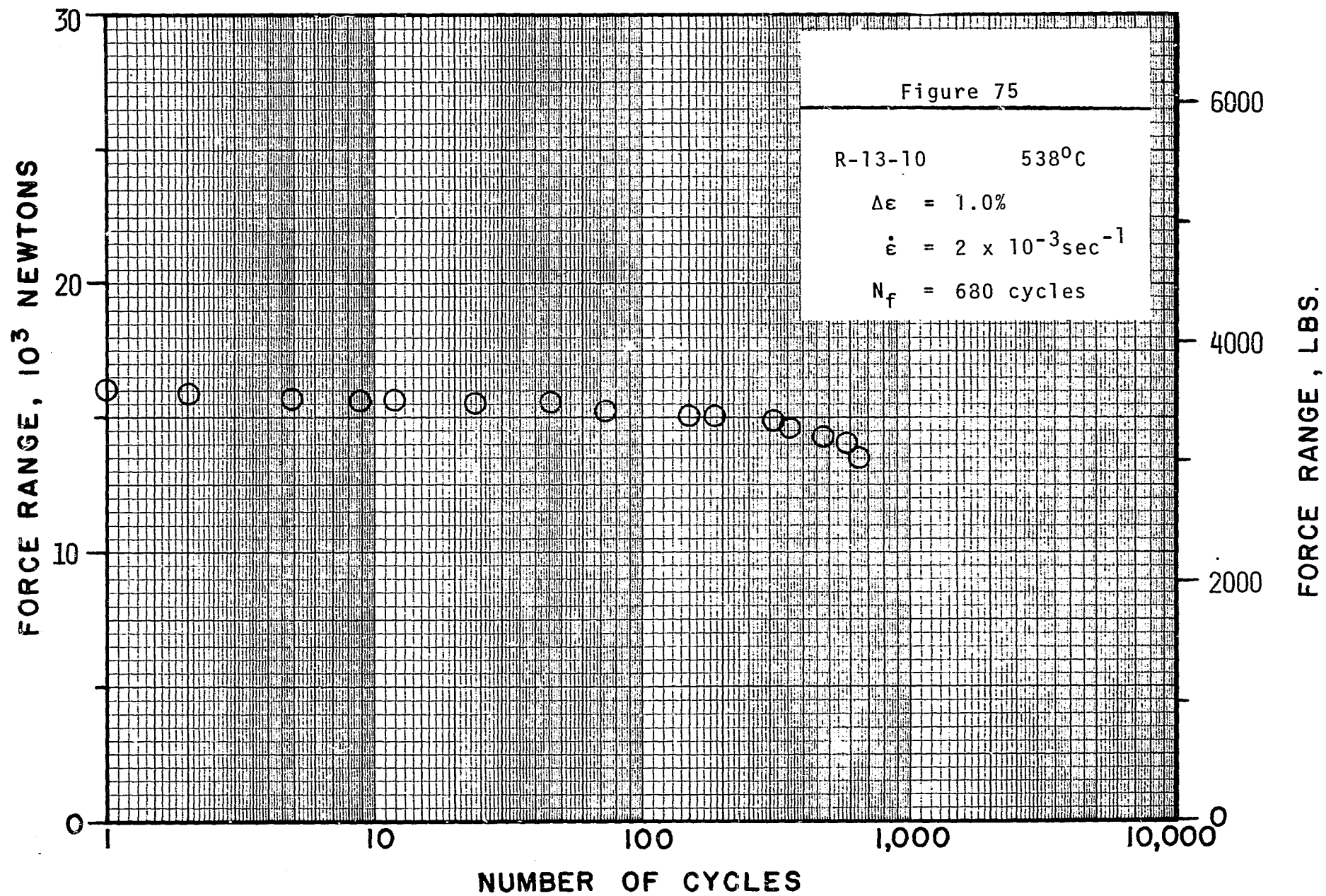


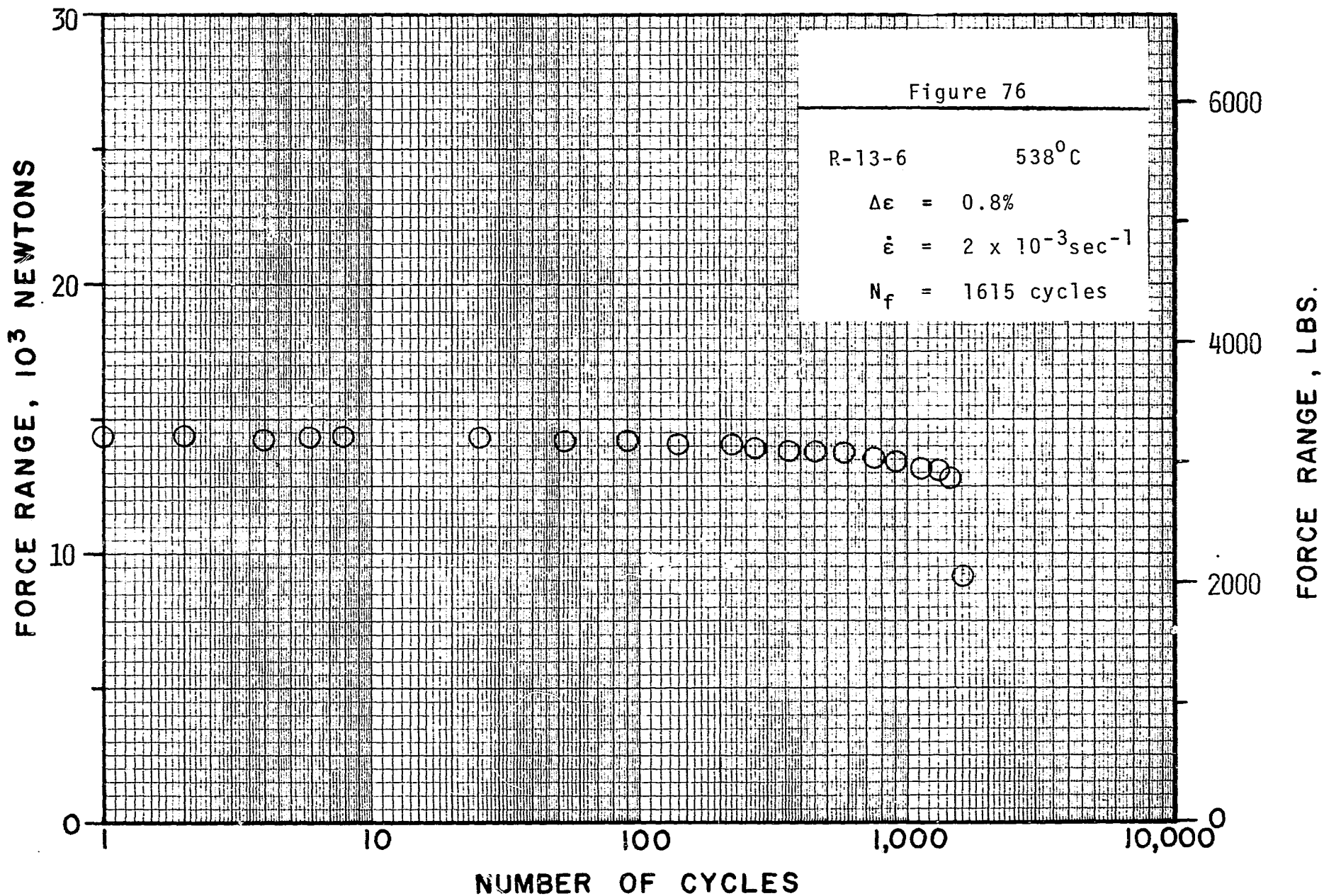


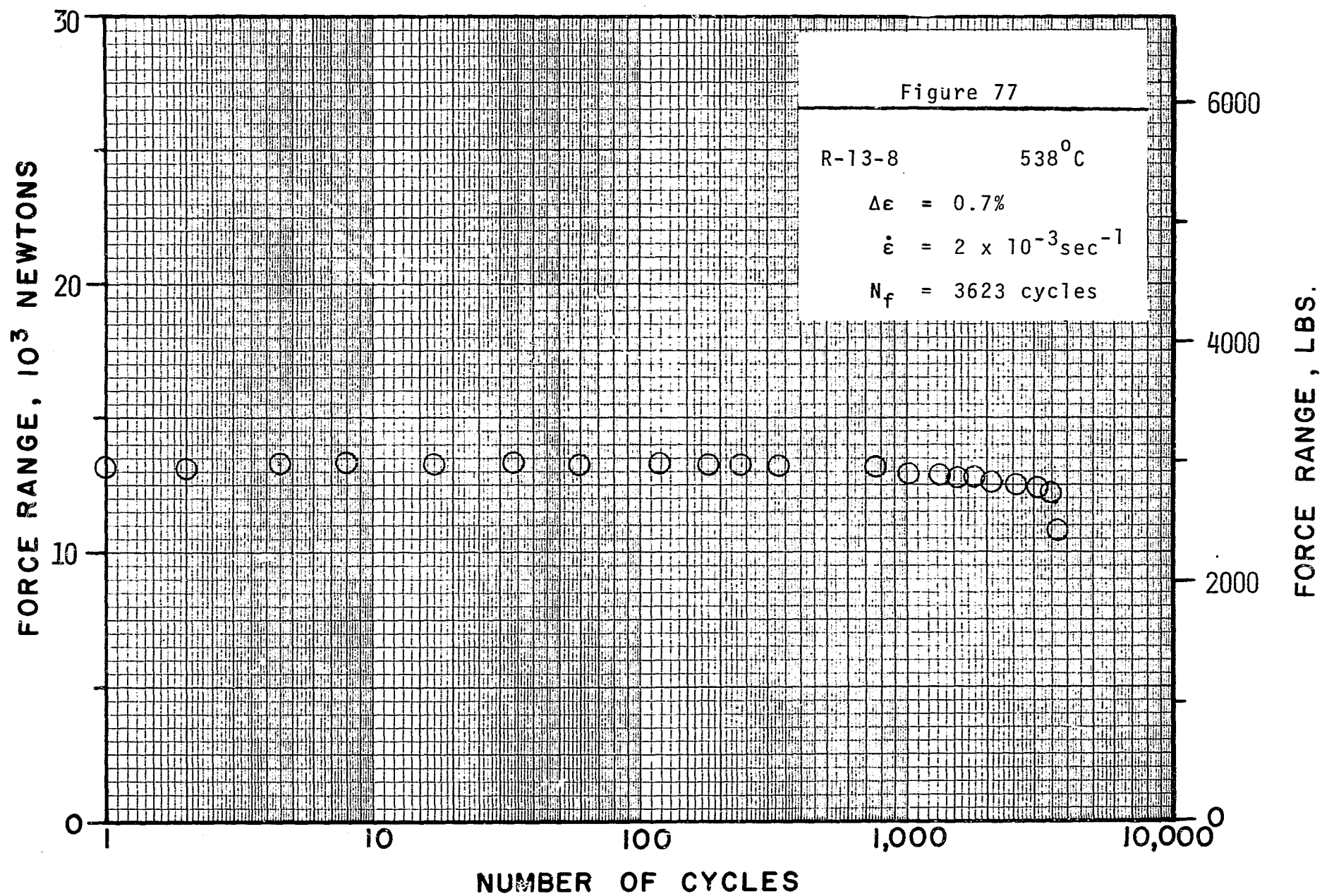


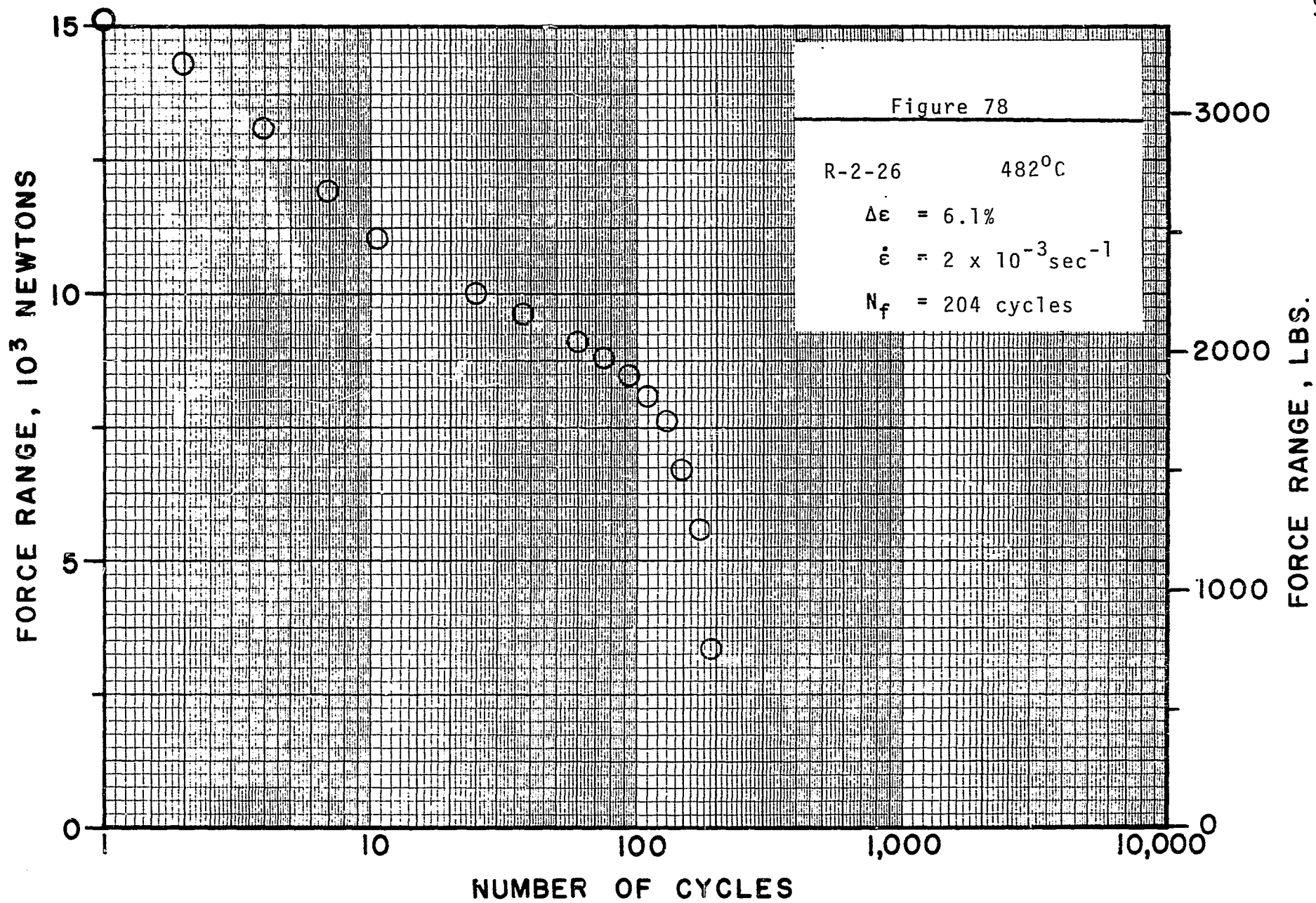


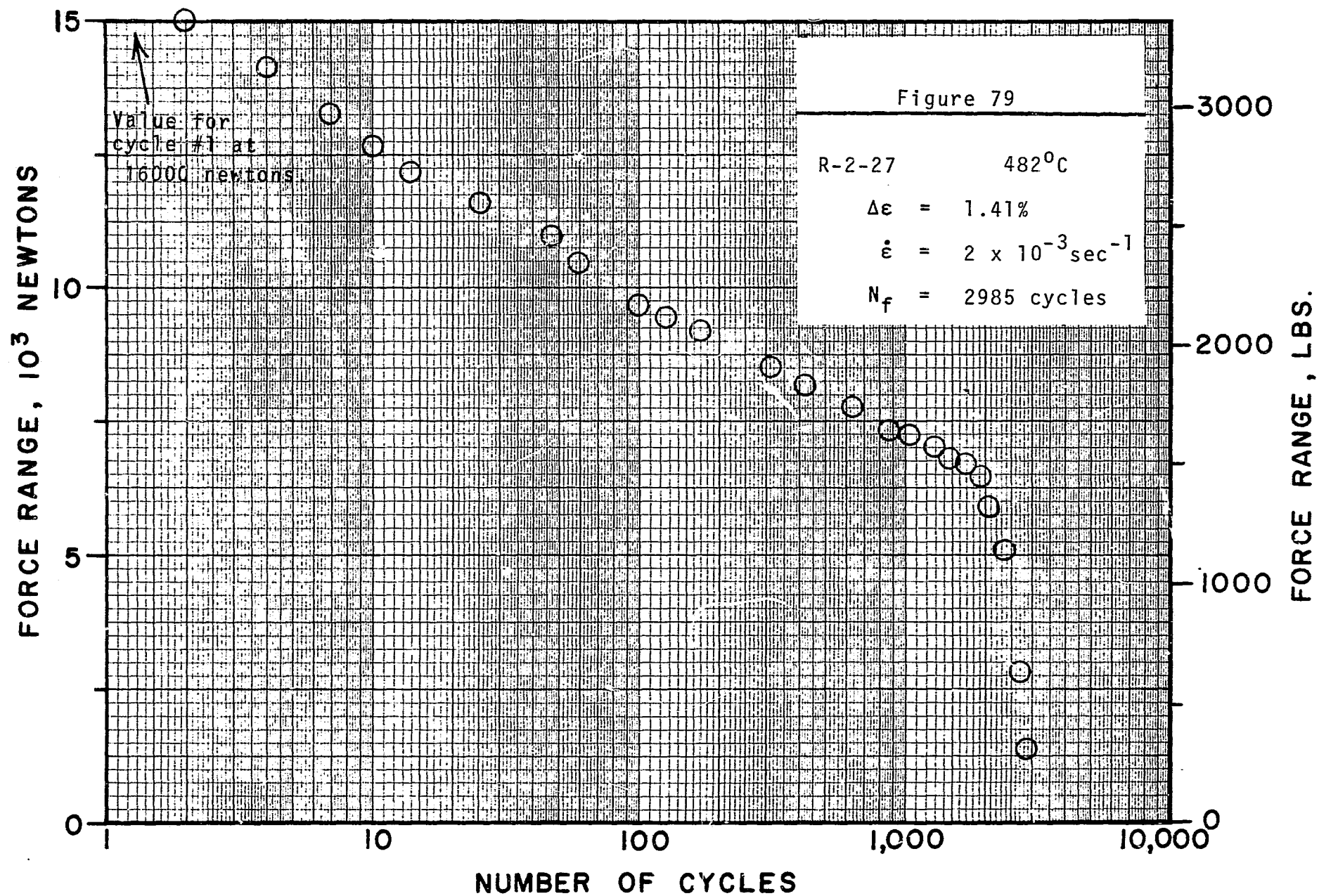


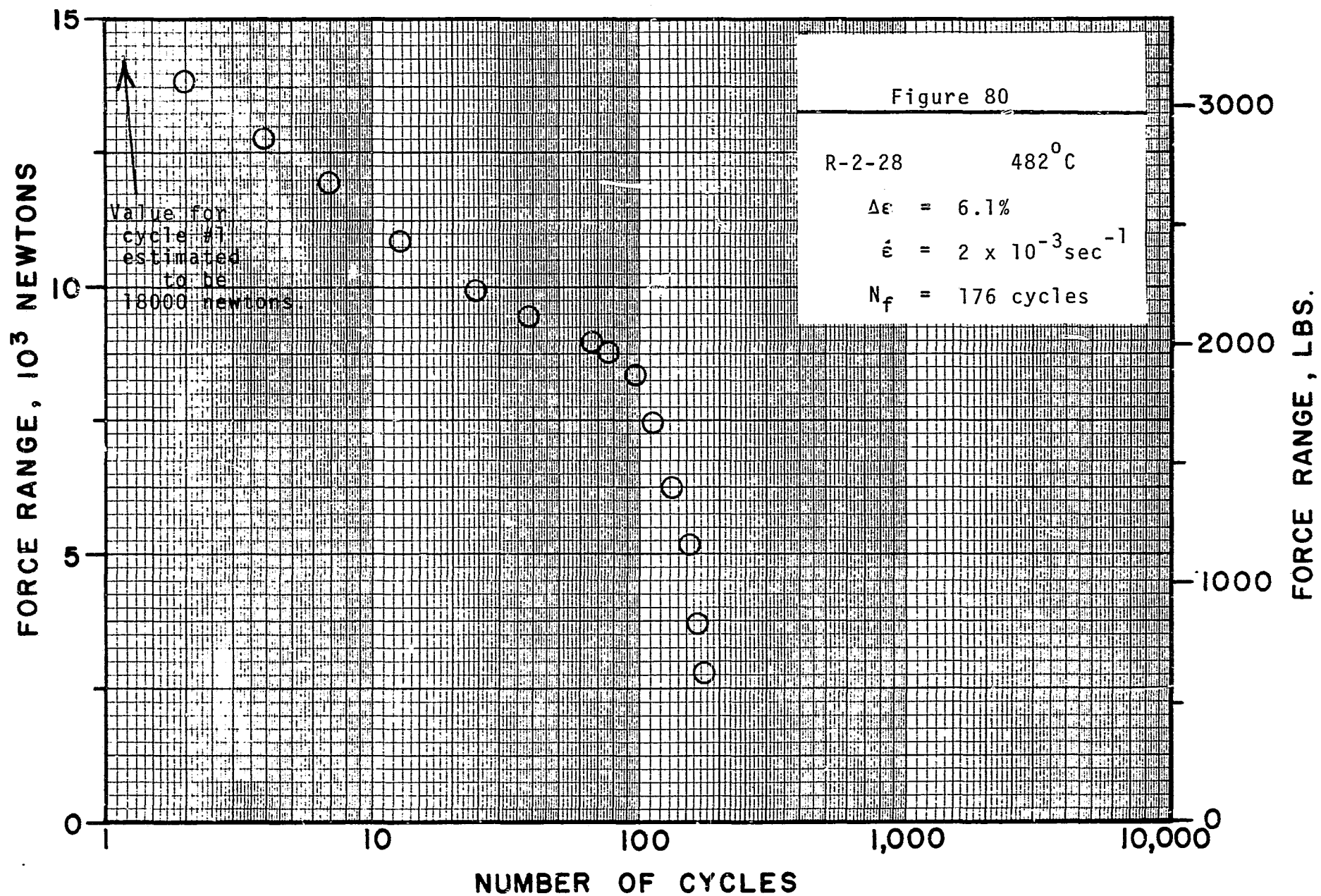


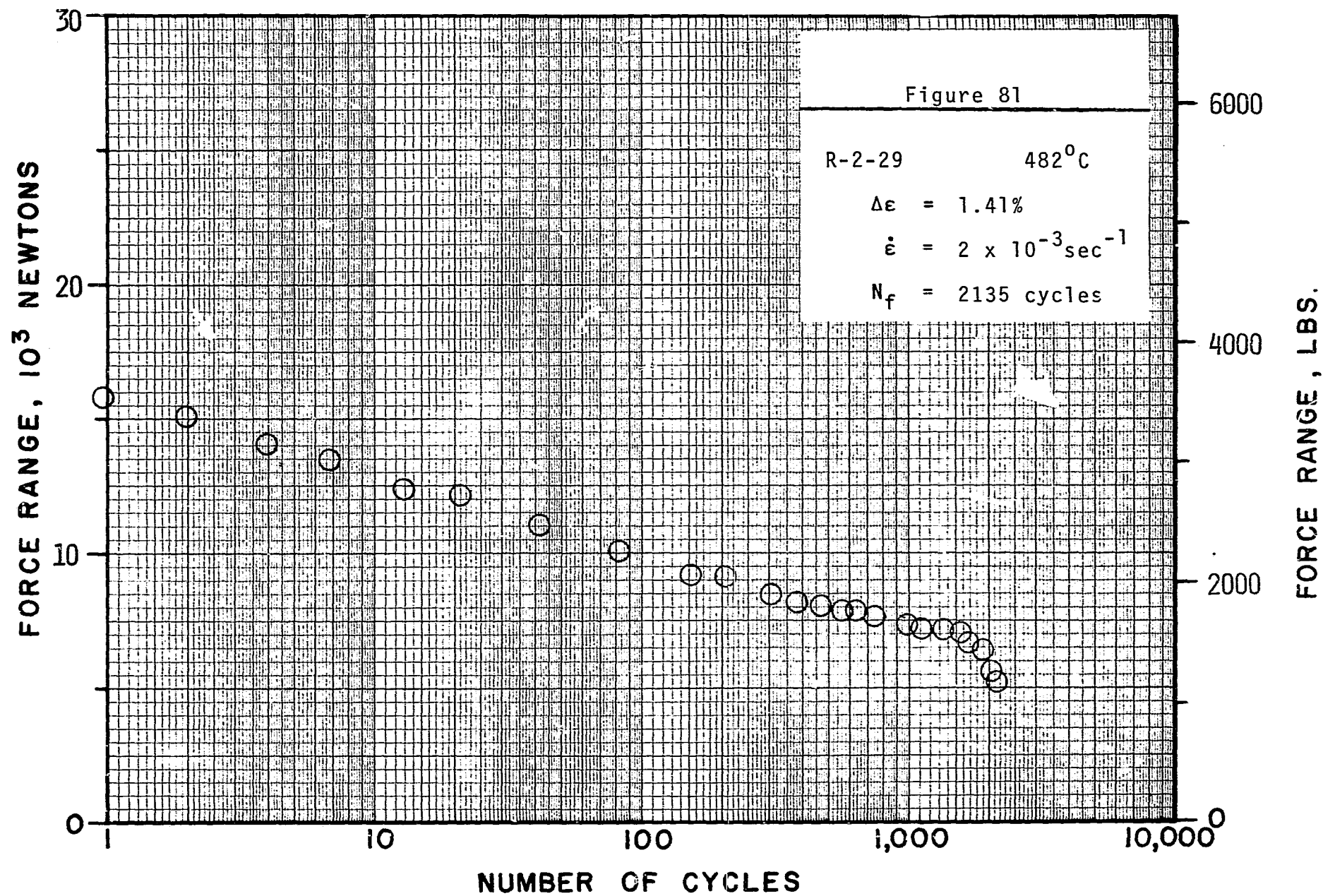


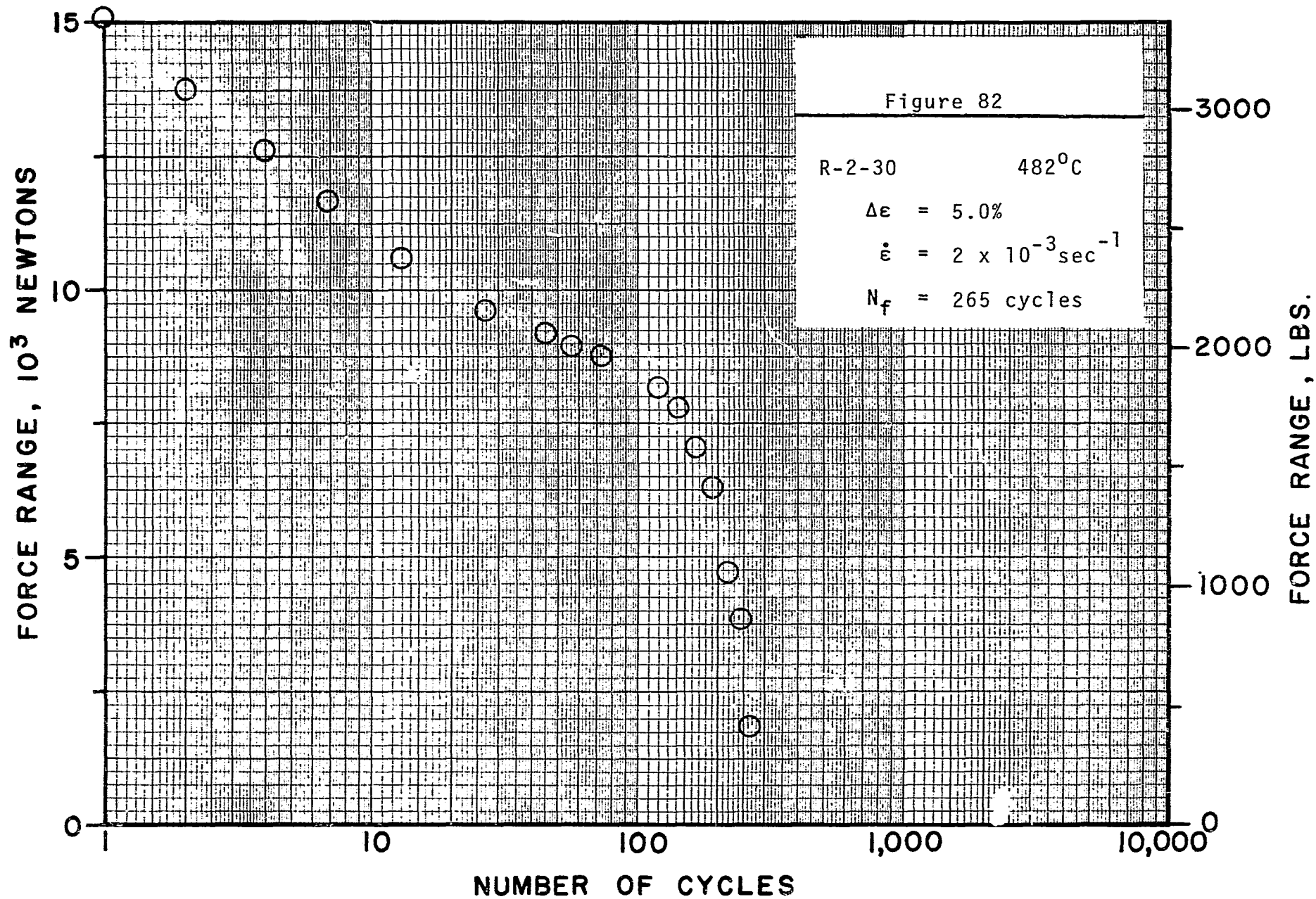


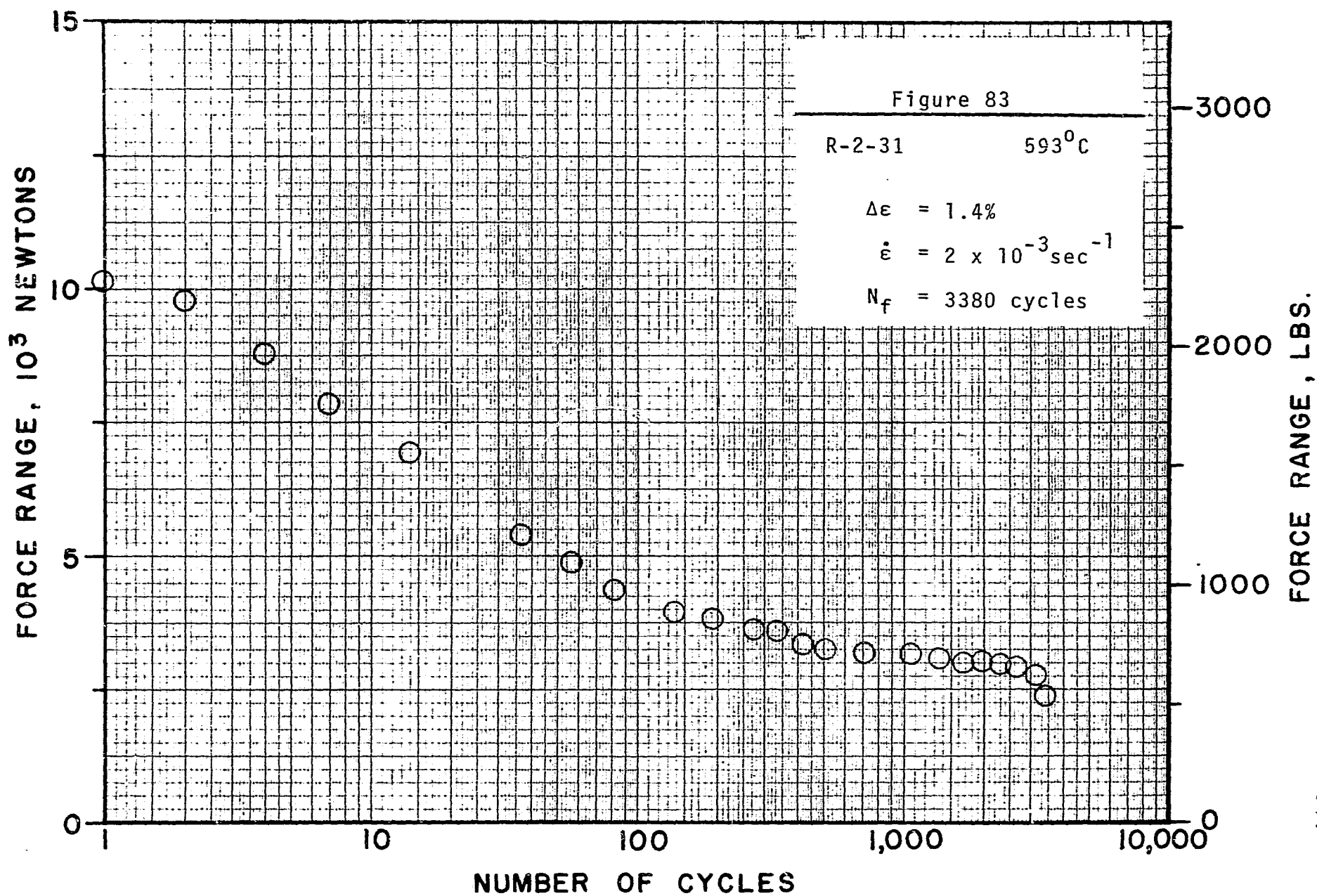


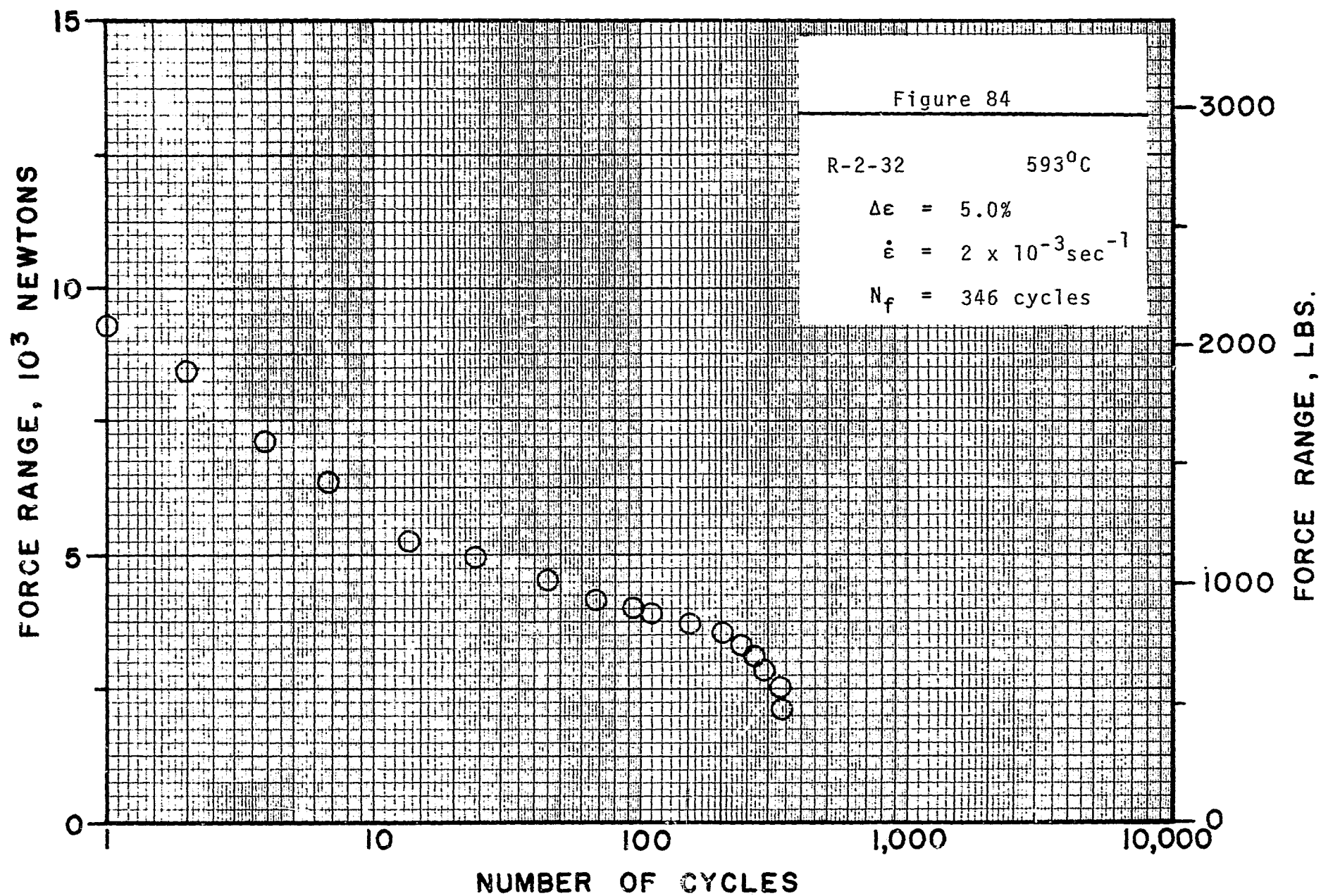


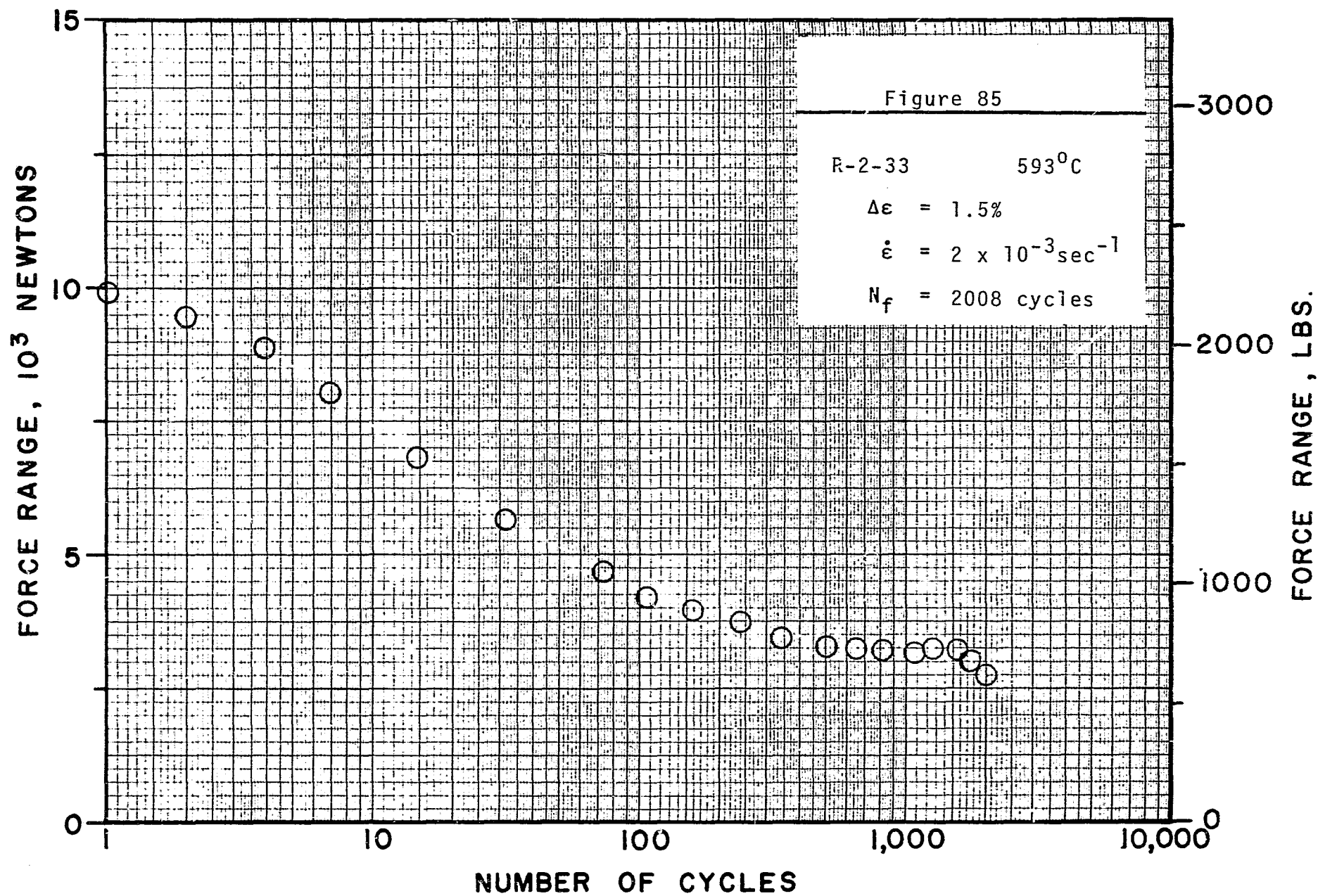


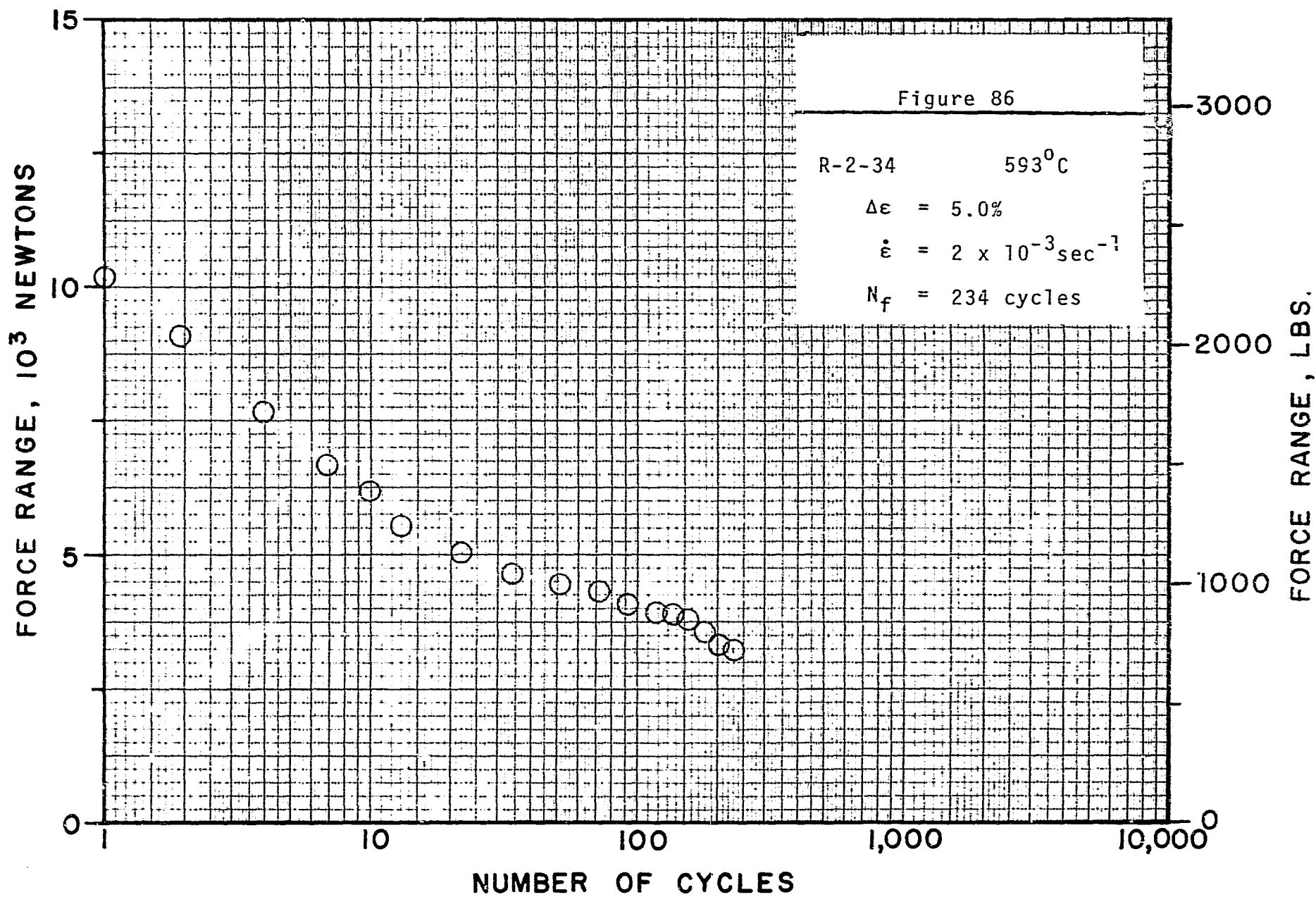


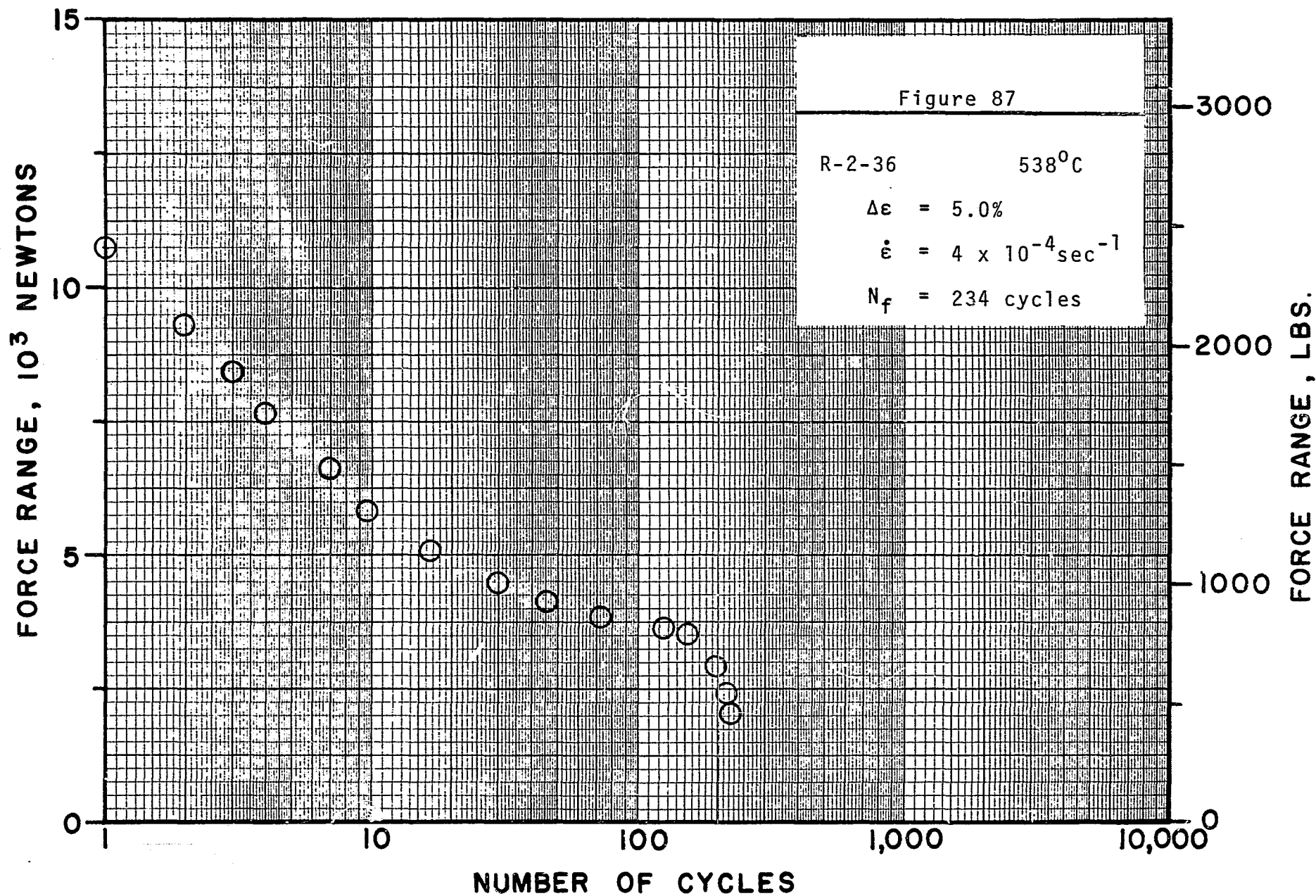


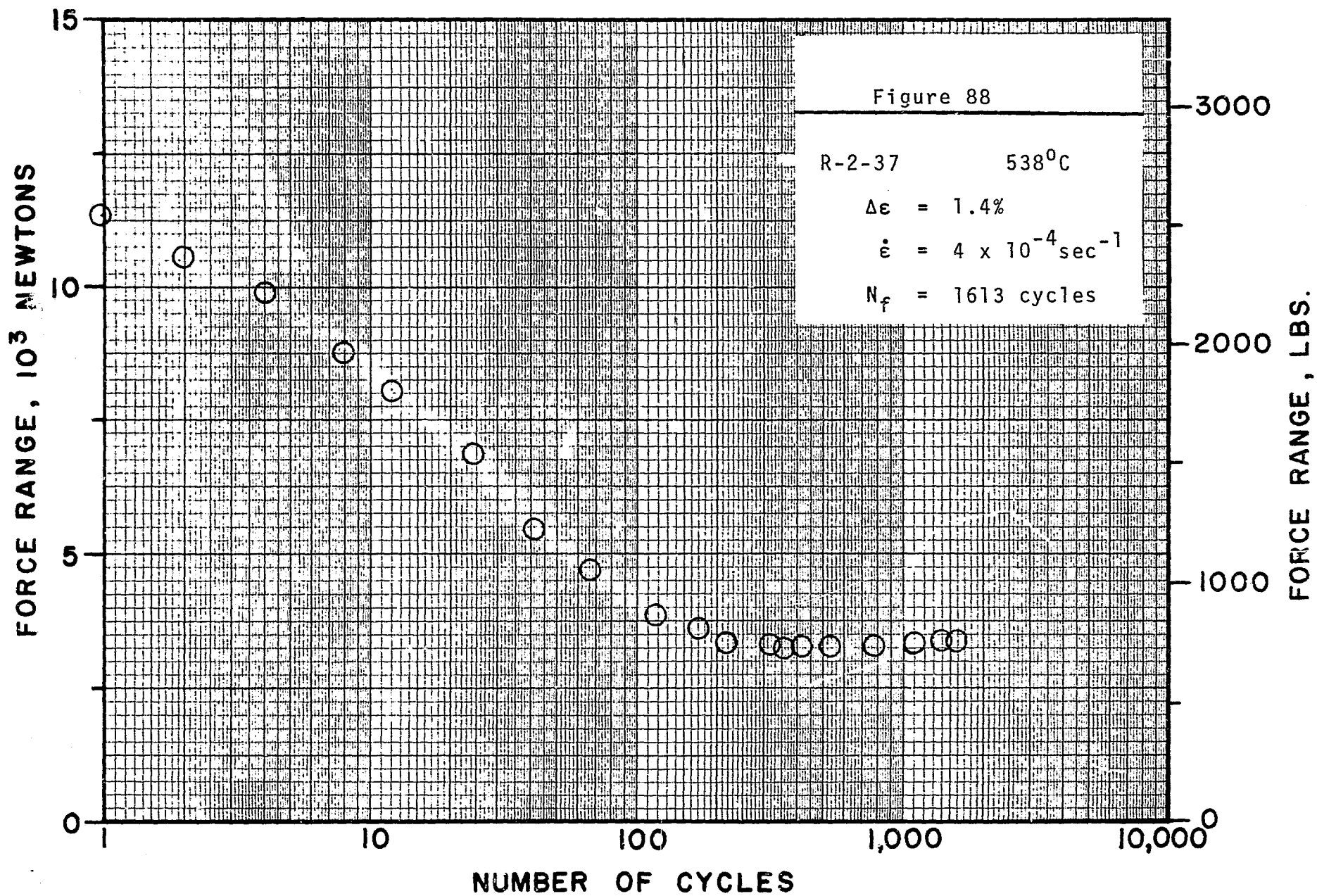


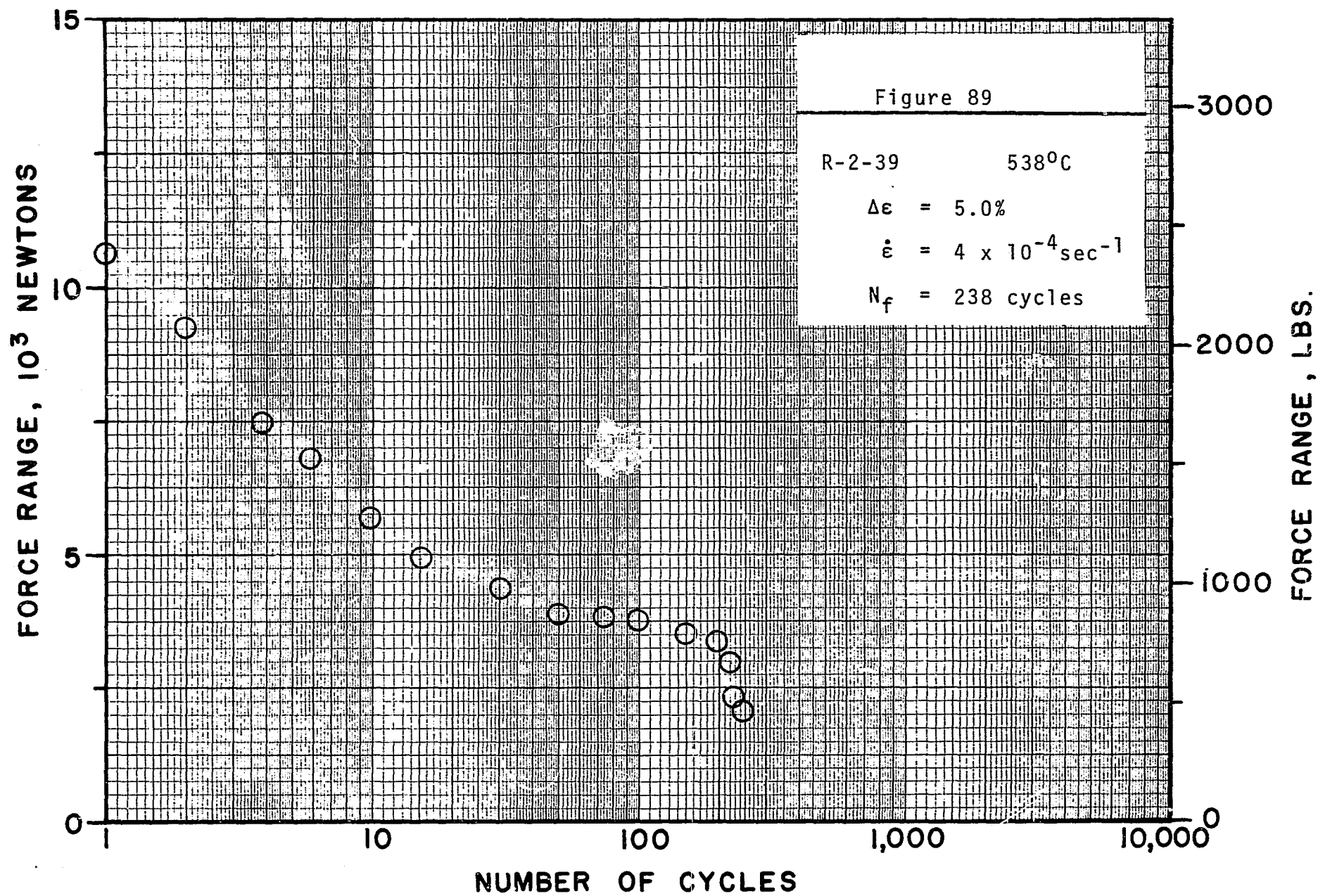


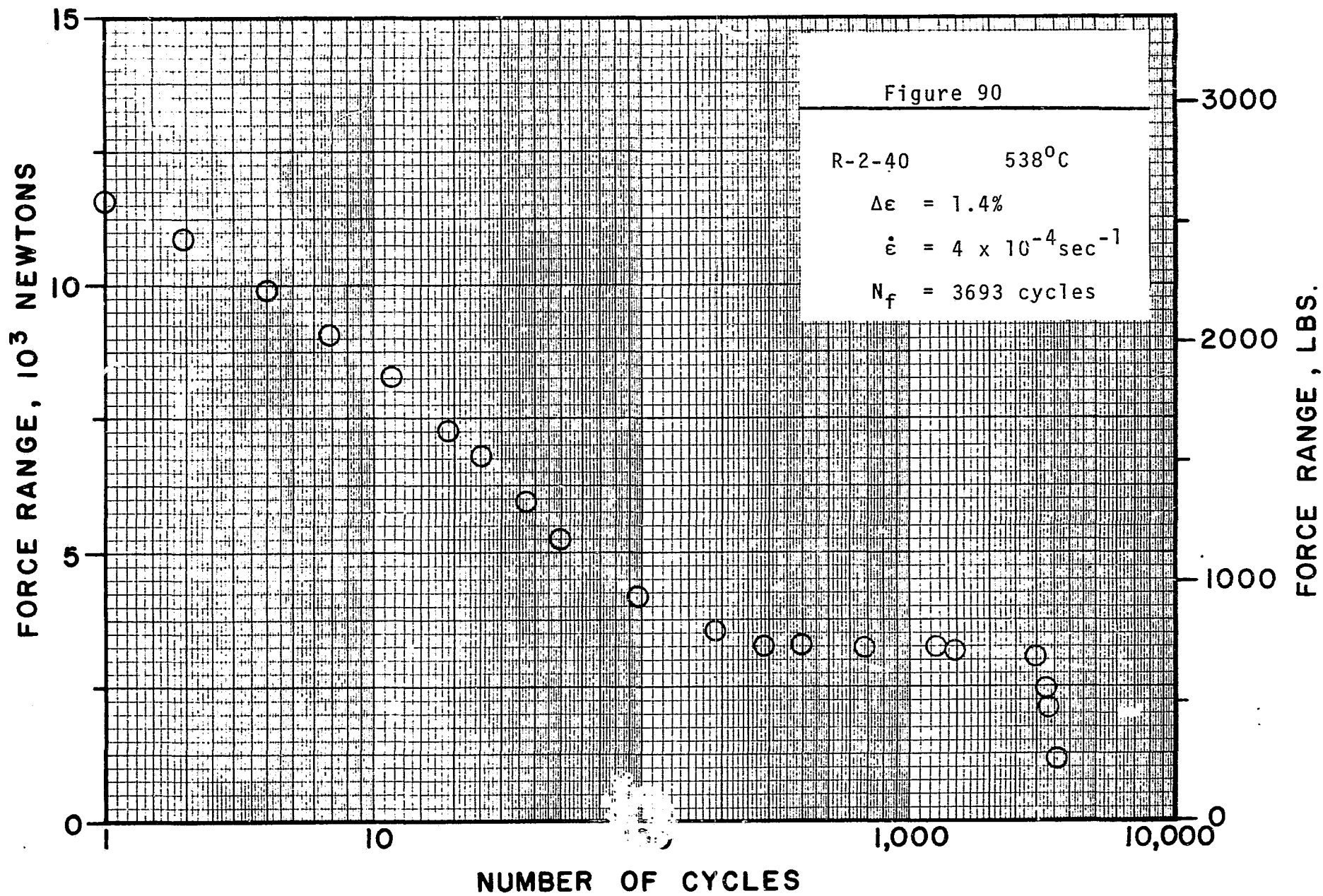


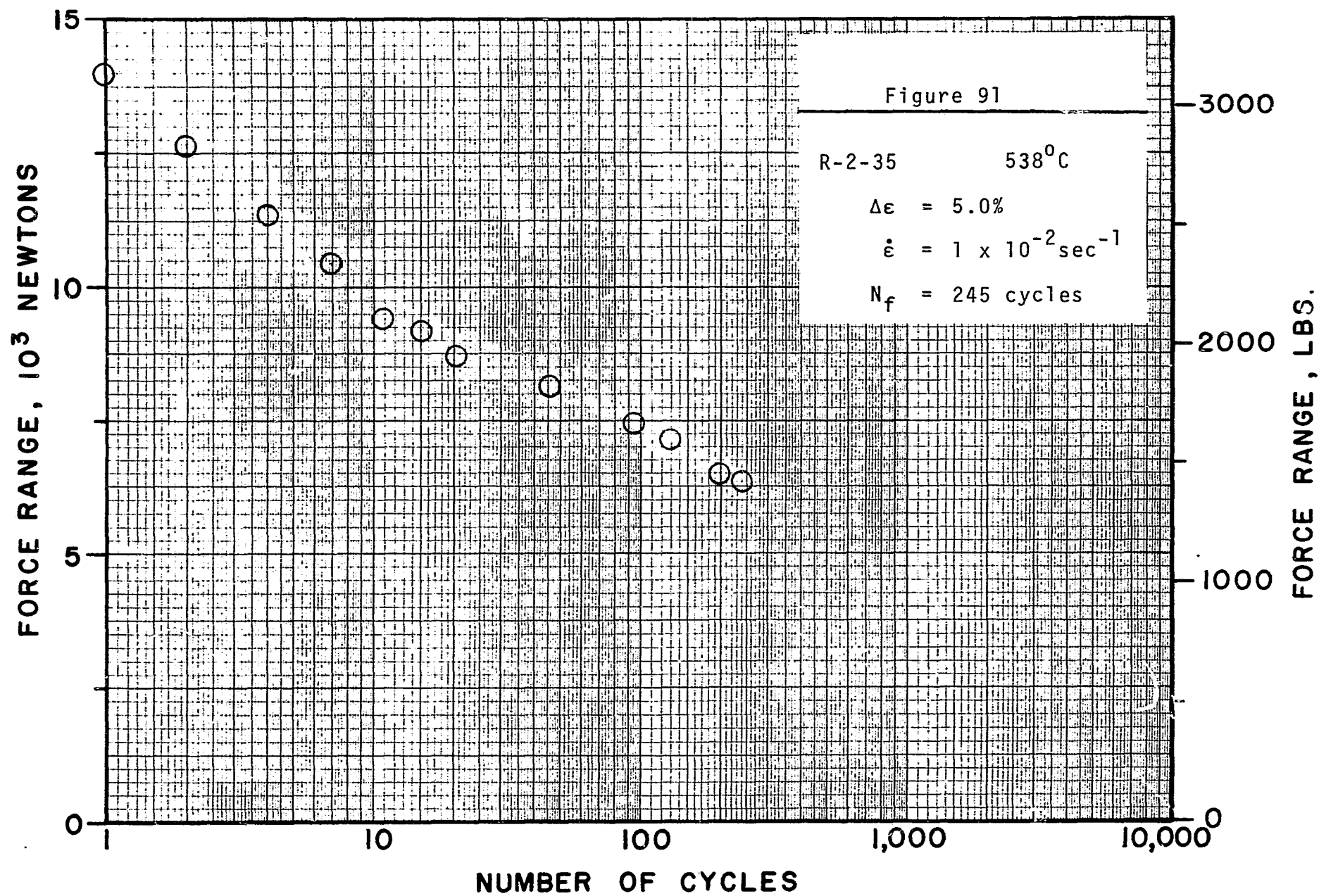


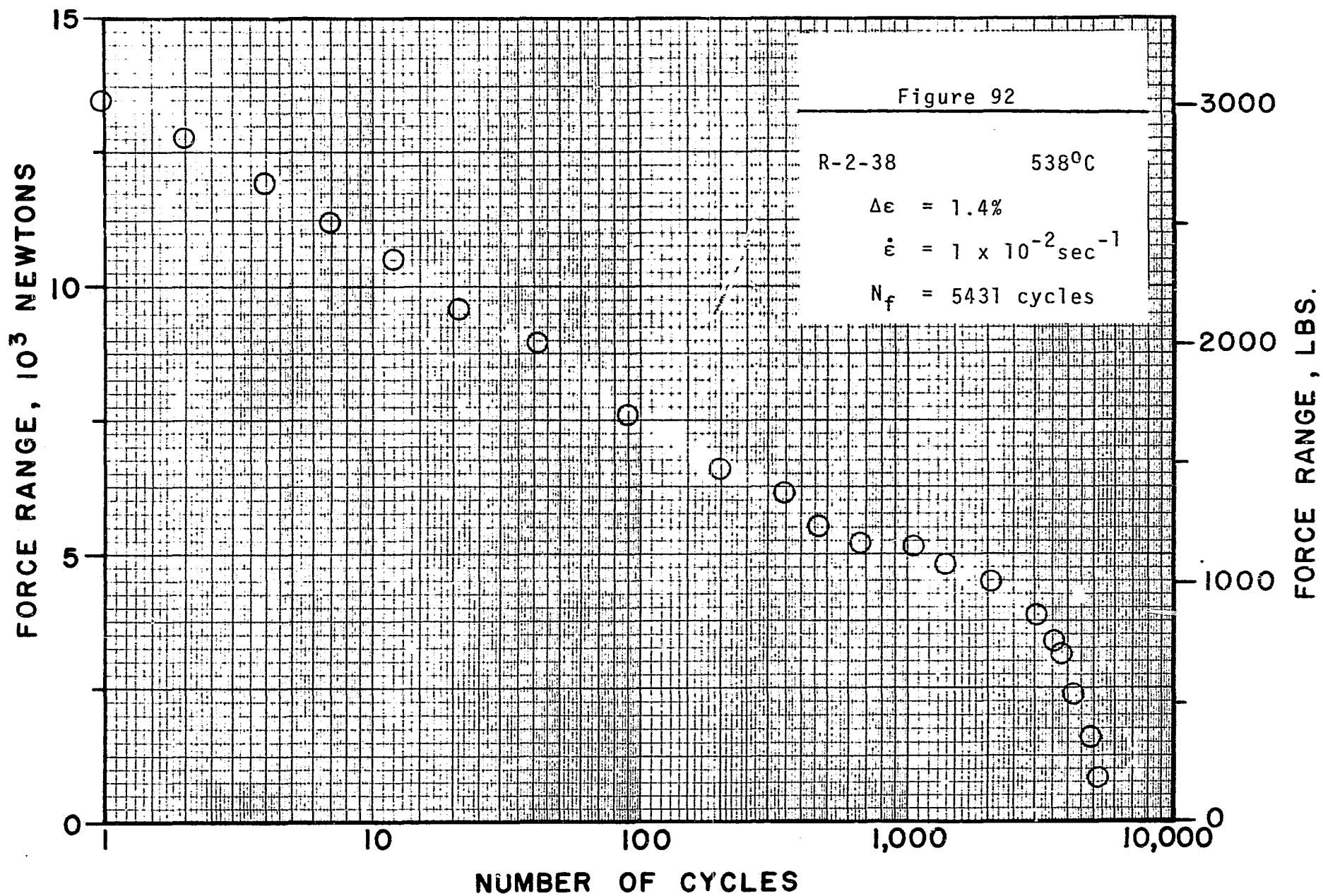


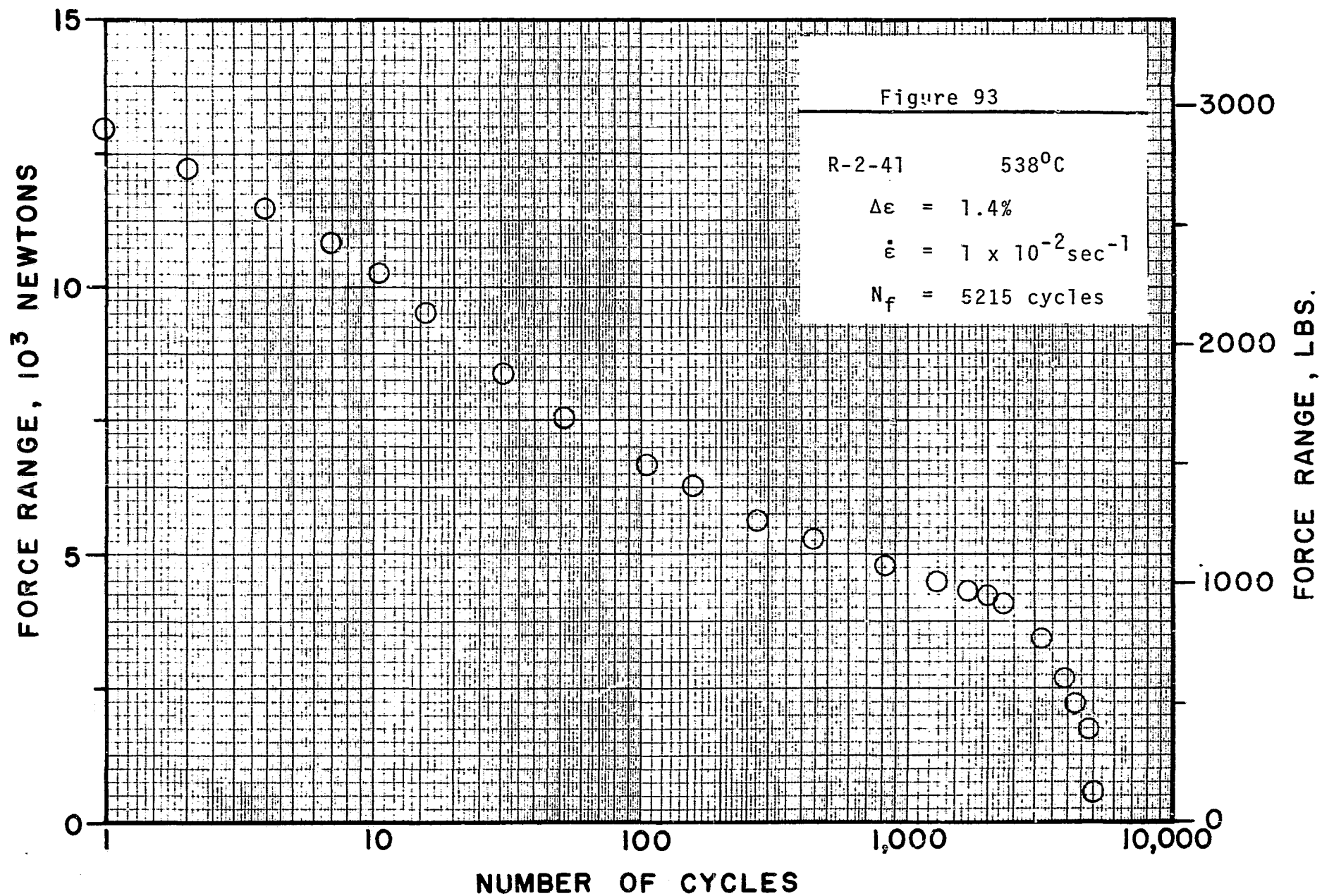


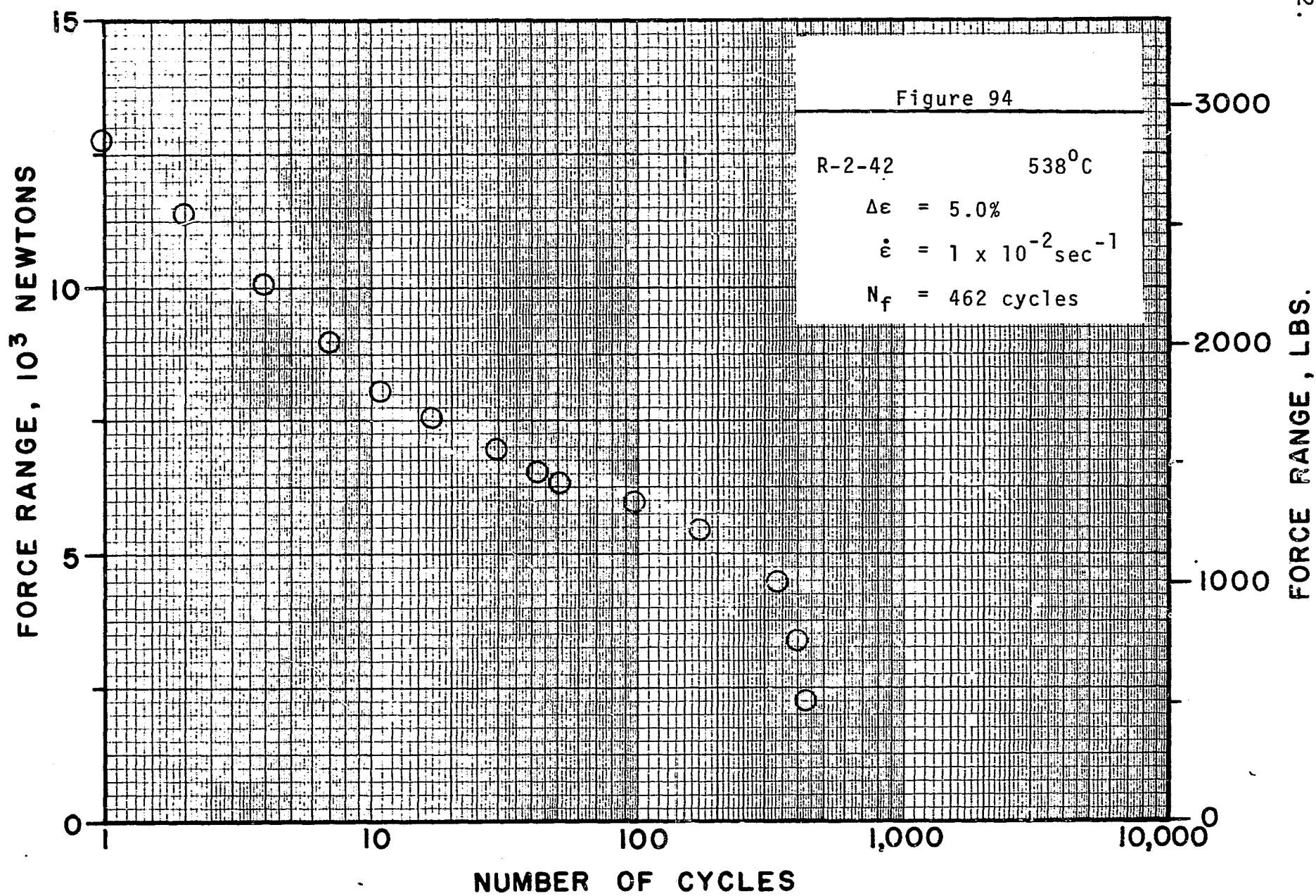


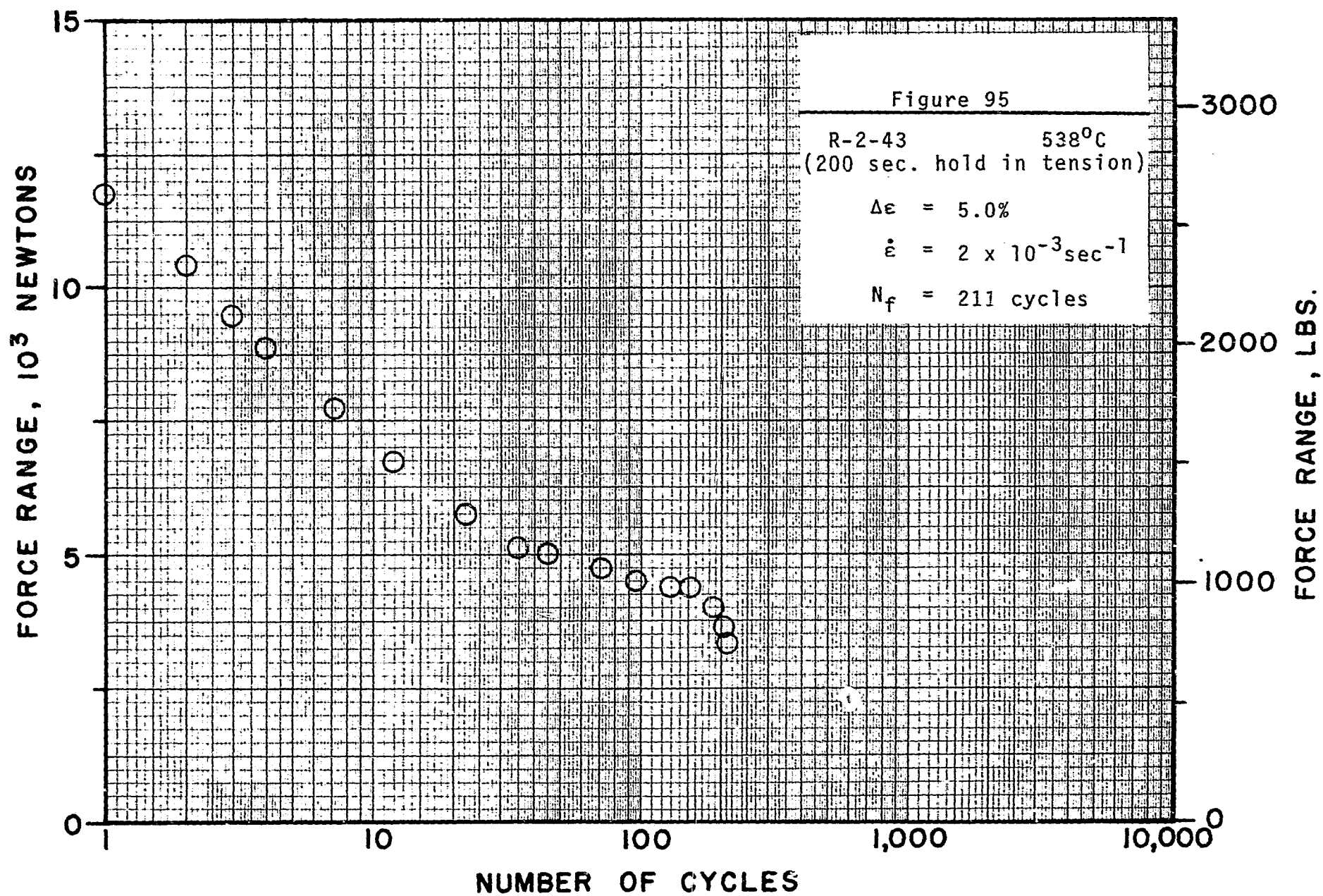


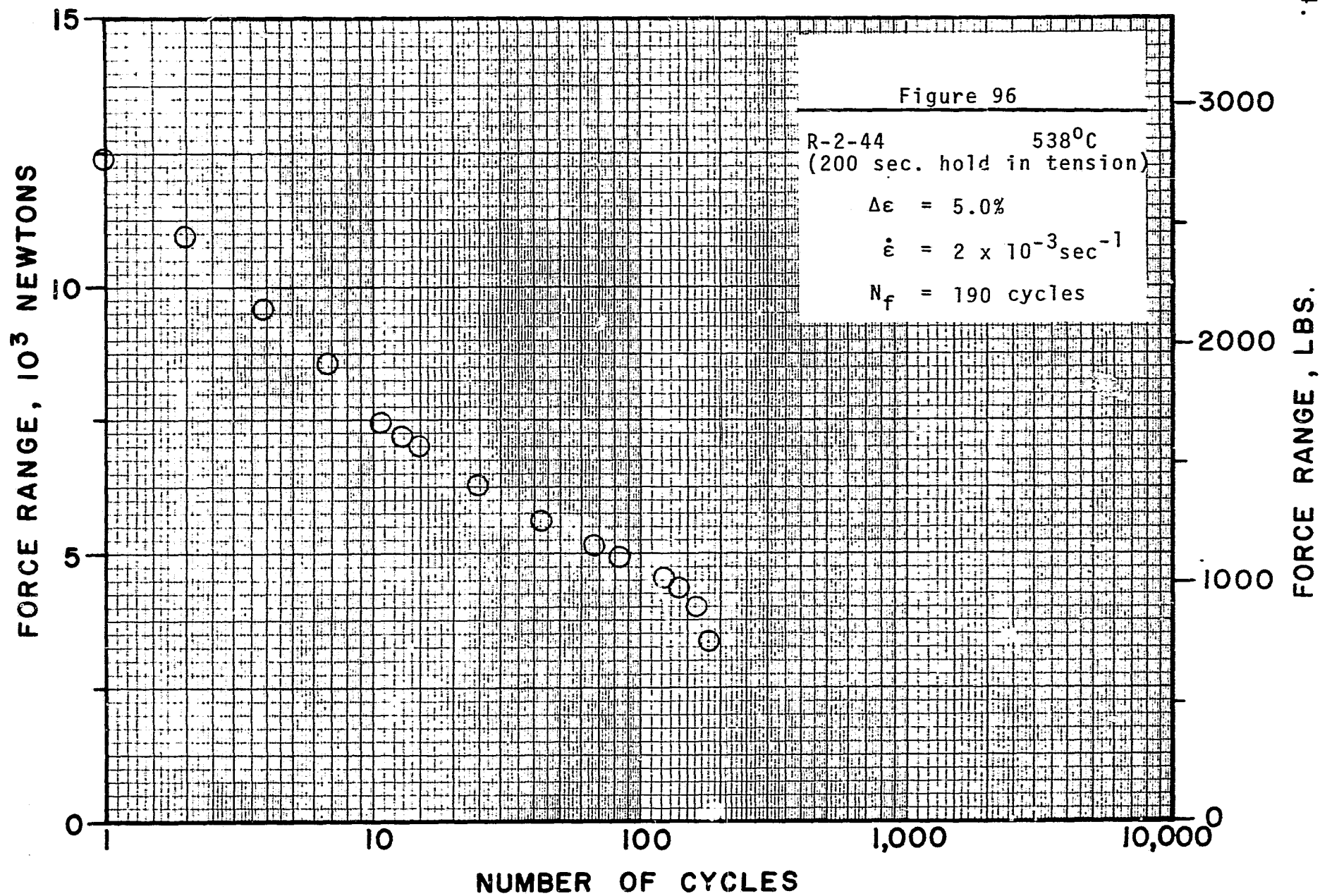


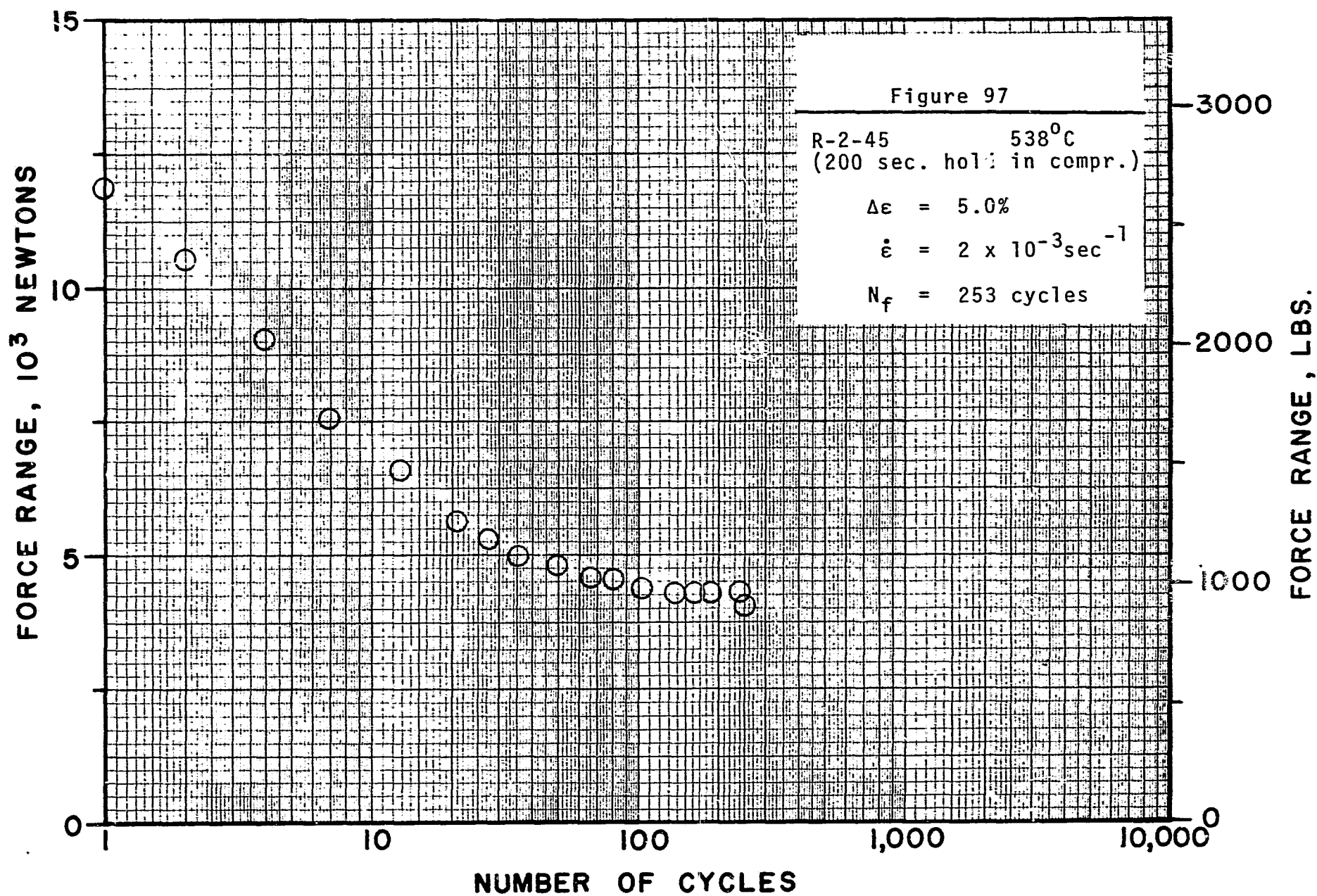


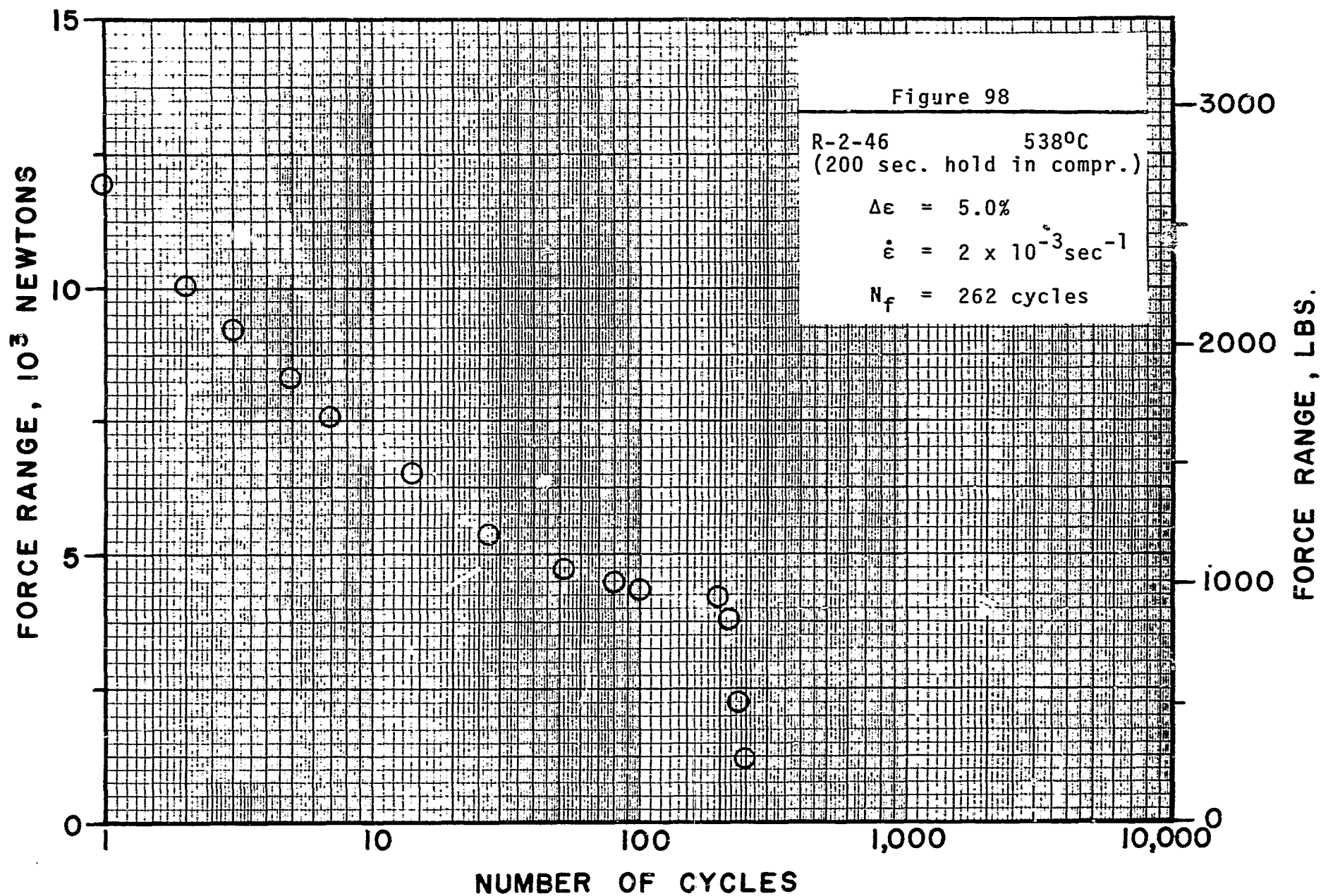


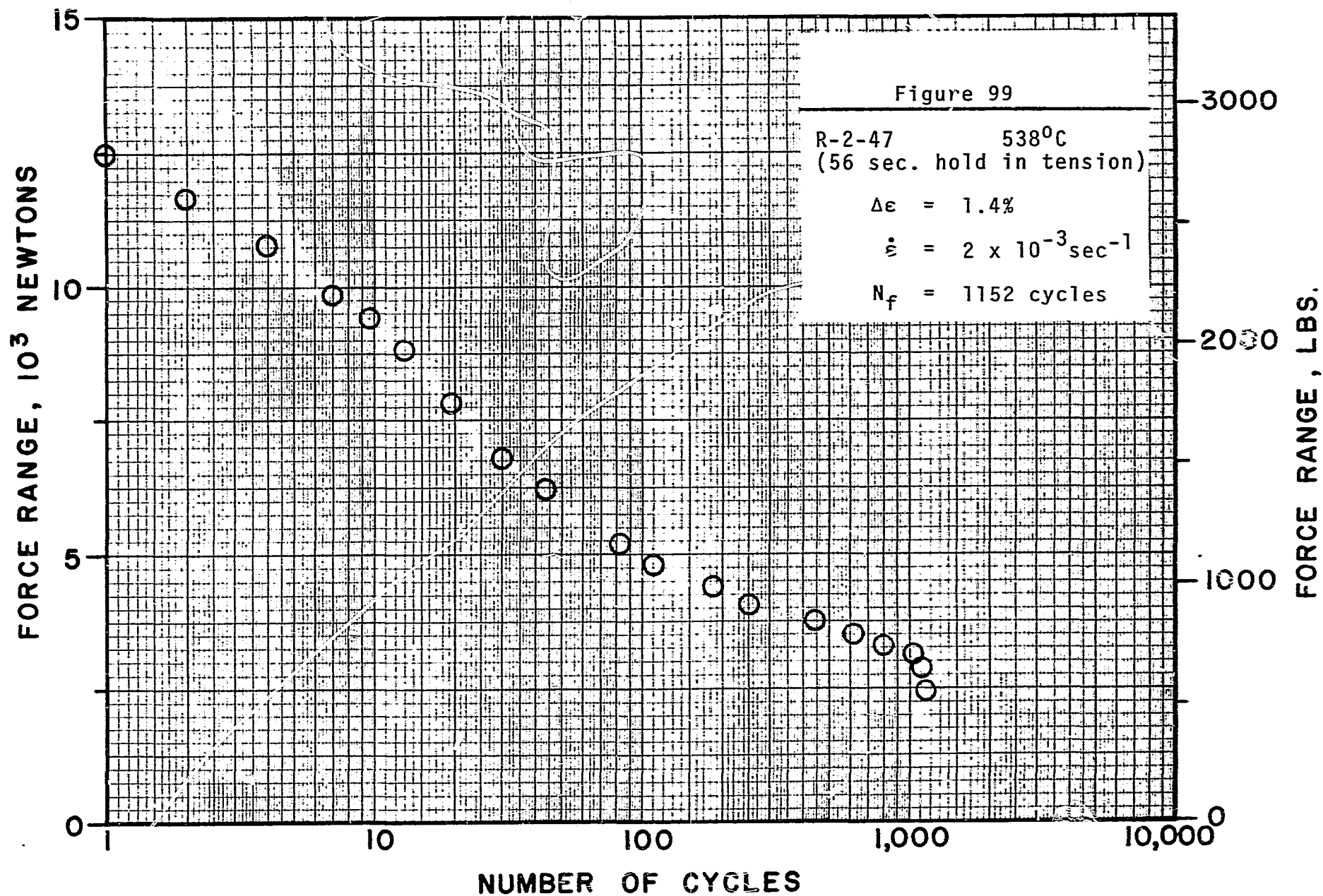


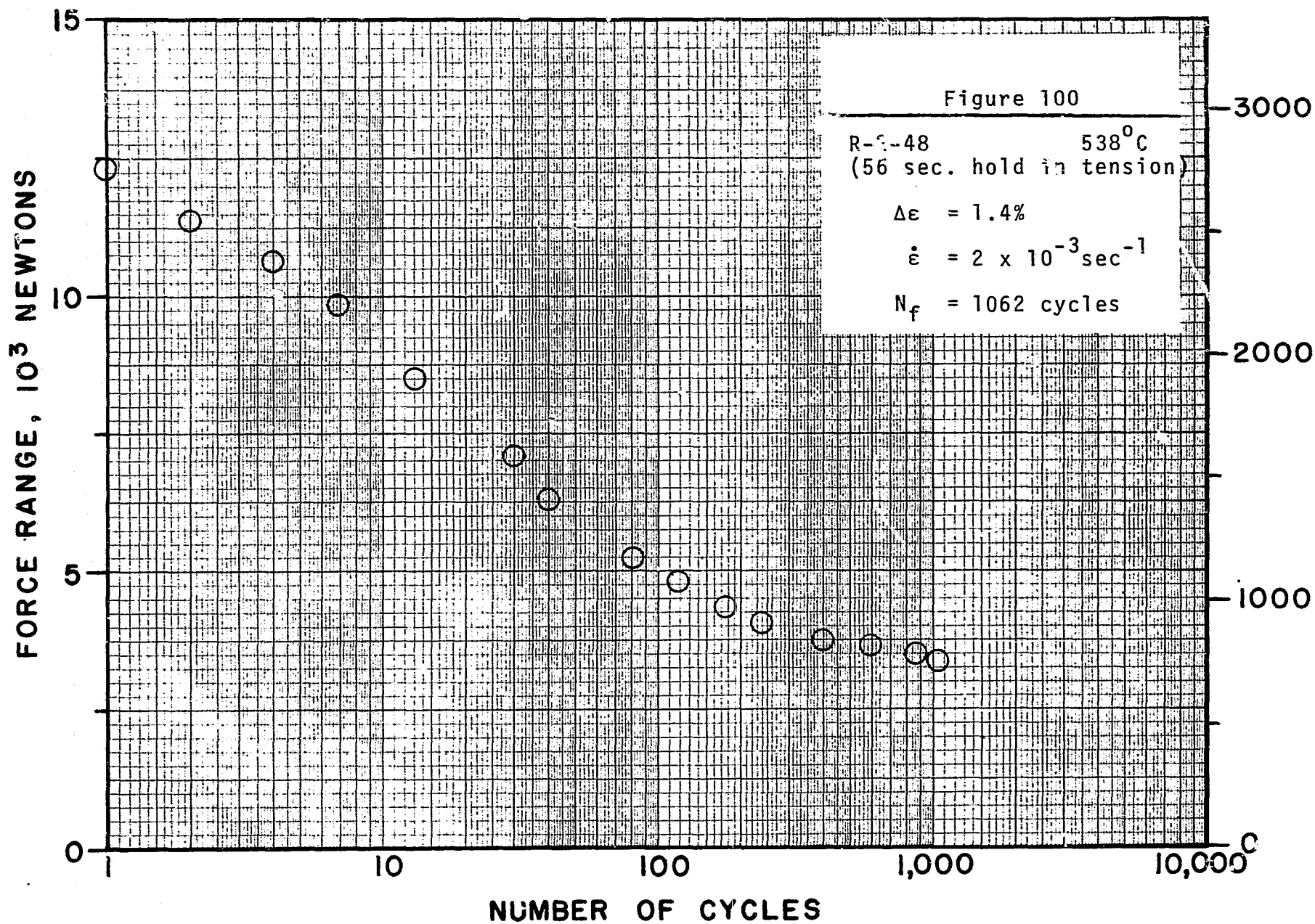


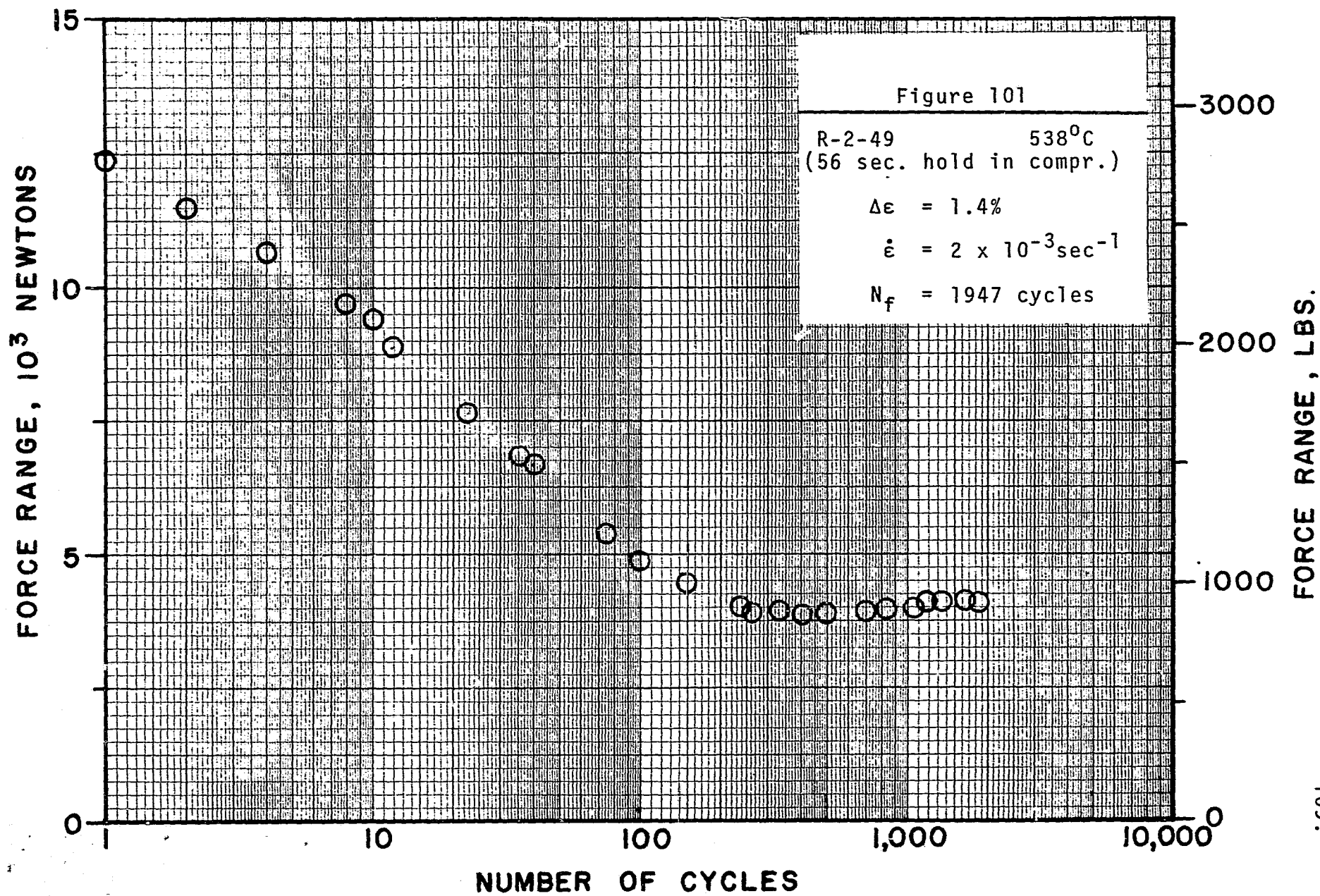


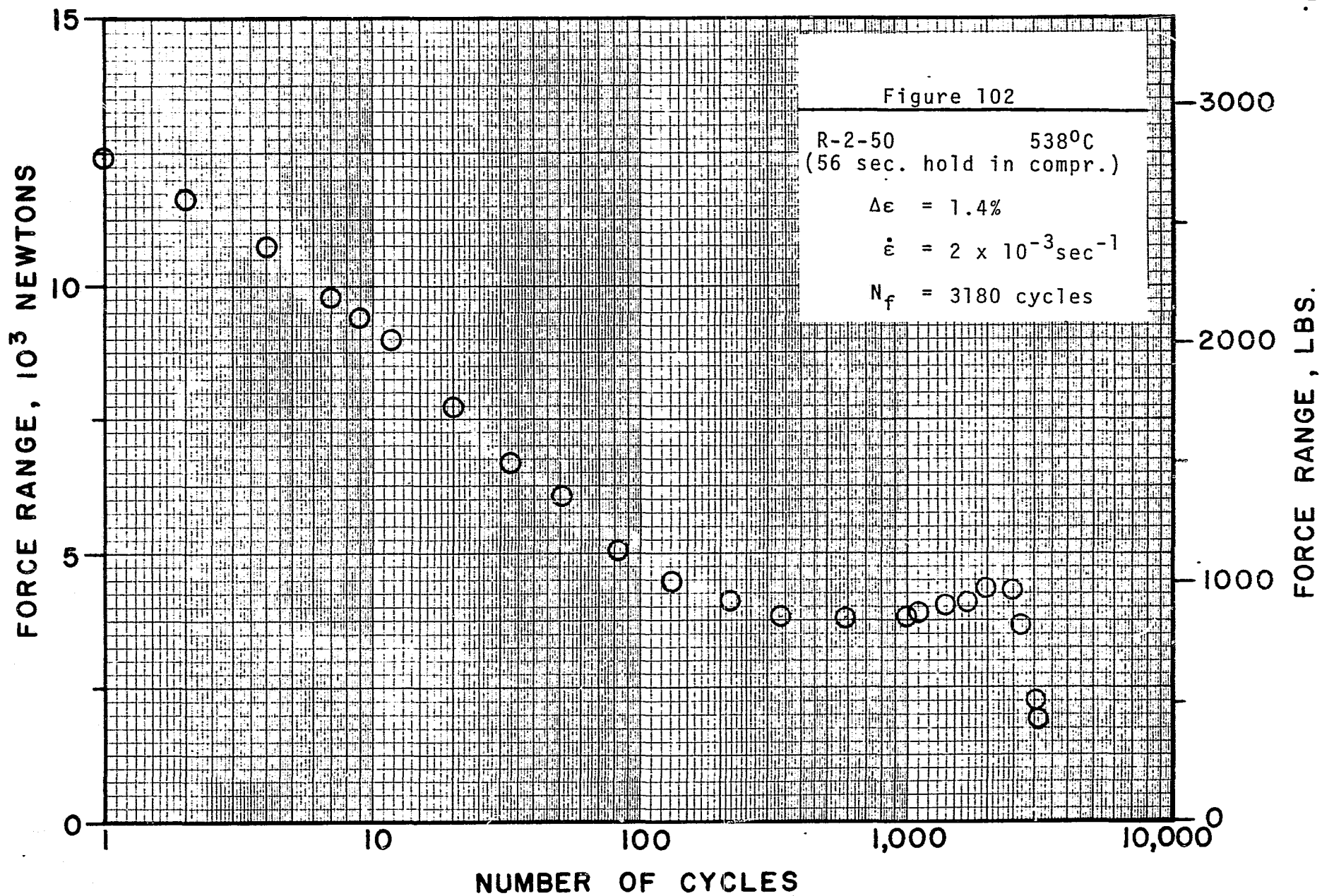


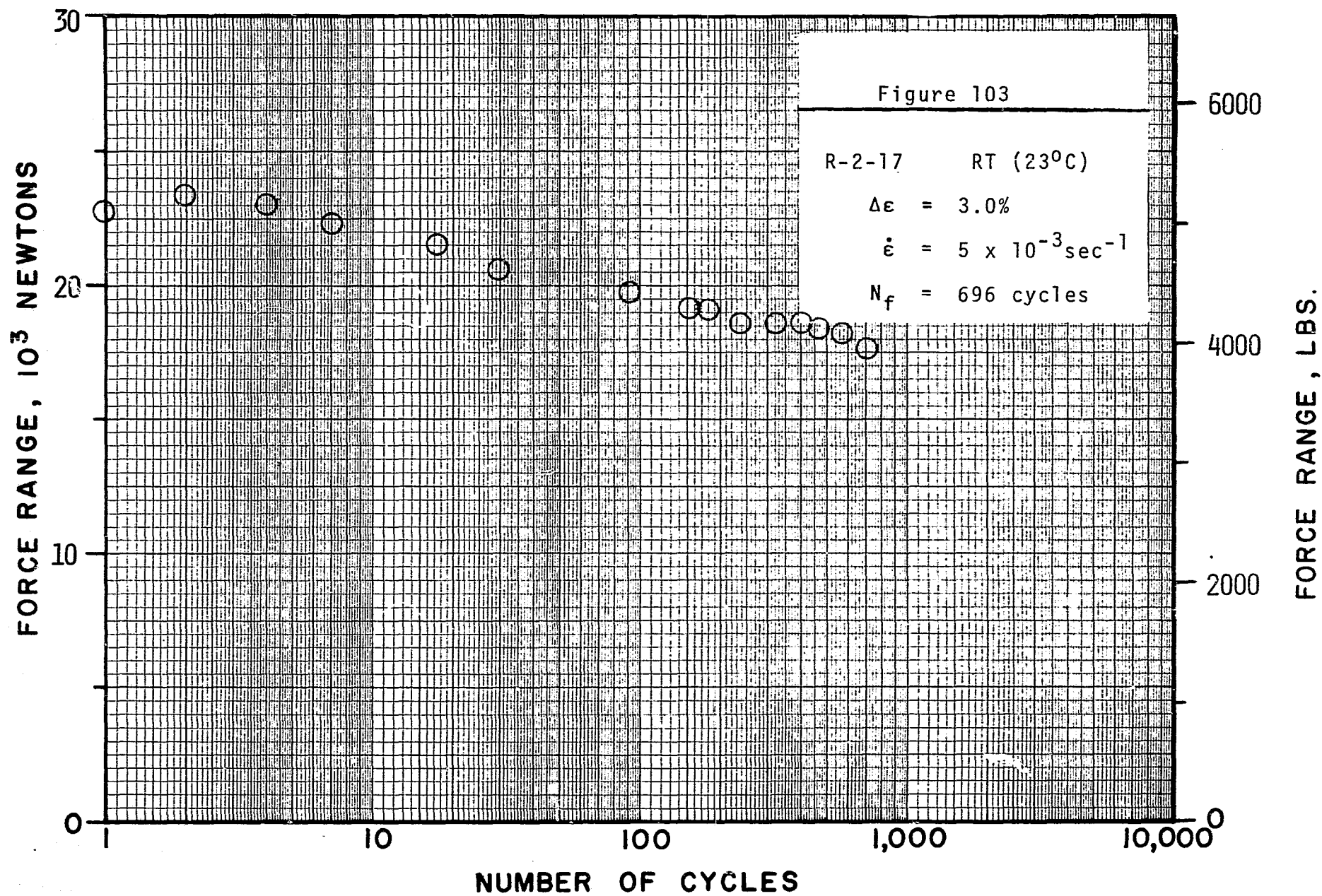


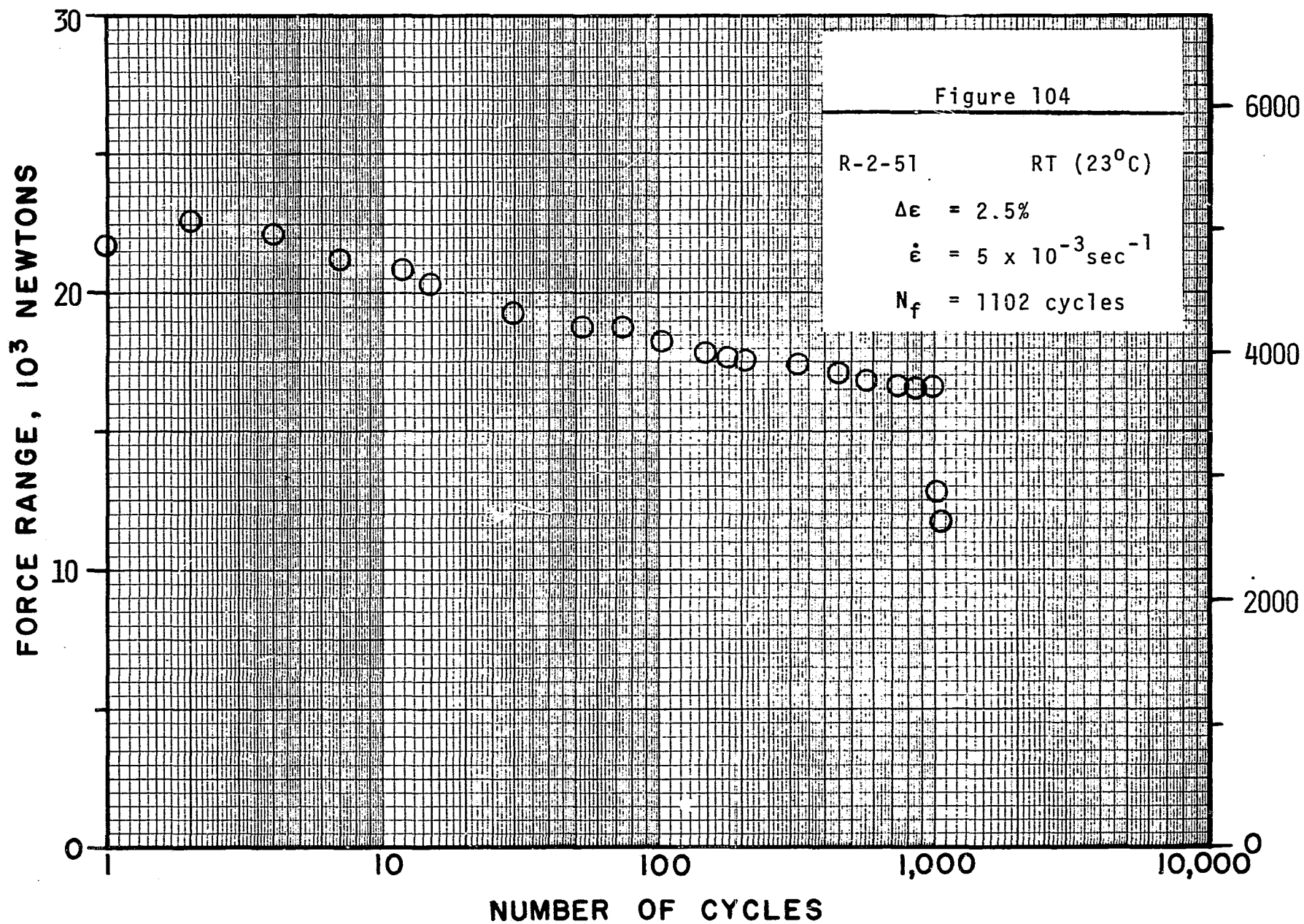


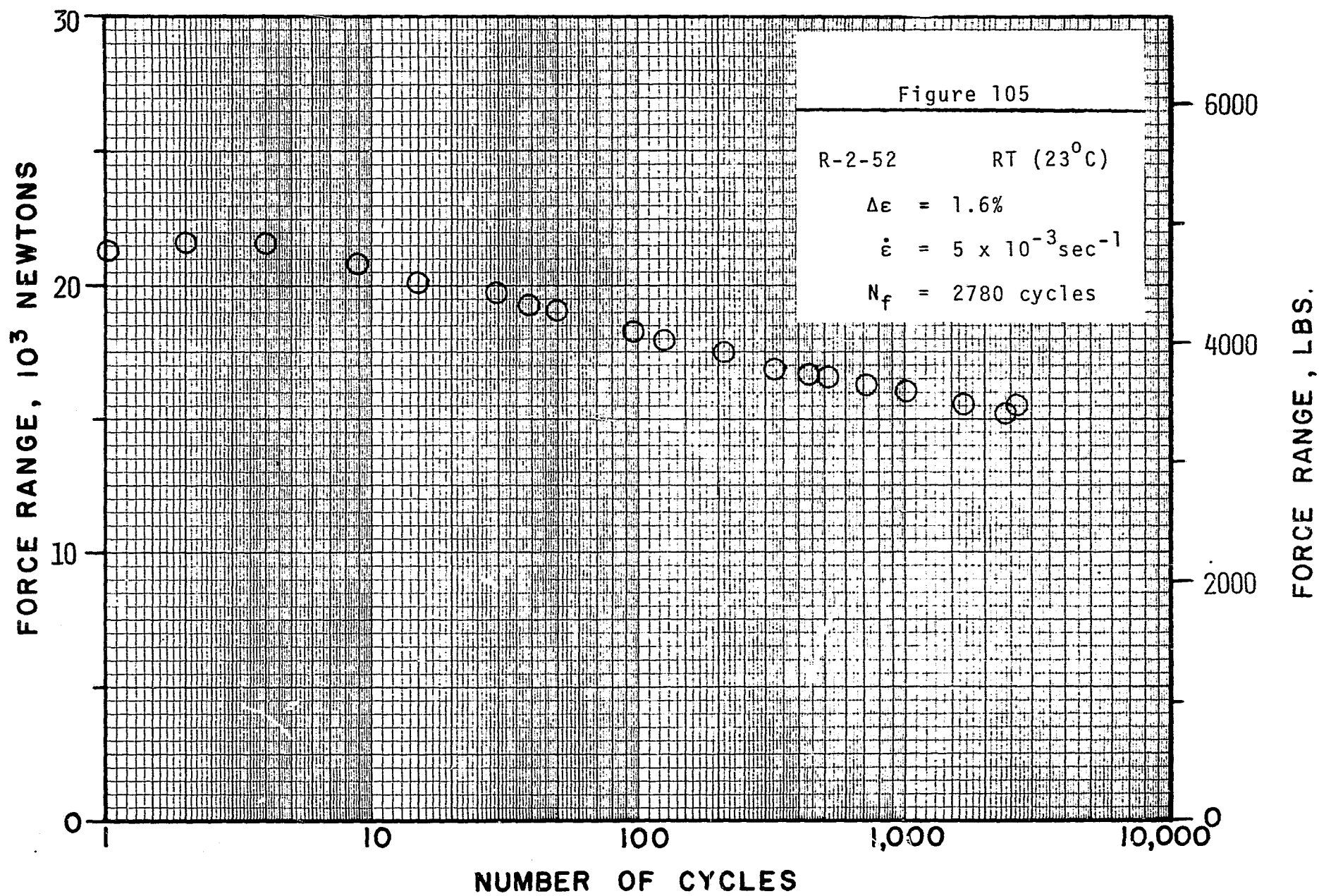


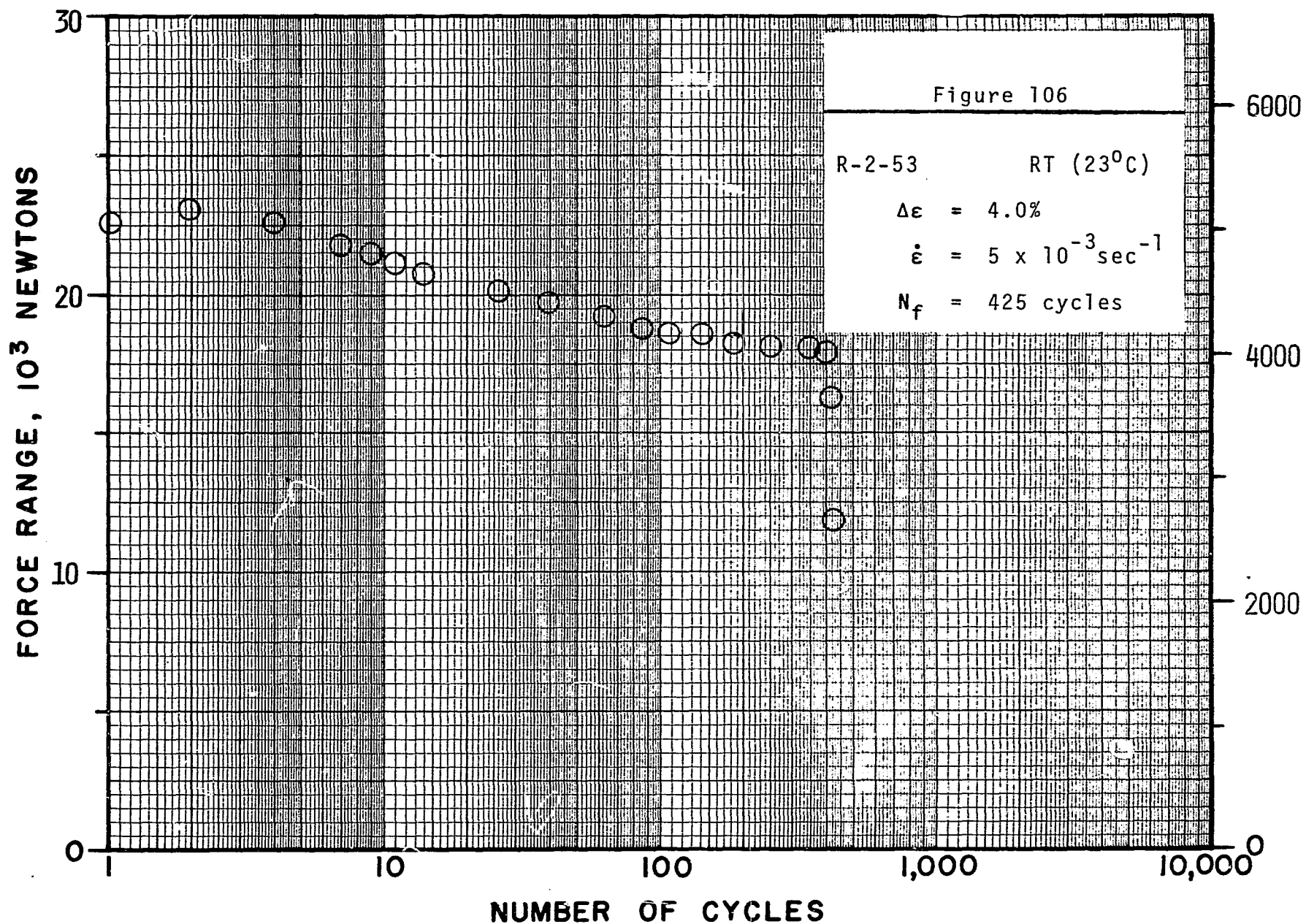


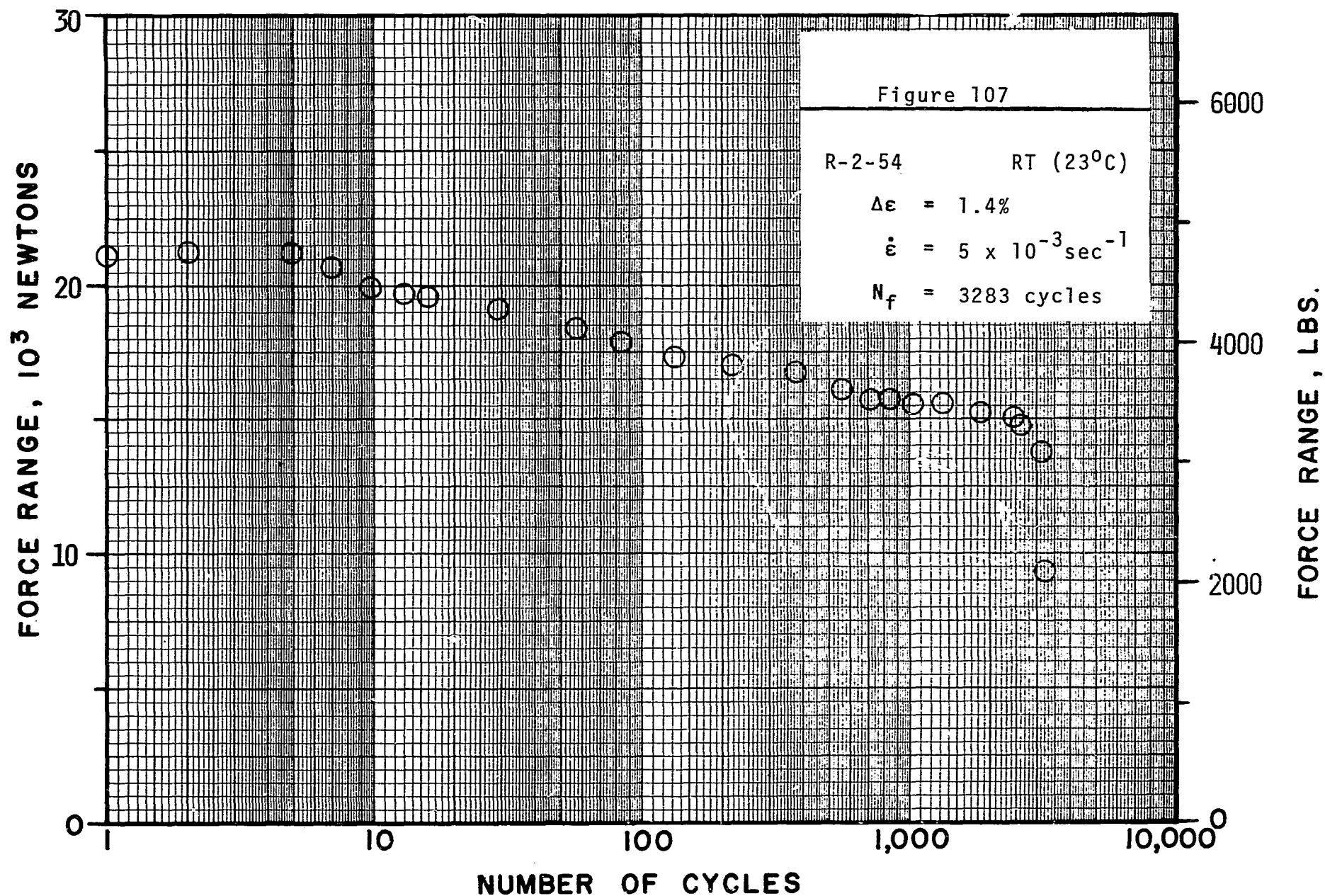


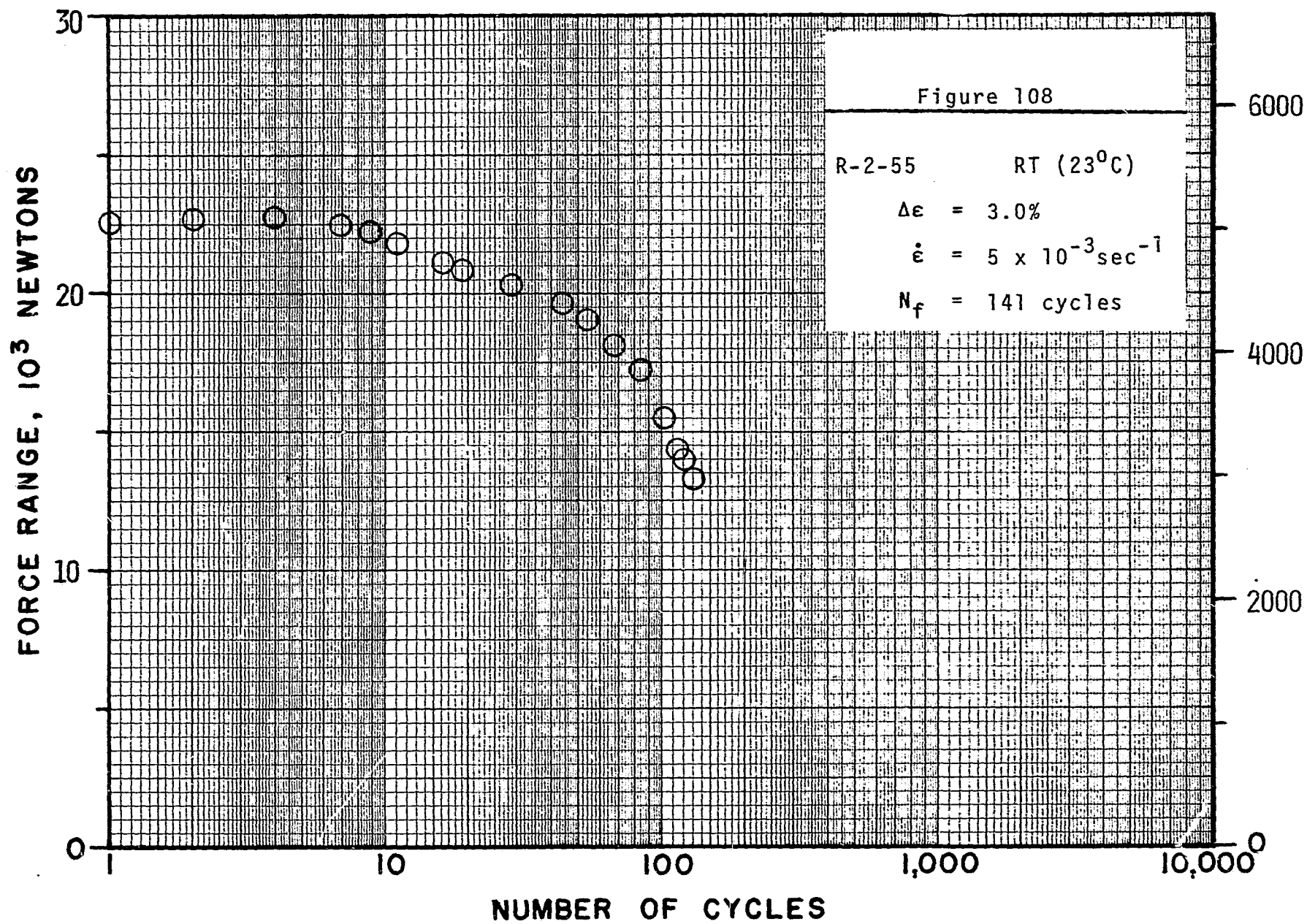


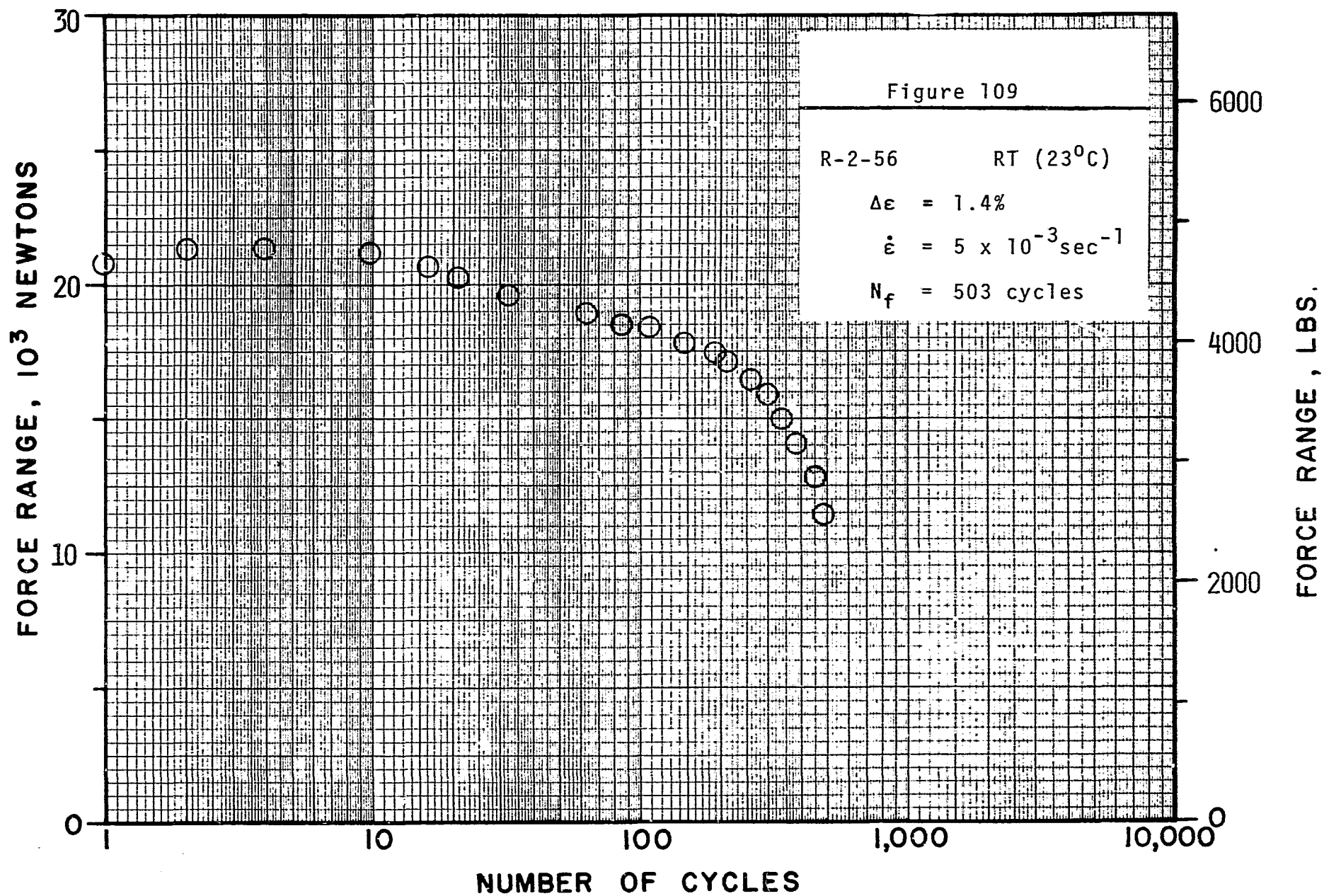


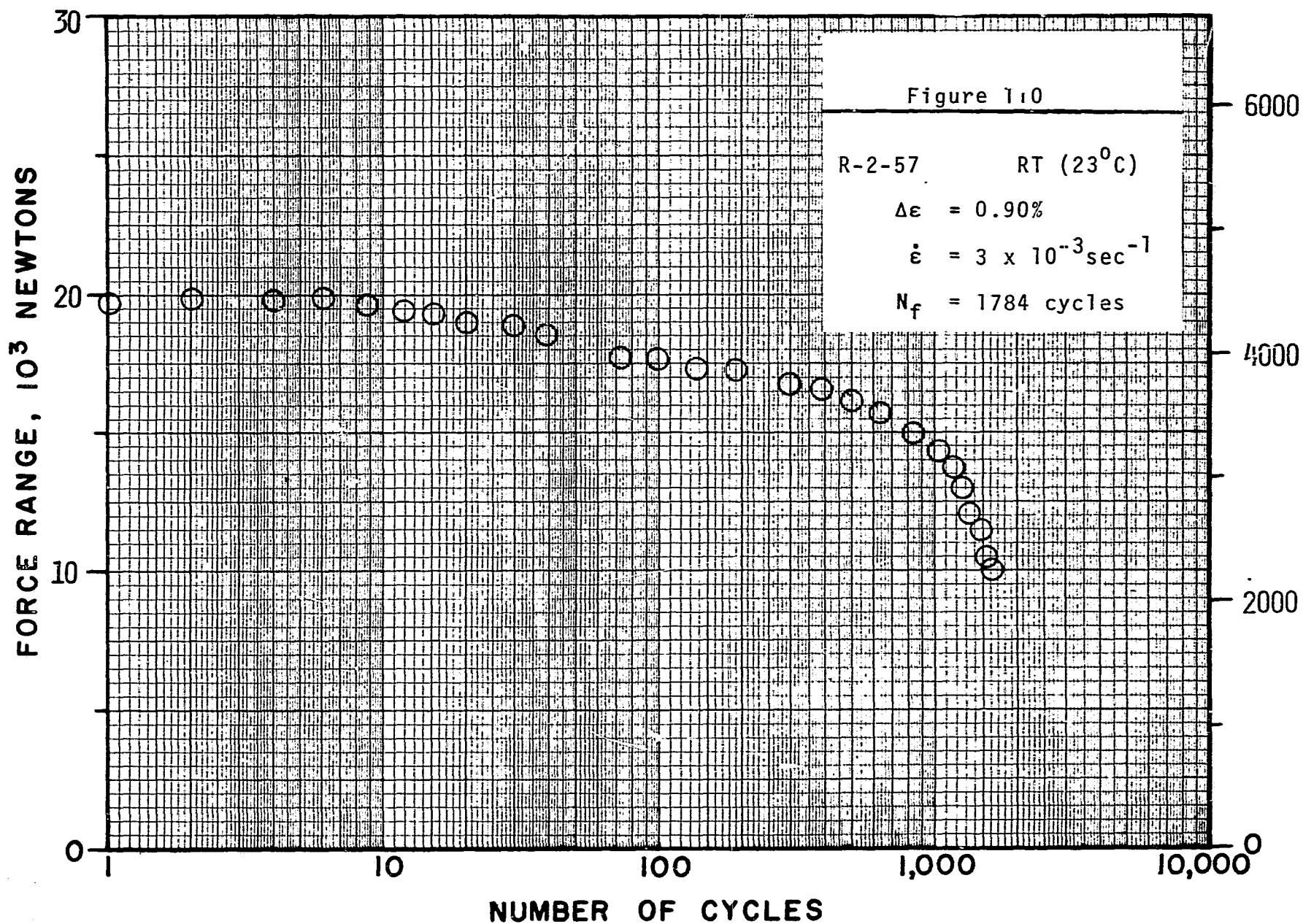


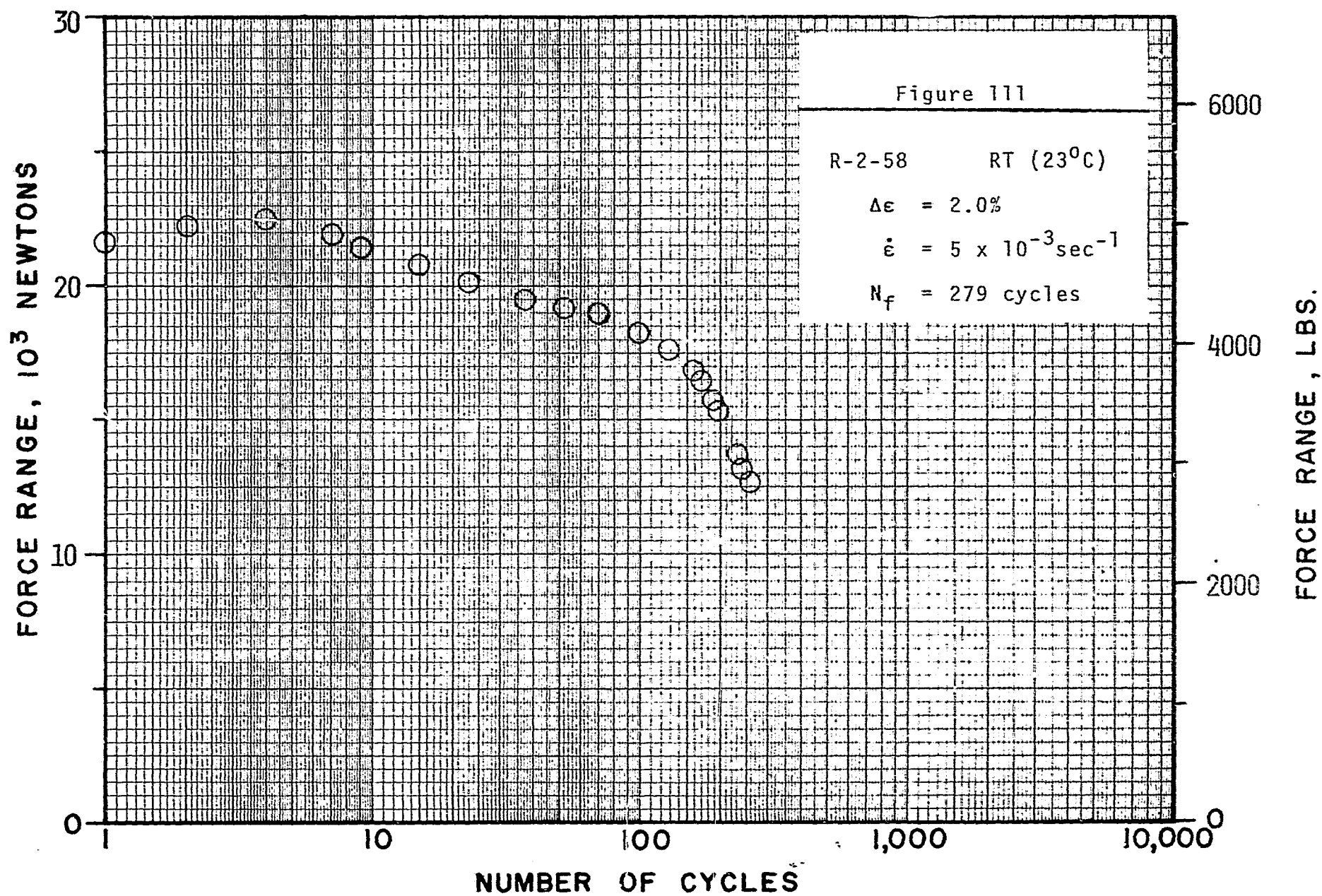


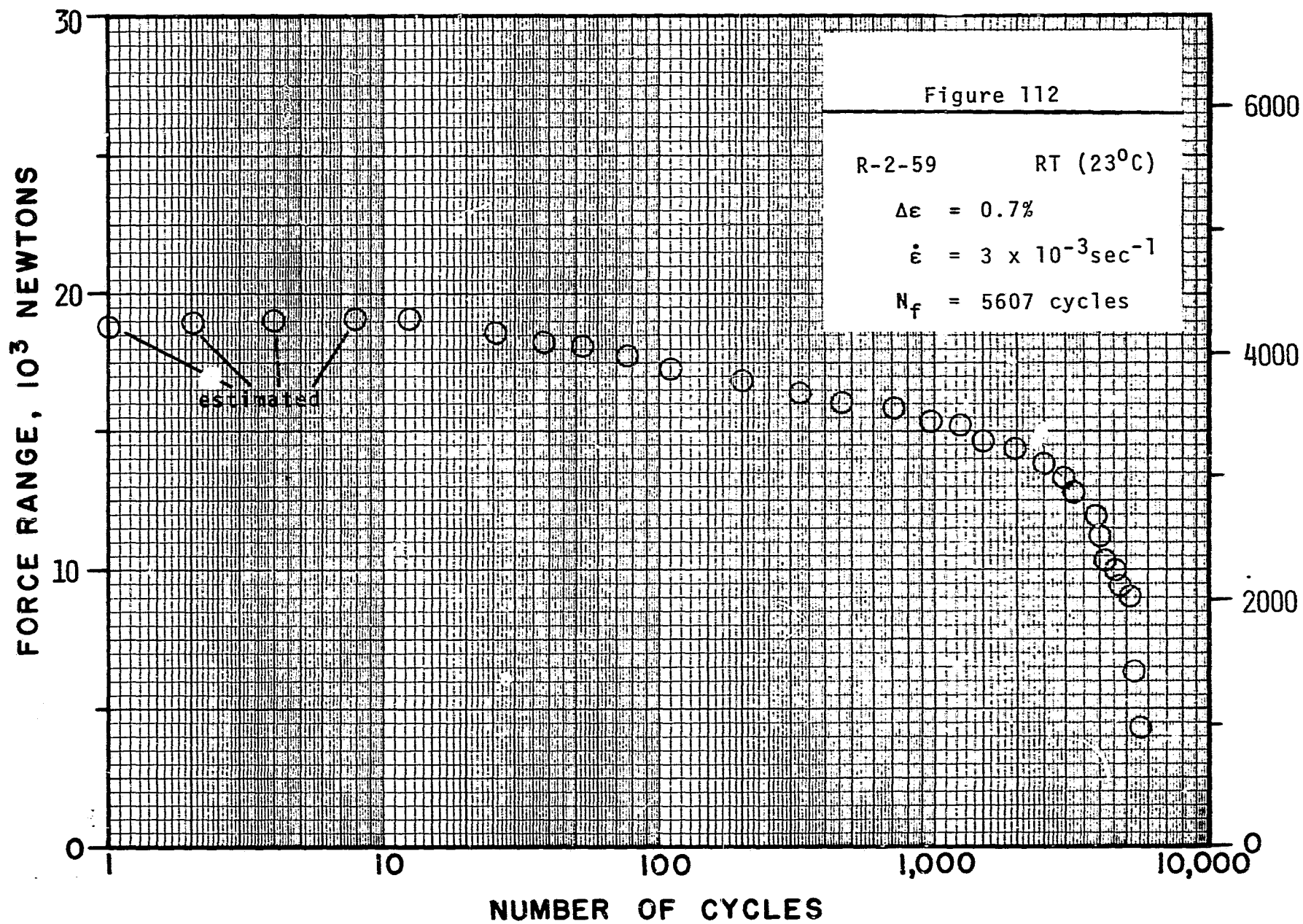


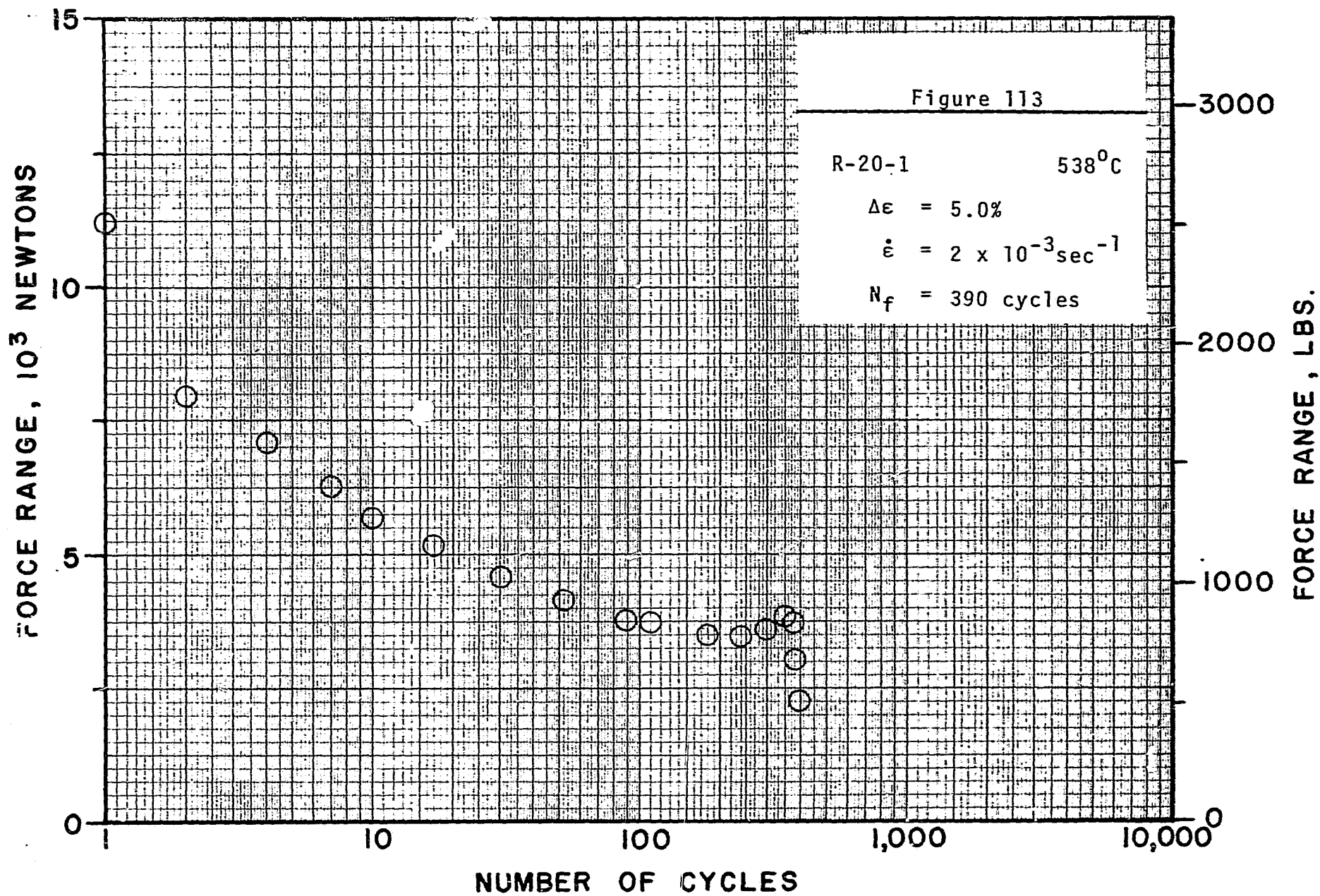


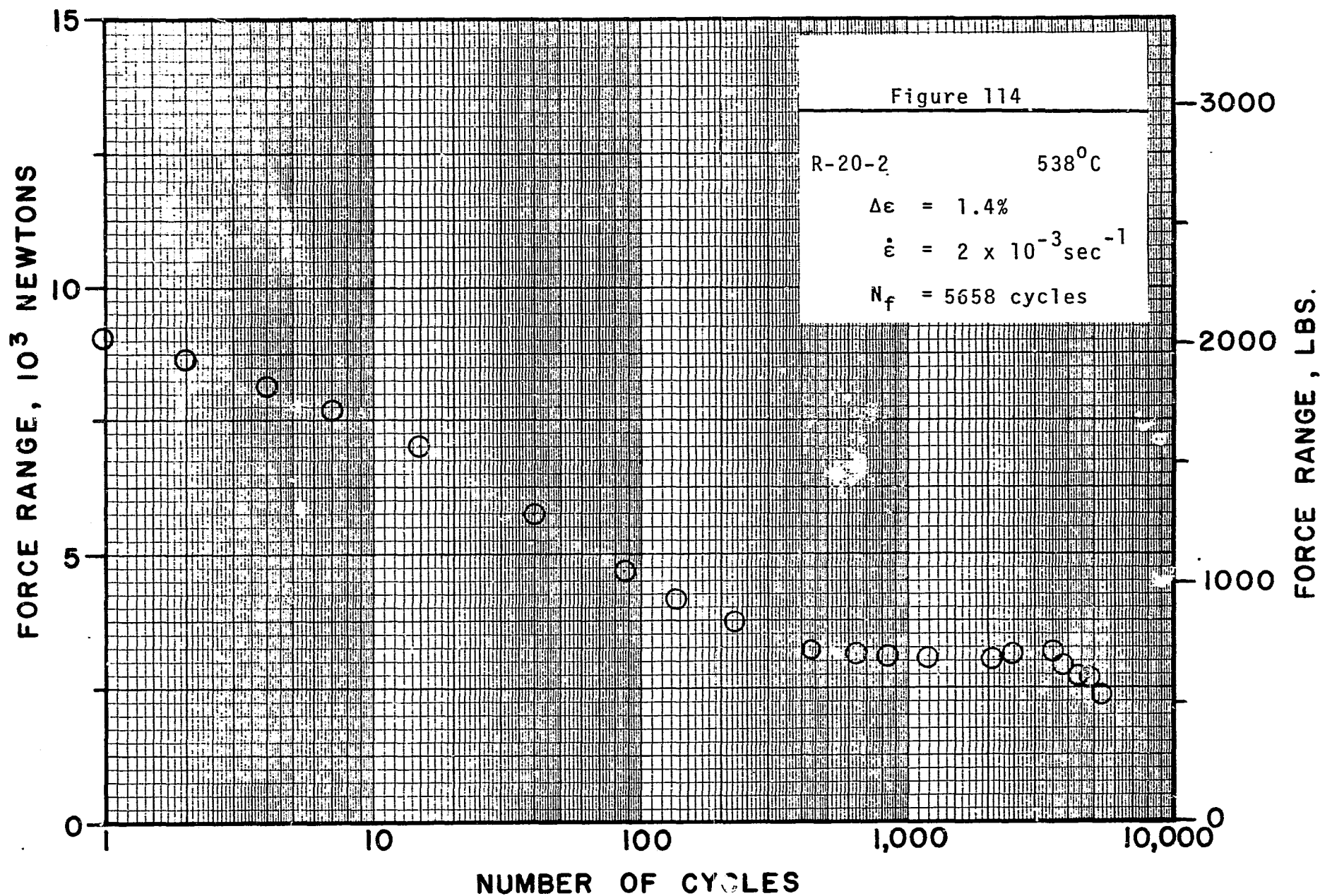


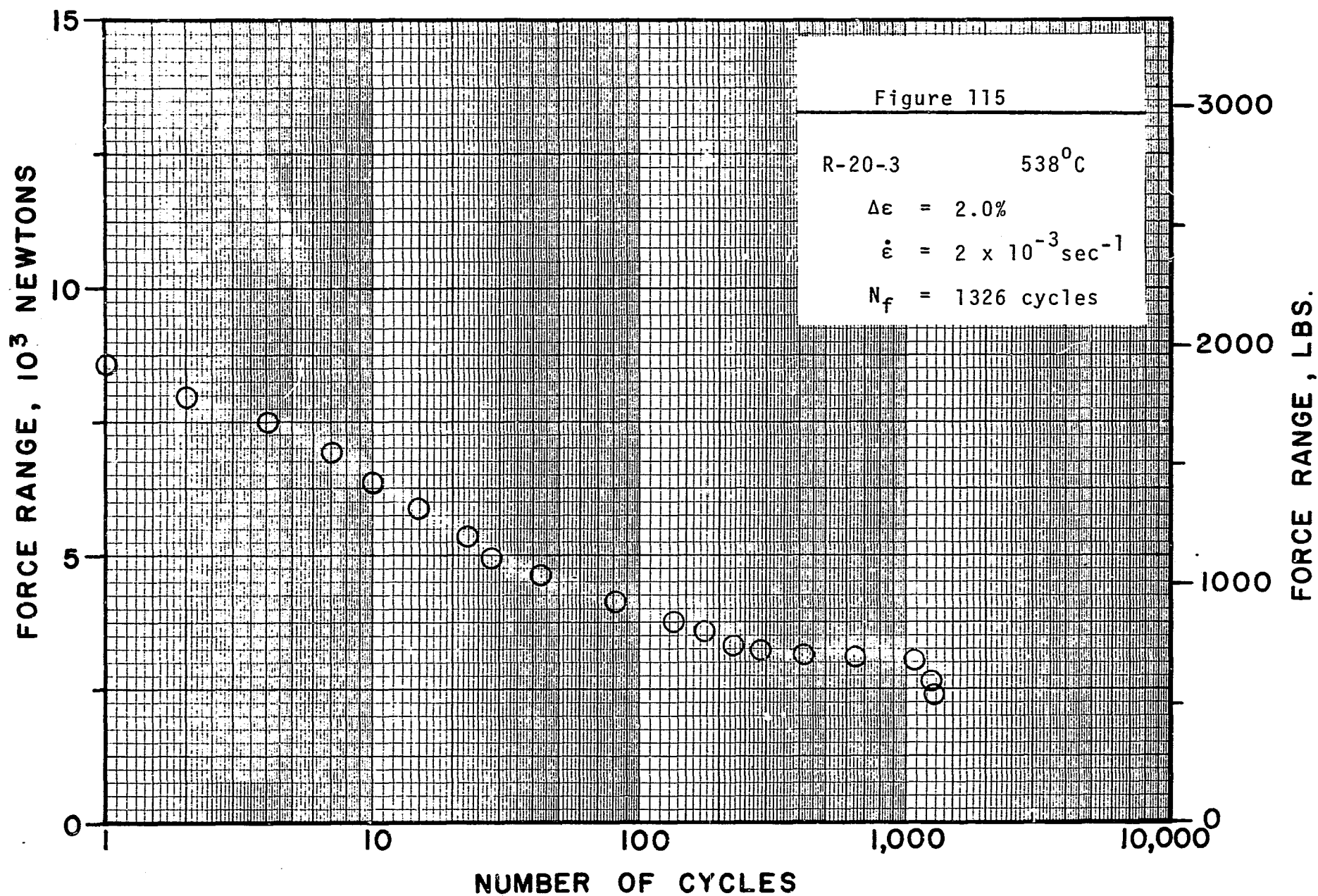


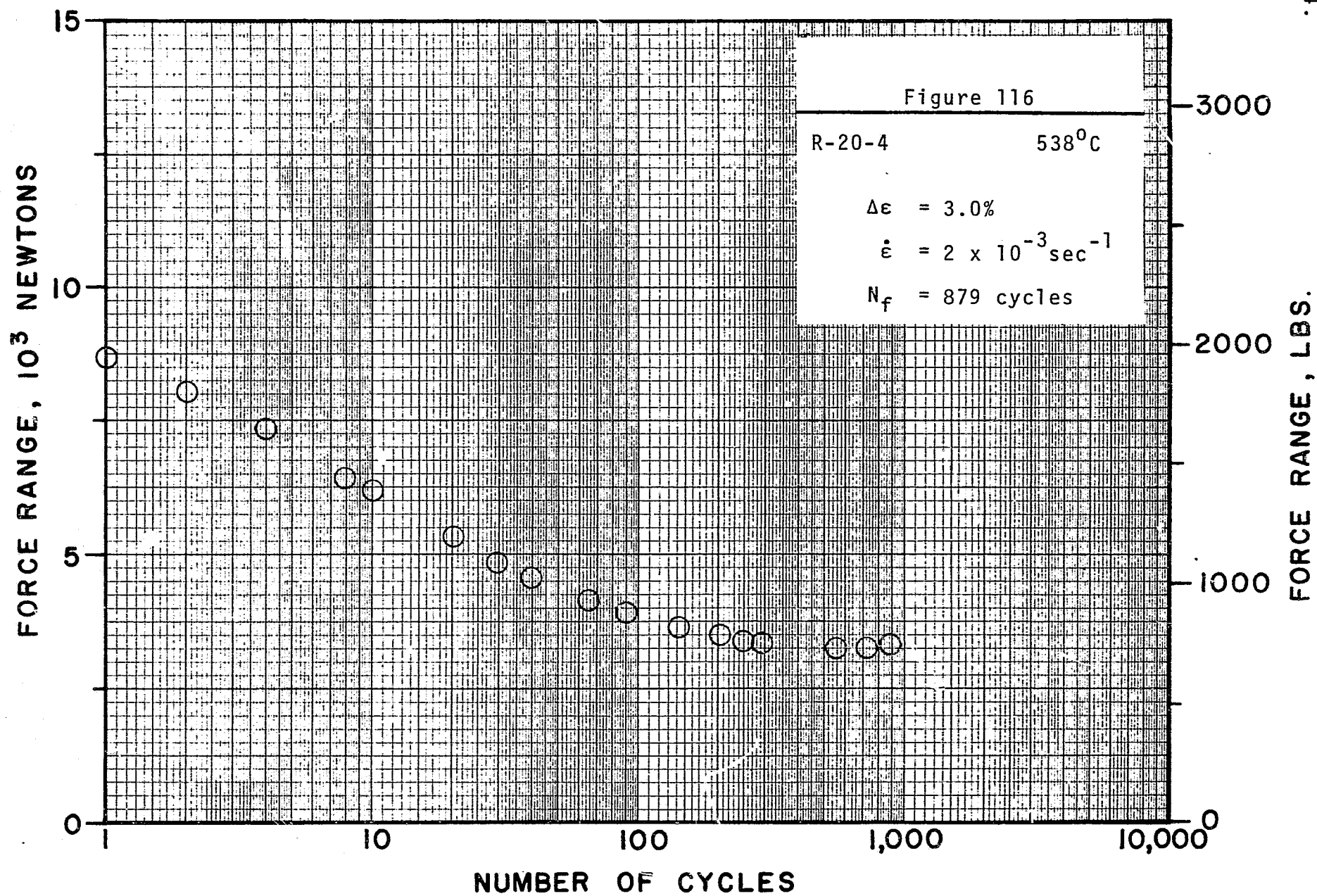


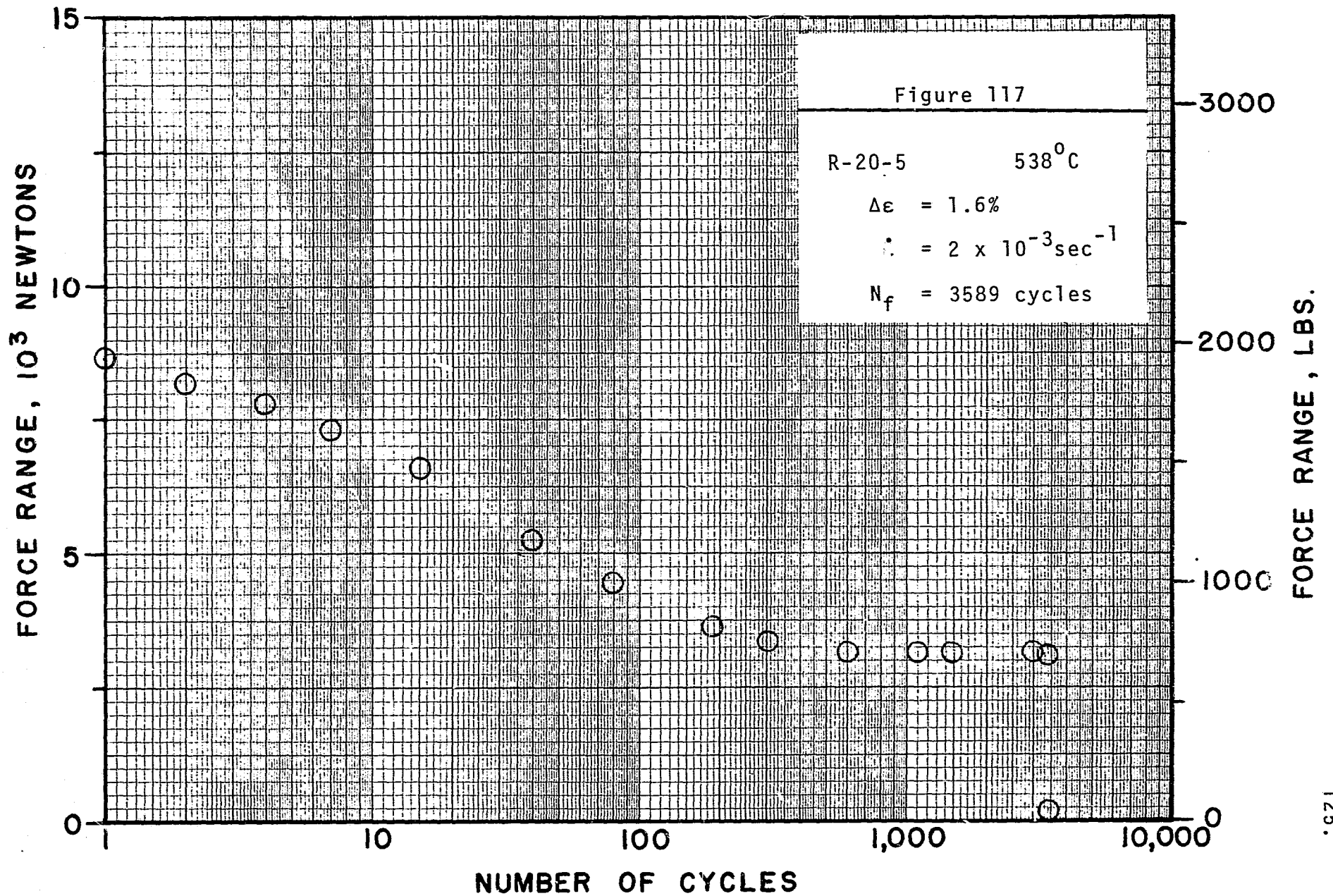


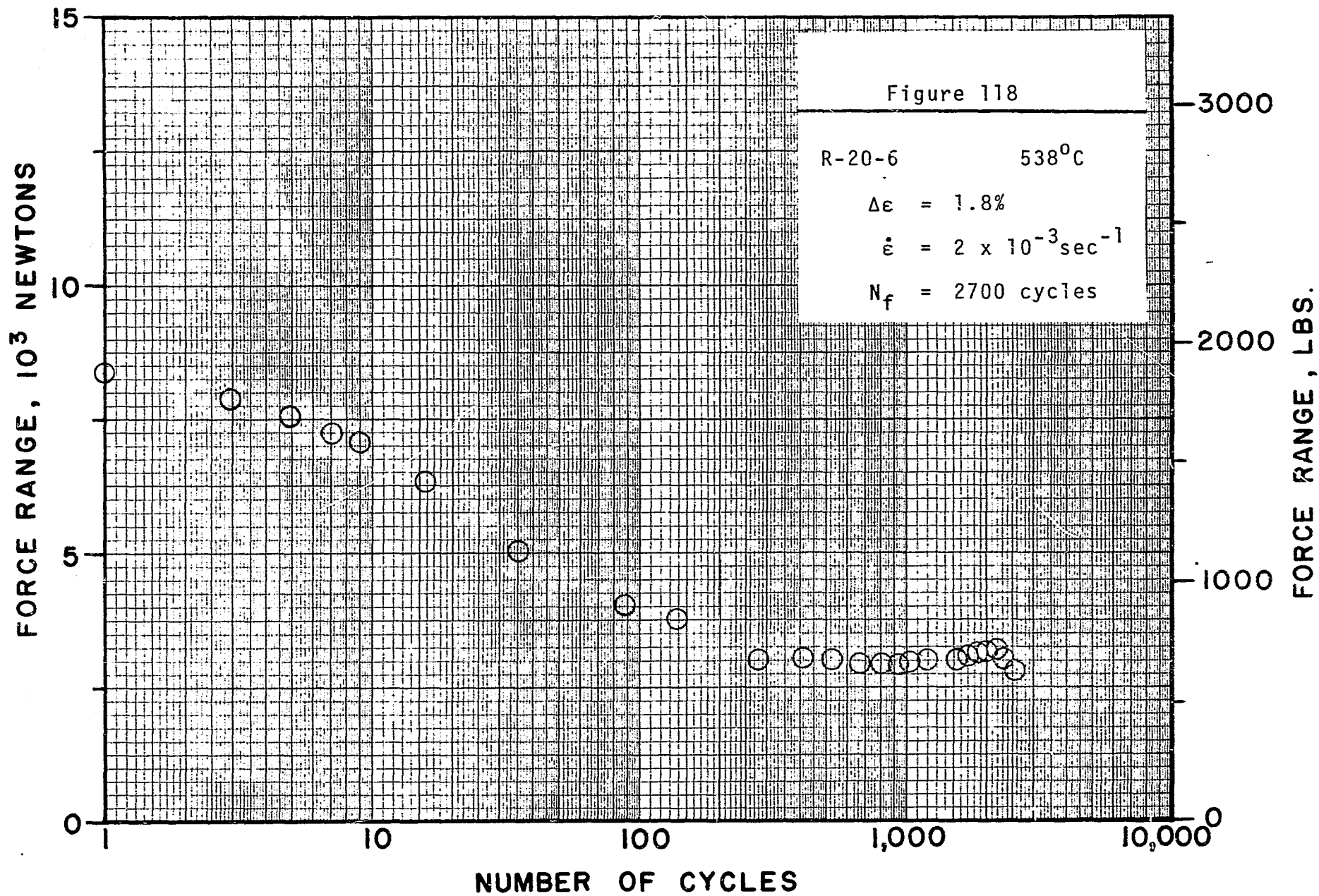


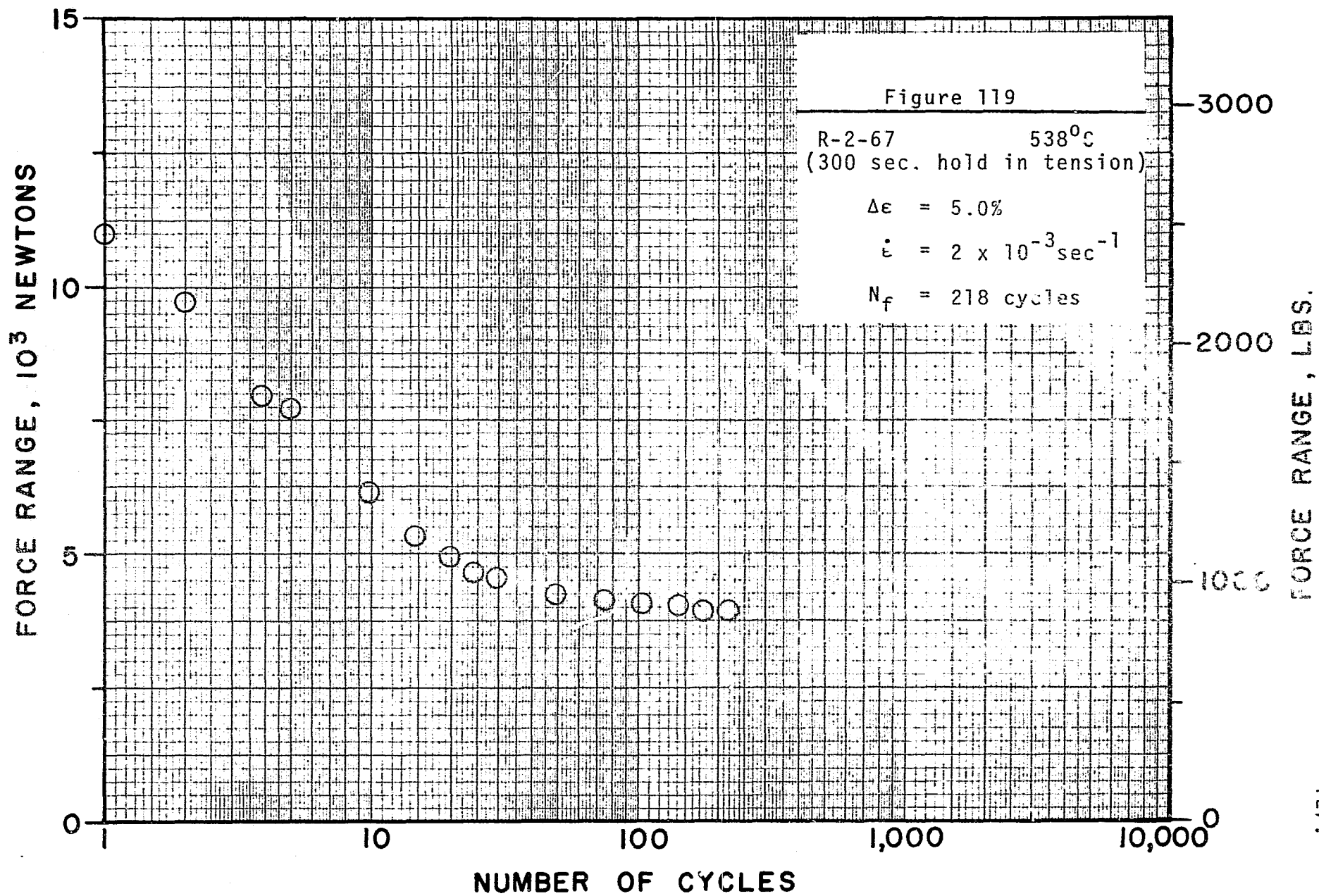


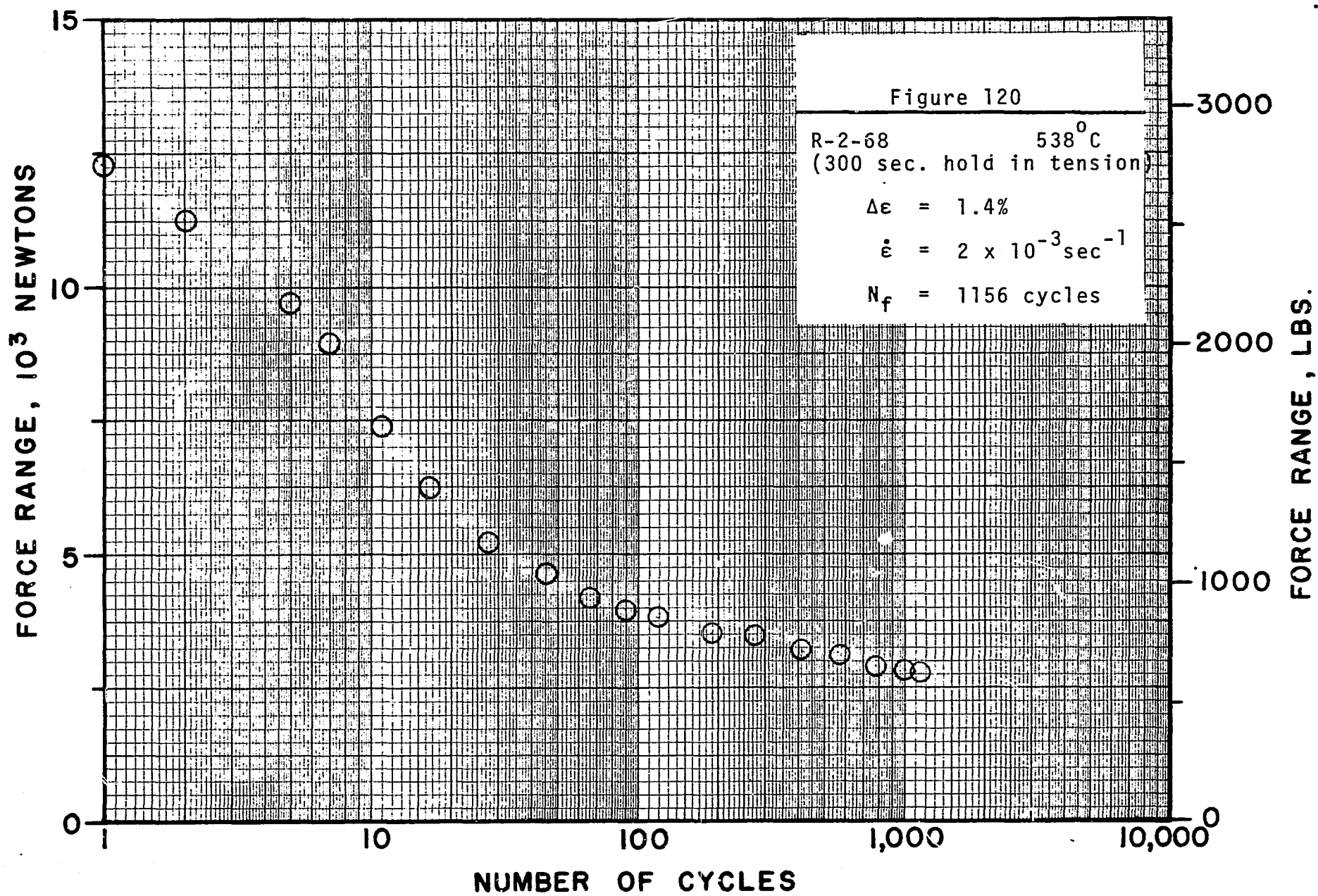


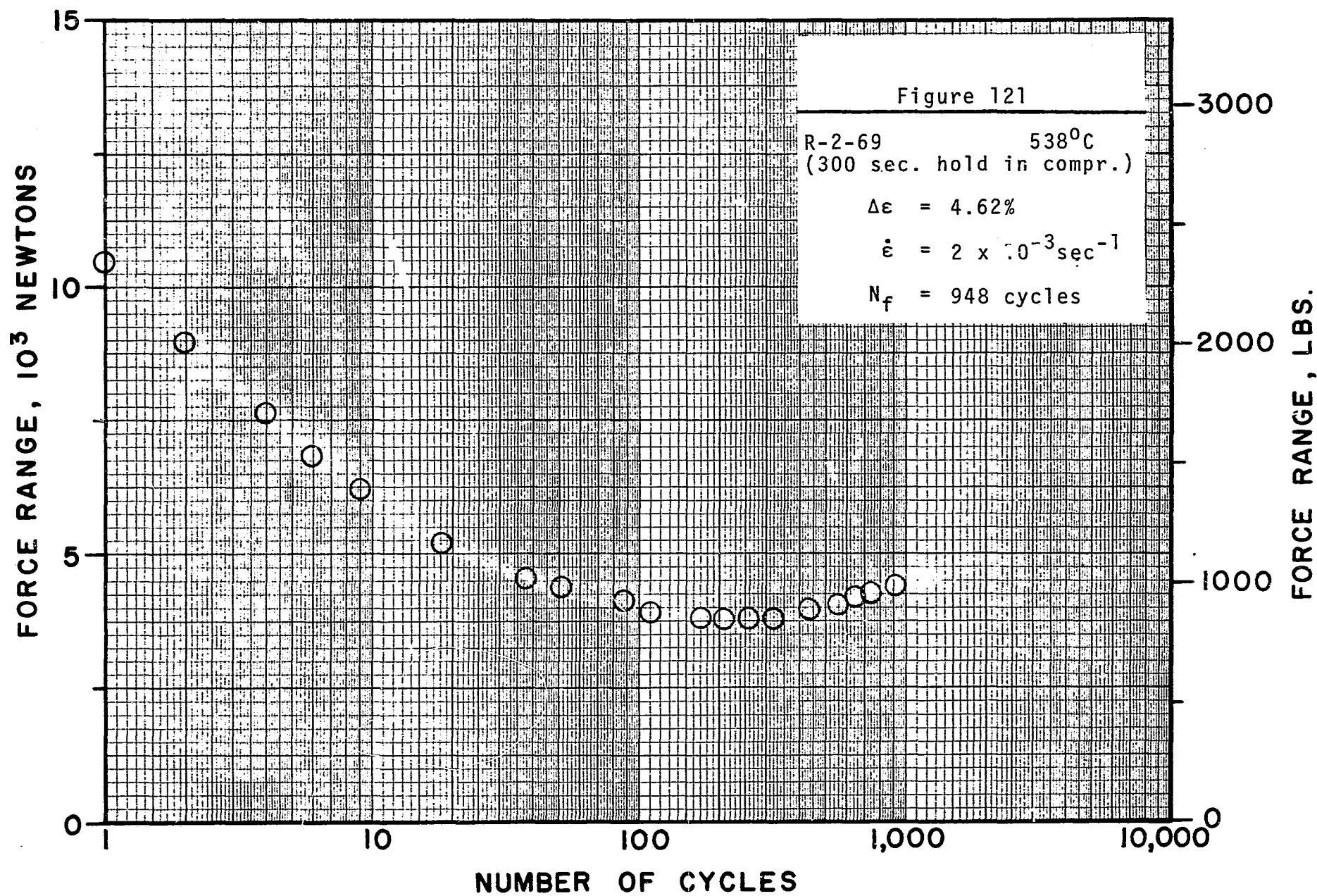


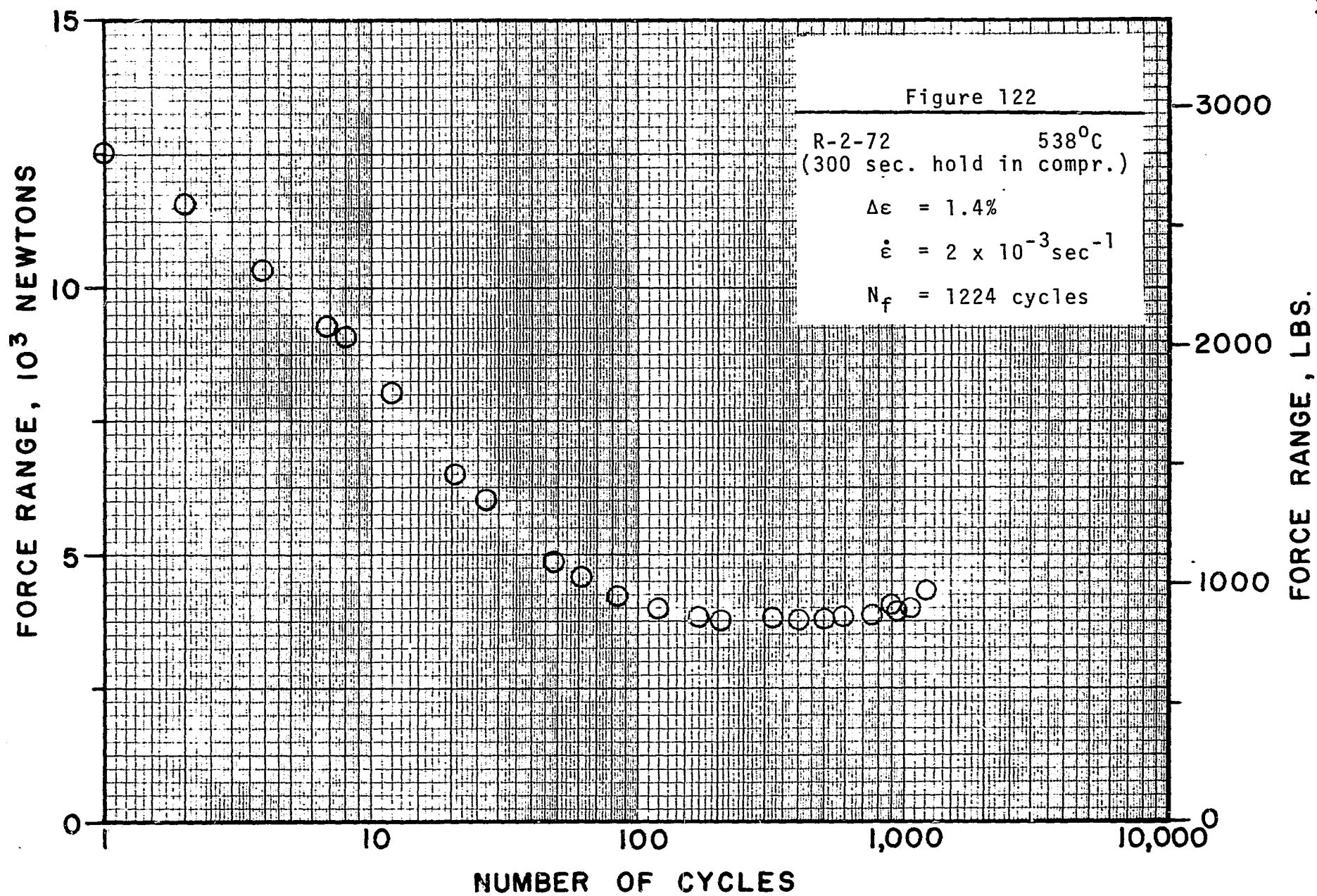


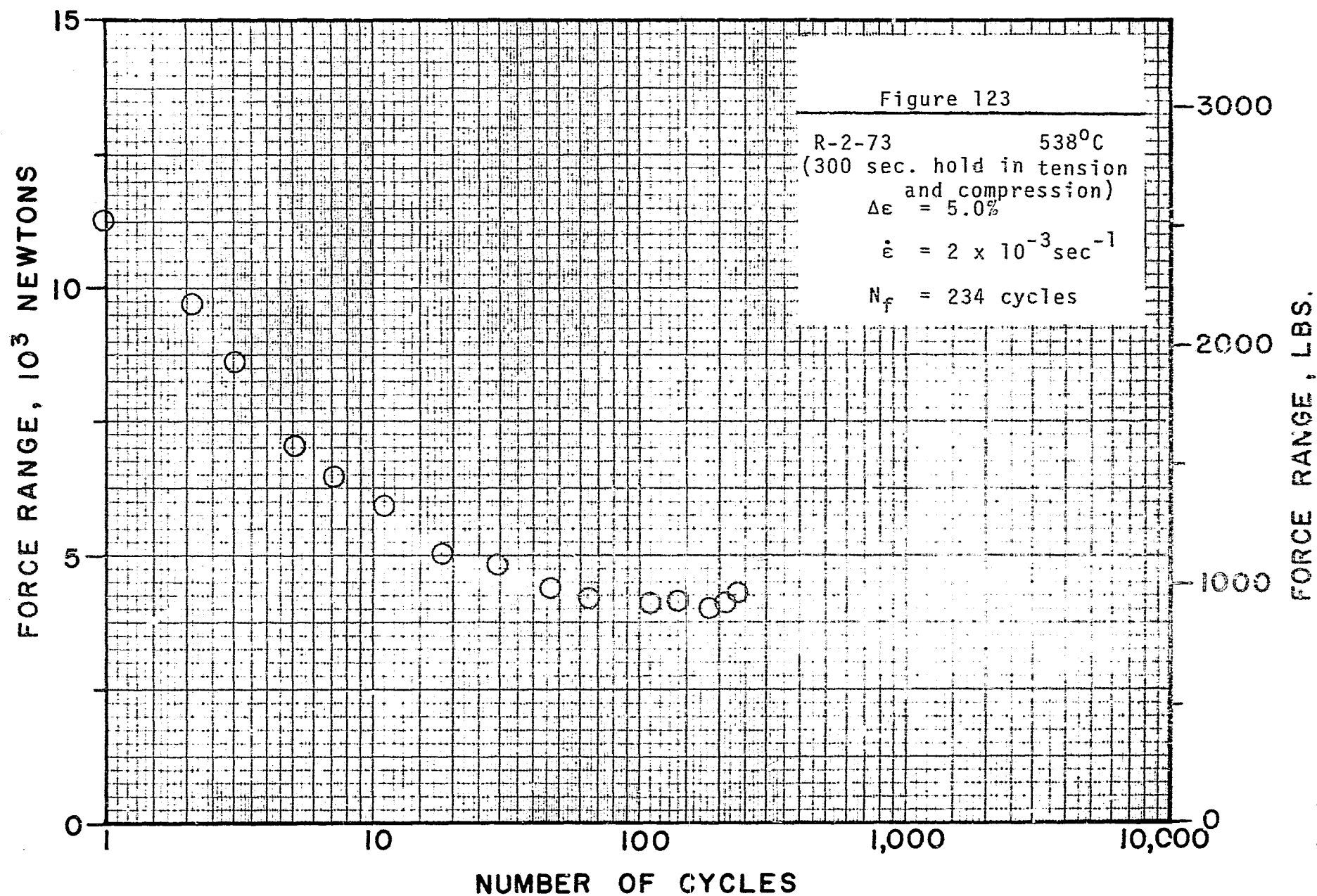


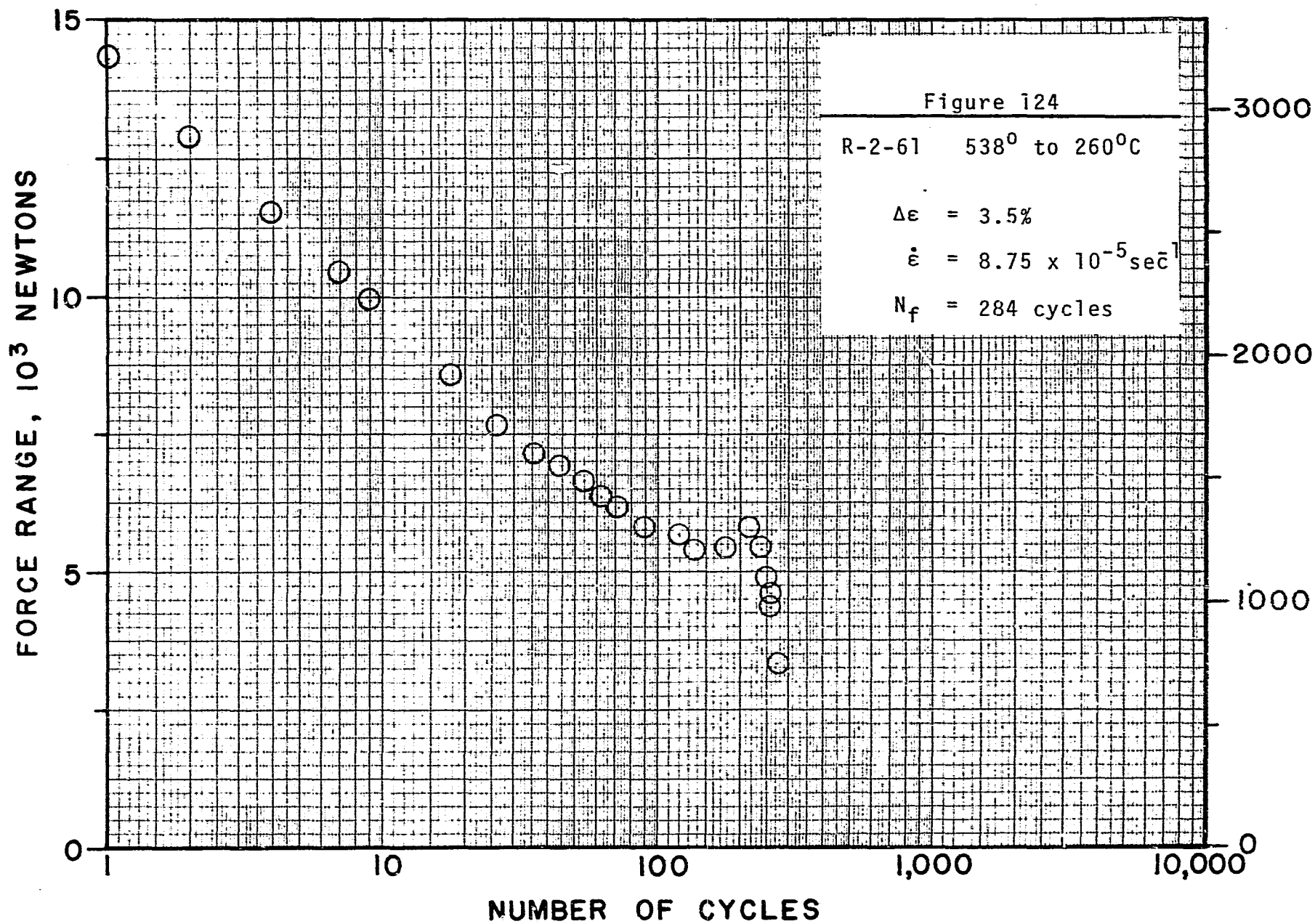


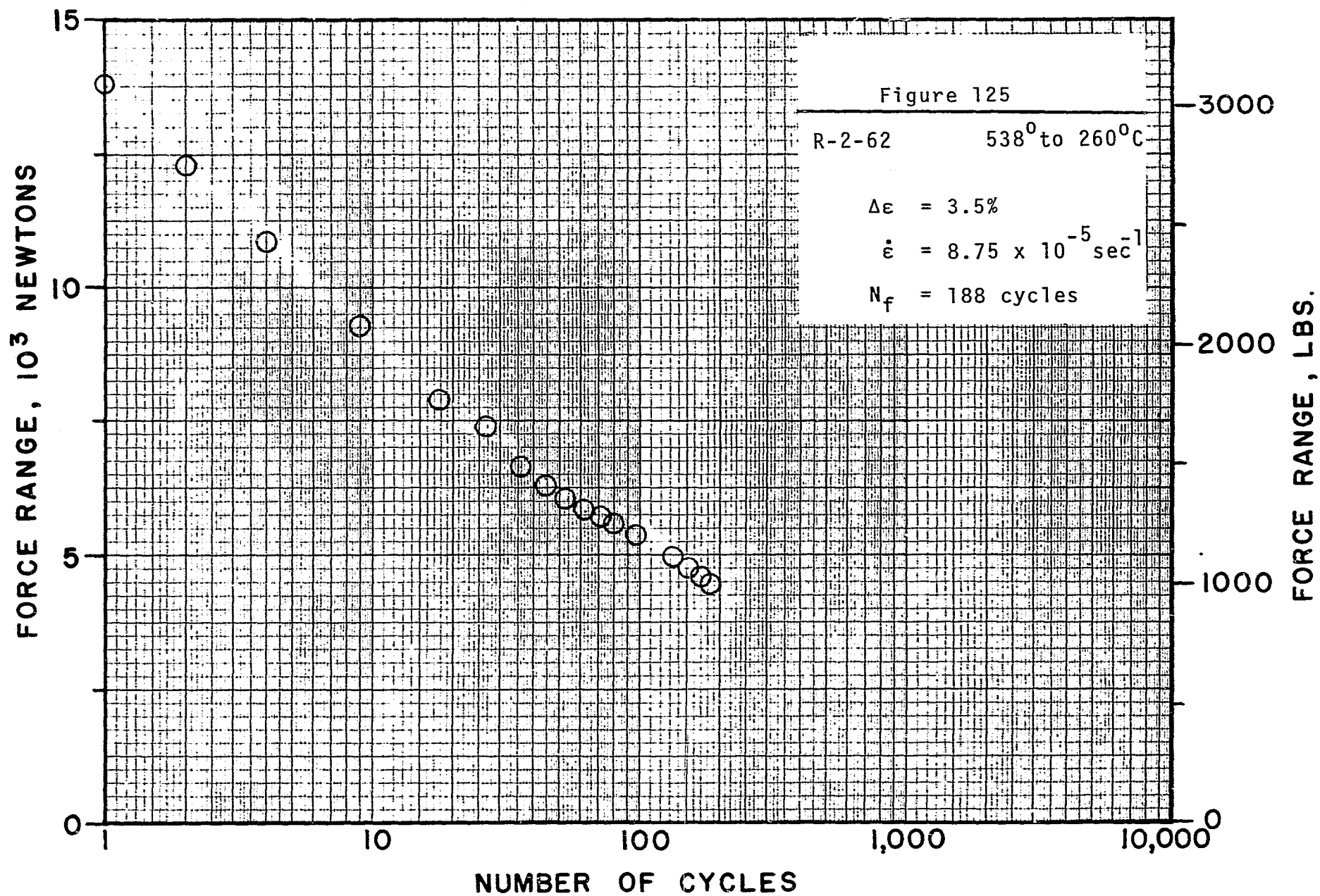


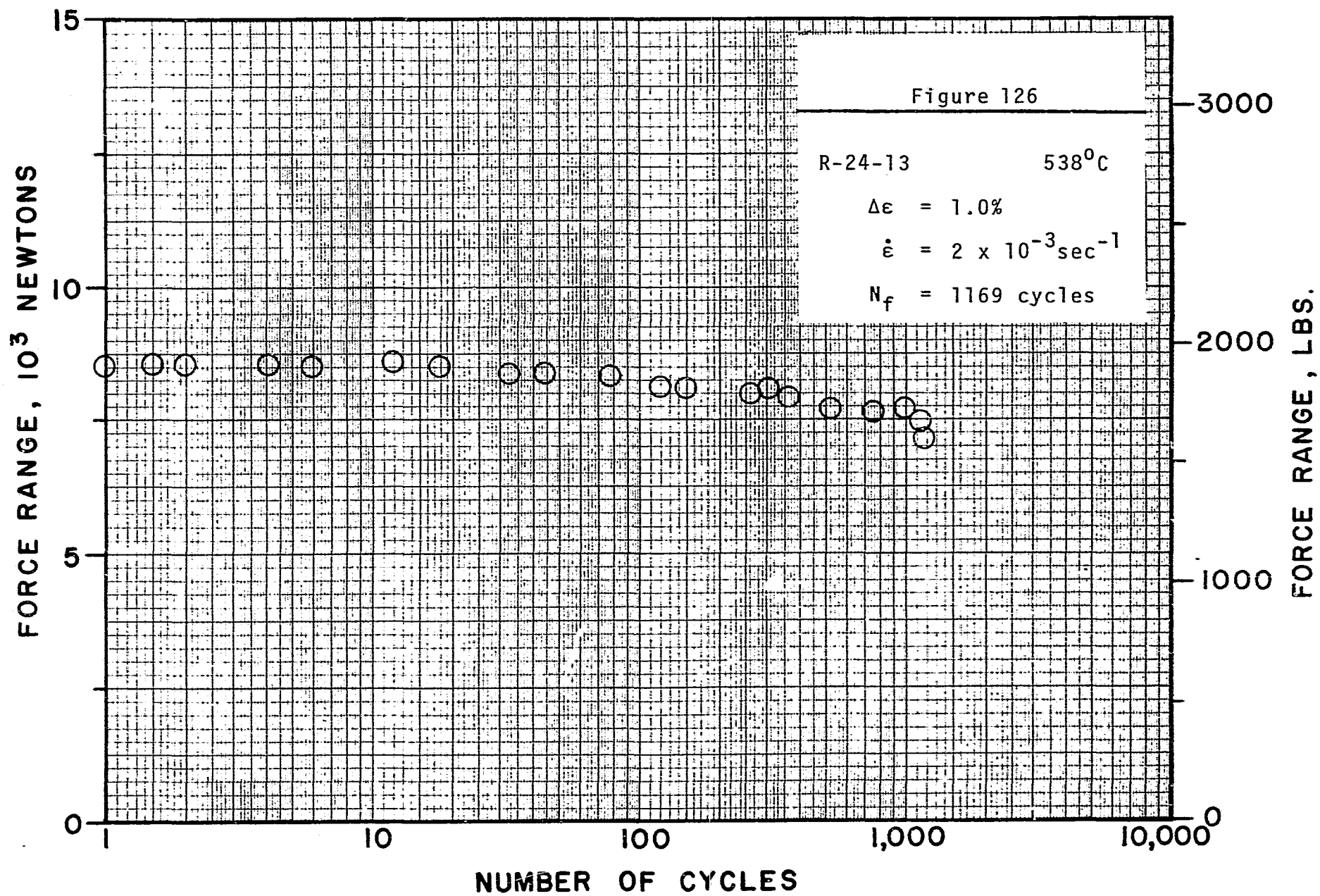


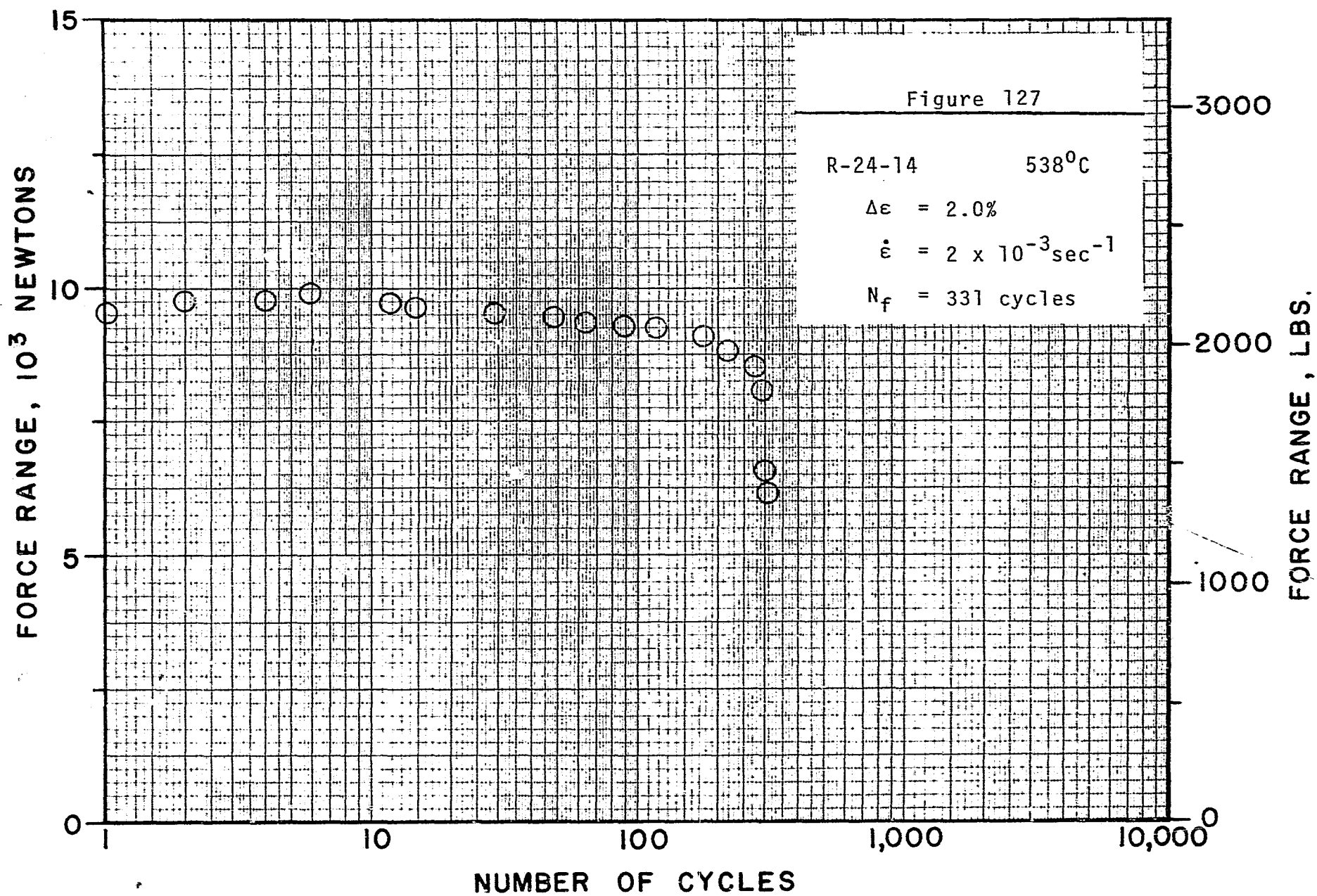


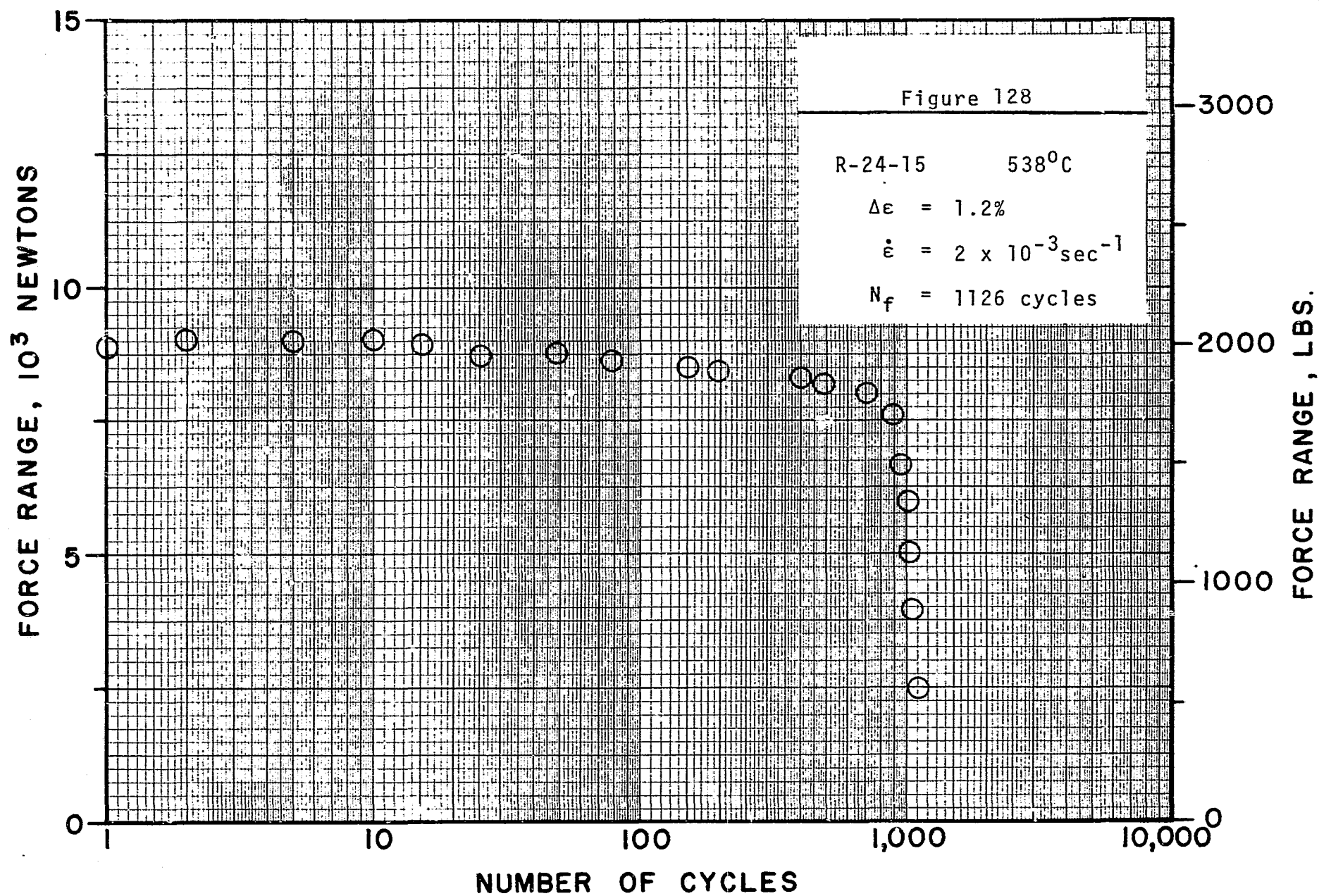


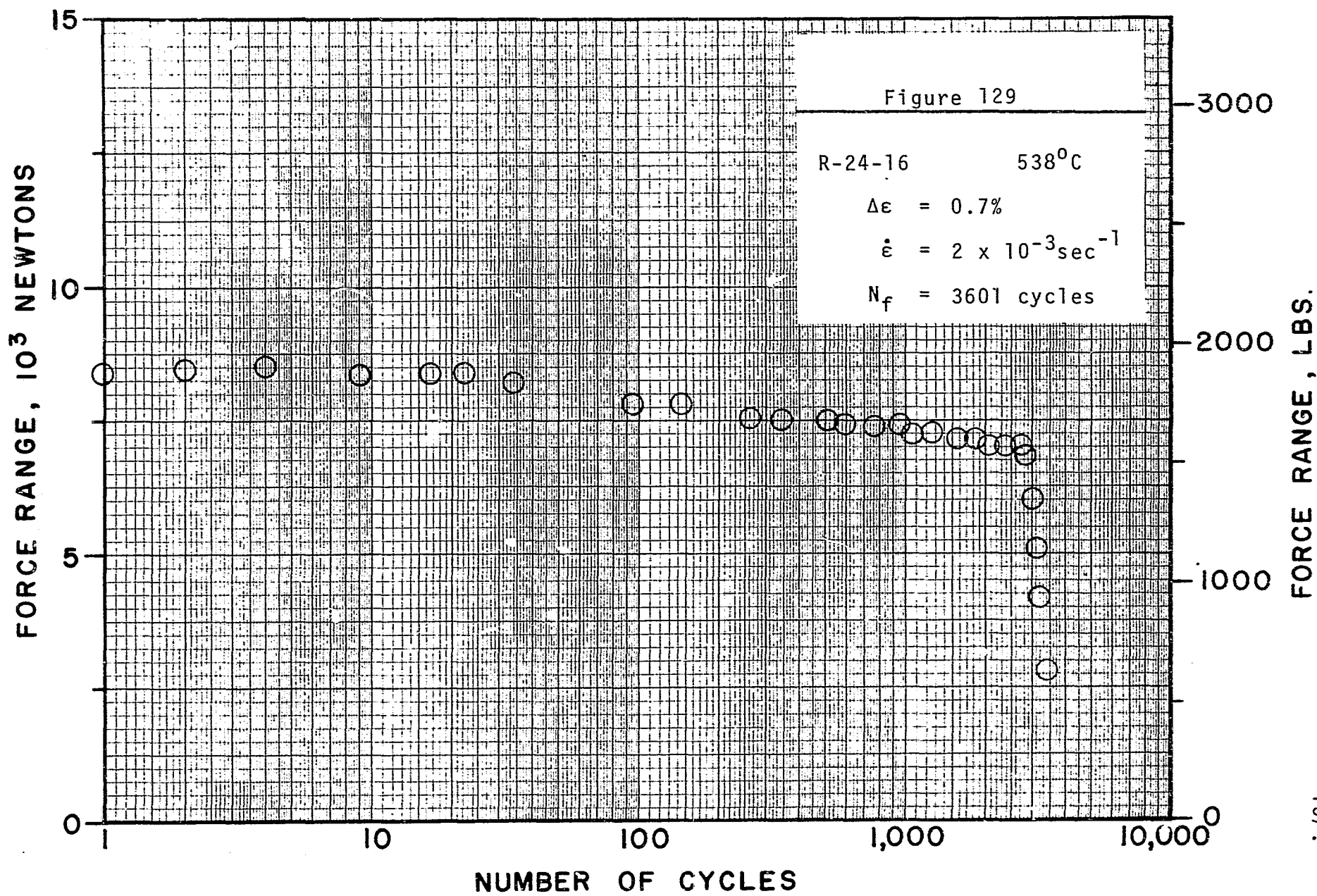


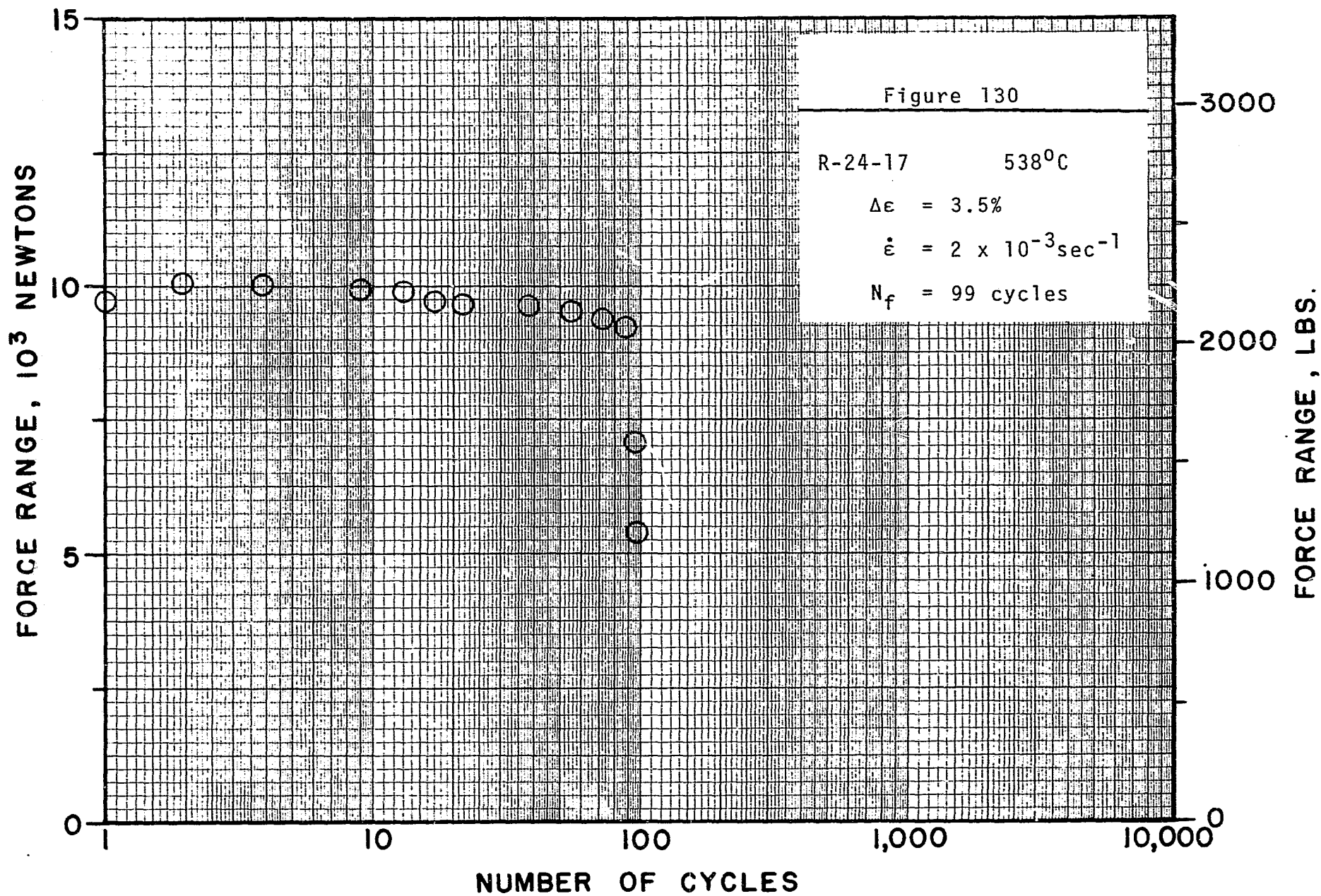


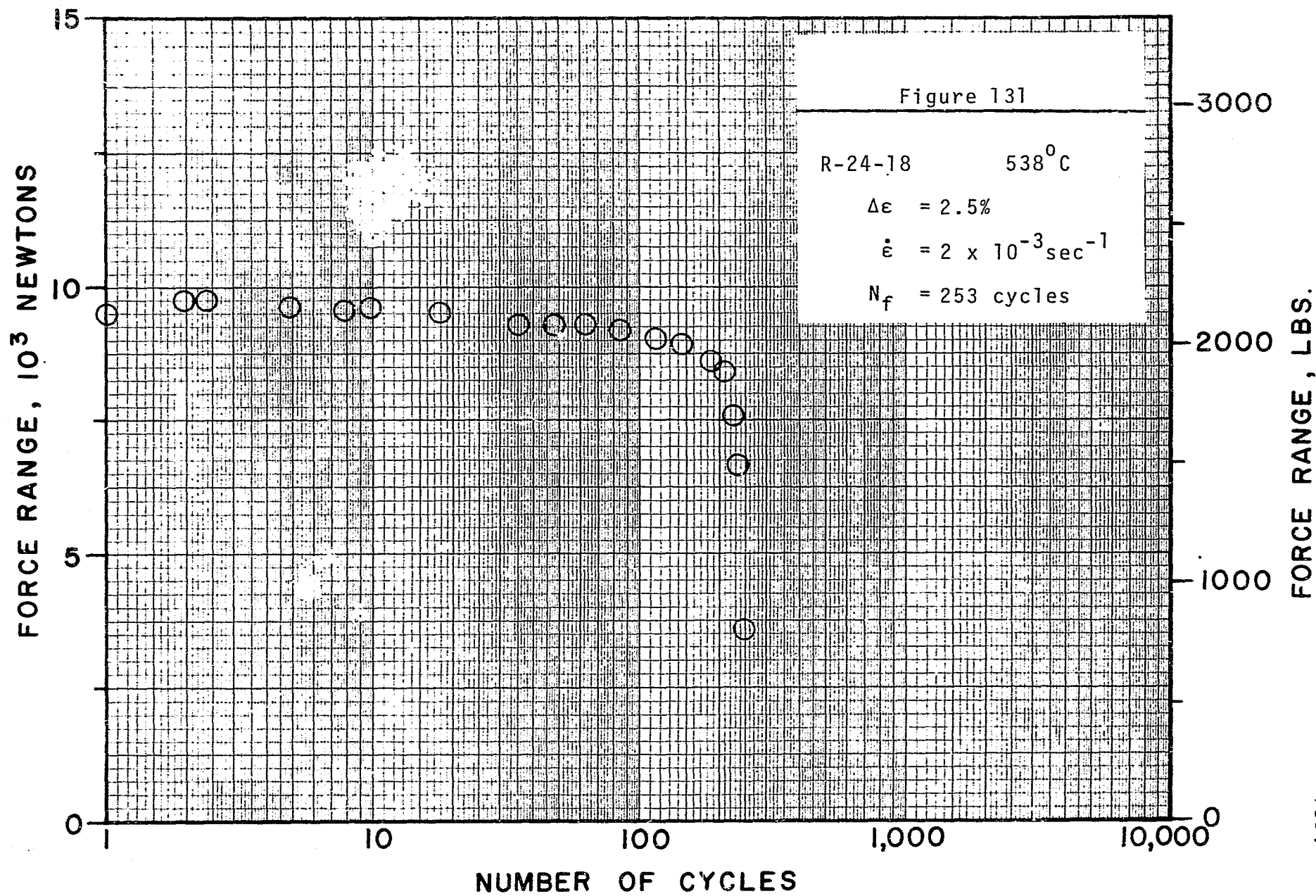


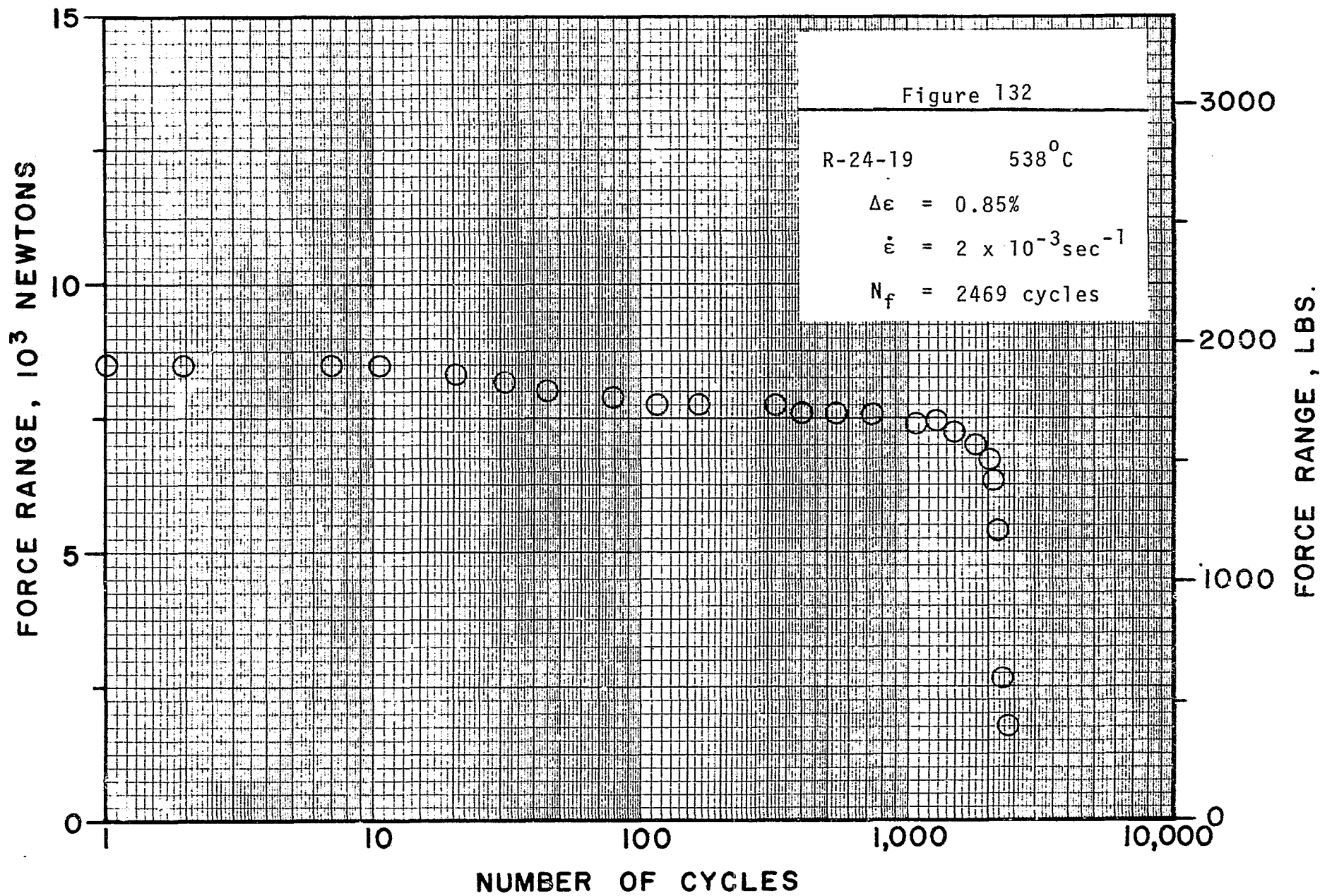


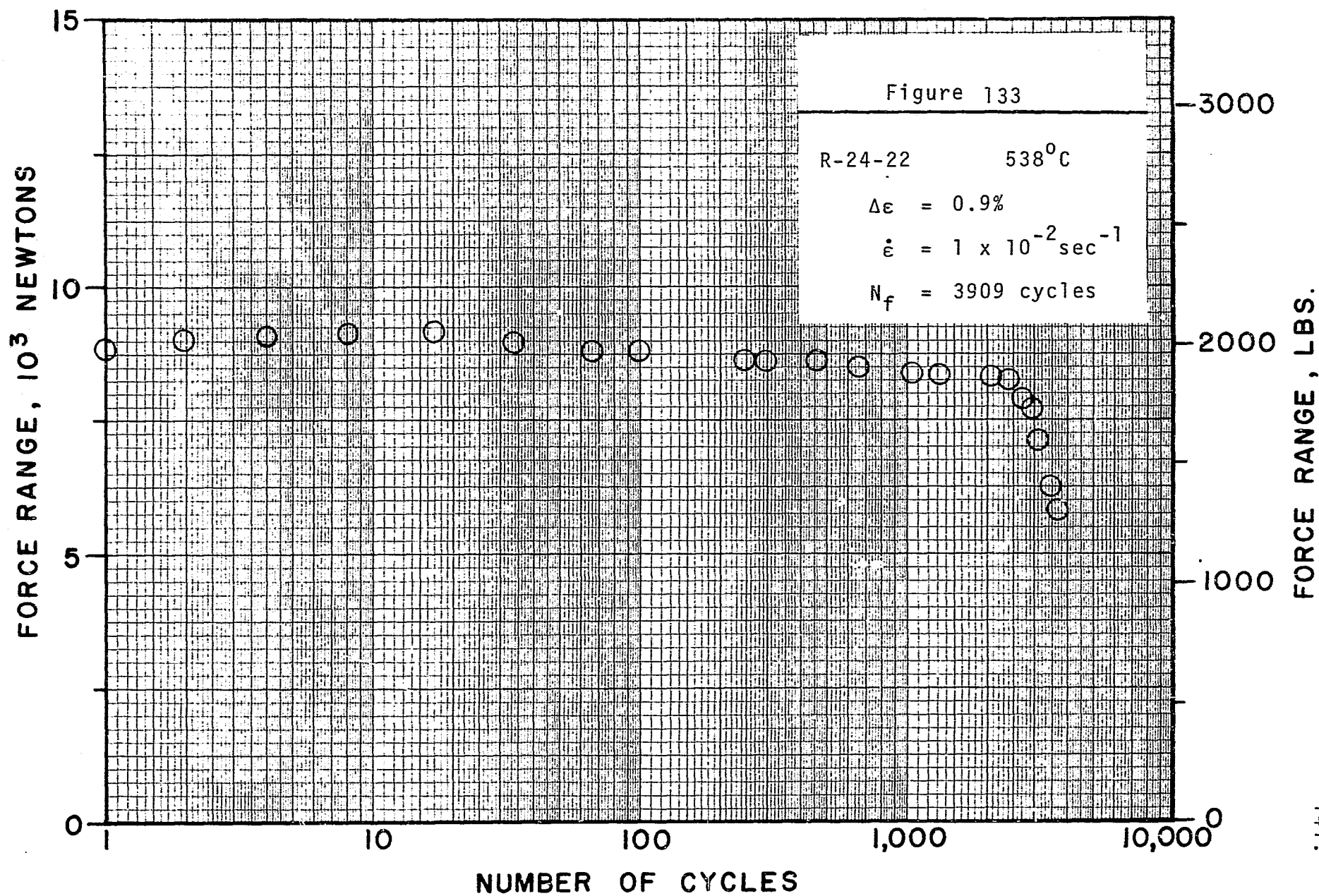


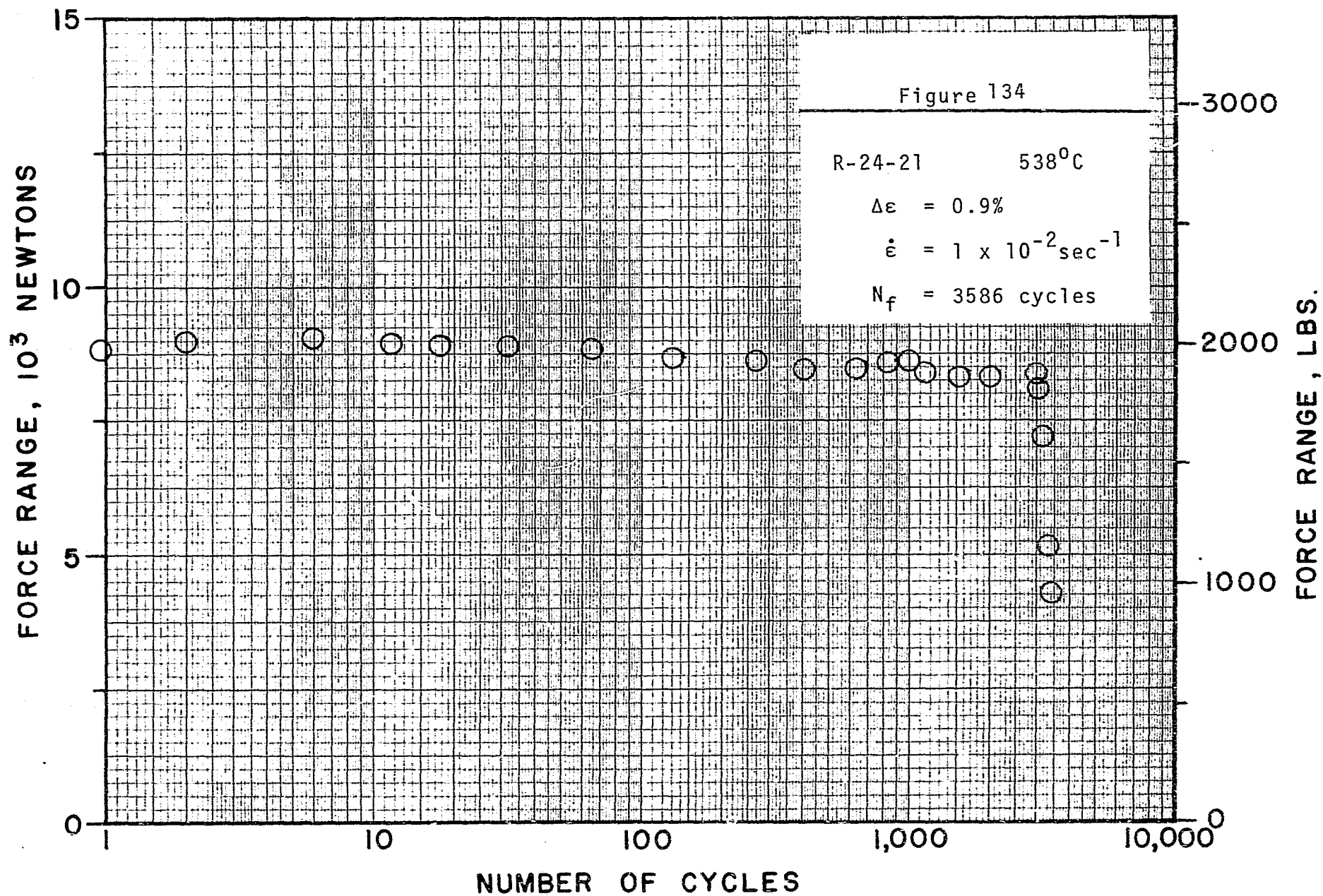


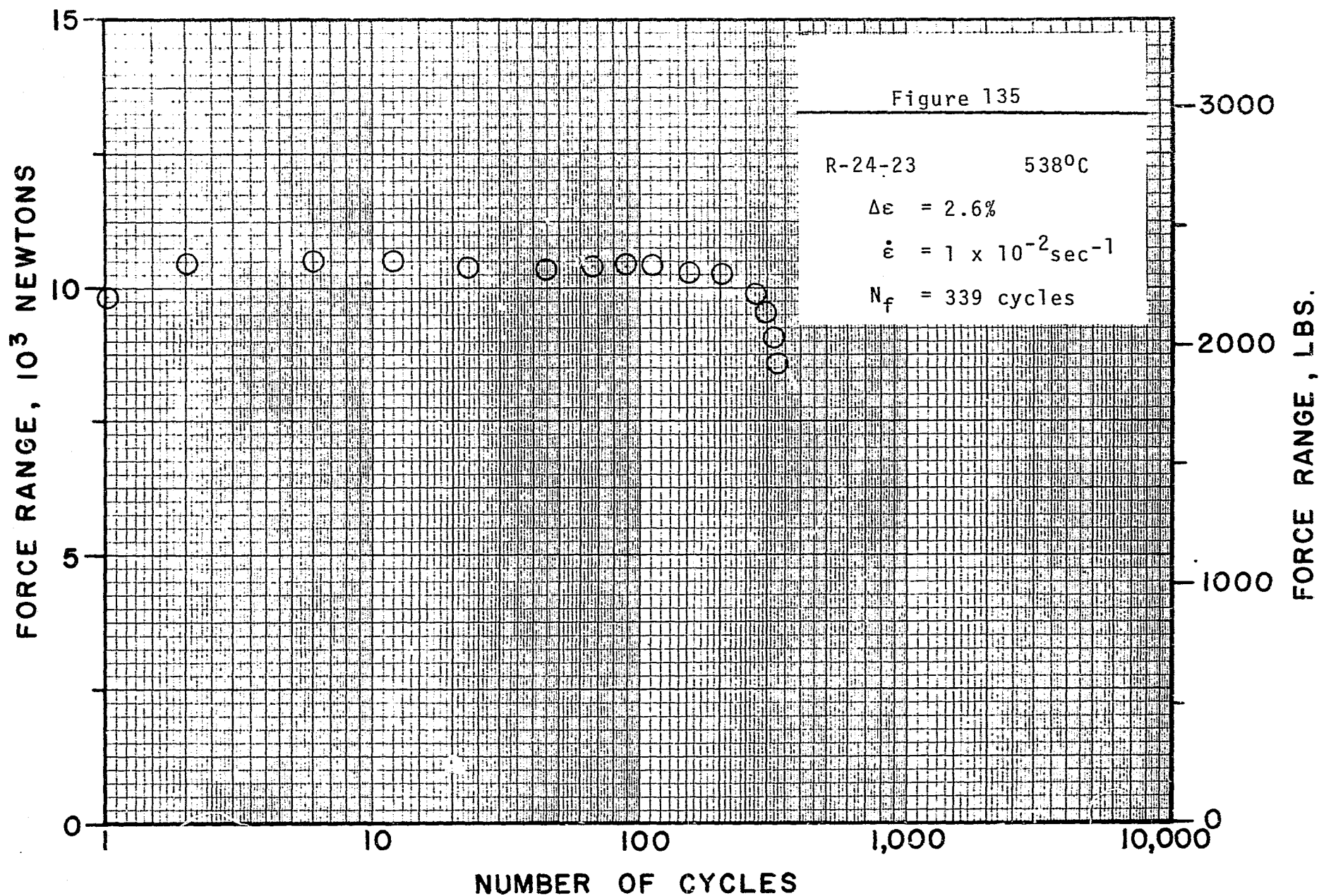


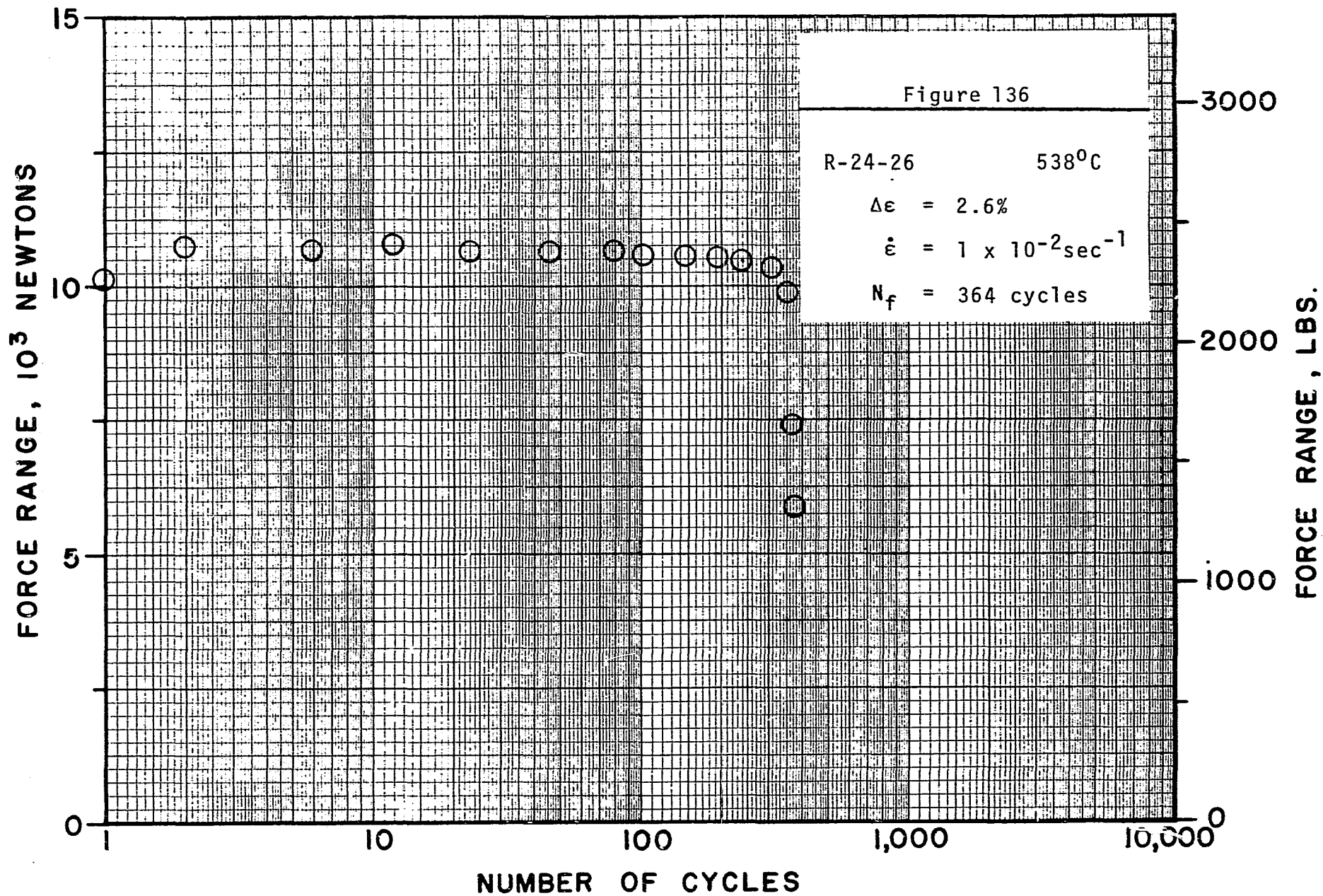


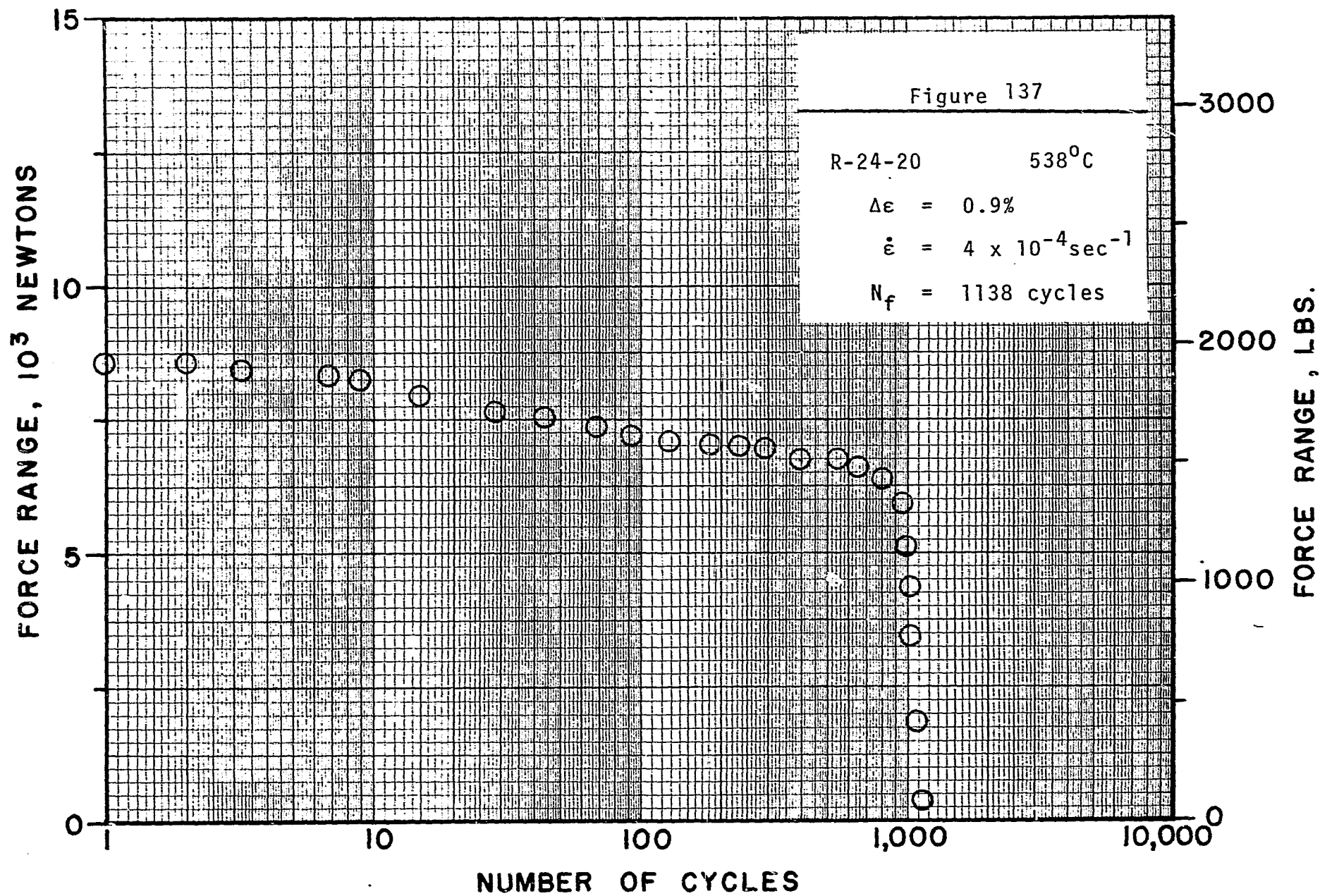


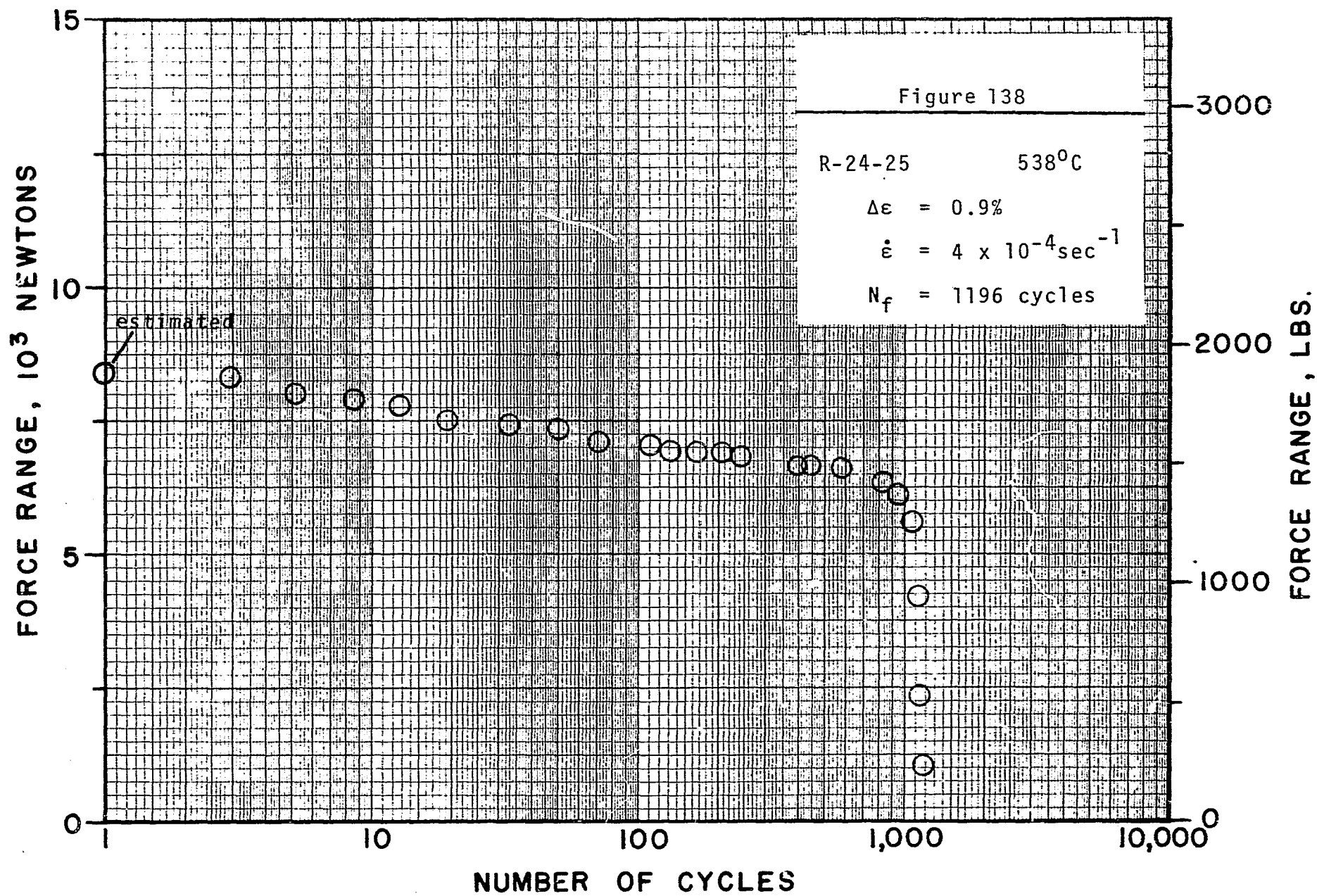


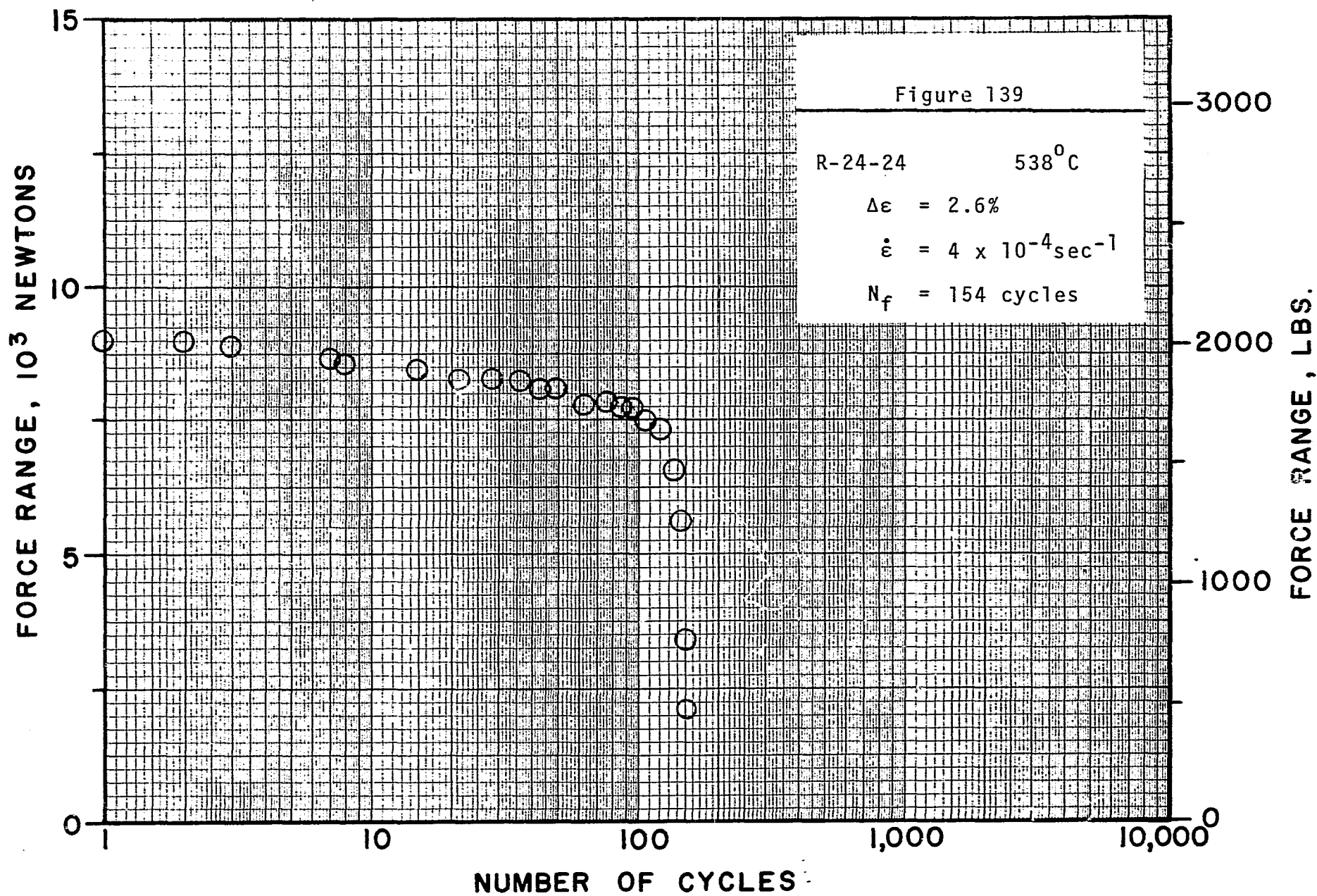


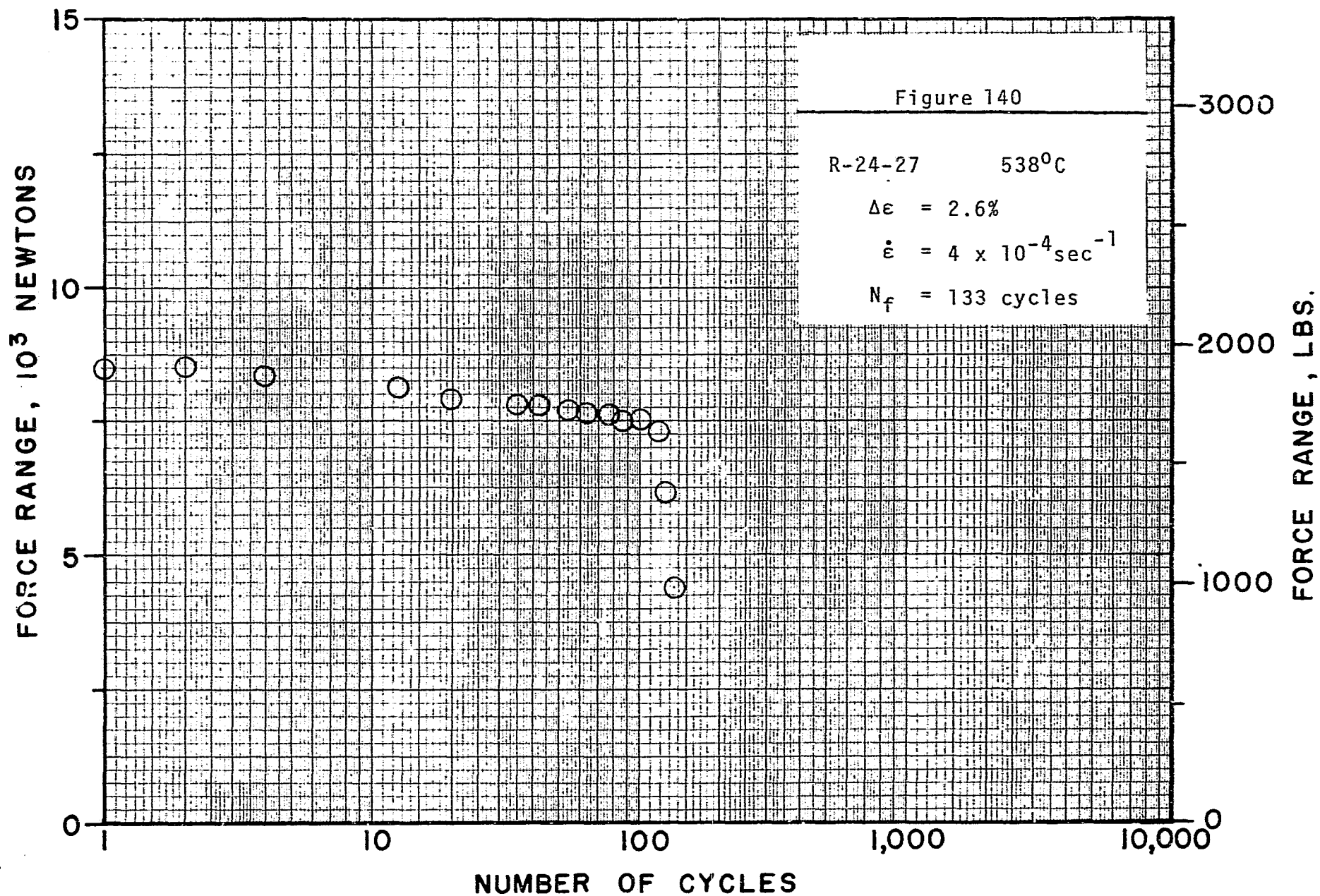


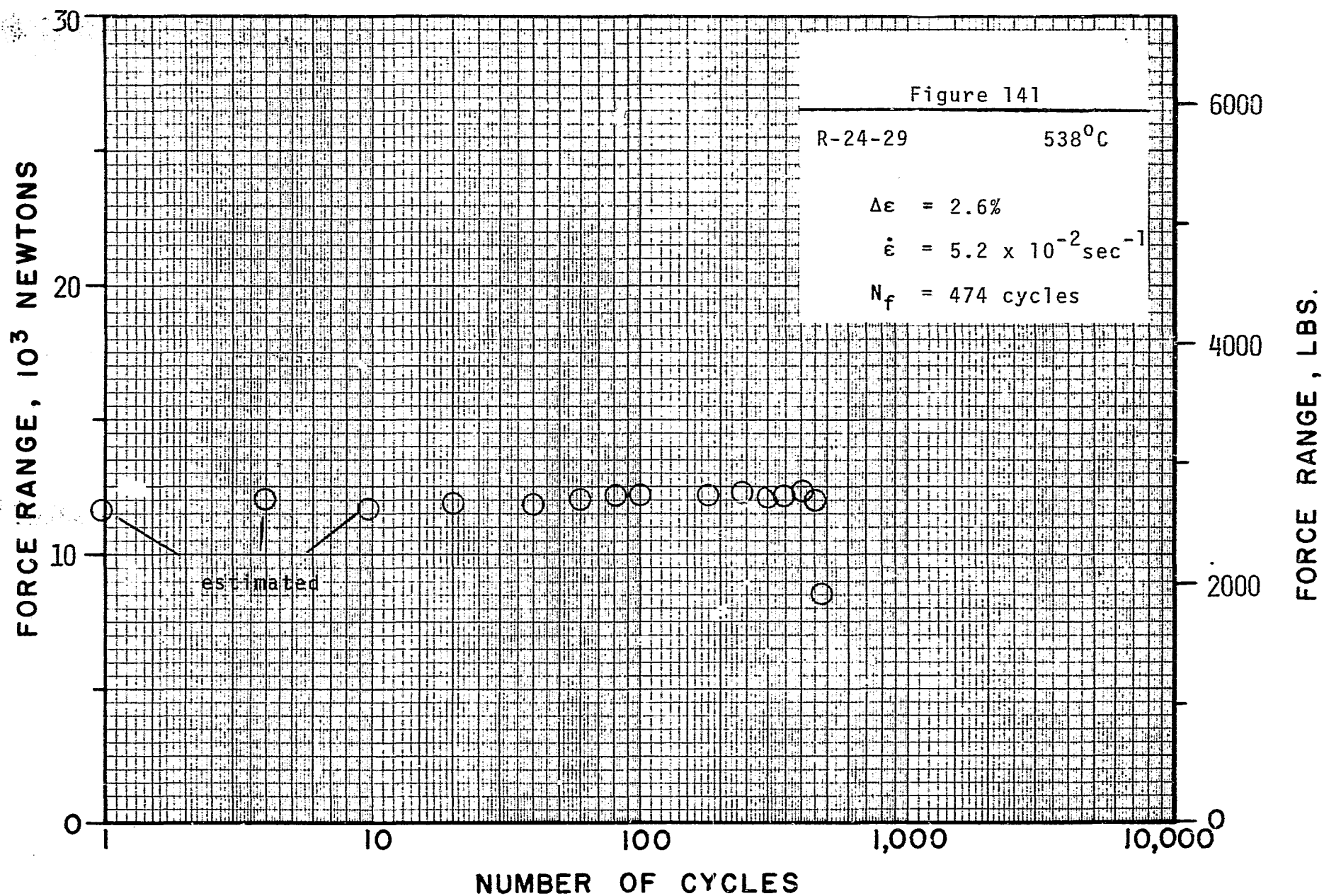


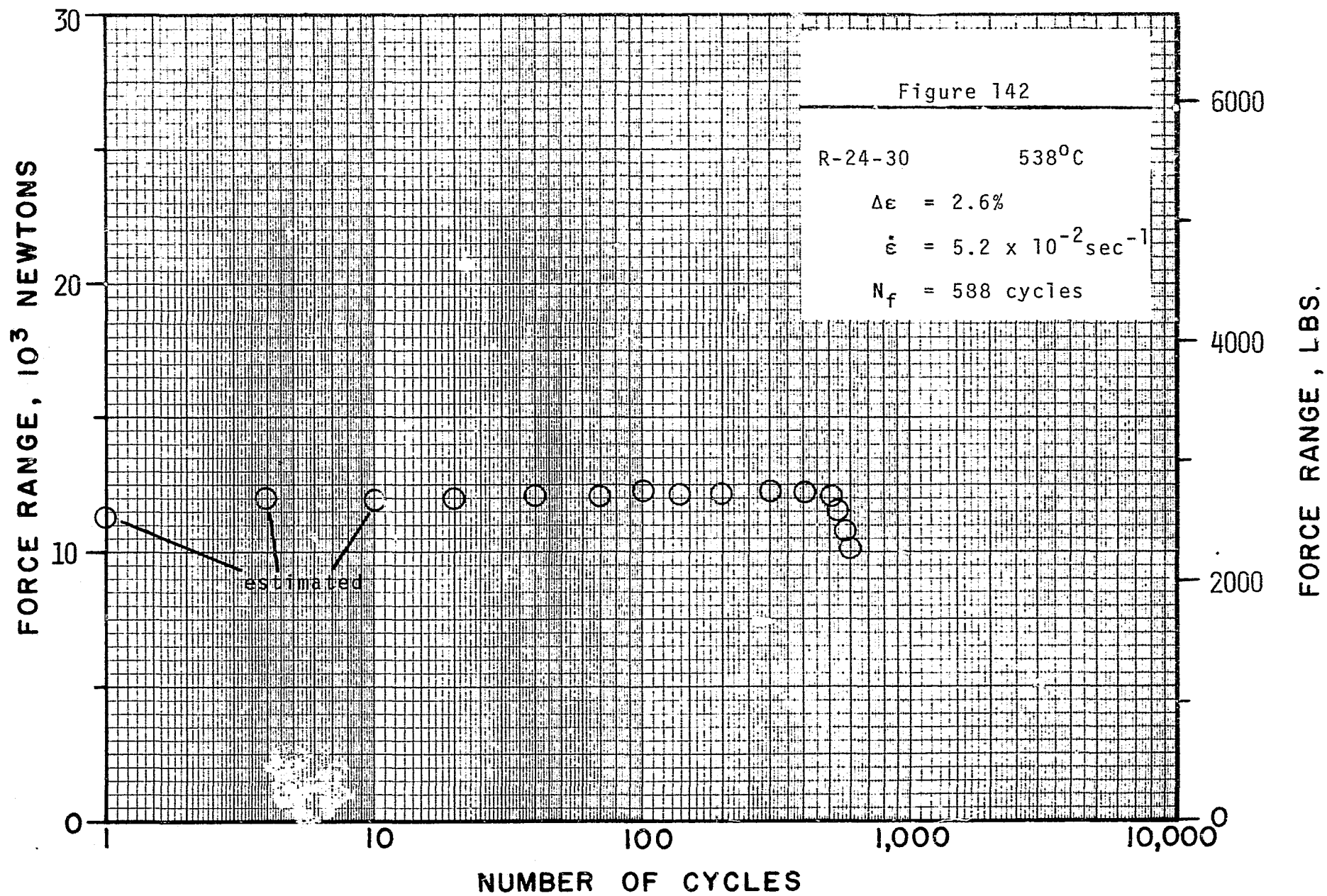


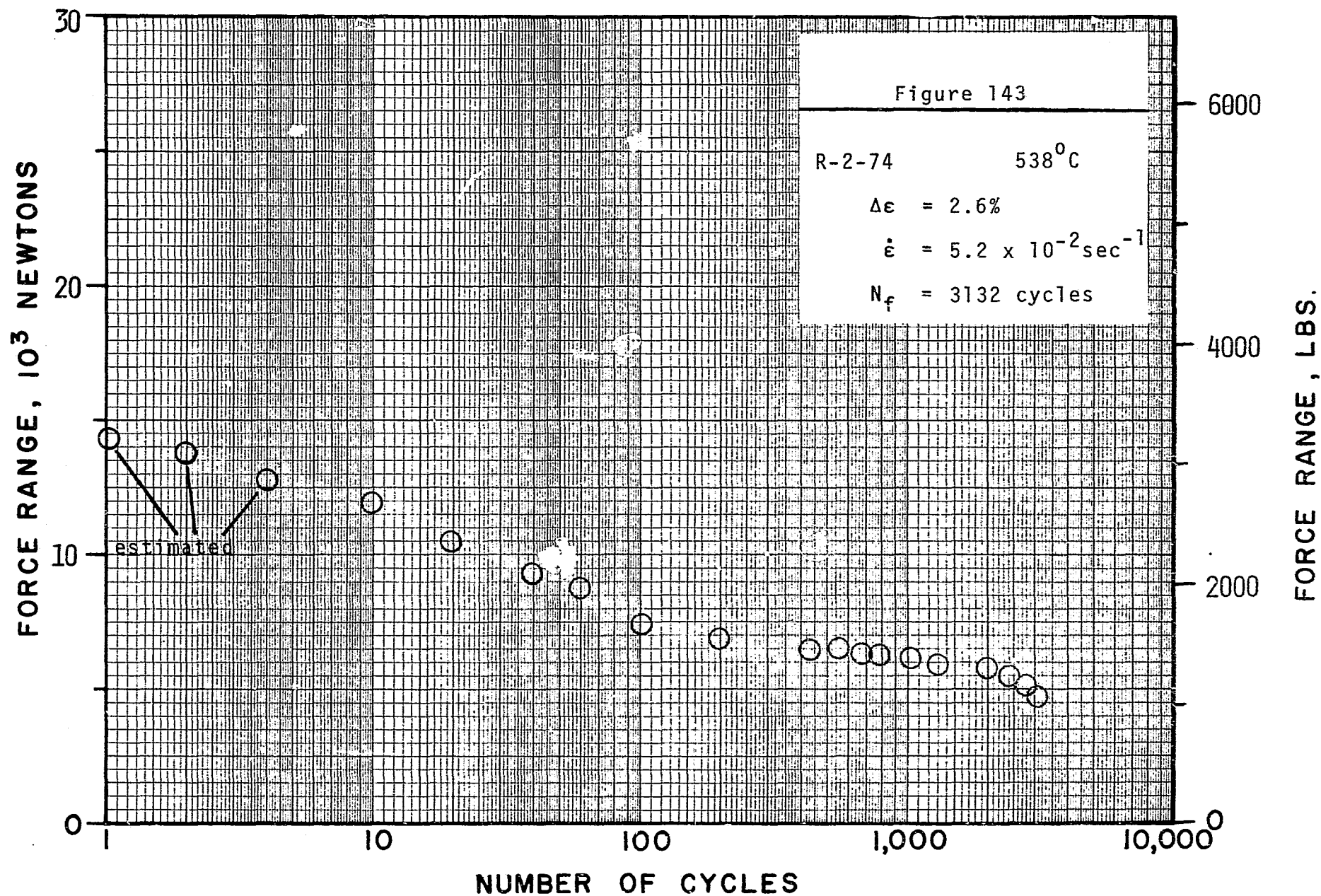


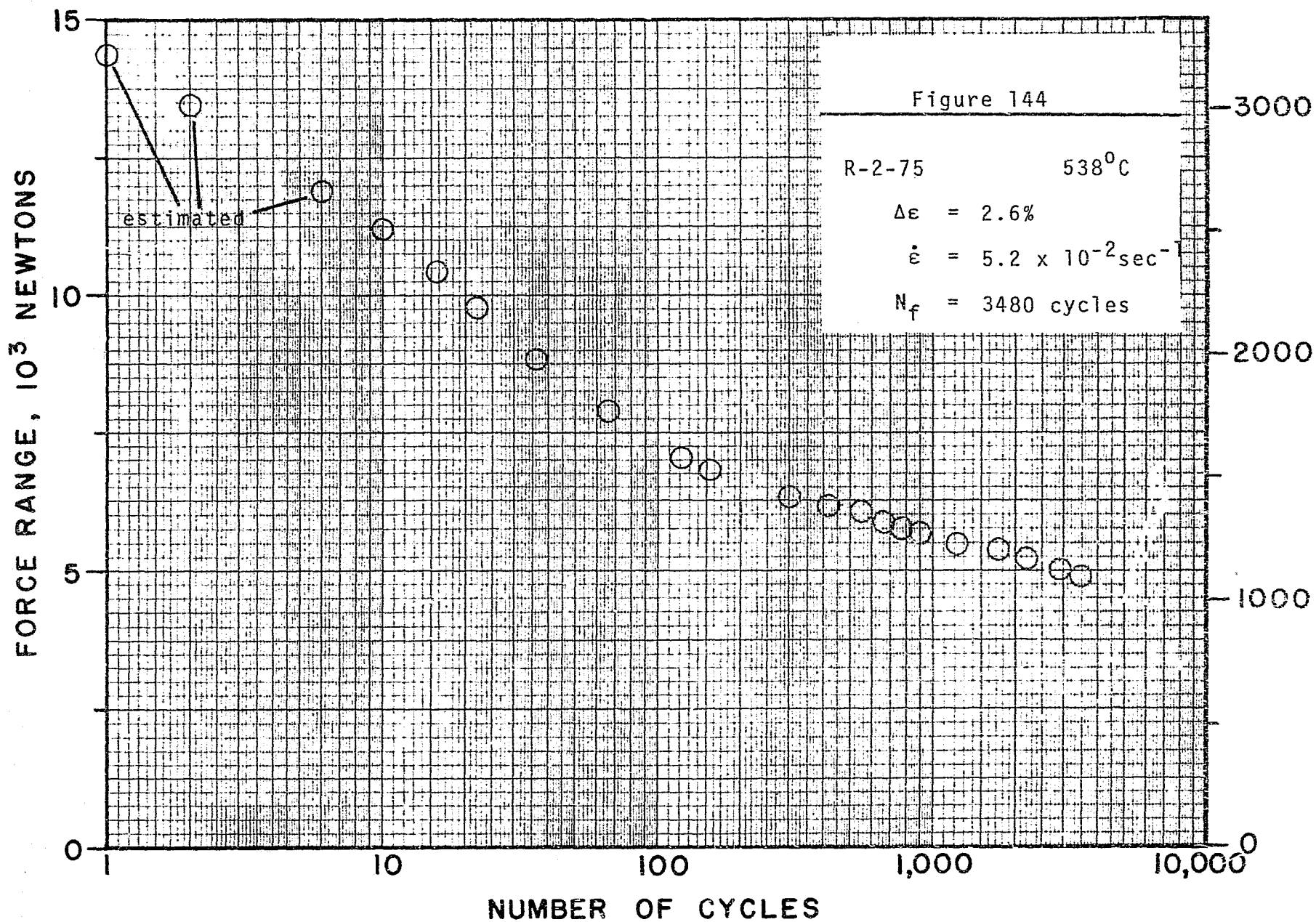


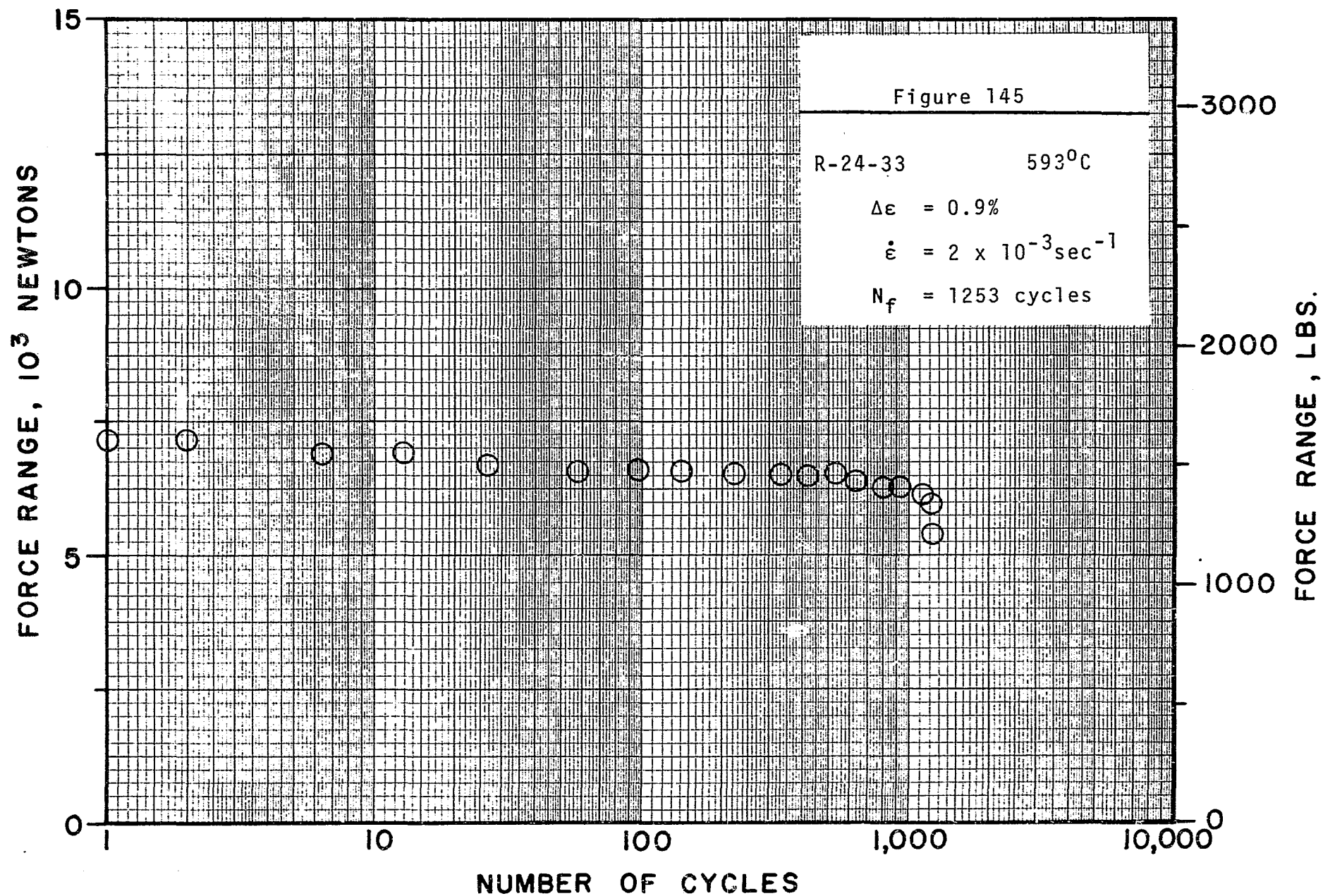


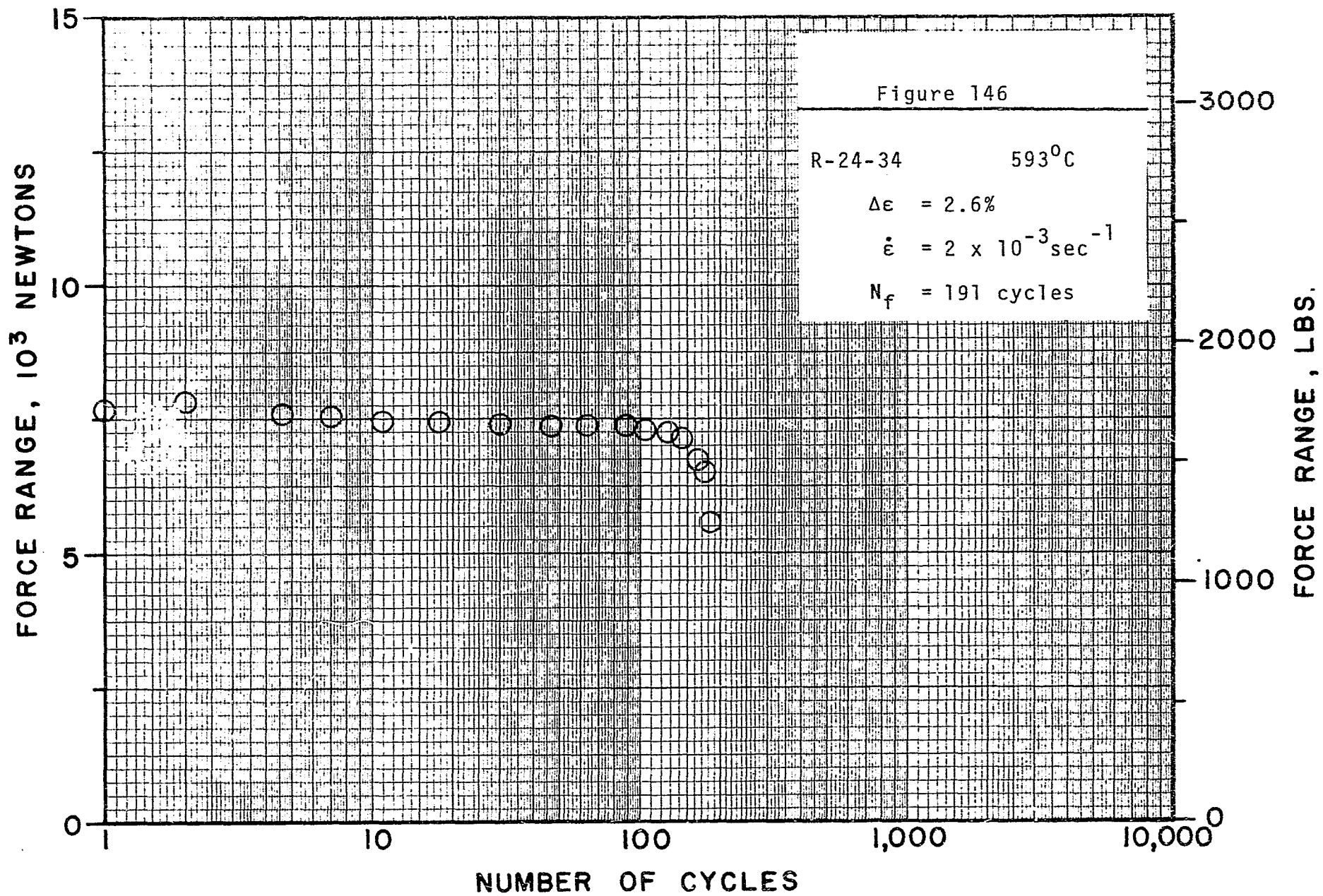


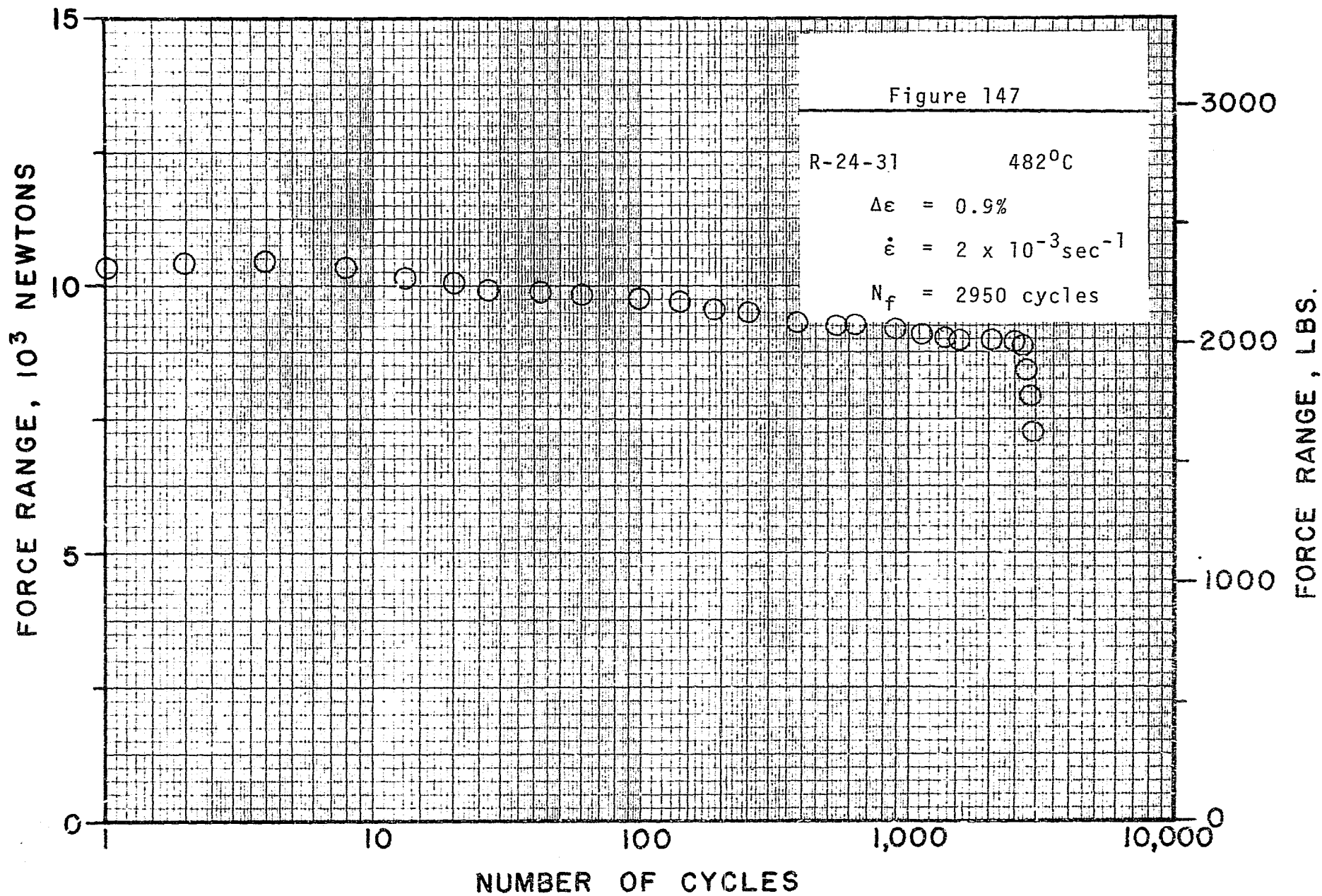


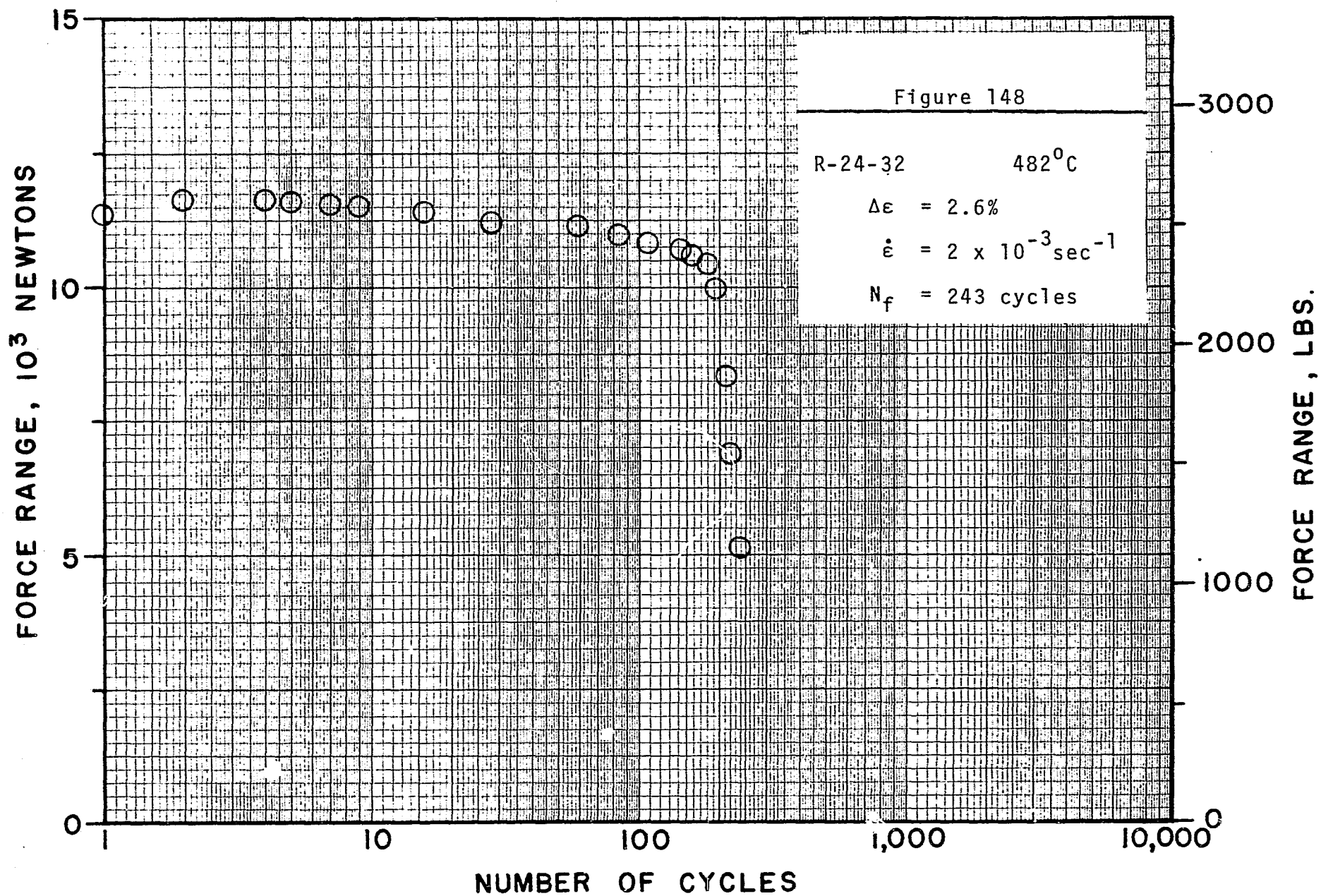


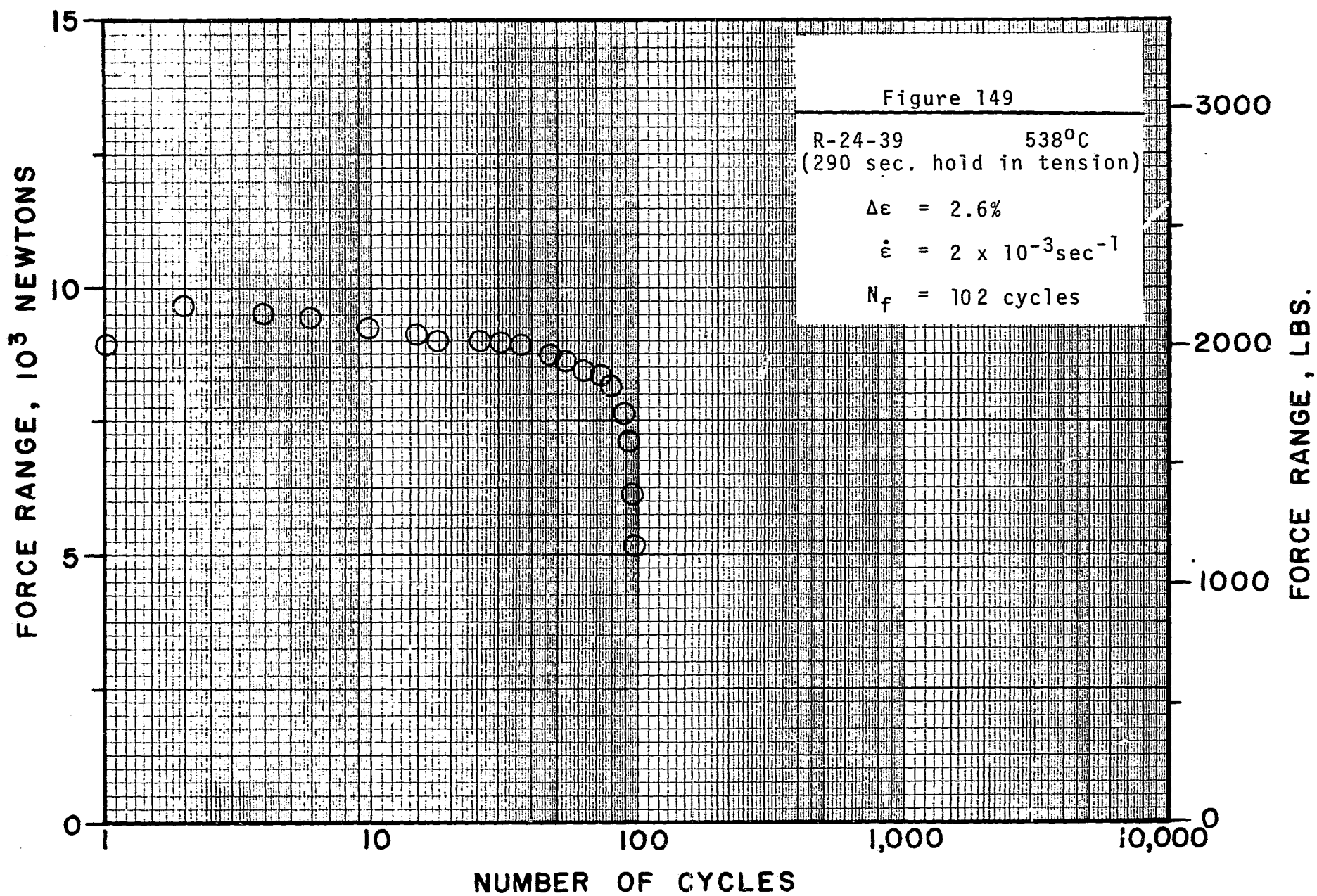


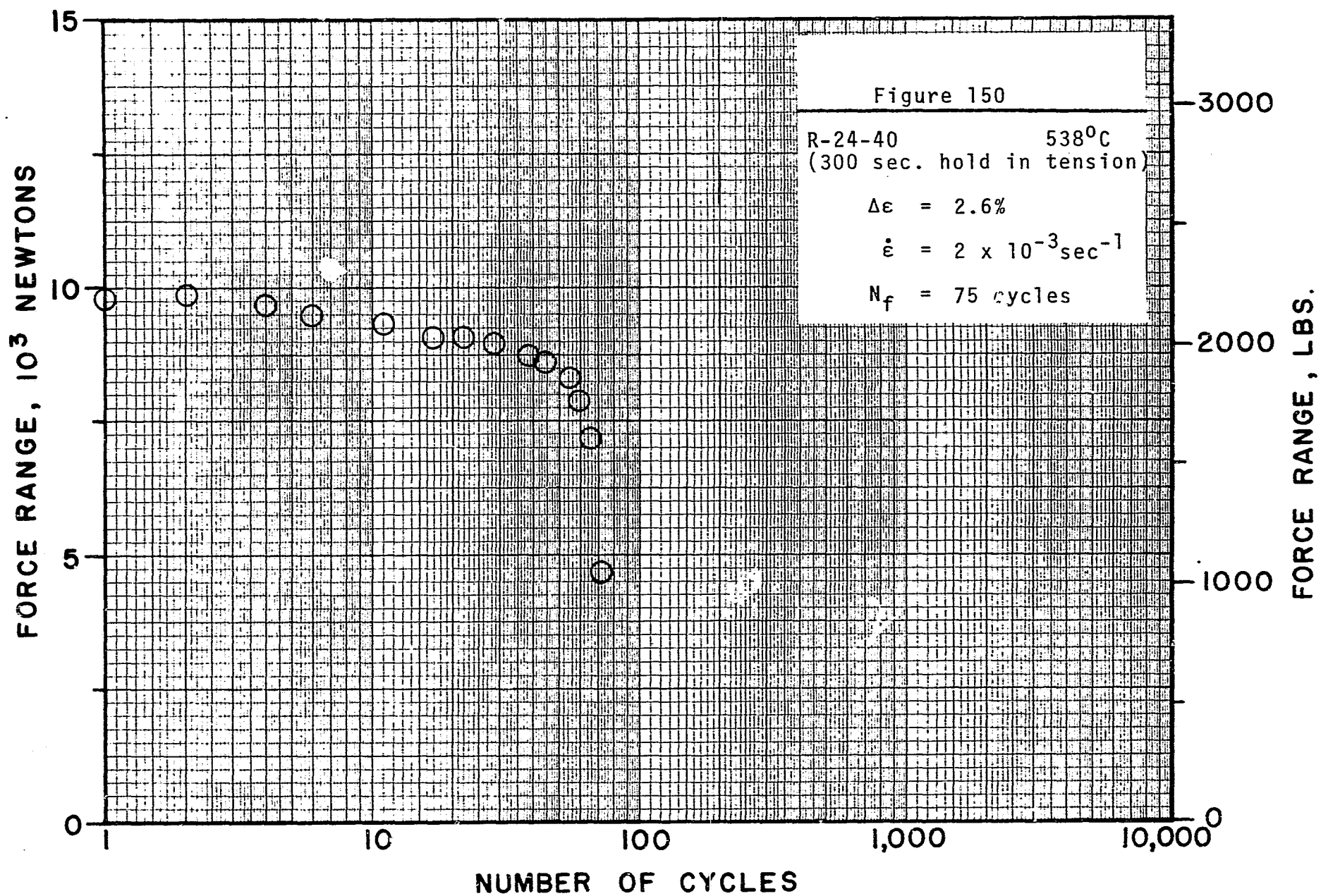


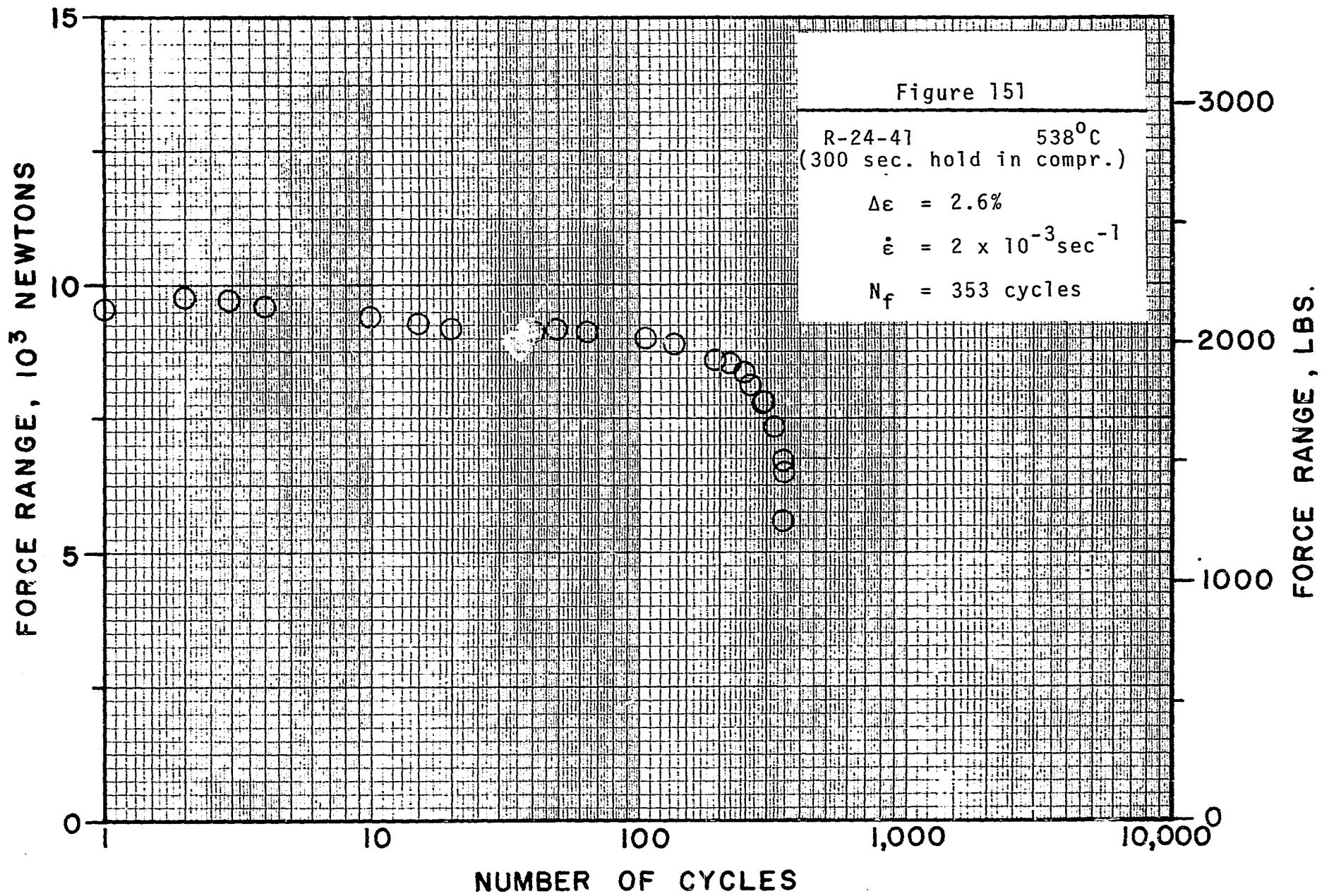


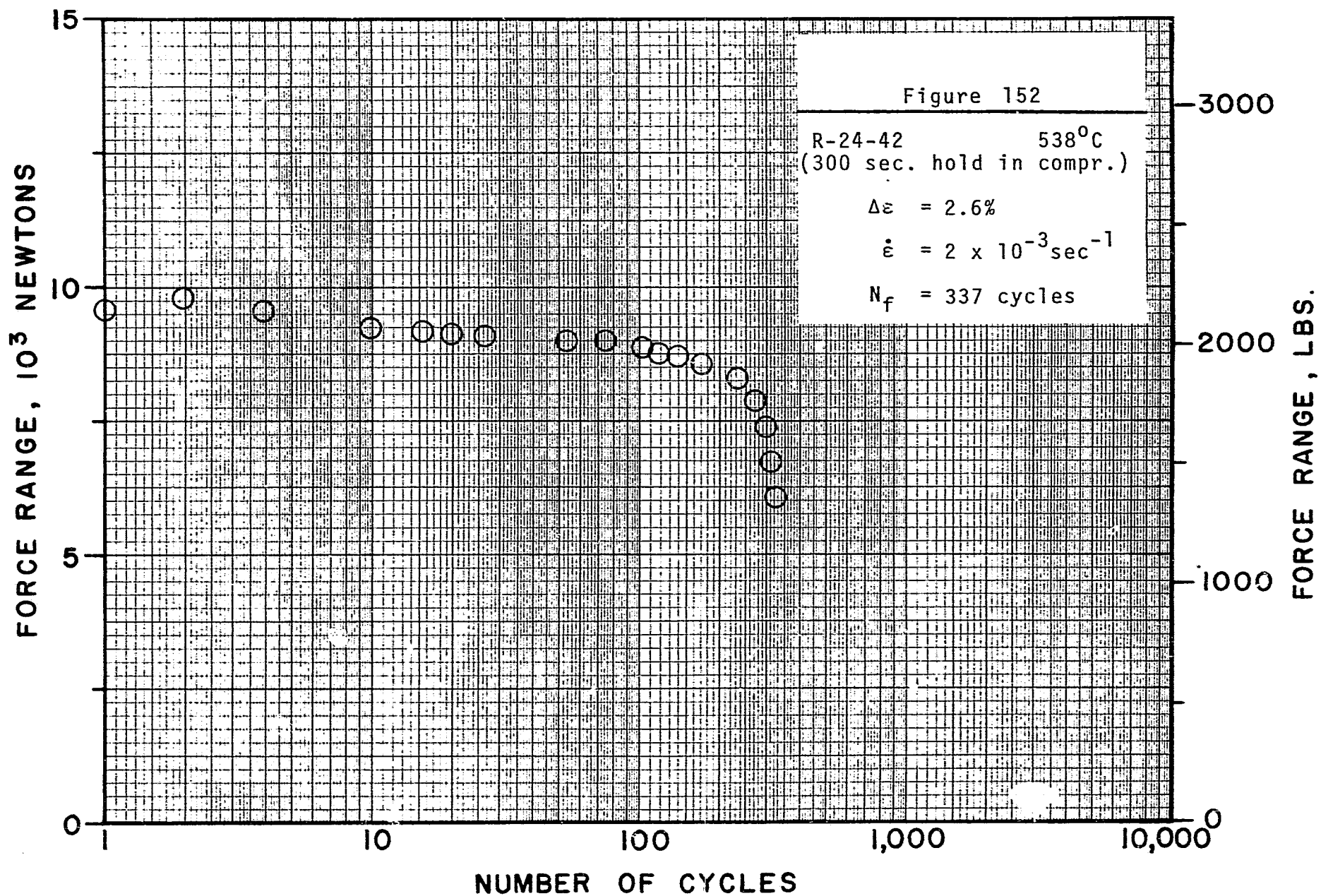


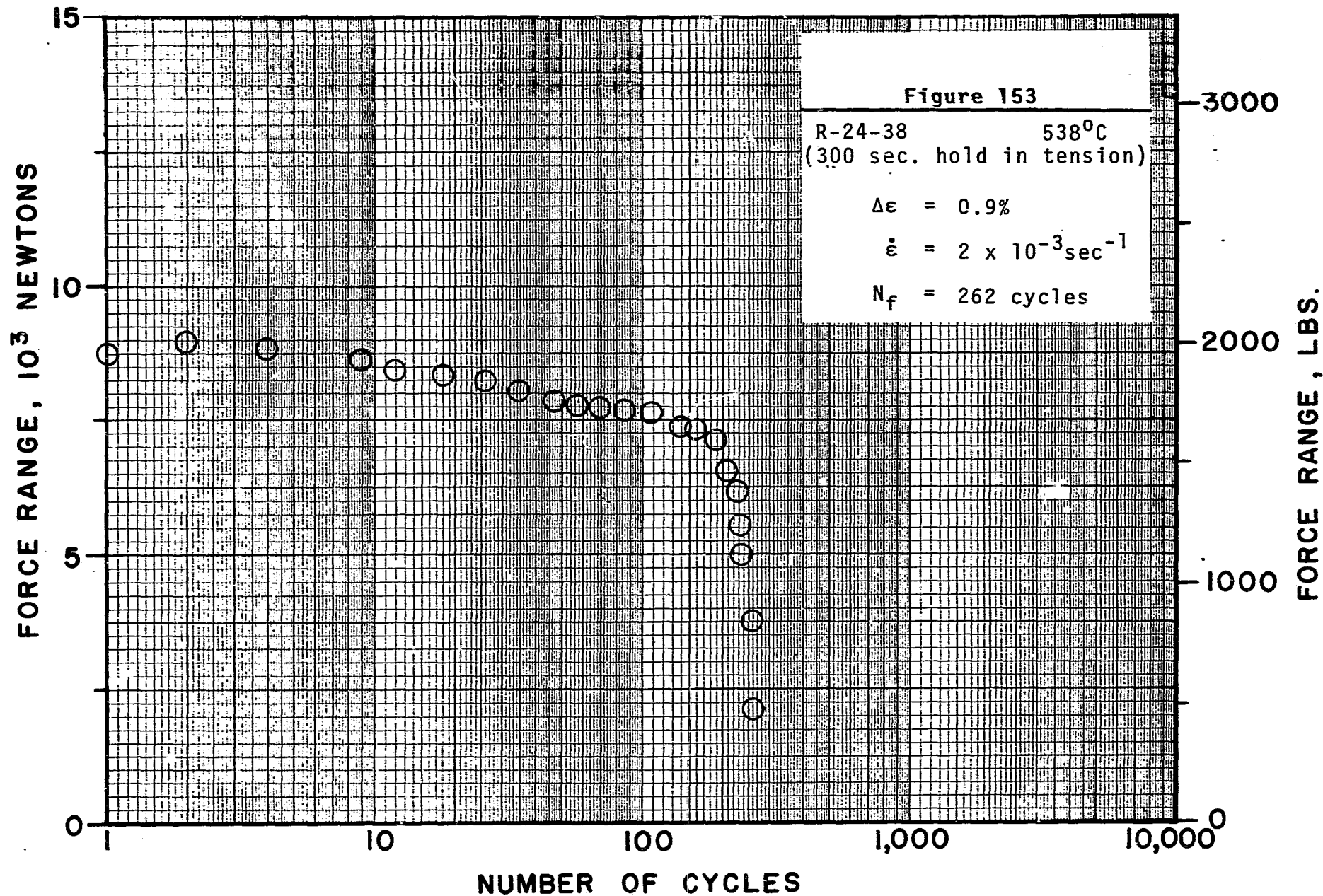


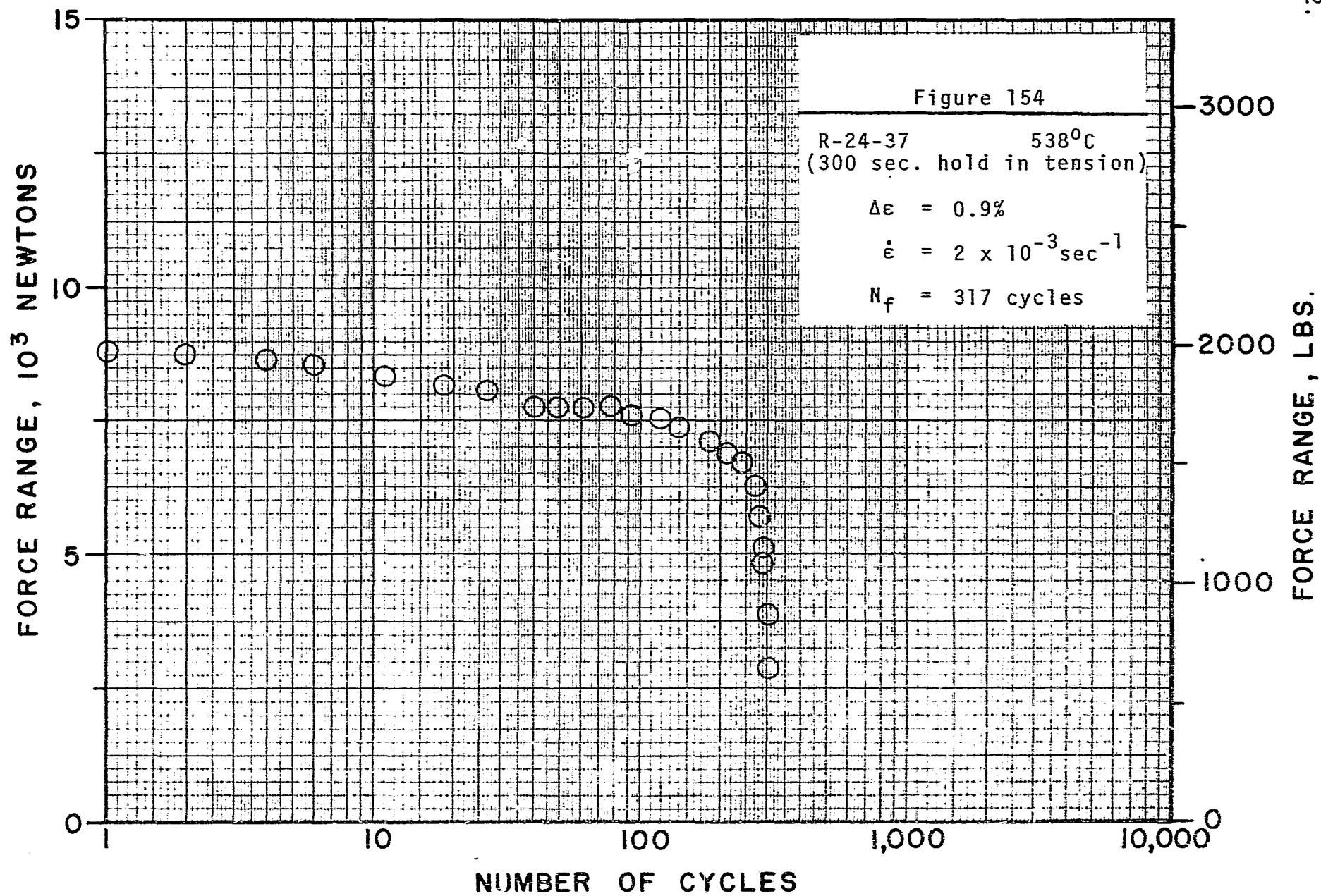


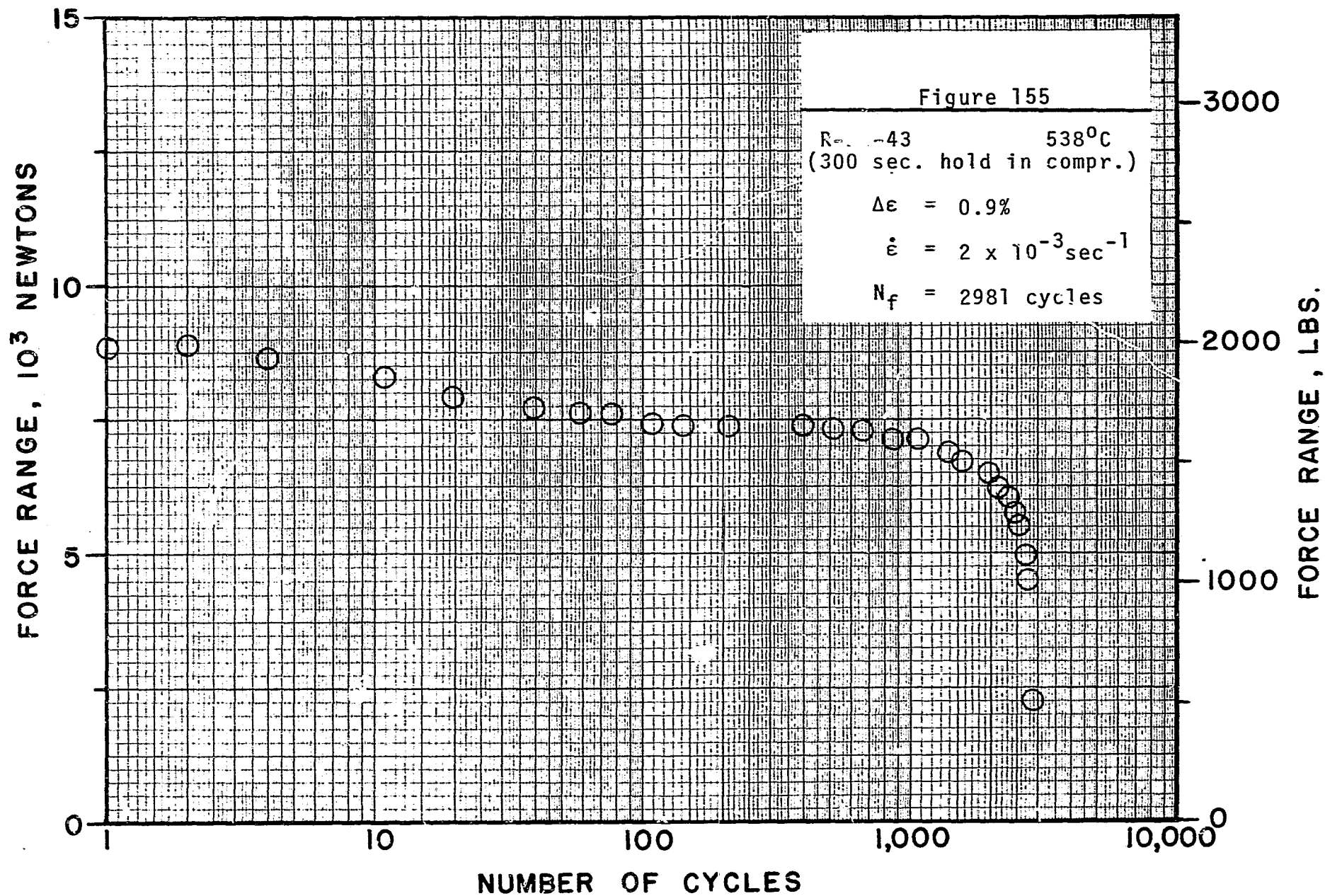


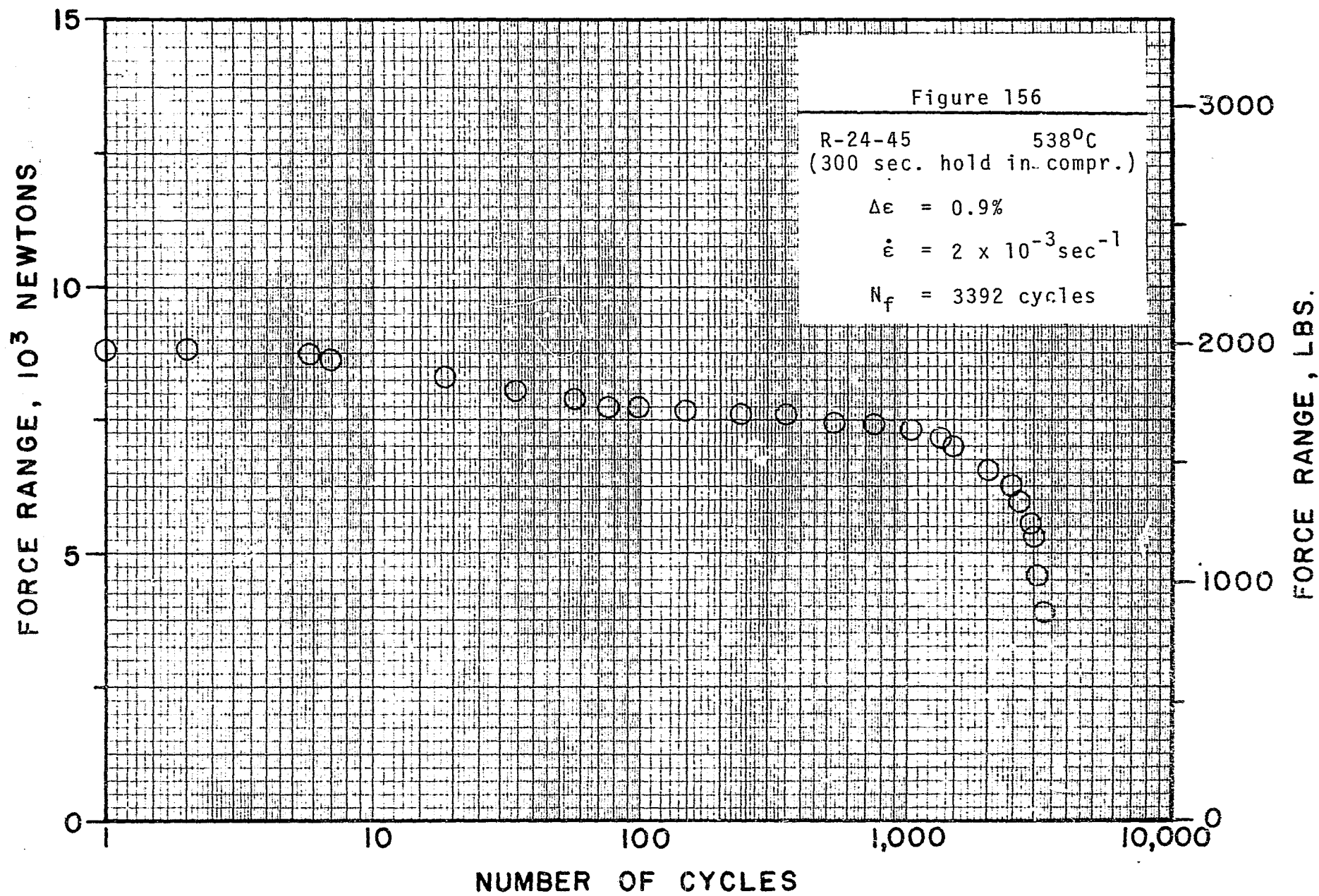


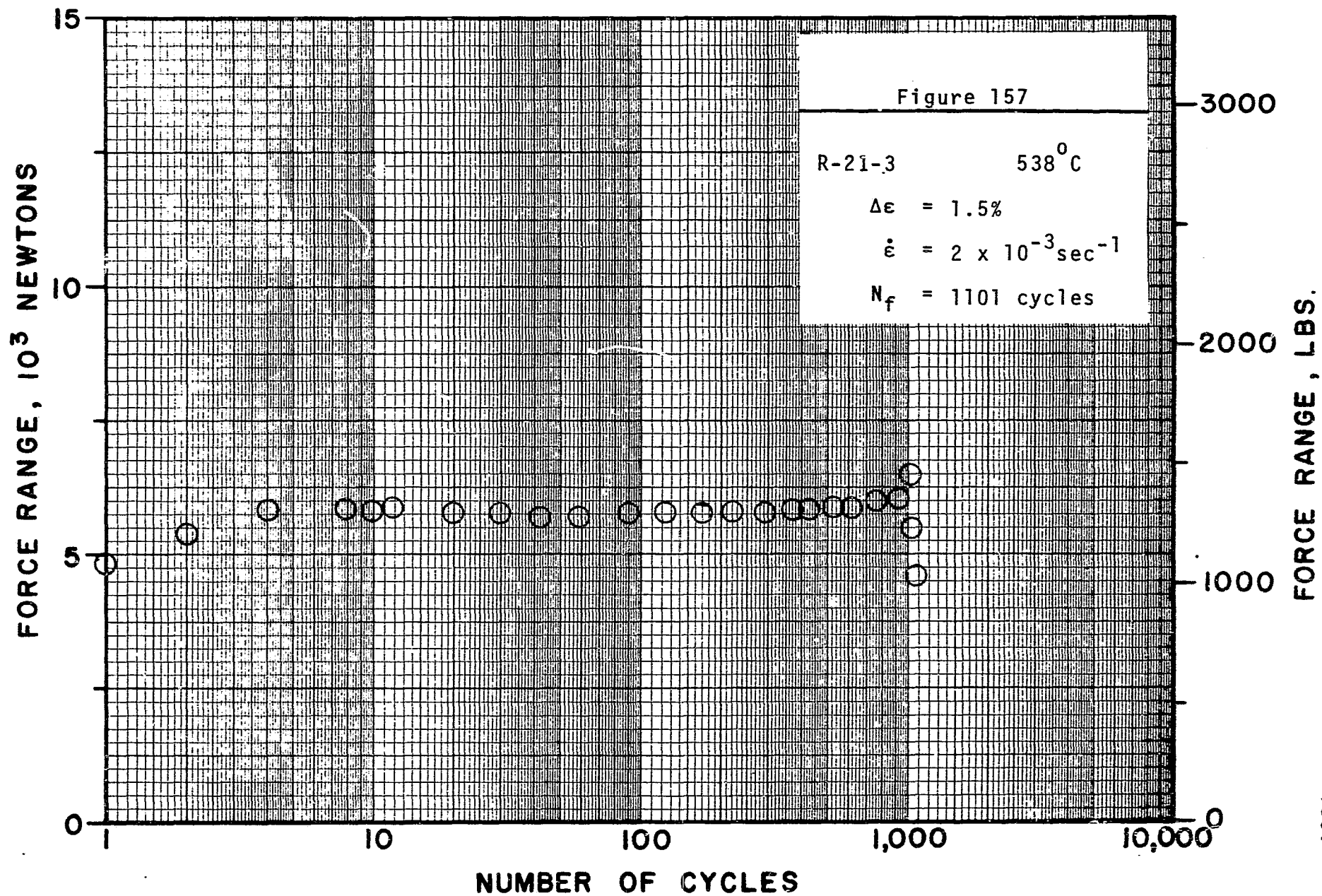


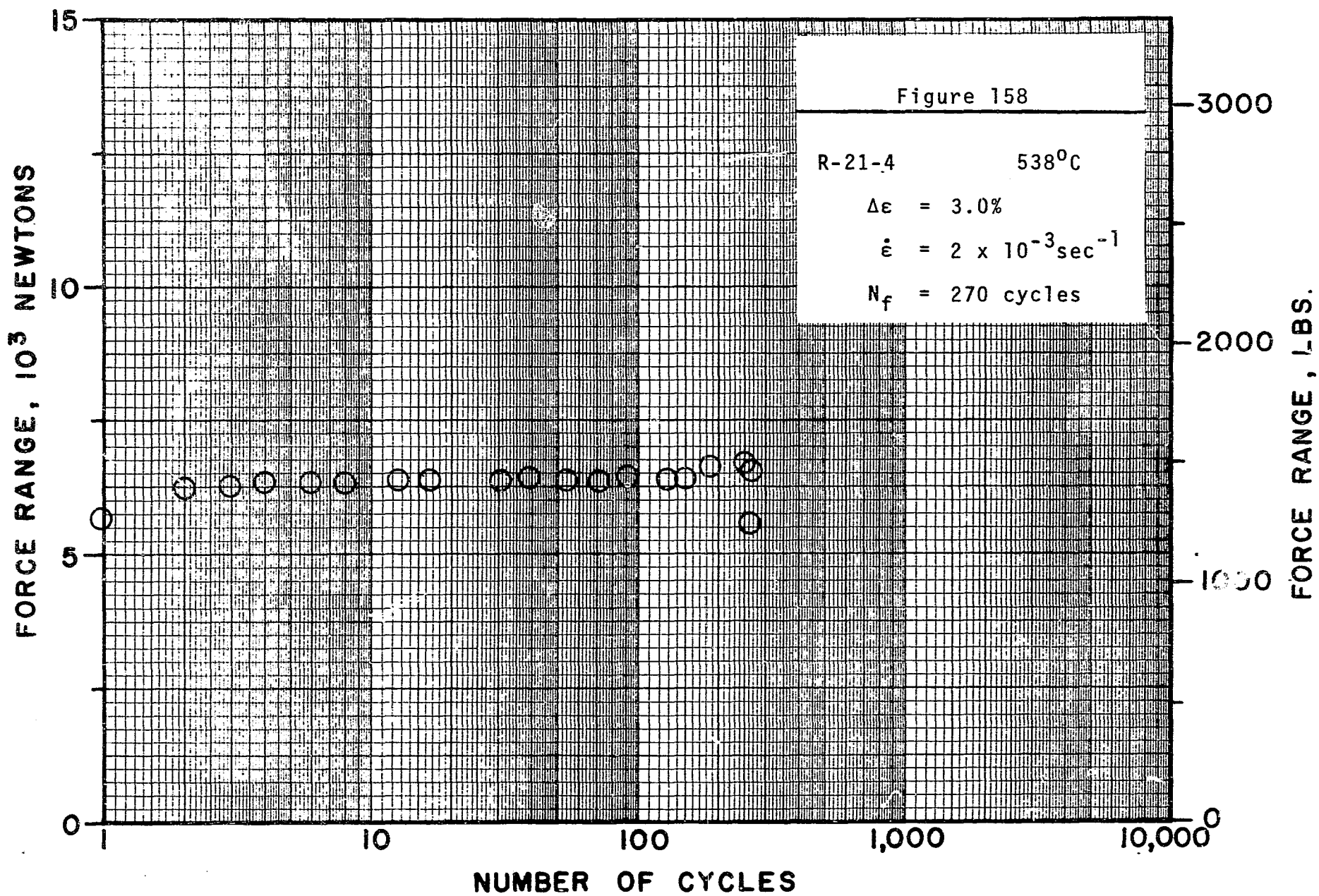


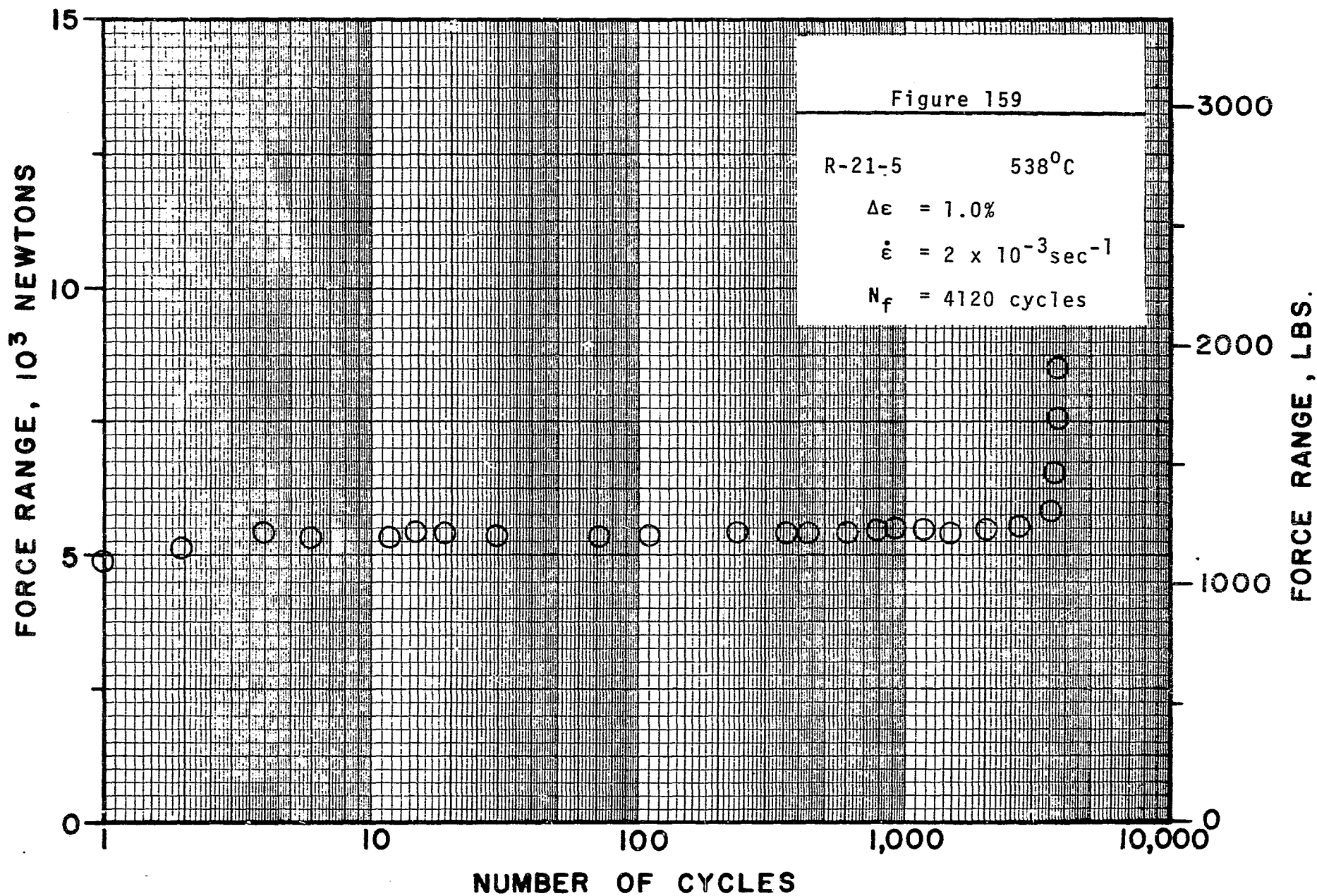


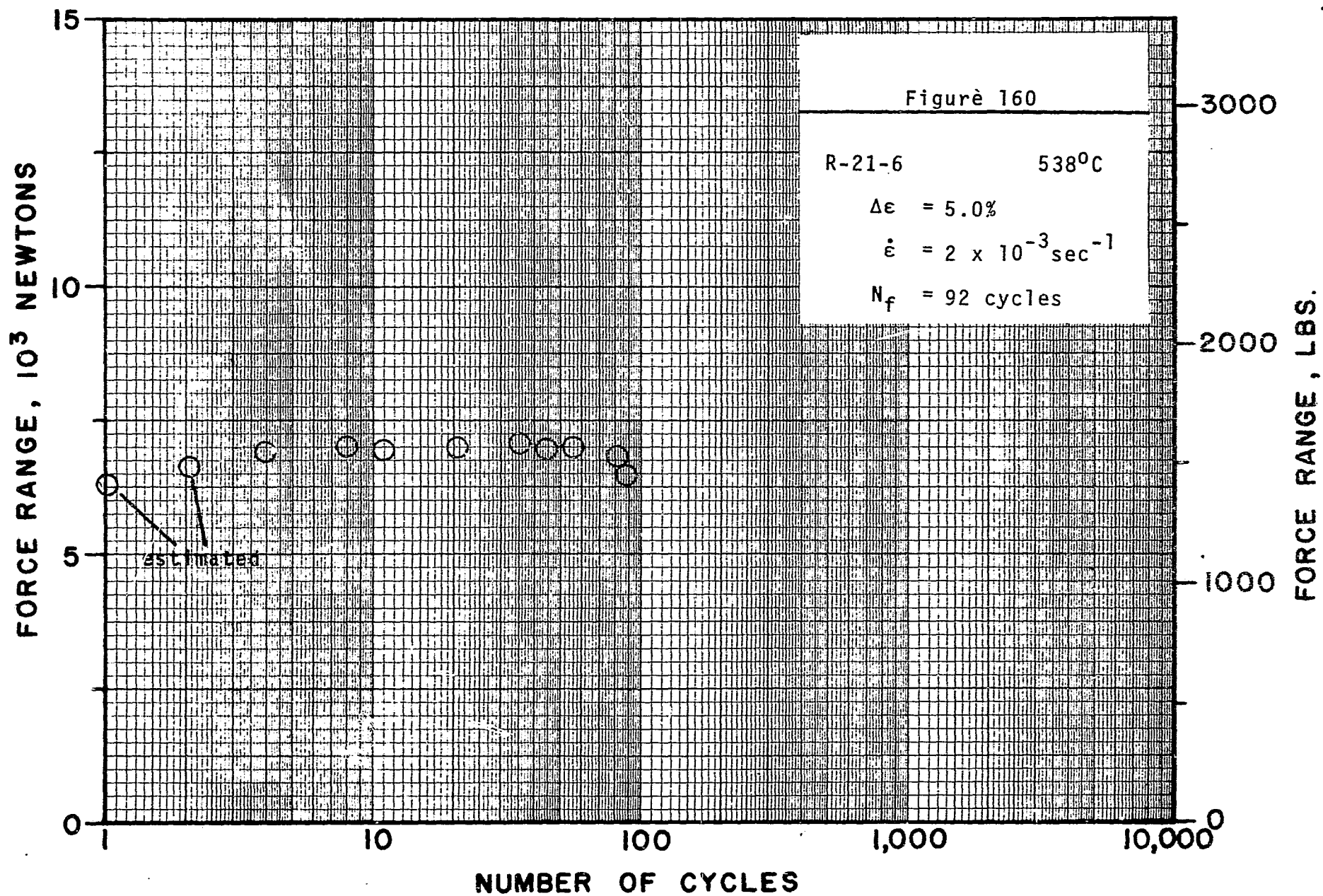


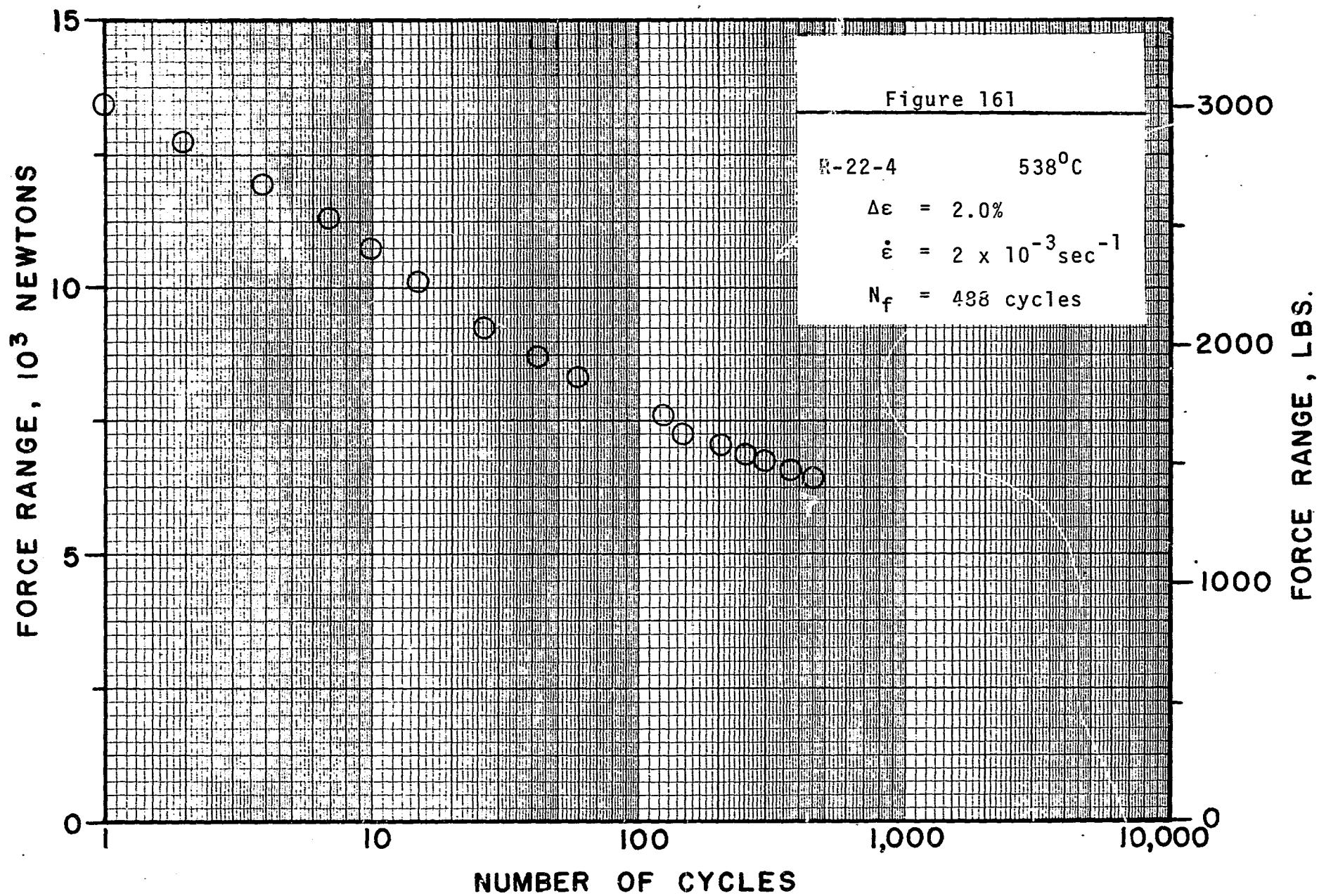


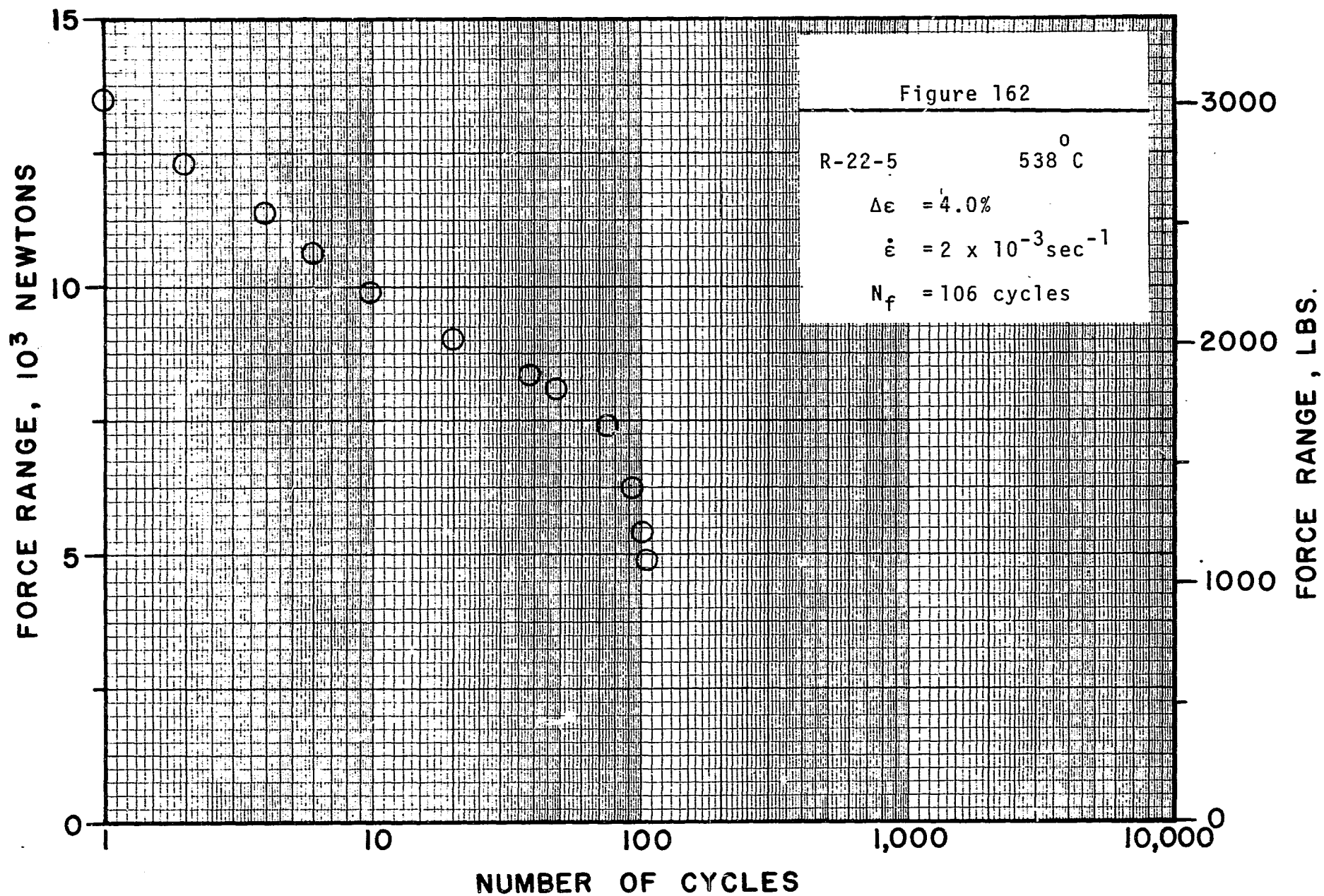


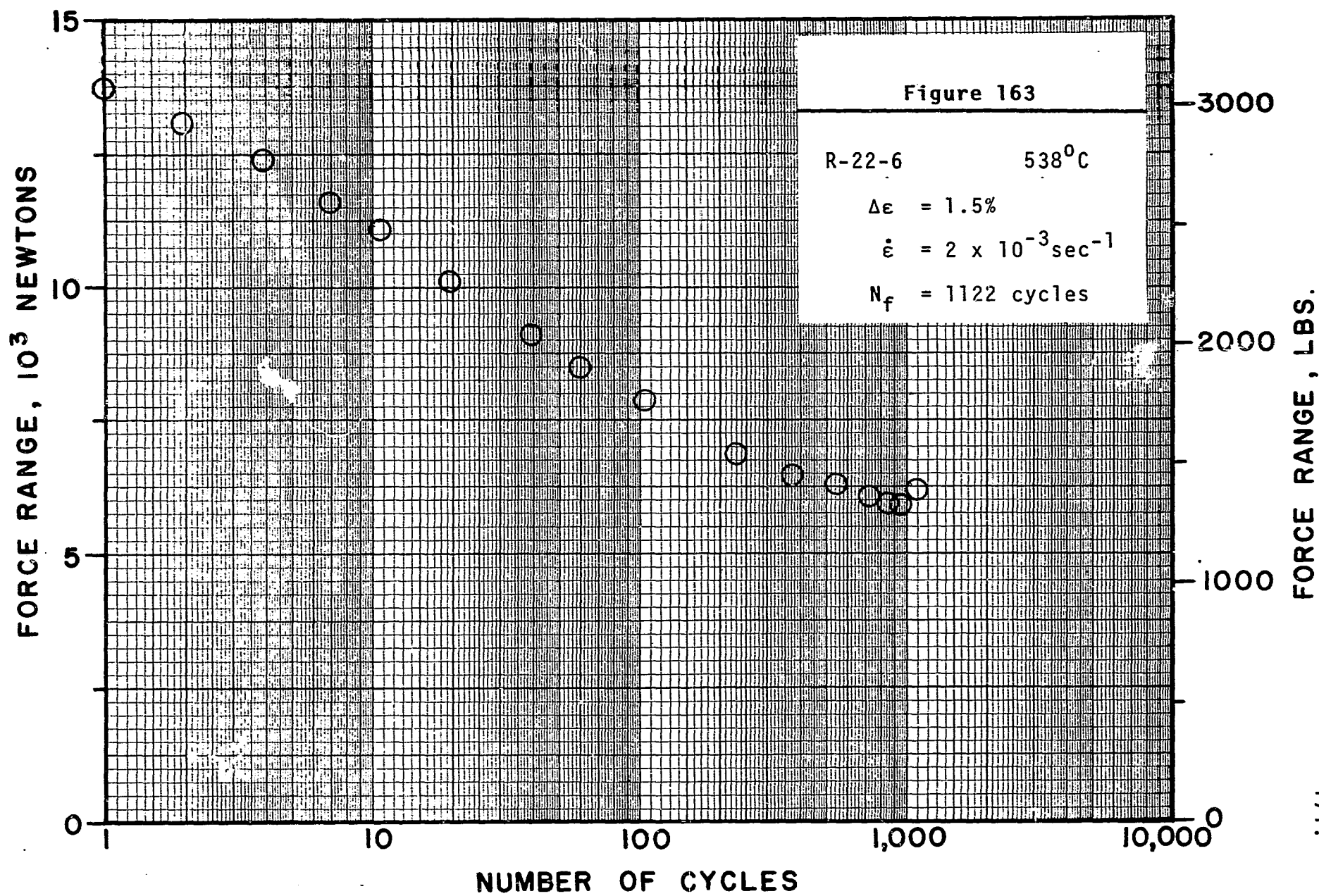


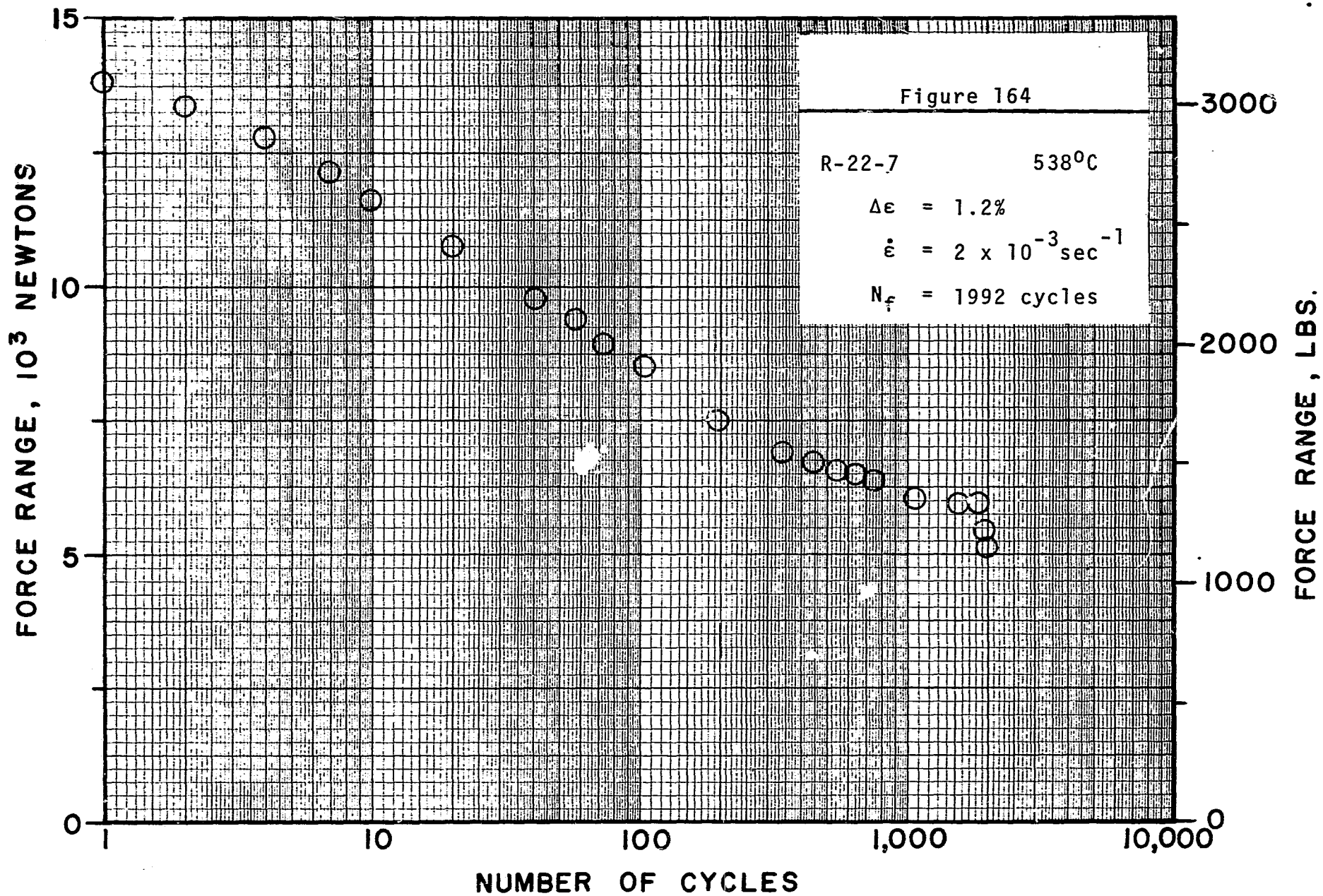


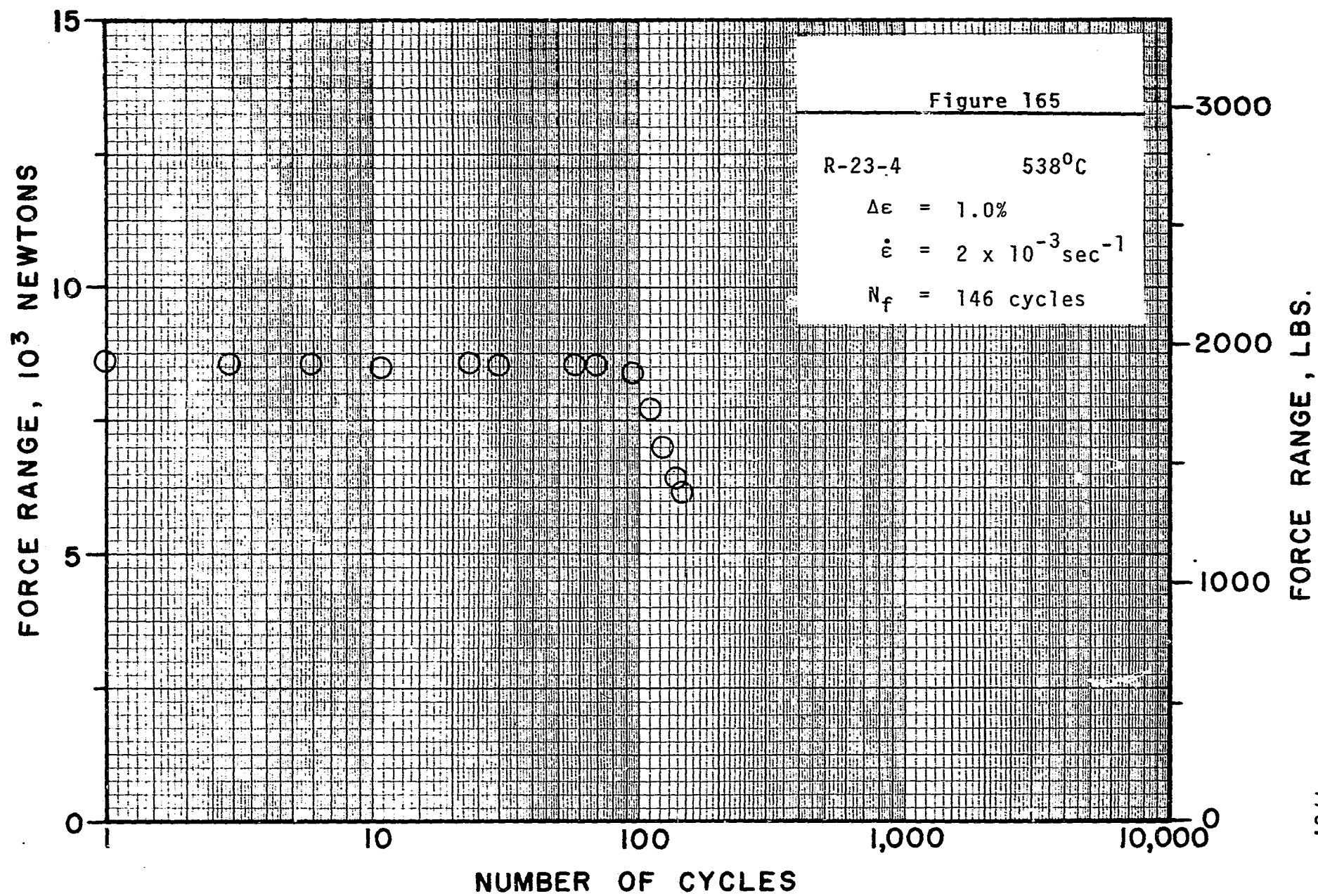


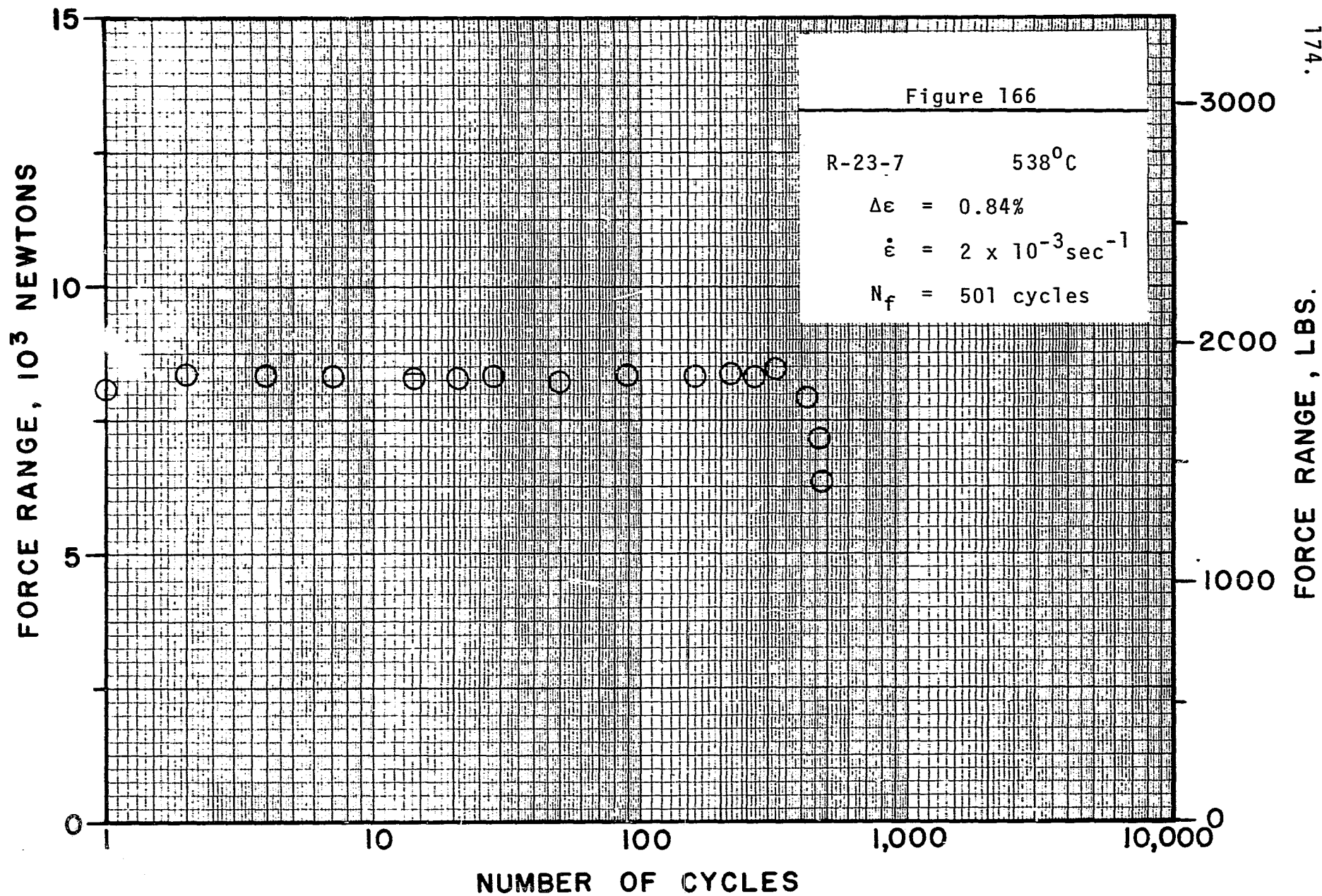


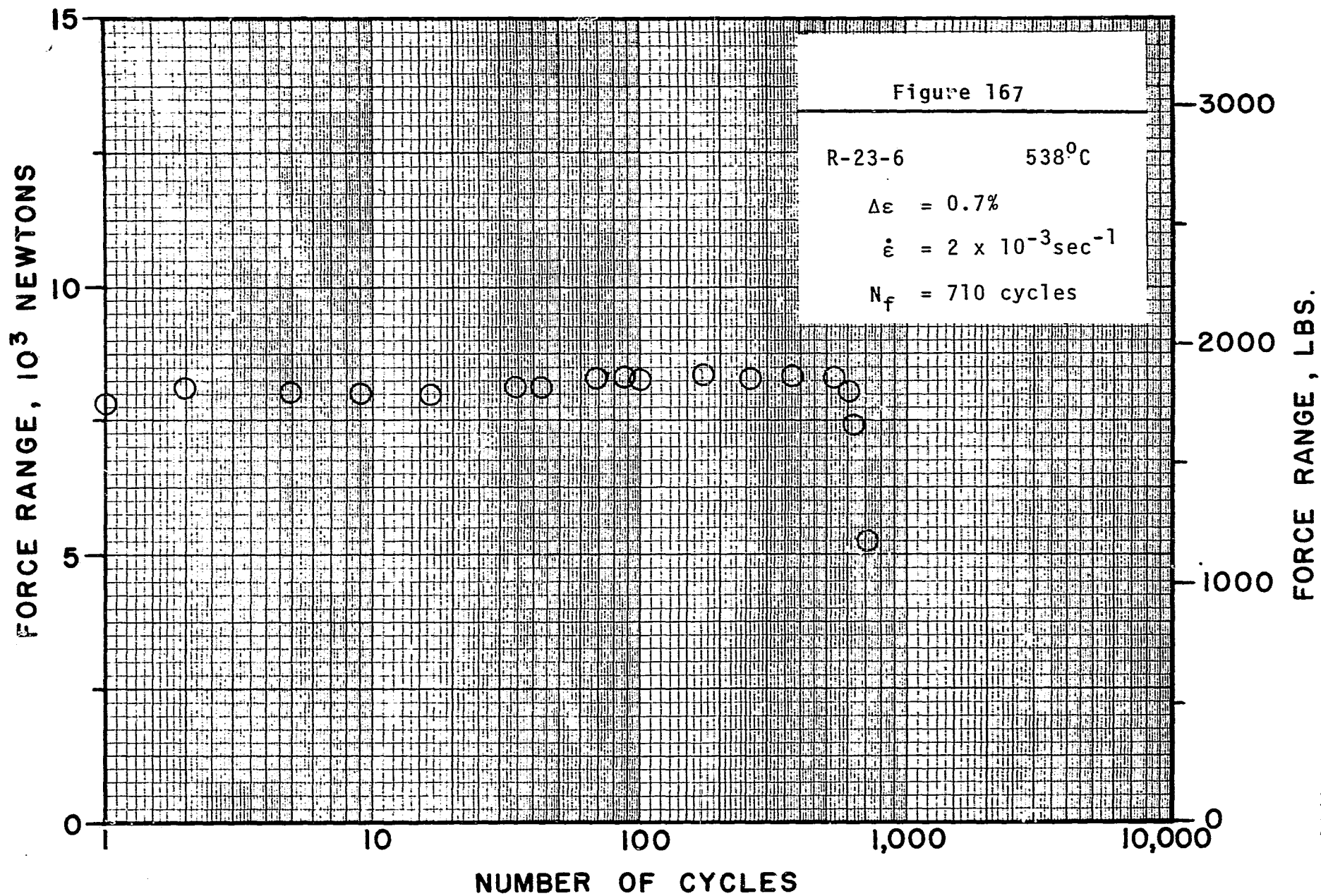


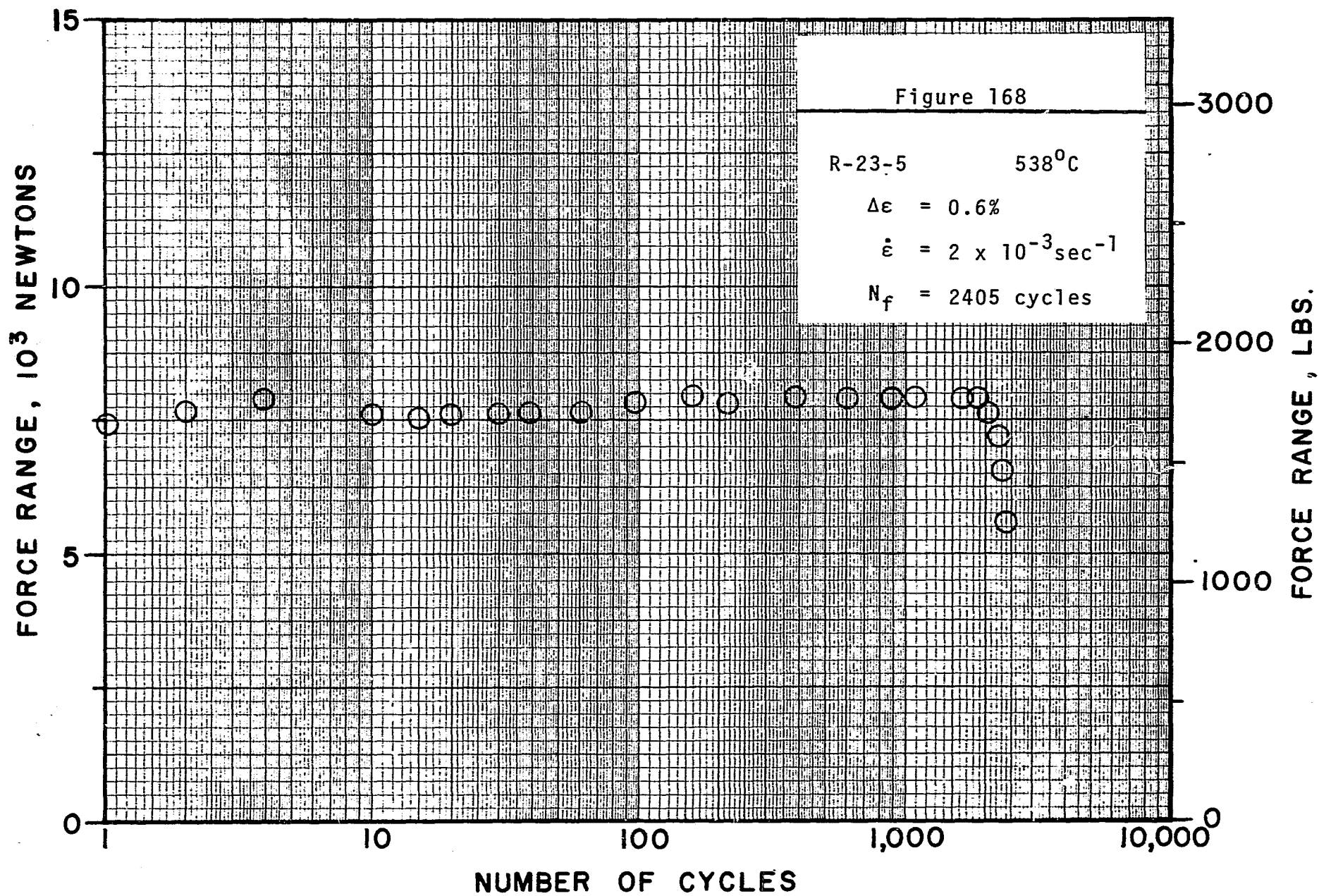


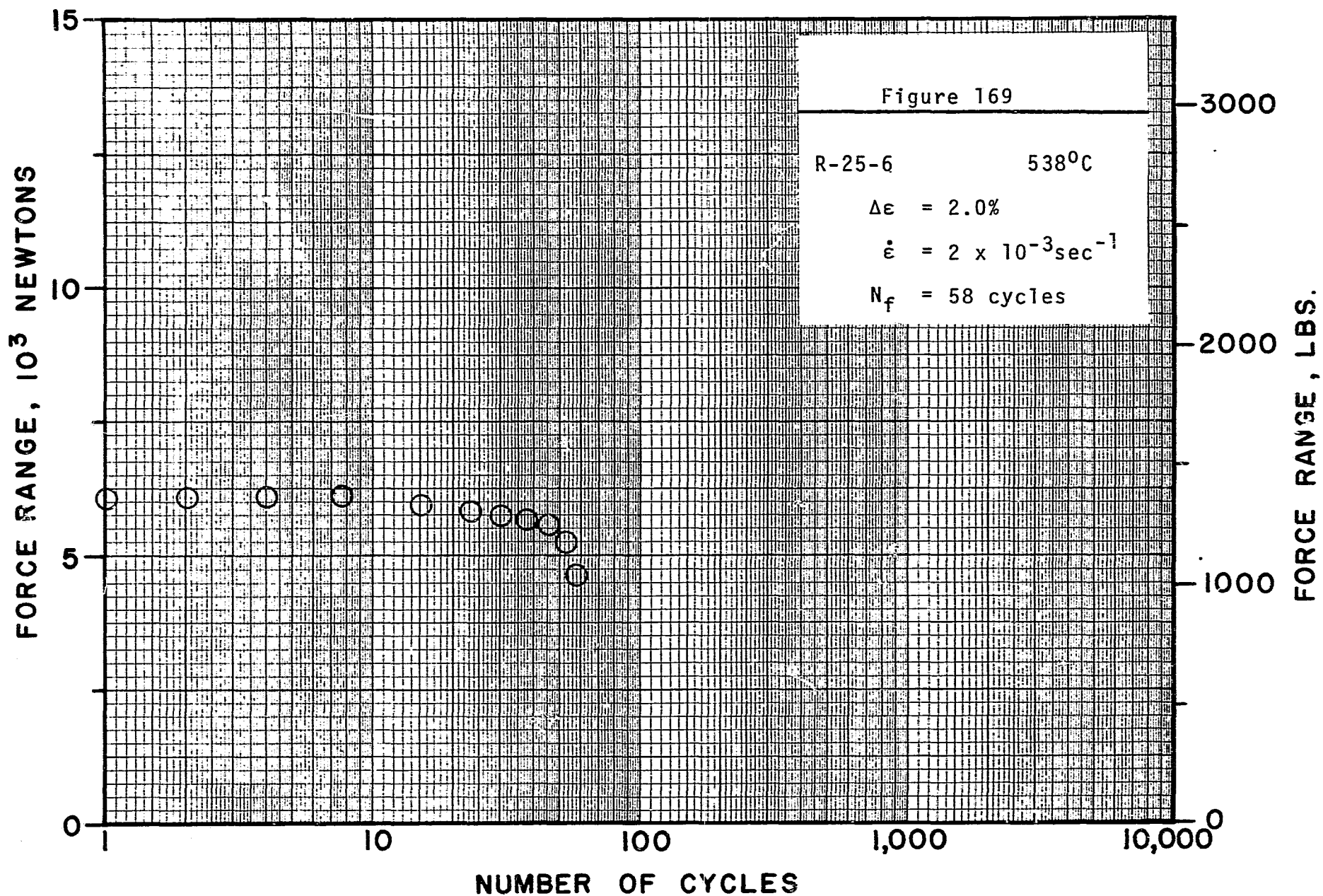


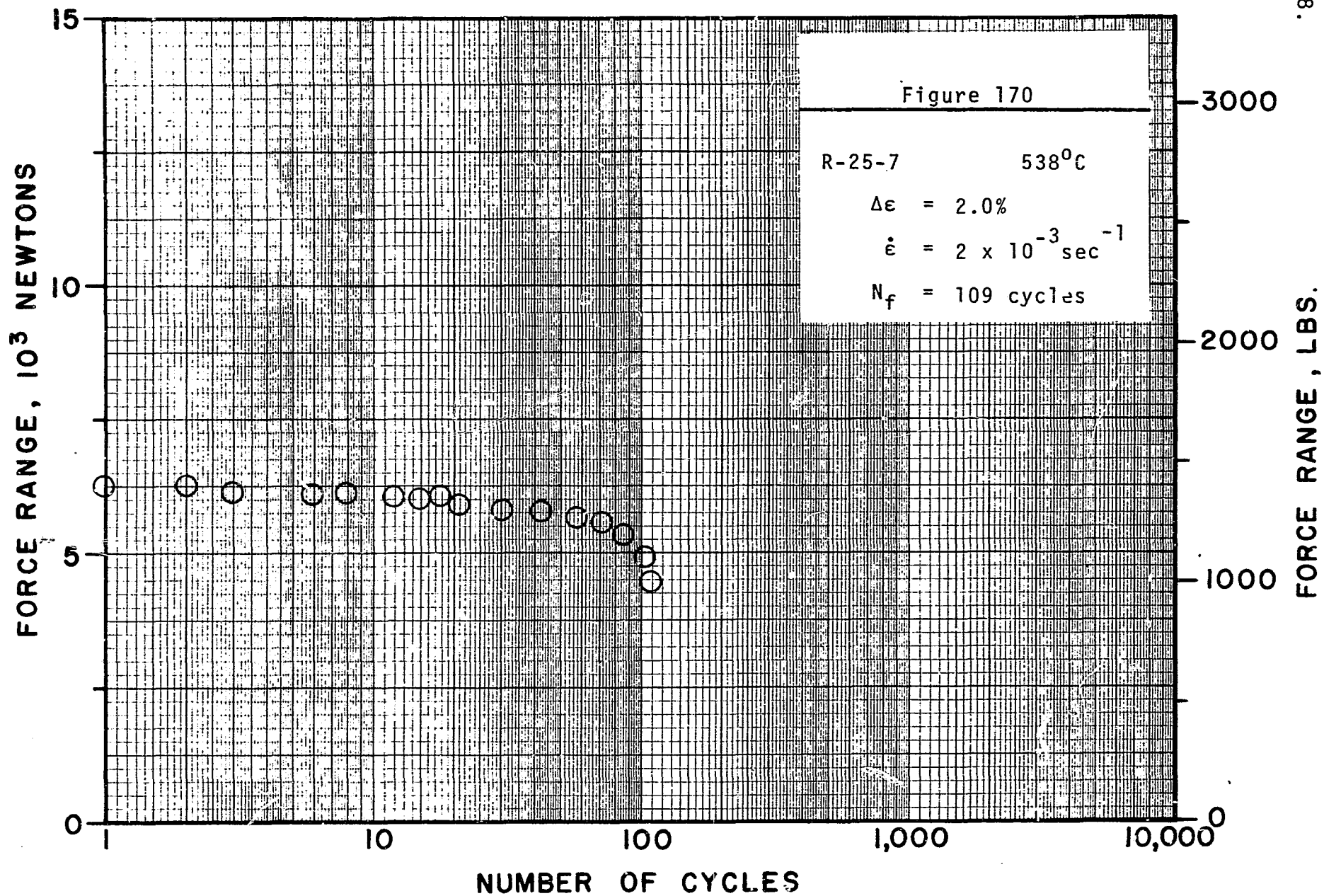


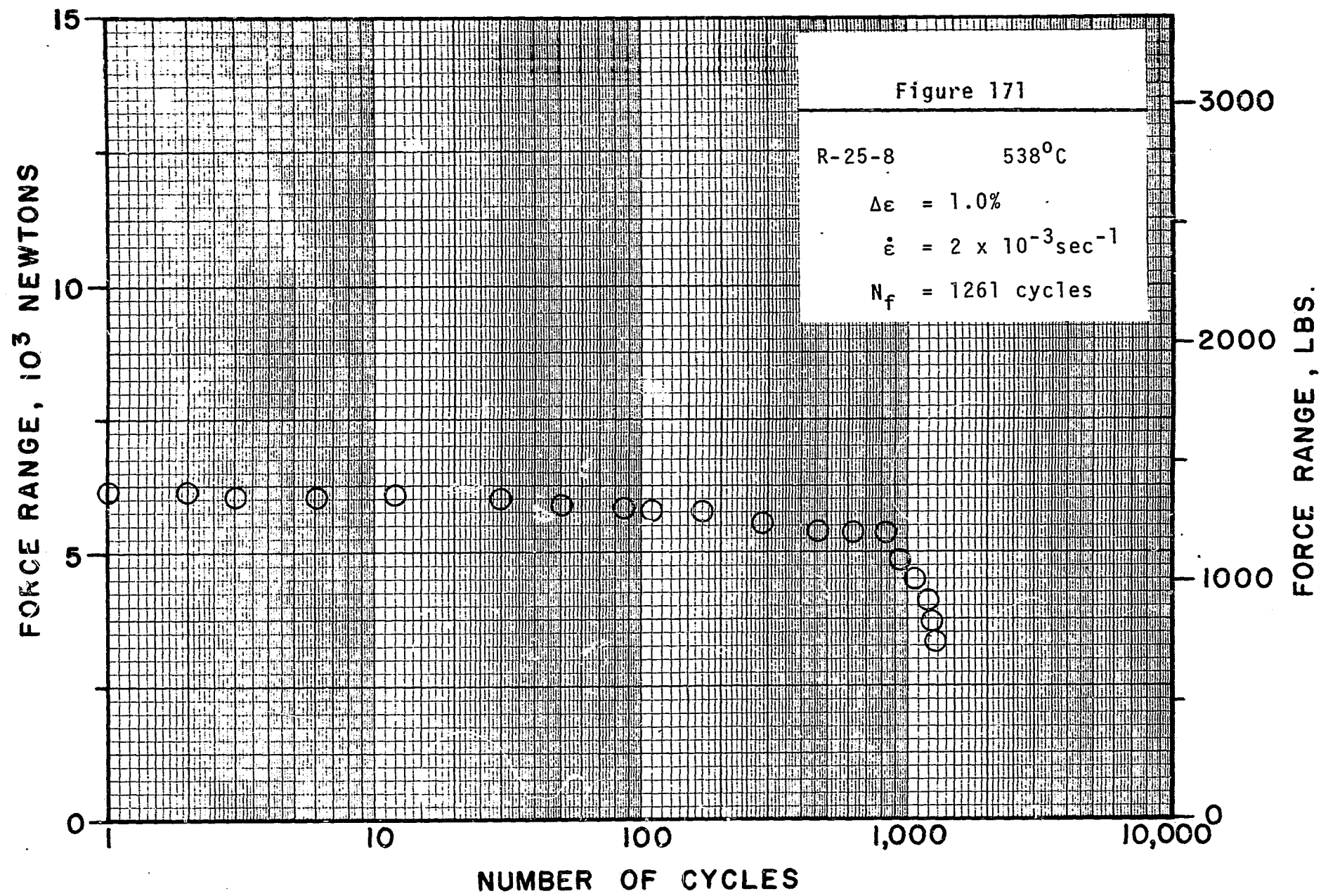


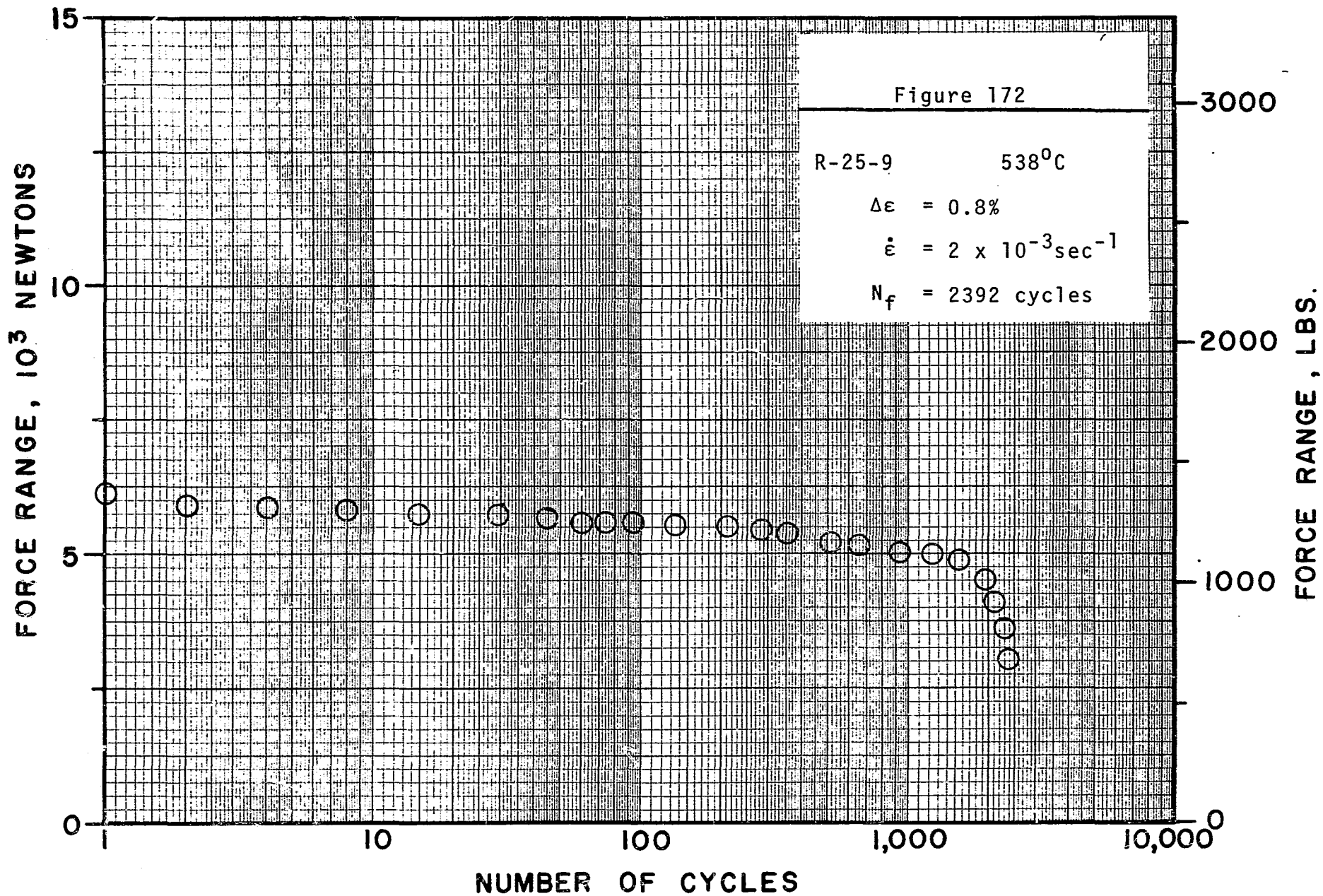


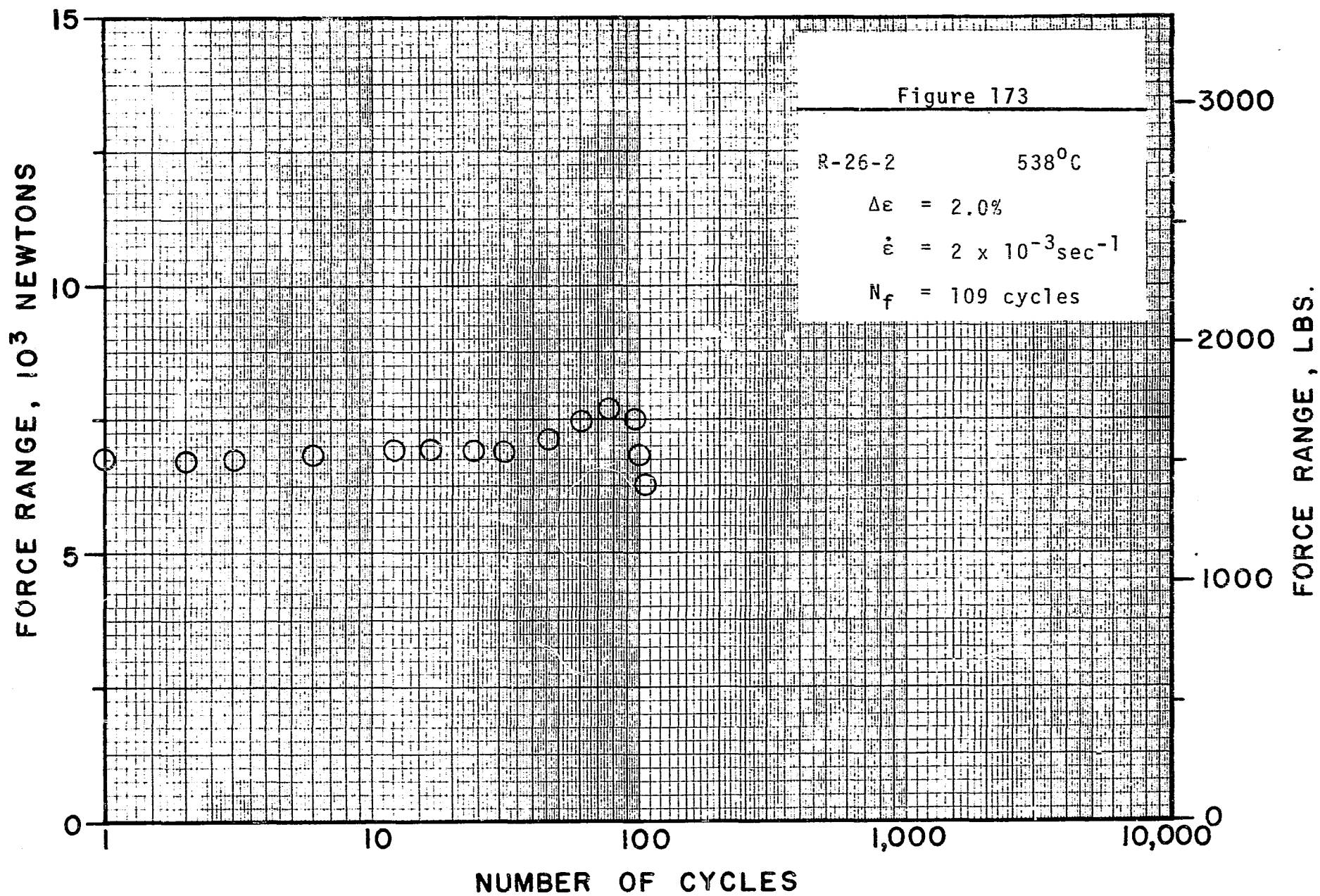


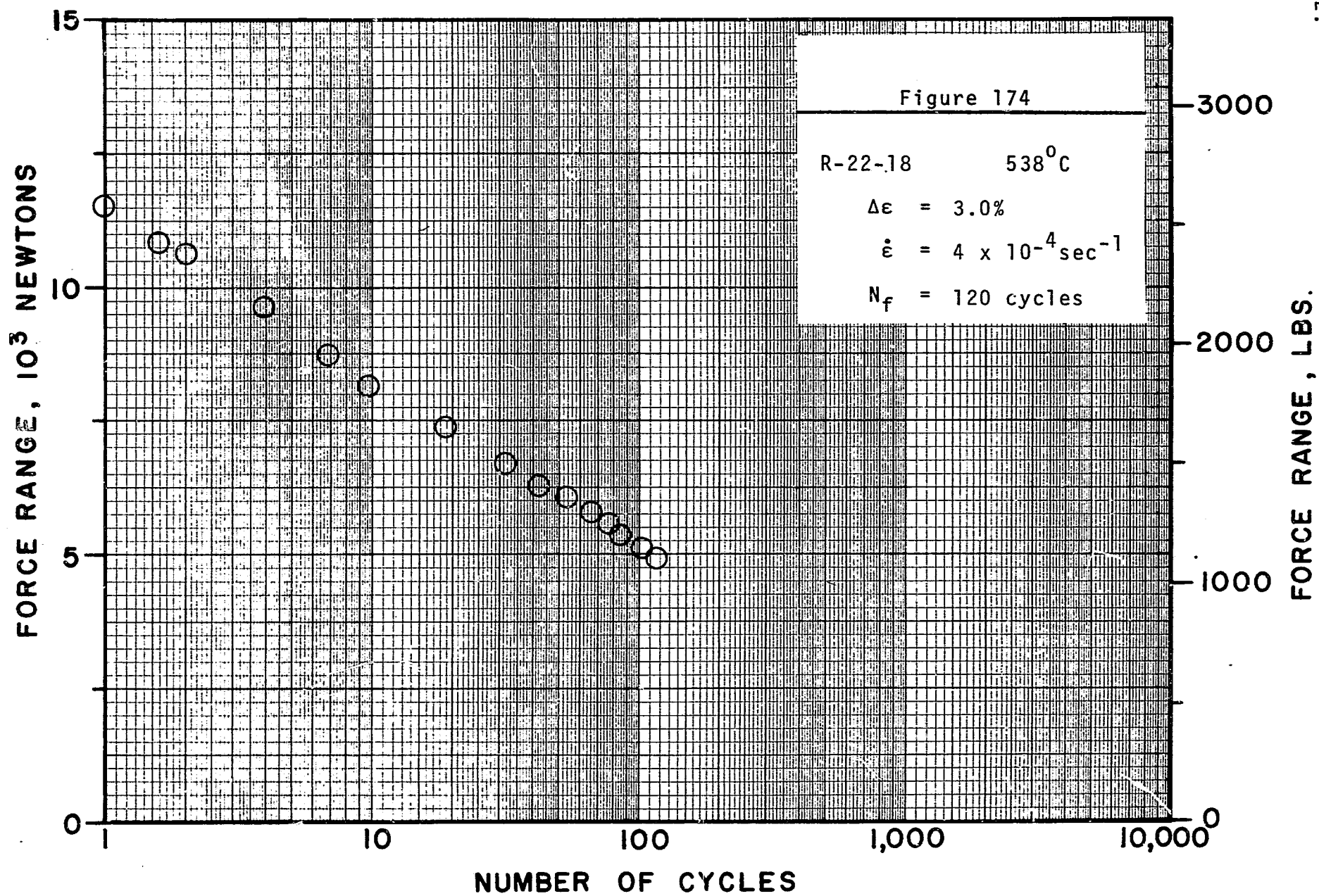


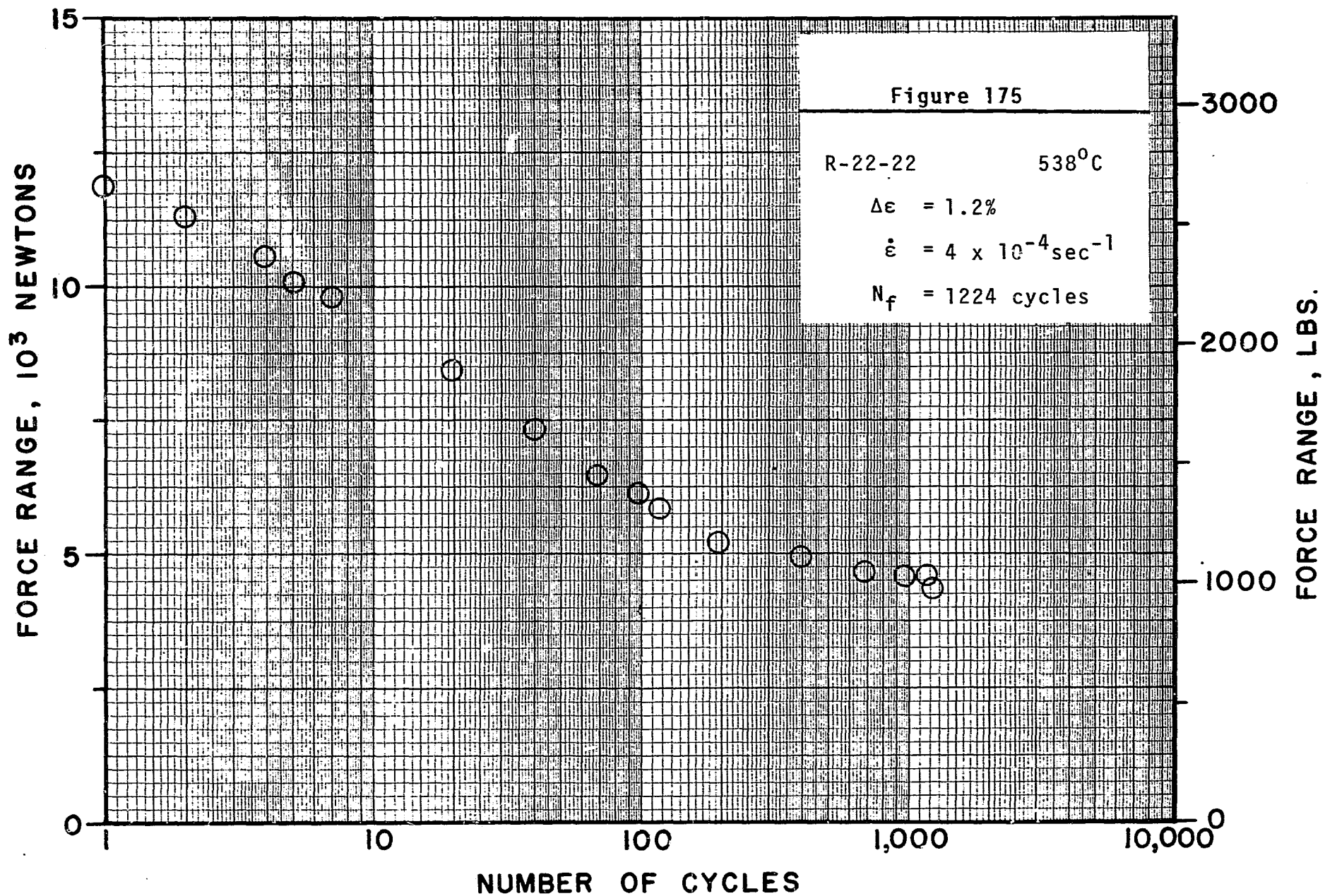


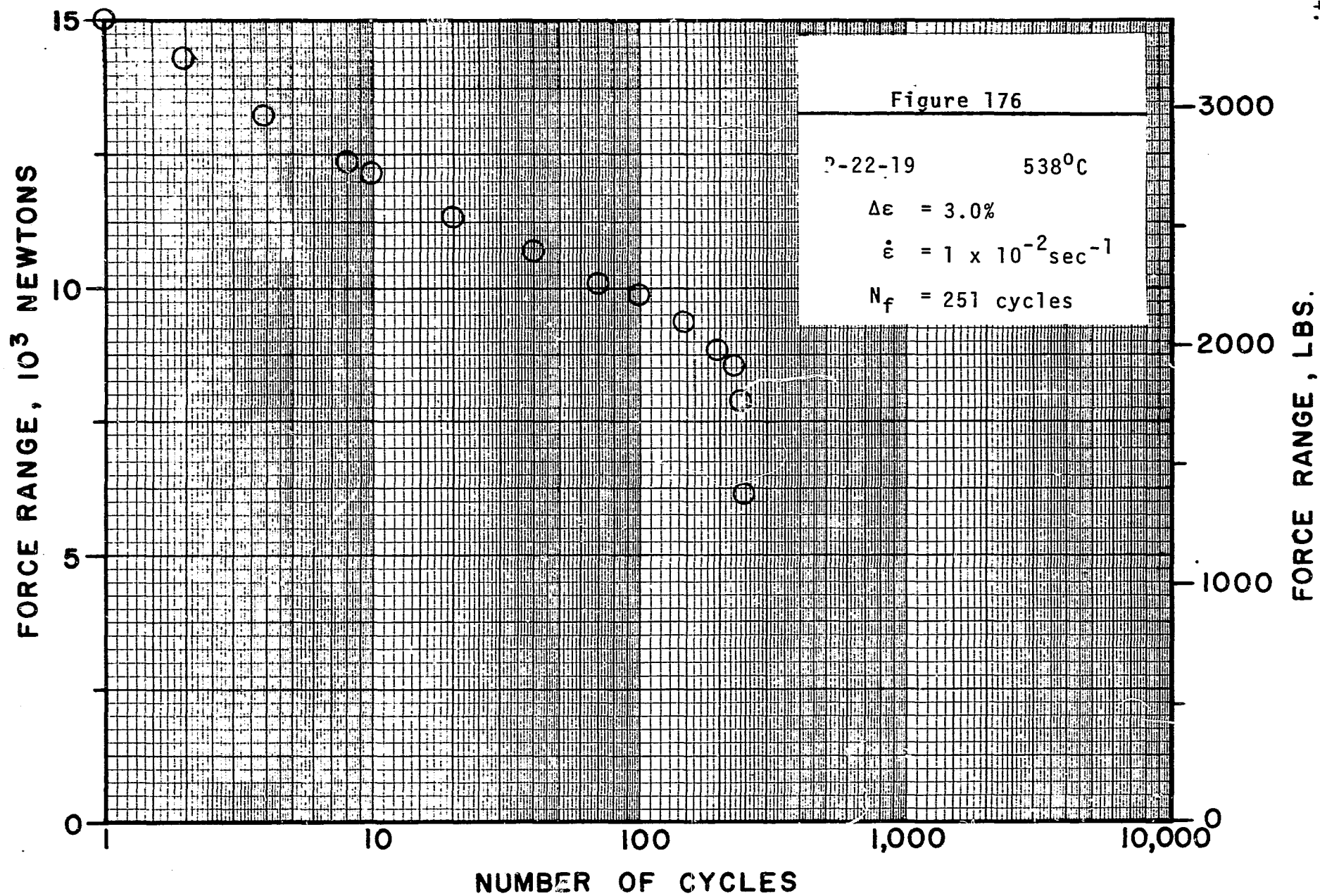


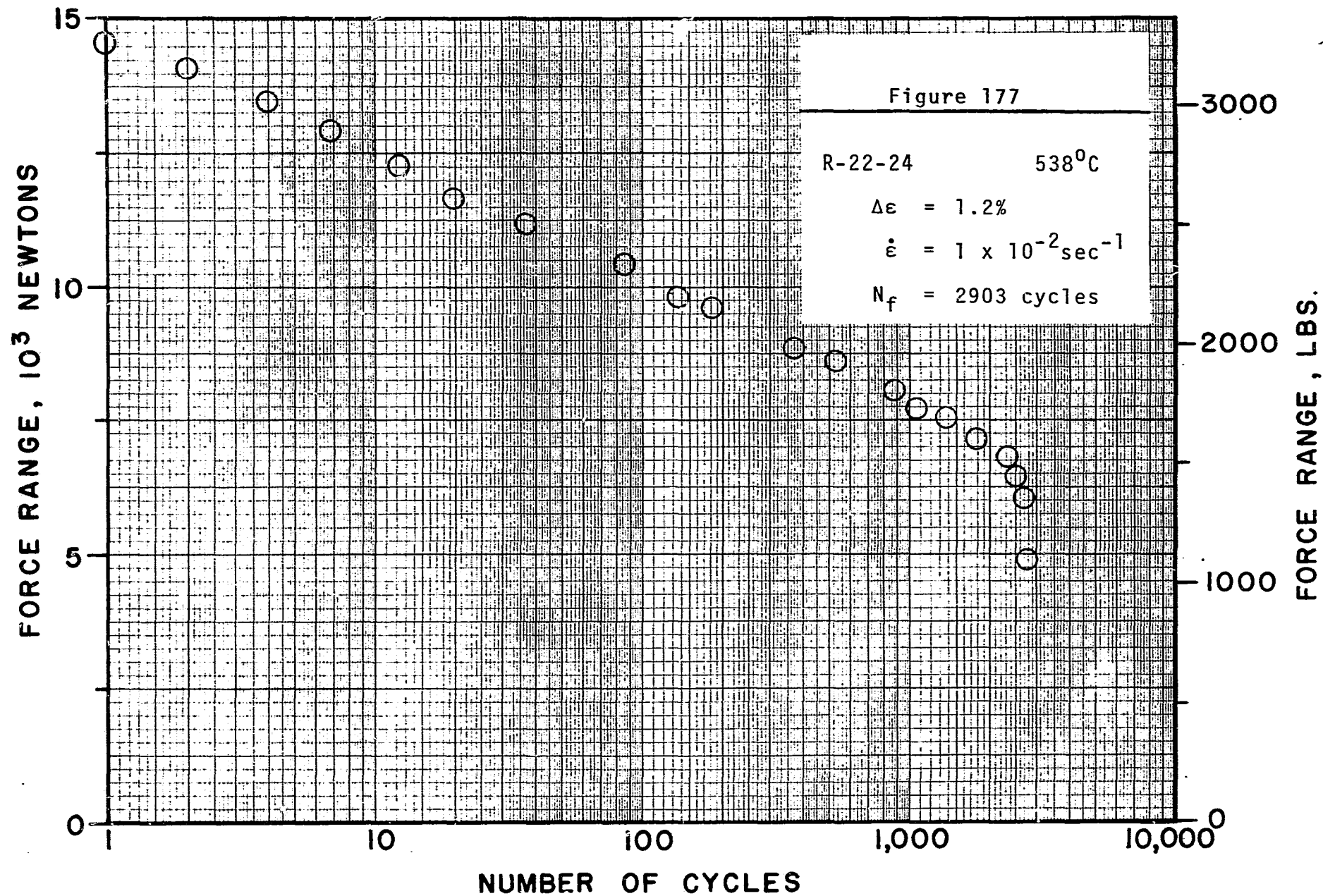


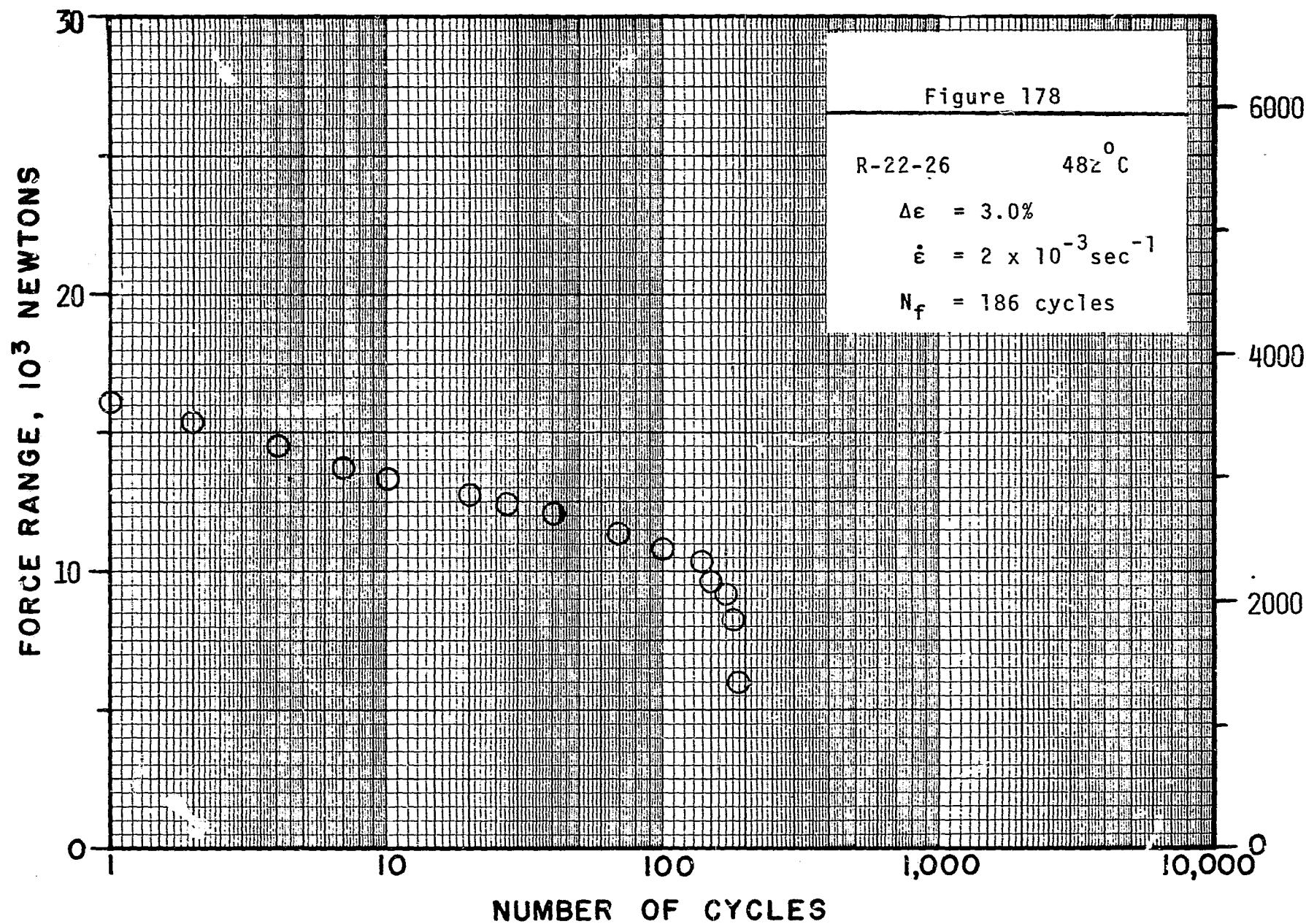




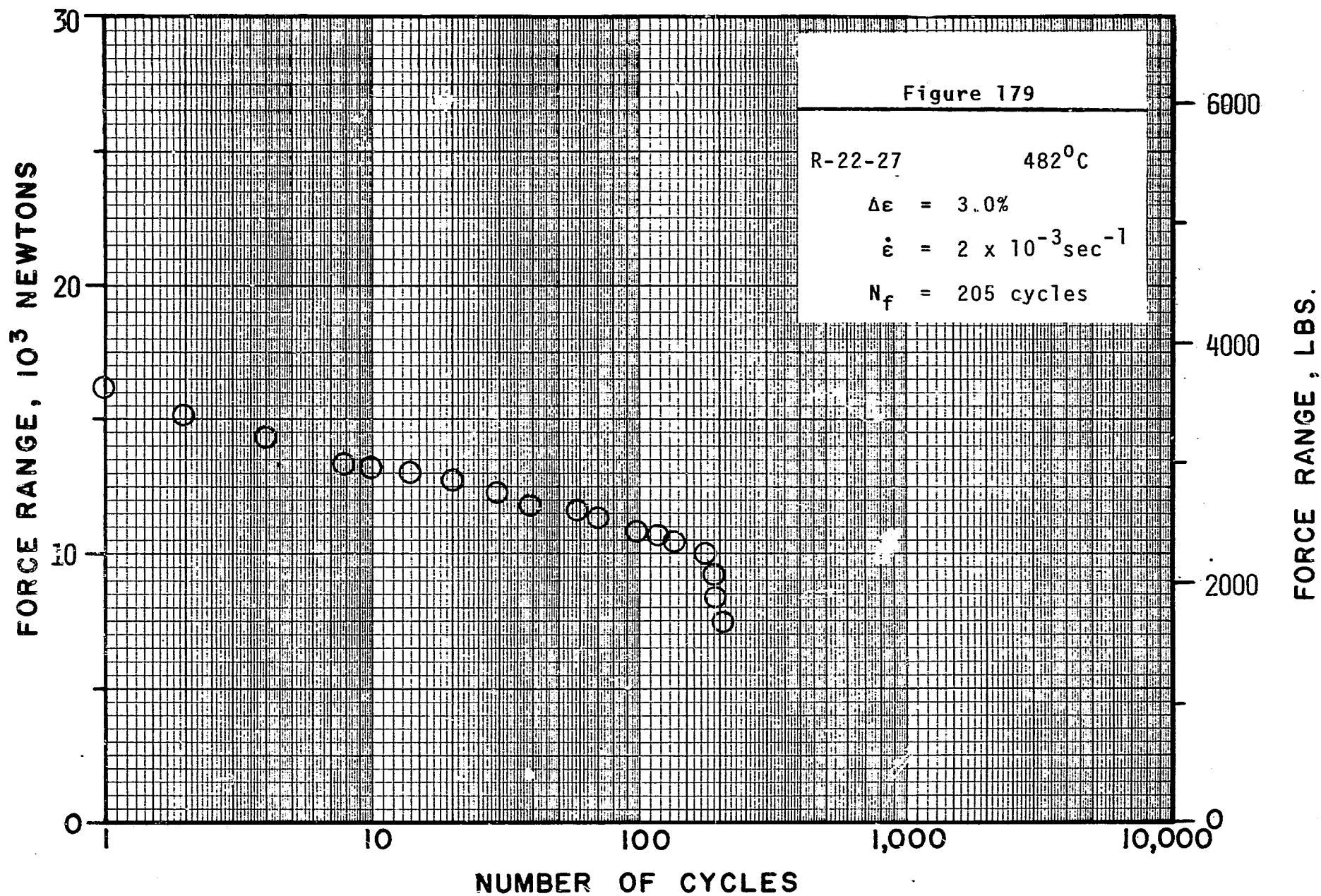


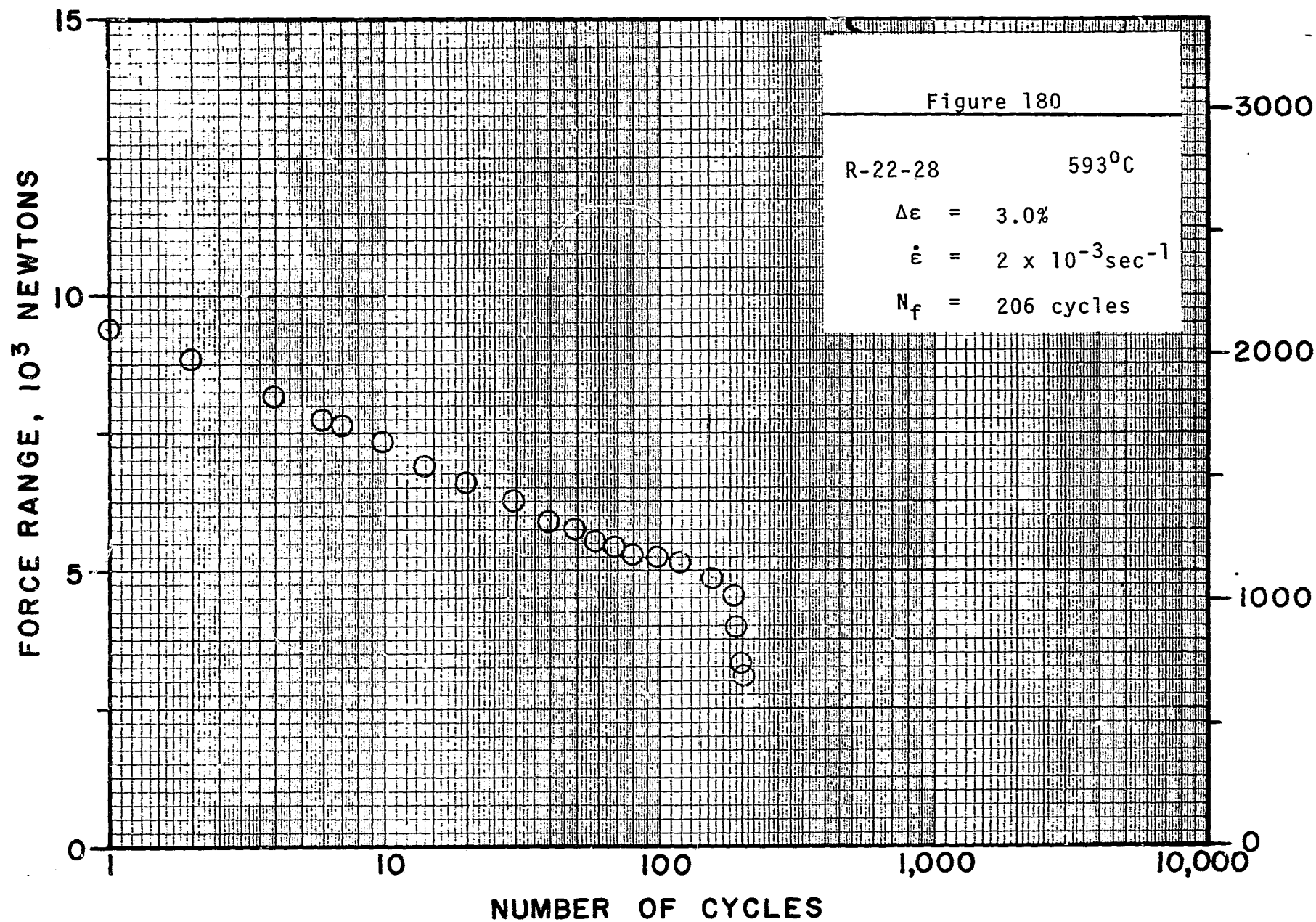


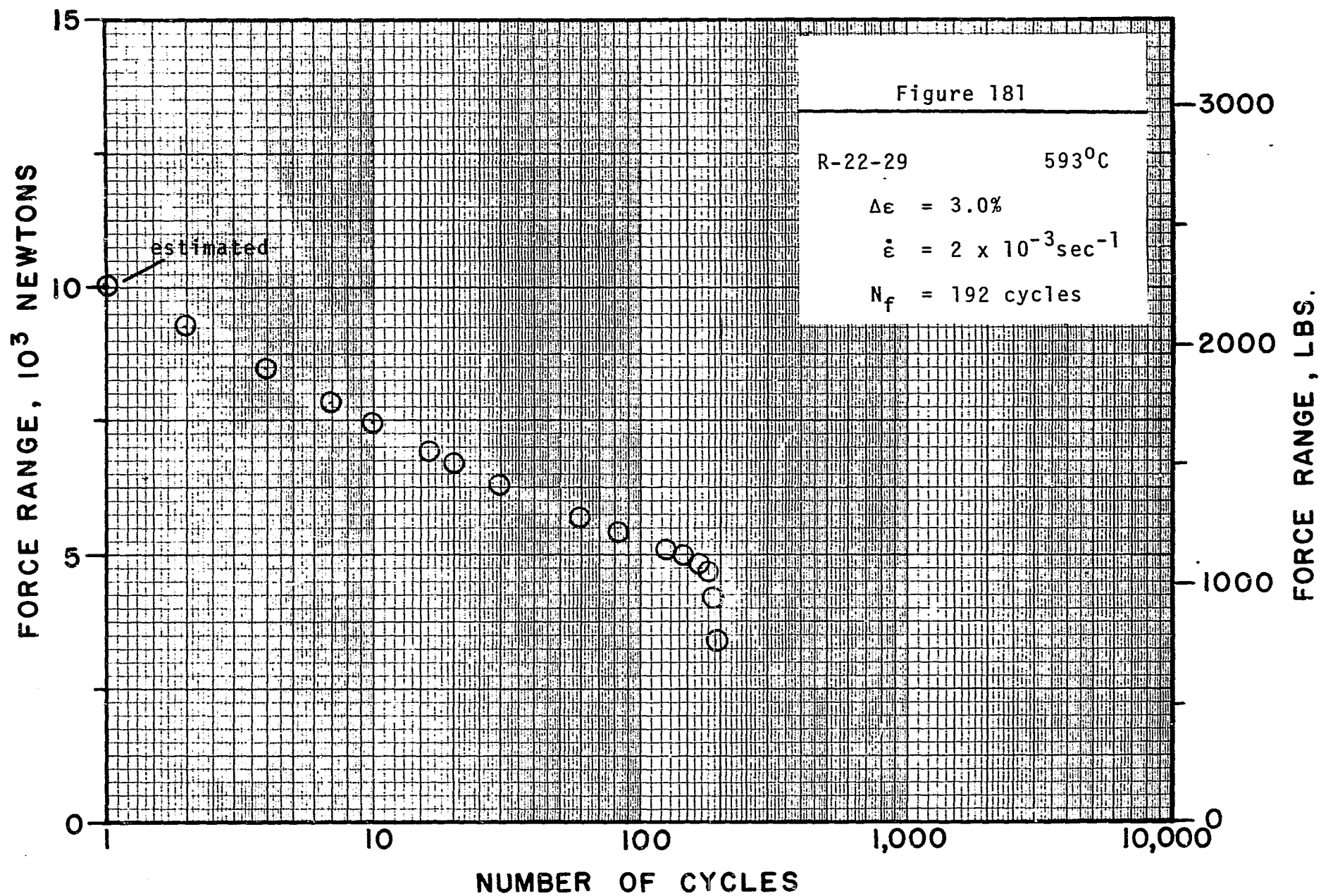


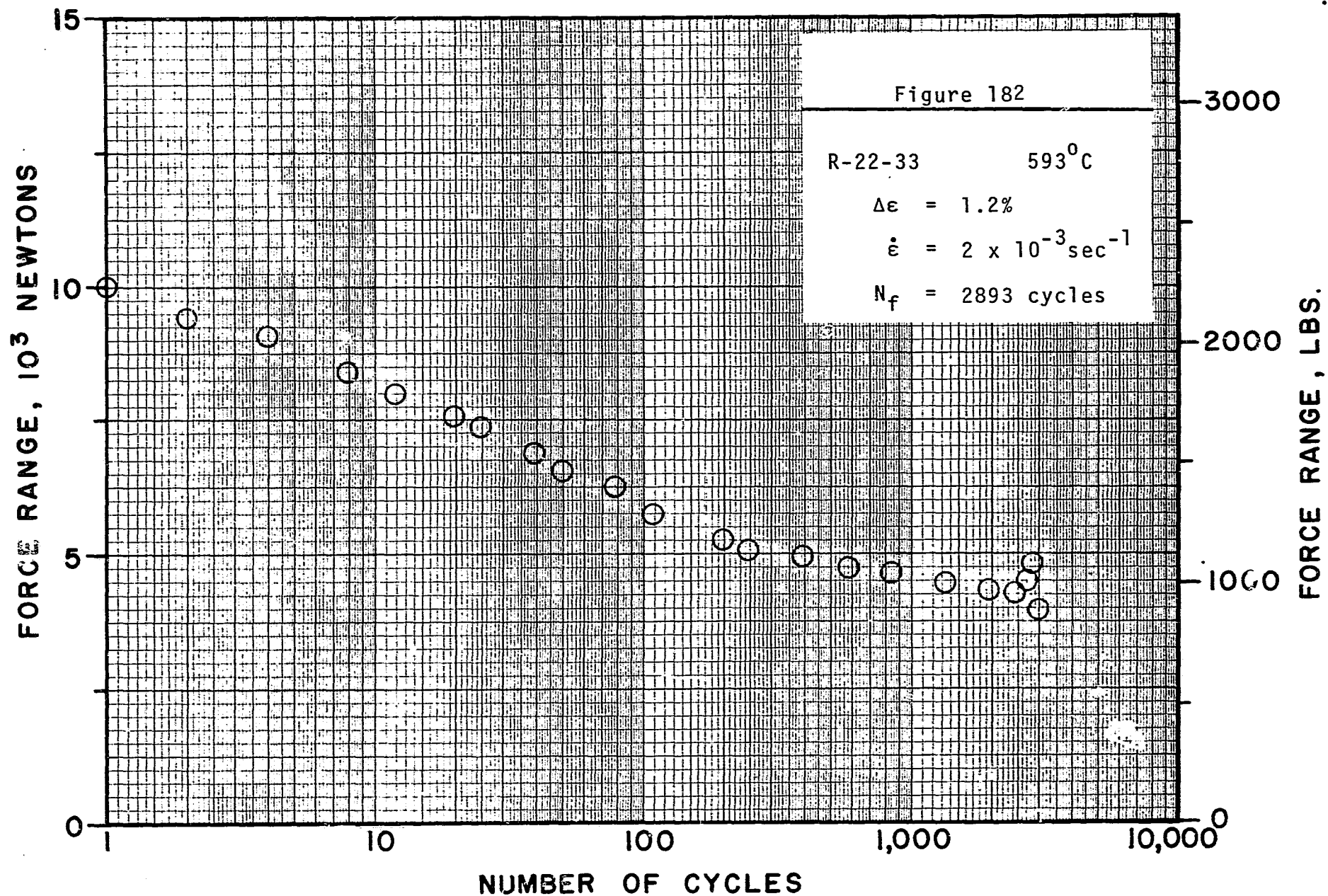


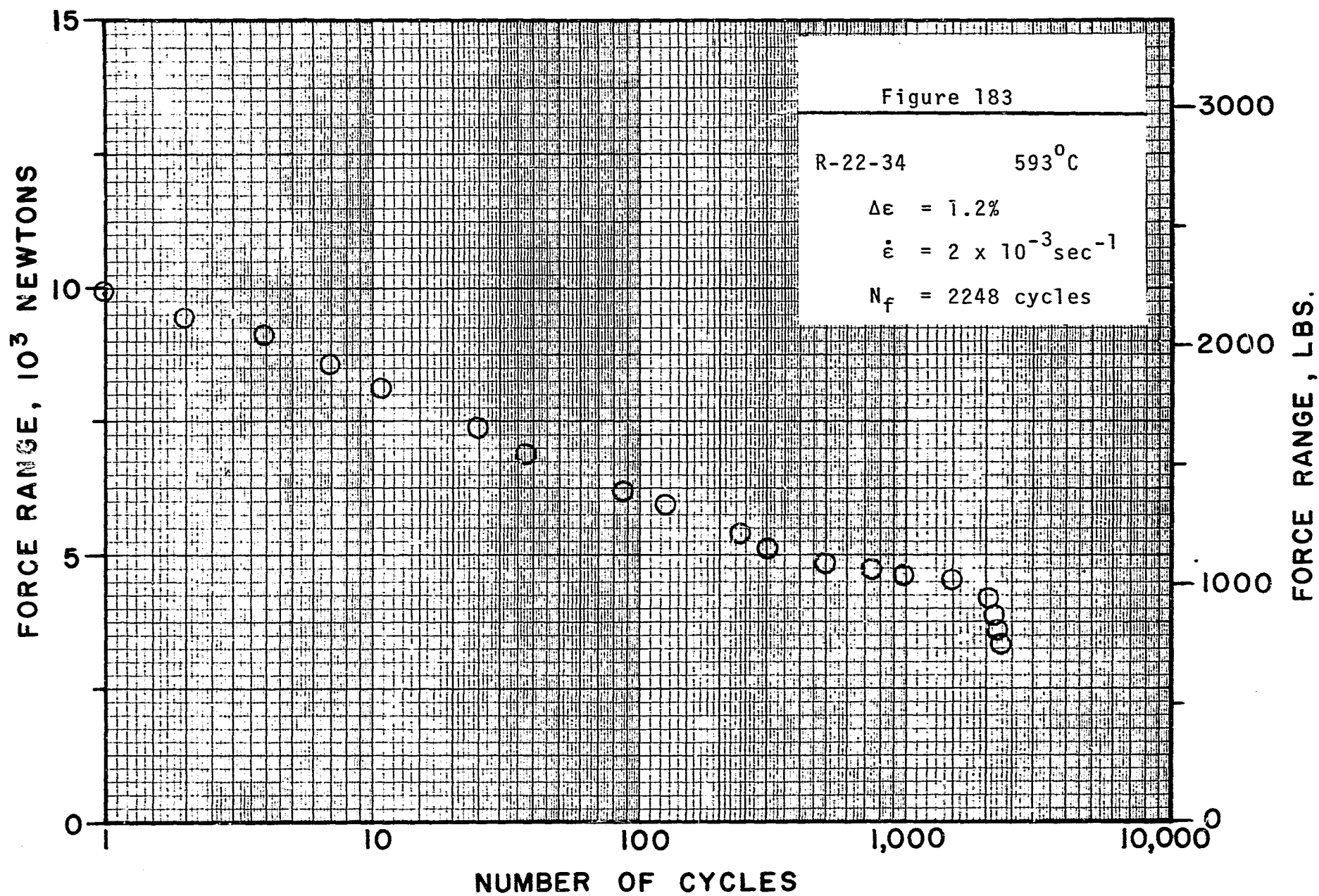
FORCE RANGE, LBS.



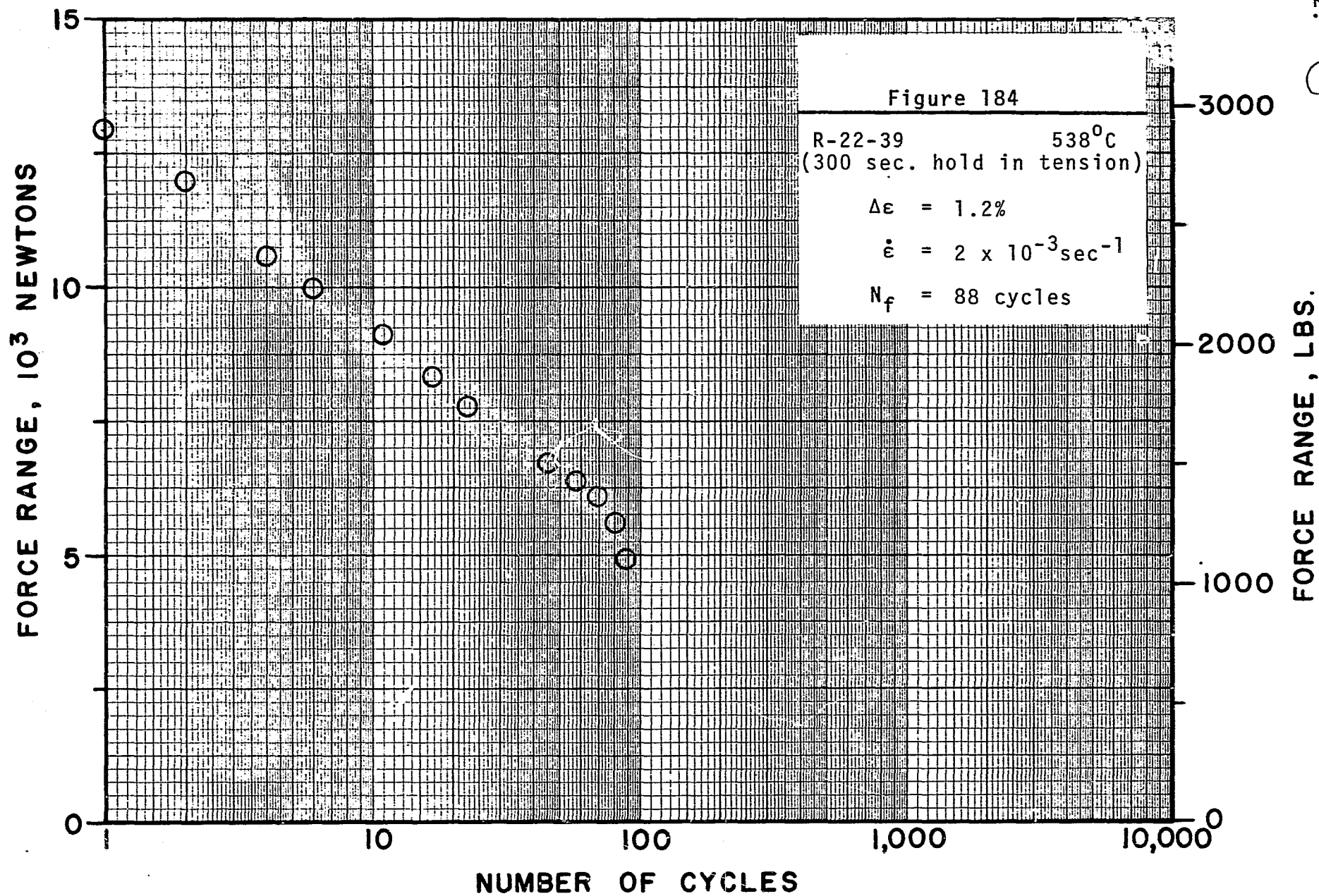


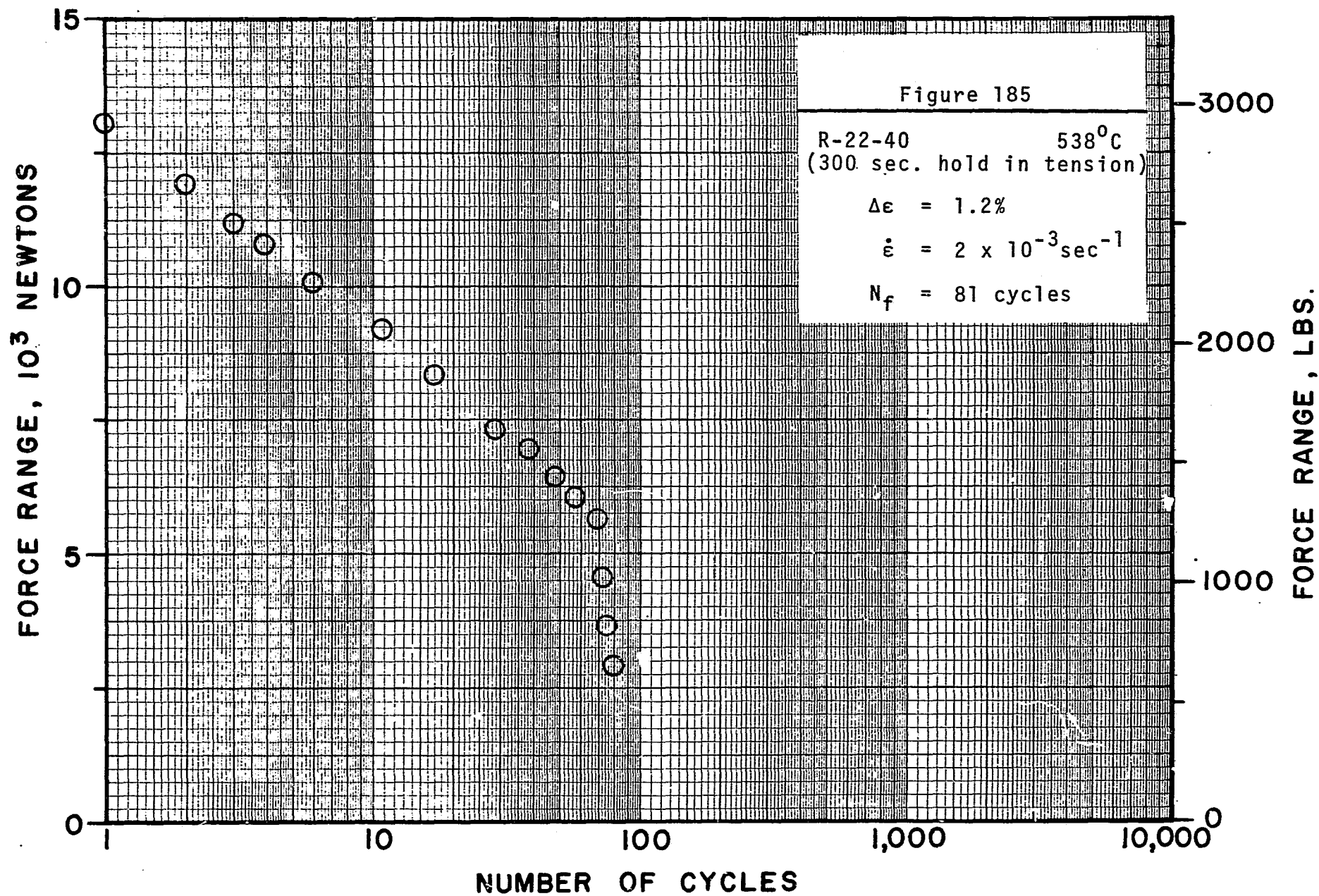


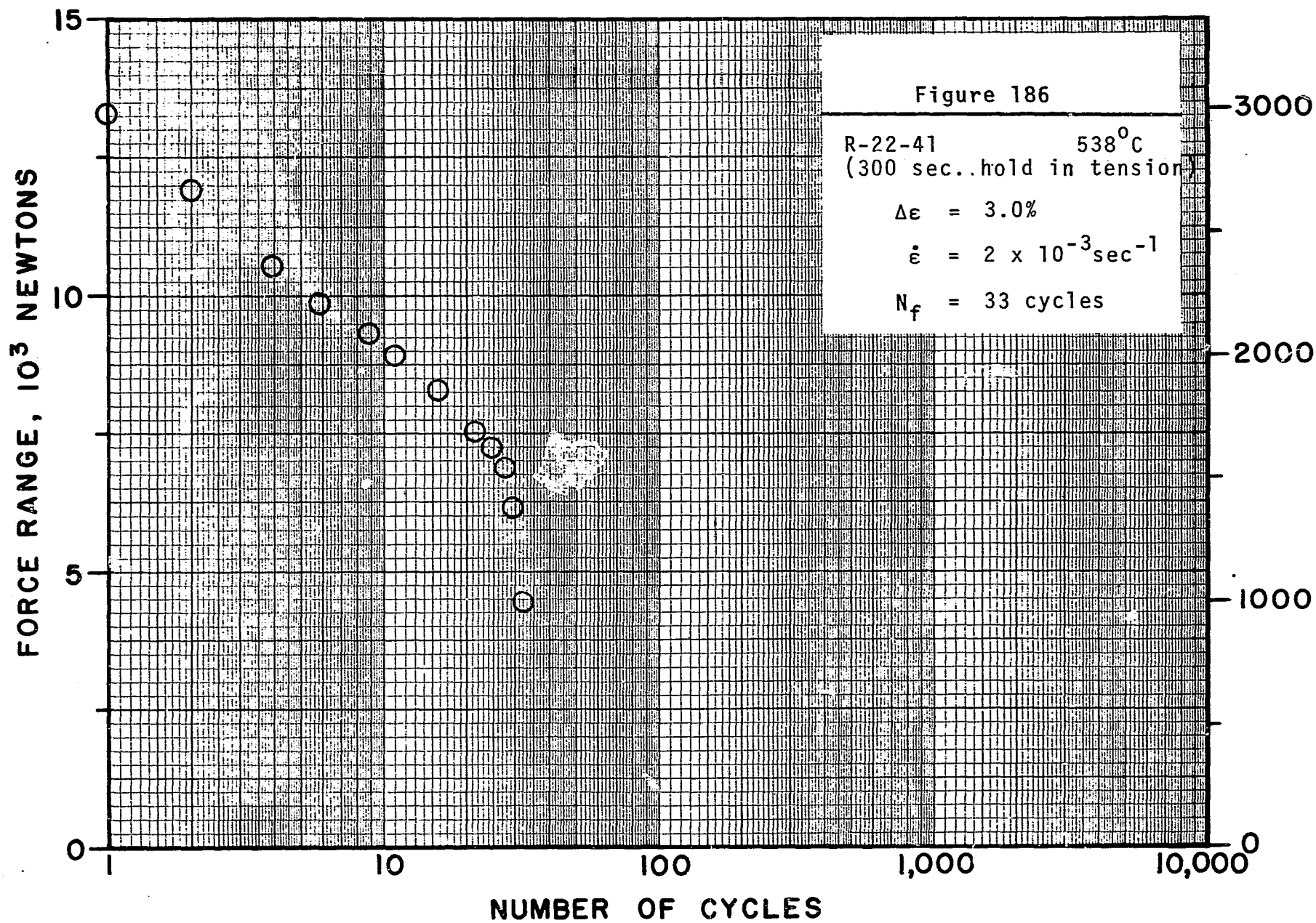


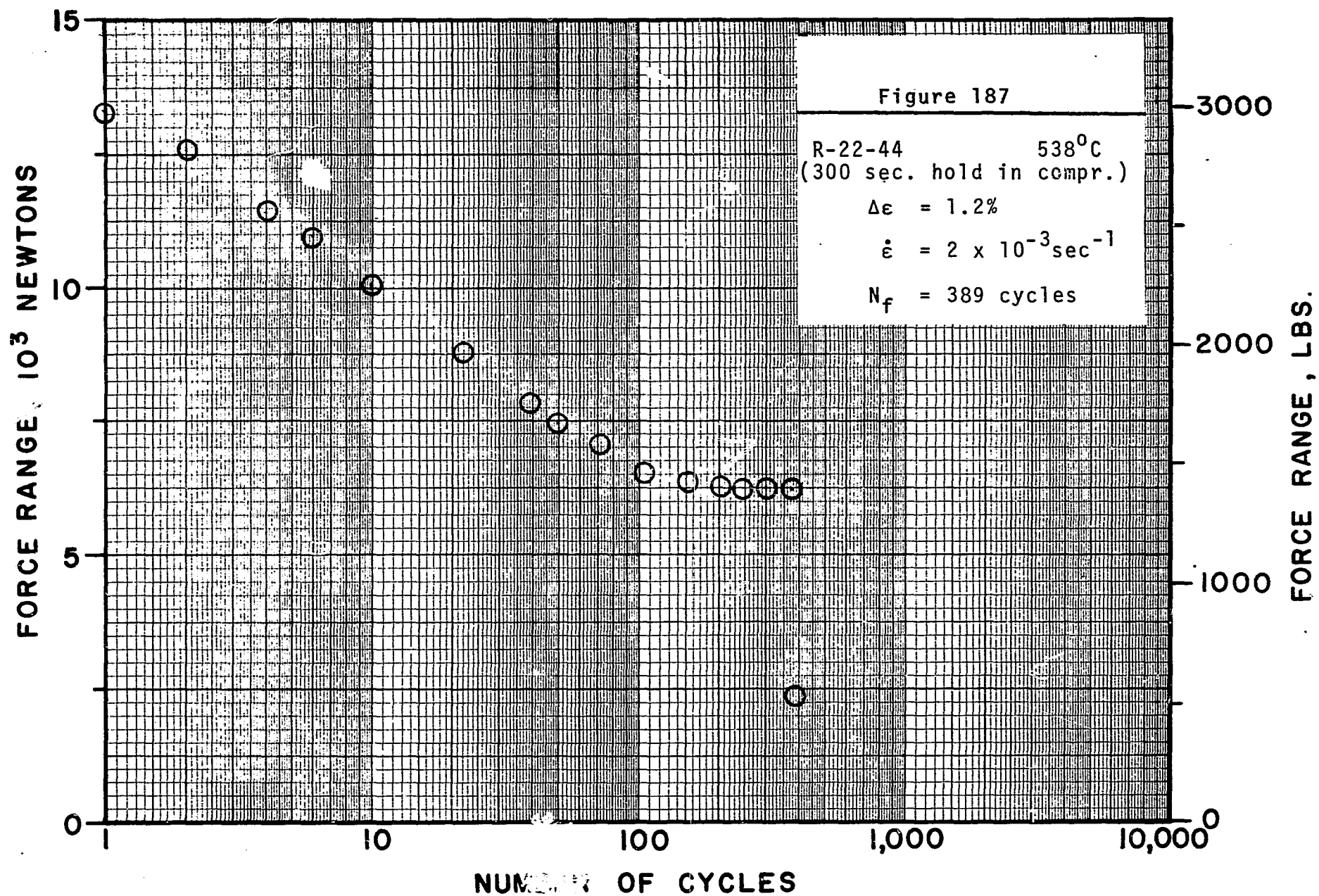


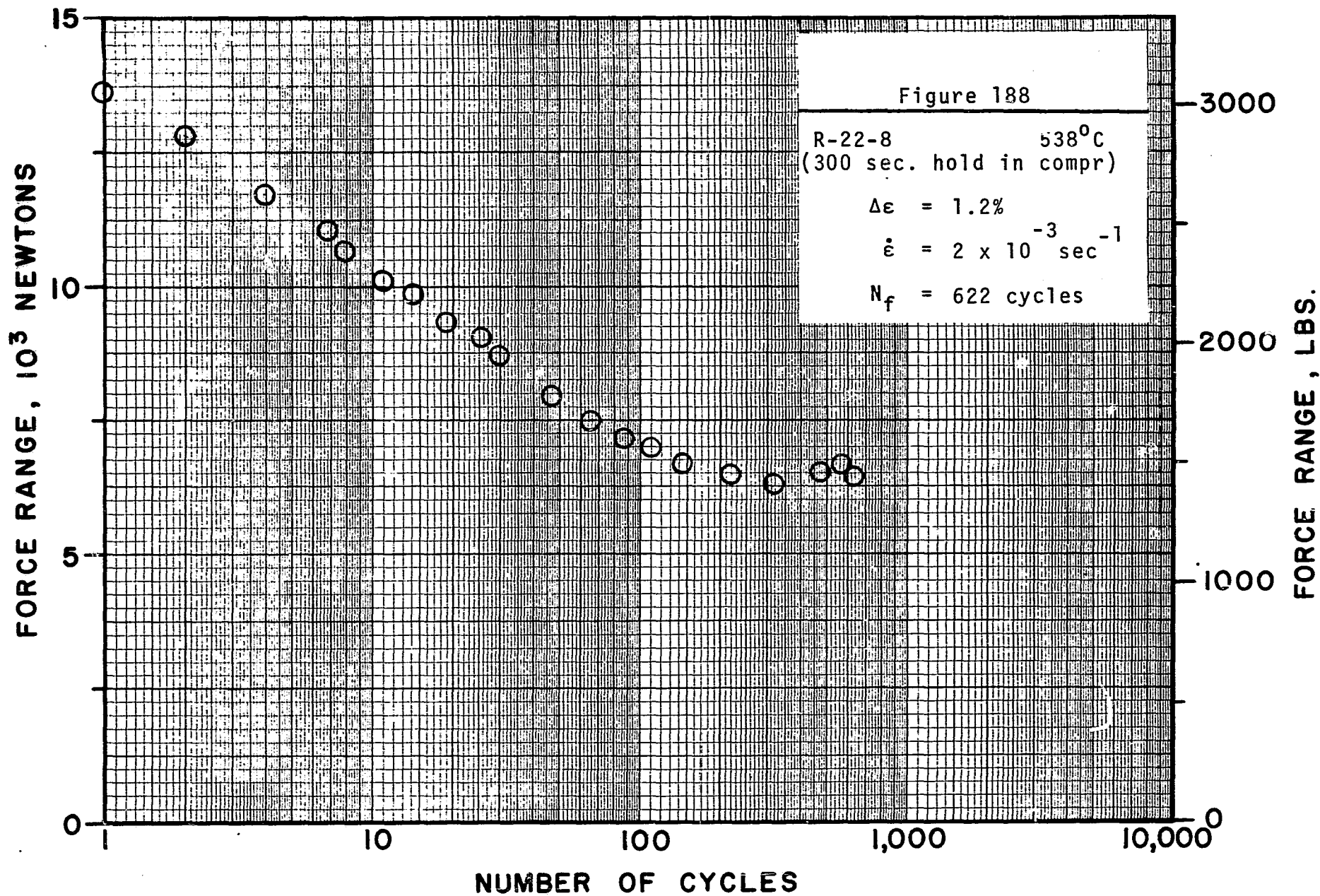
C-3

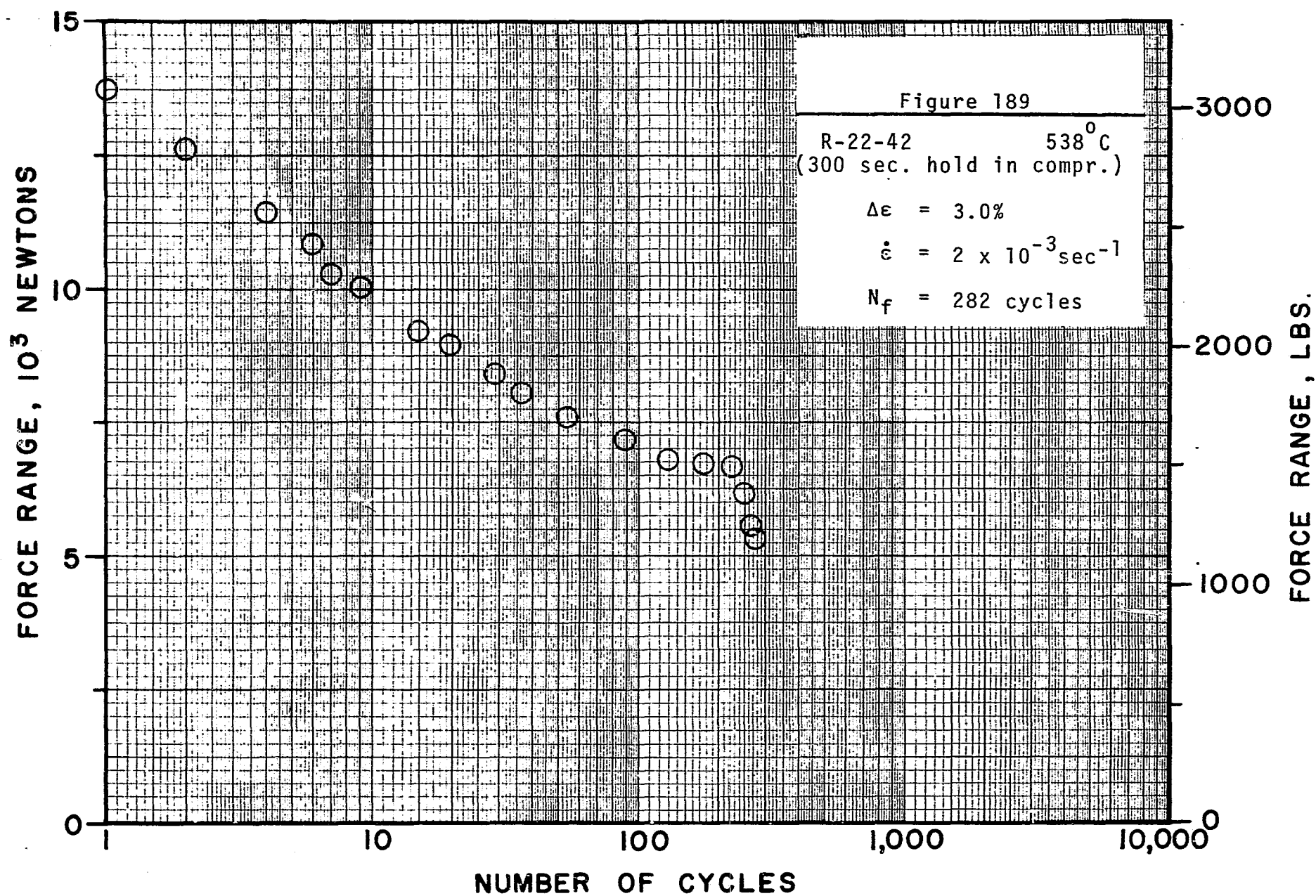












b) HYSTERESIS LOOPS

Figures 190 through 288.

Note: A frequency of 60 cpm was employed in the testing of Specimen Numbers R-24-29, R-24-30, R-2-74 and R-2-75. Since this frequency was beyond the response characteristics of the x-y recorder, no hysteresis loops could be obtained for these tests.

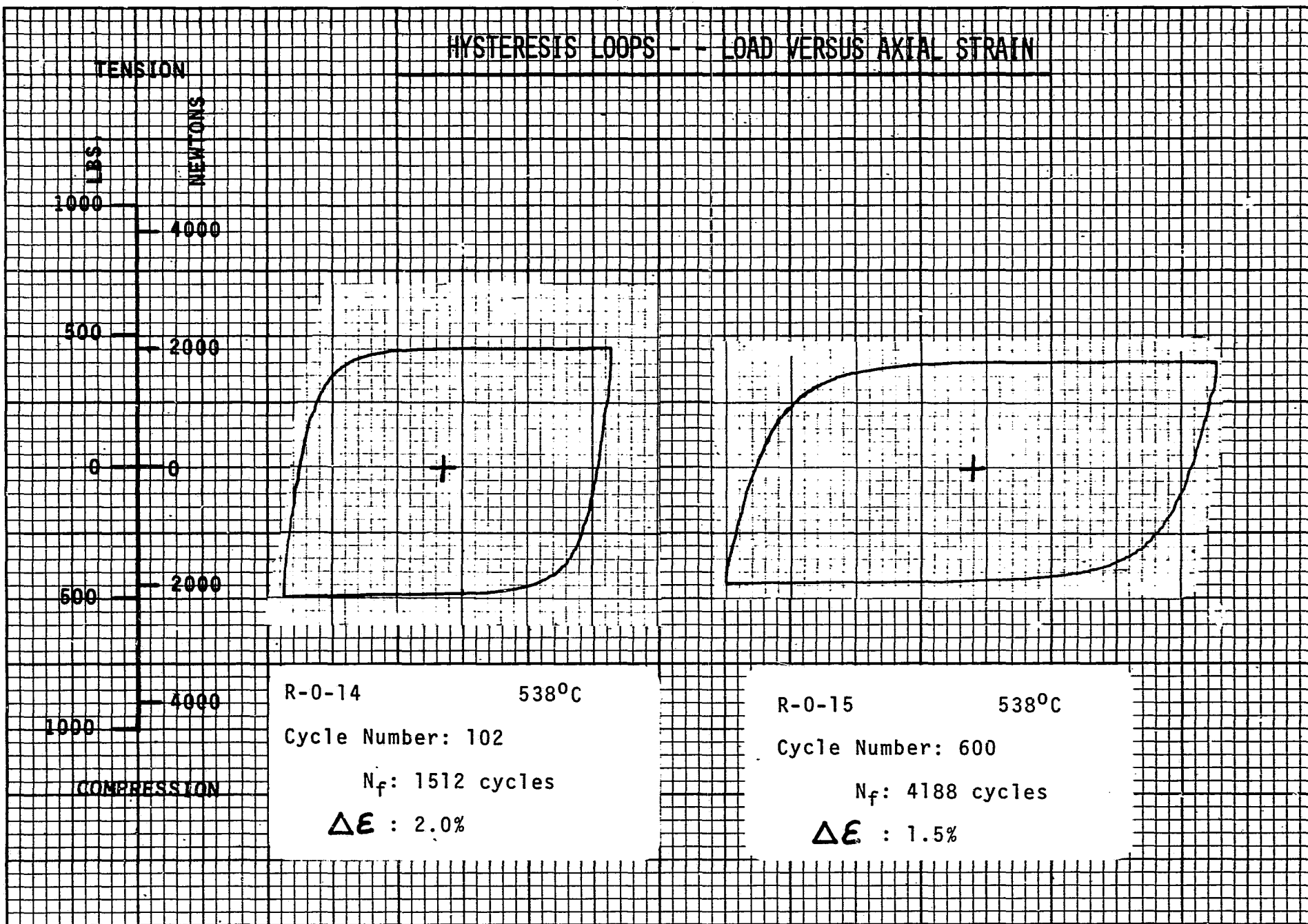
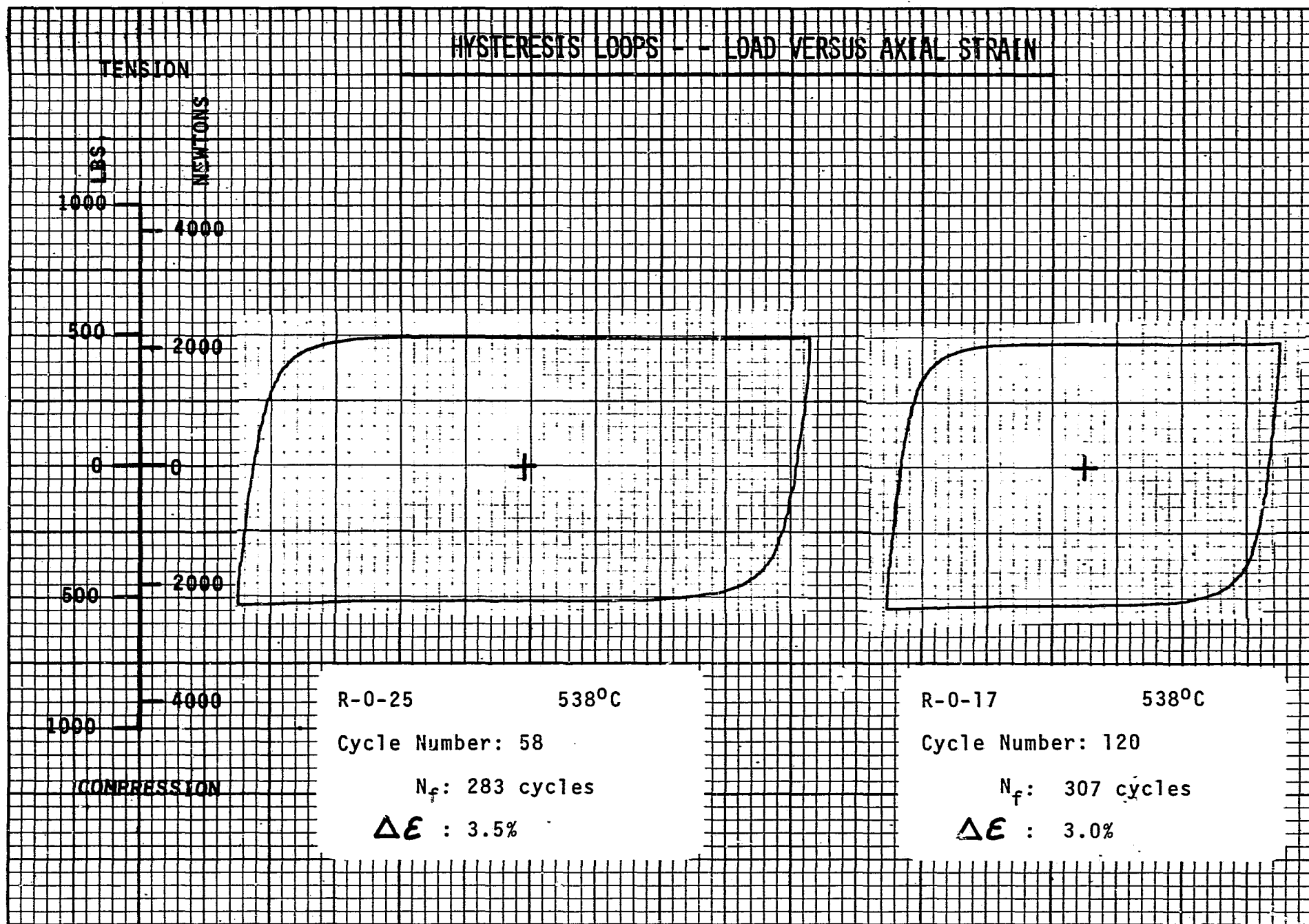


Figure 190



HYSTERESIS LOOPS - - LOAD VERSUS AXIAL STRAIN

TENSION

LBS

NEWTONS

1000

500

0

500

1000

4000

2000

0

2000

4000

1000

COMPRESSION

R-0-19

538°C

Cycle Number: 211

N_f : 2300 cycles

$\Delta \epsilon$: 1.7%

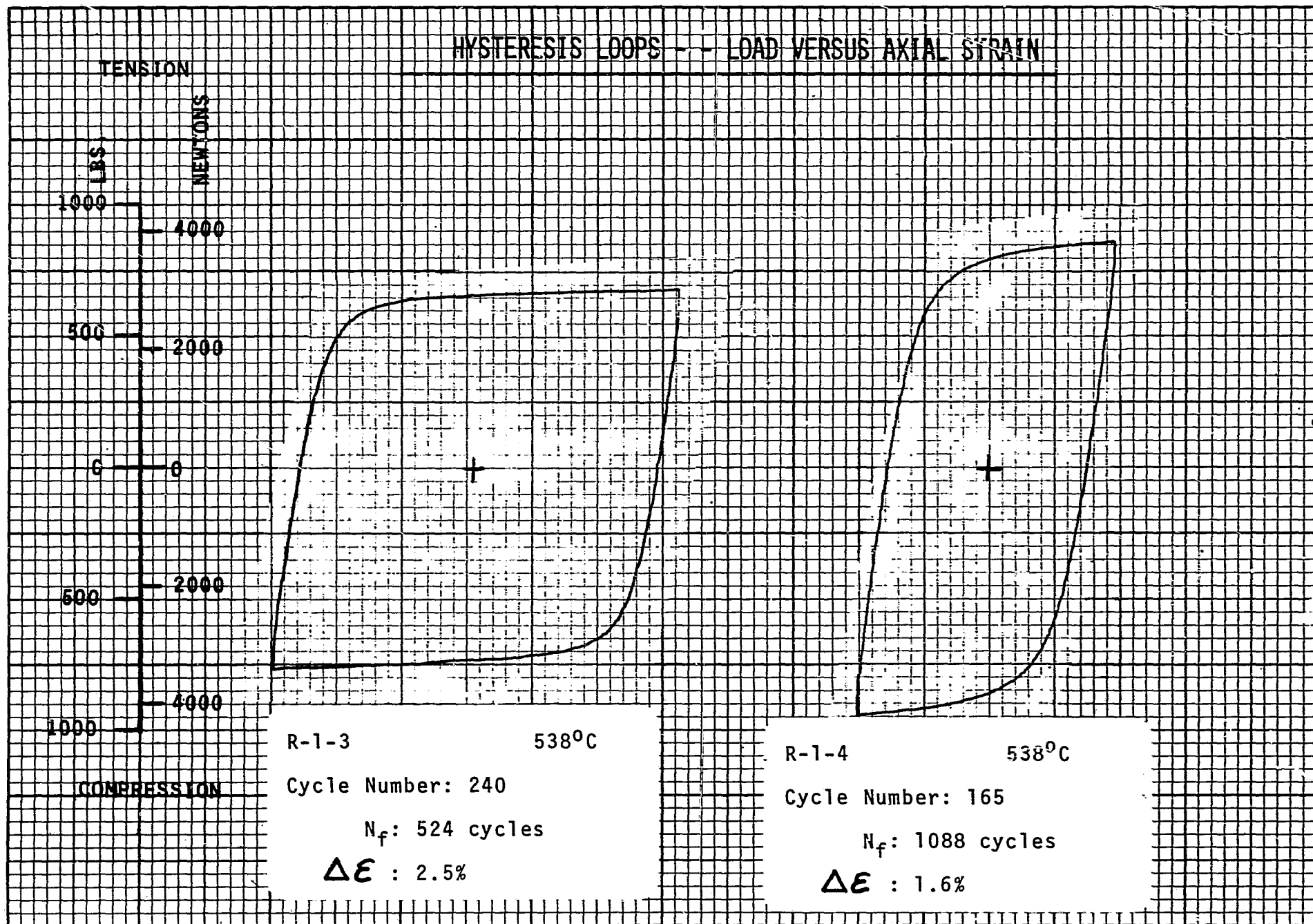
R-0-21

538°C

Cycle Number: 53

N_f : 418 cycles

$\Delta \epsilon$: 2.5%



HYSTERESIS LOOPS - - LOAD VERSUS AXIAL STRAIN

TENSION

LBS

NEWTONS

1000

500

0

500

1000

4000

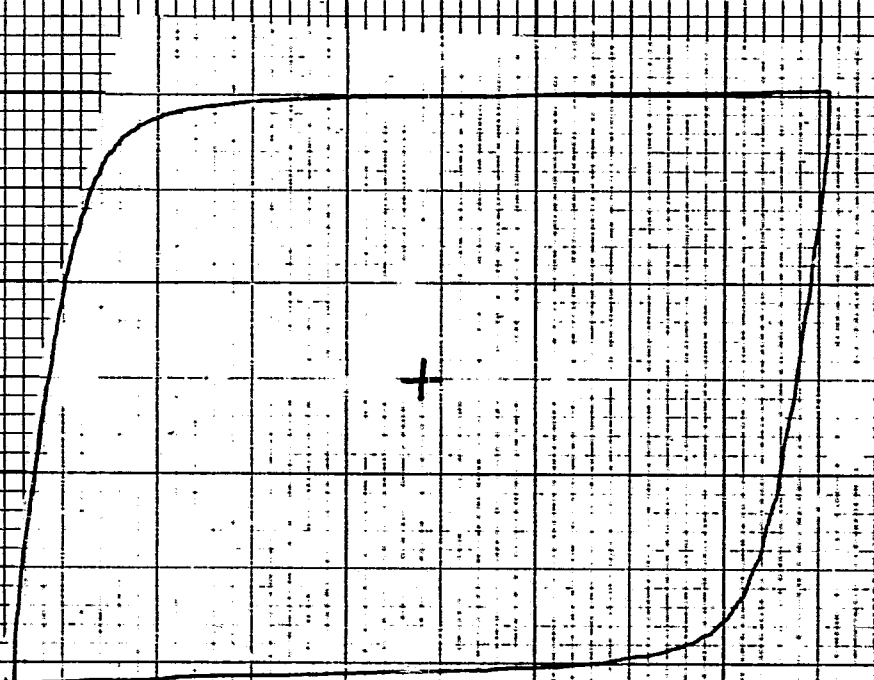
2000

0

2000

4000

COMPRESSION



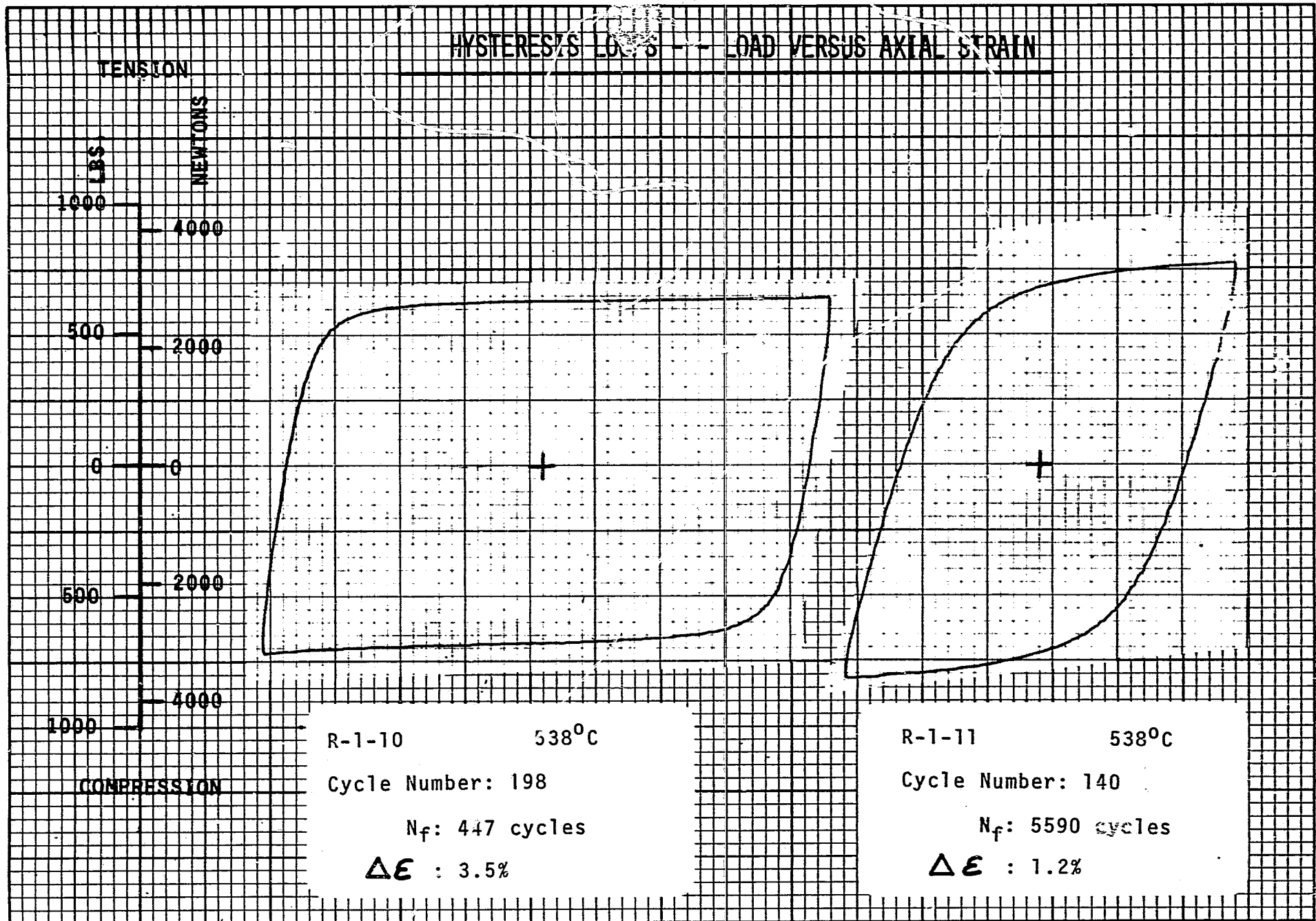
R-1-9

538°C

Cycle Number: 64

N_f : 562 cycles

$\Delta \epsilon$: 3.5%



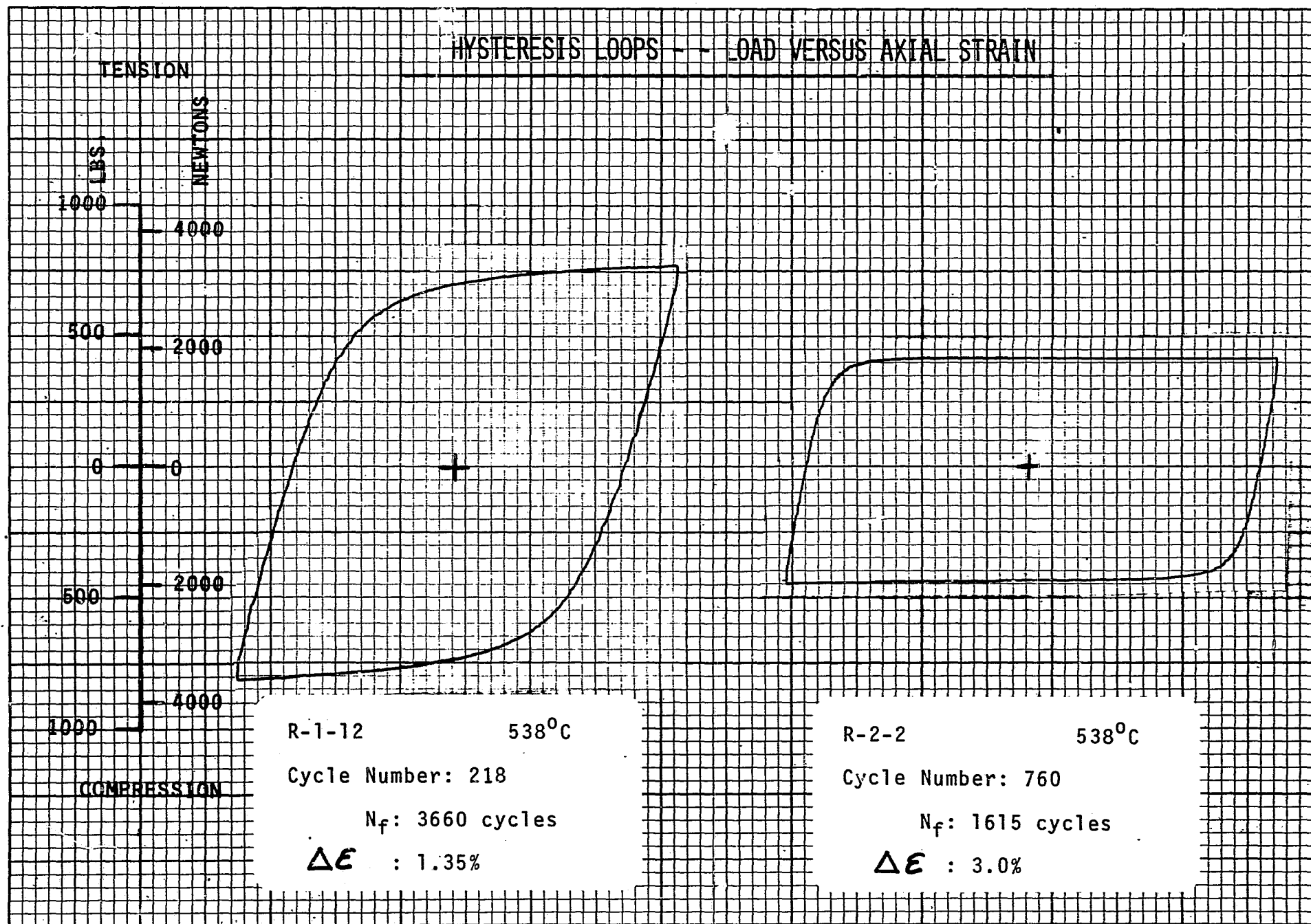
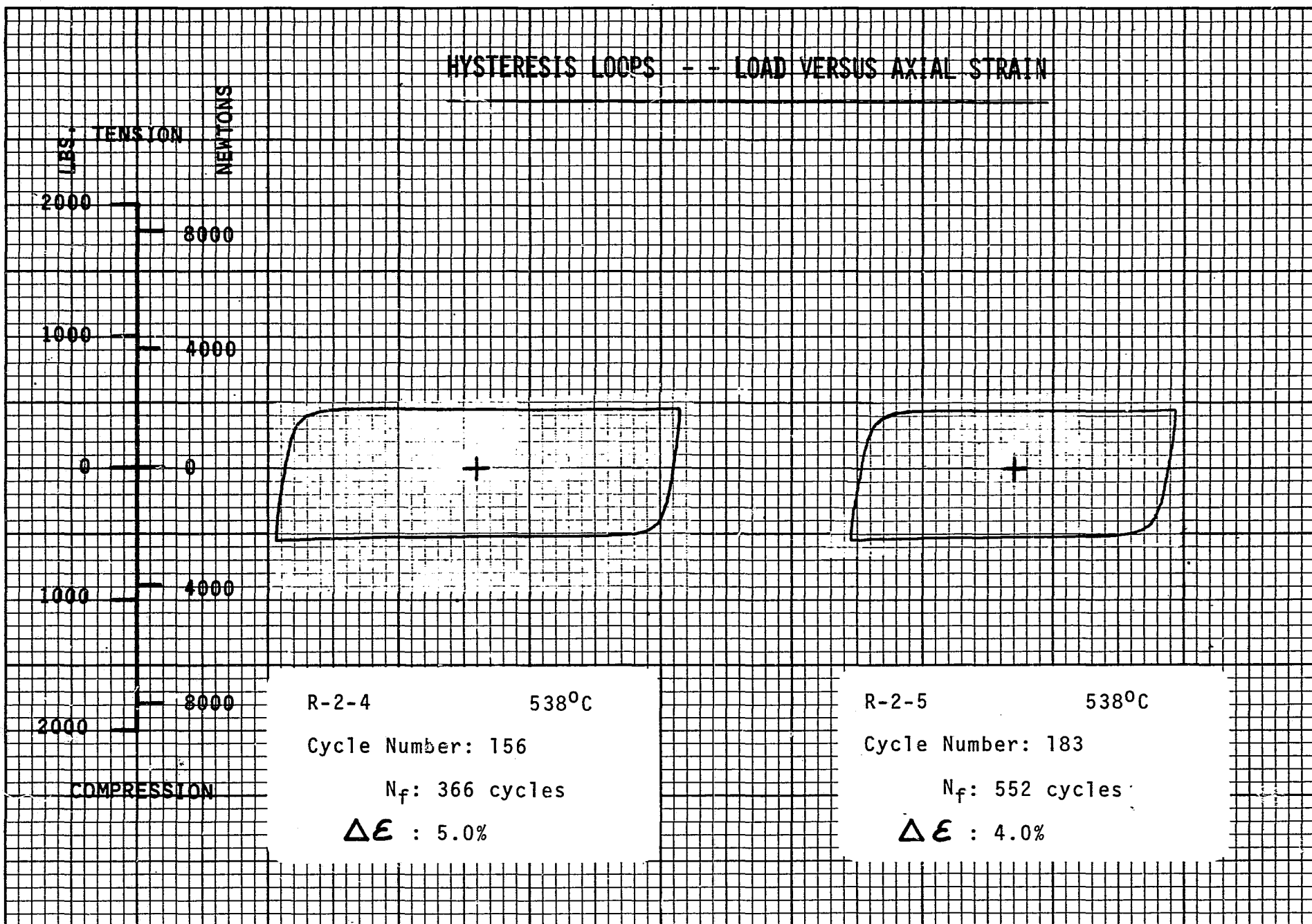


Figure 196



HYSTERESIS LOOPS - - LOAD VERSUS AXIAL STRAIN

LBS
 TENSION
 2000
 1000
 0
 1000
 2000
 COMPRESSION

NEWTONS

8000

4000

0

4000

8000

R-2-10 538°C

Cycle Number: 321

N_f : 1055 cycles

$\Delta \epsilon$: 2.8%

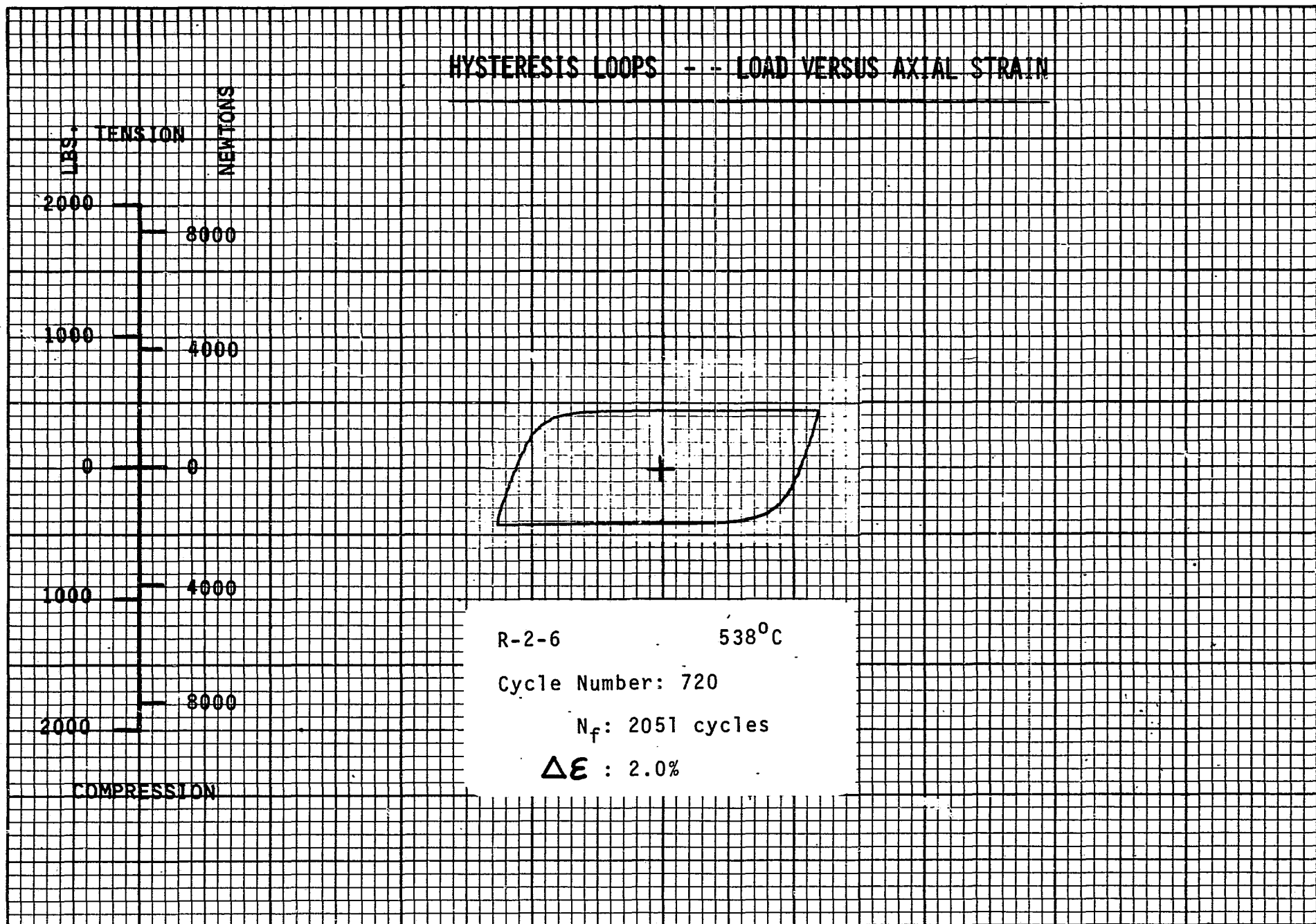
R-2-11 538°C

Cycle Number: 222

N_f : 1239 cycles

$\Delta \epsilon$: 2.0%

Figure 198



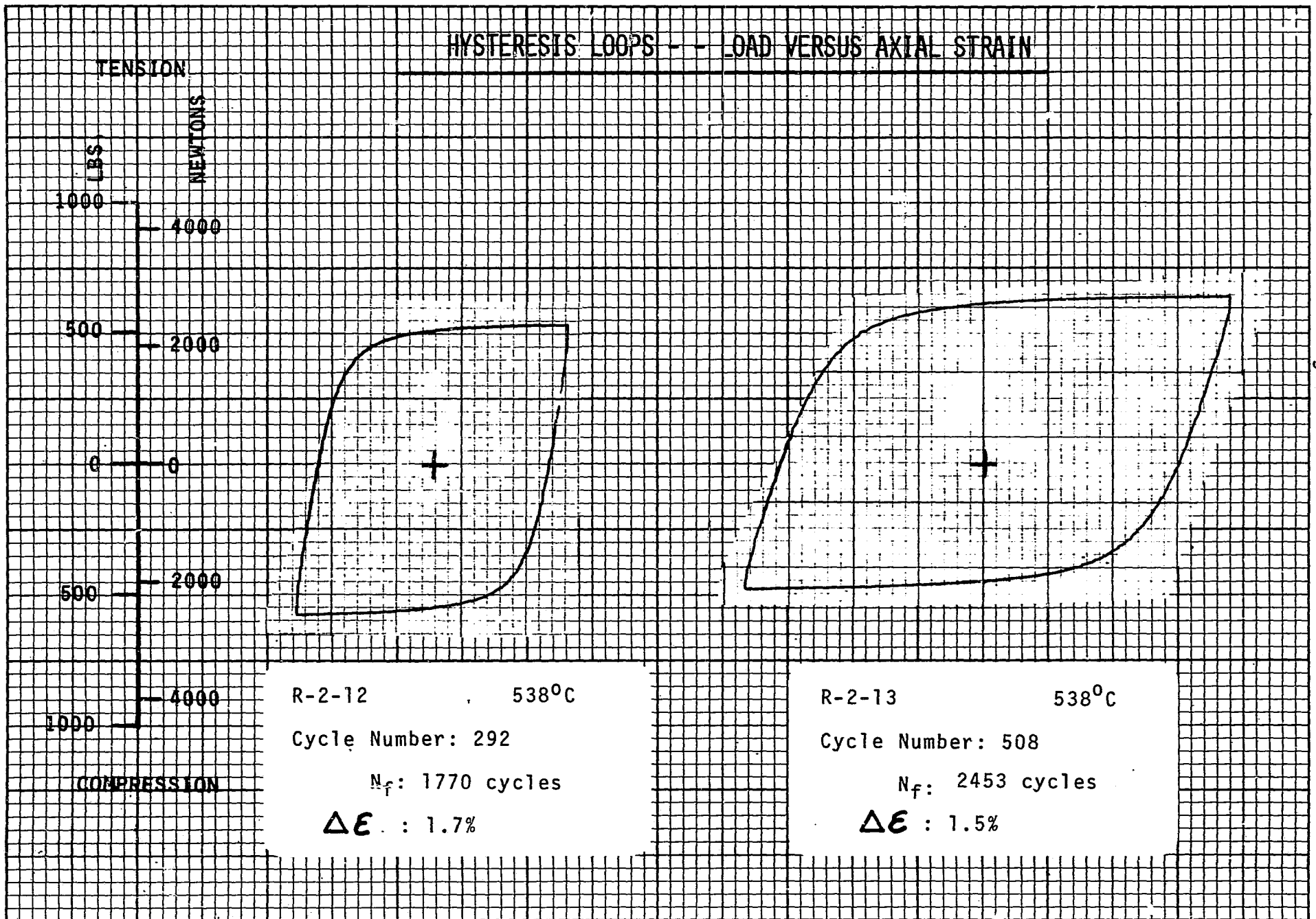
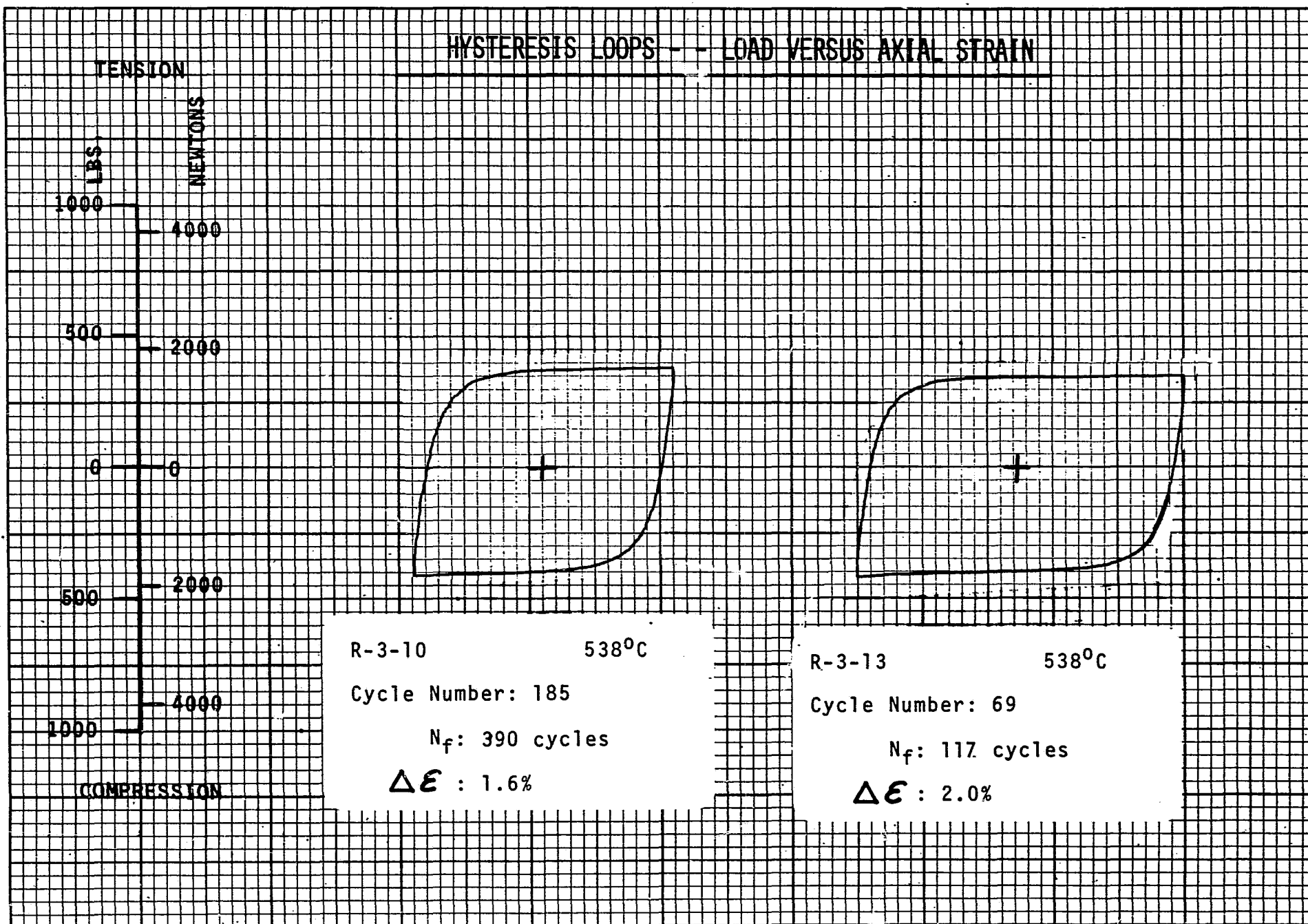


Figure 200



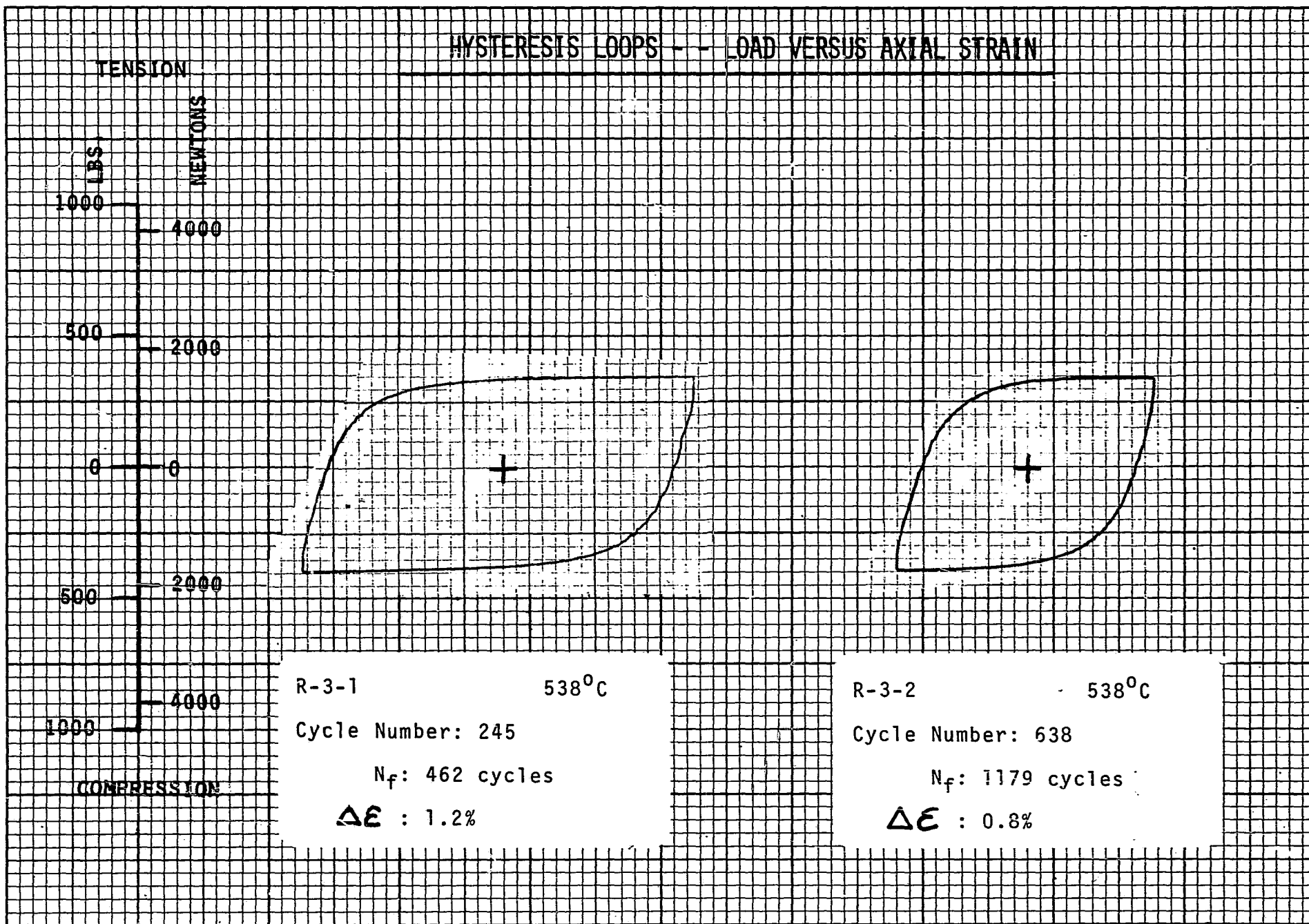
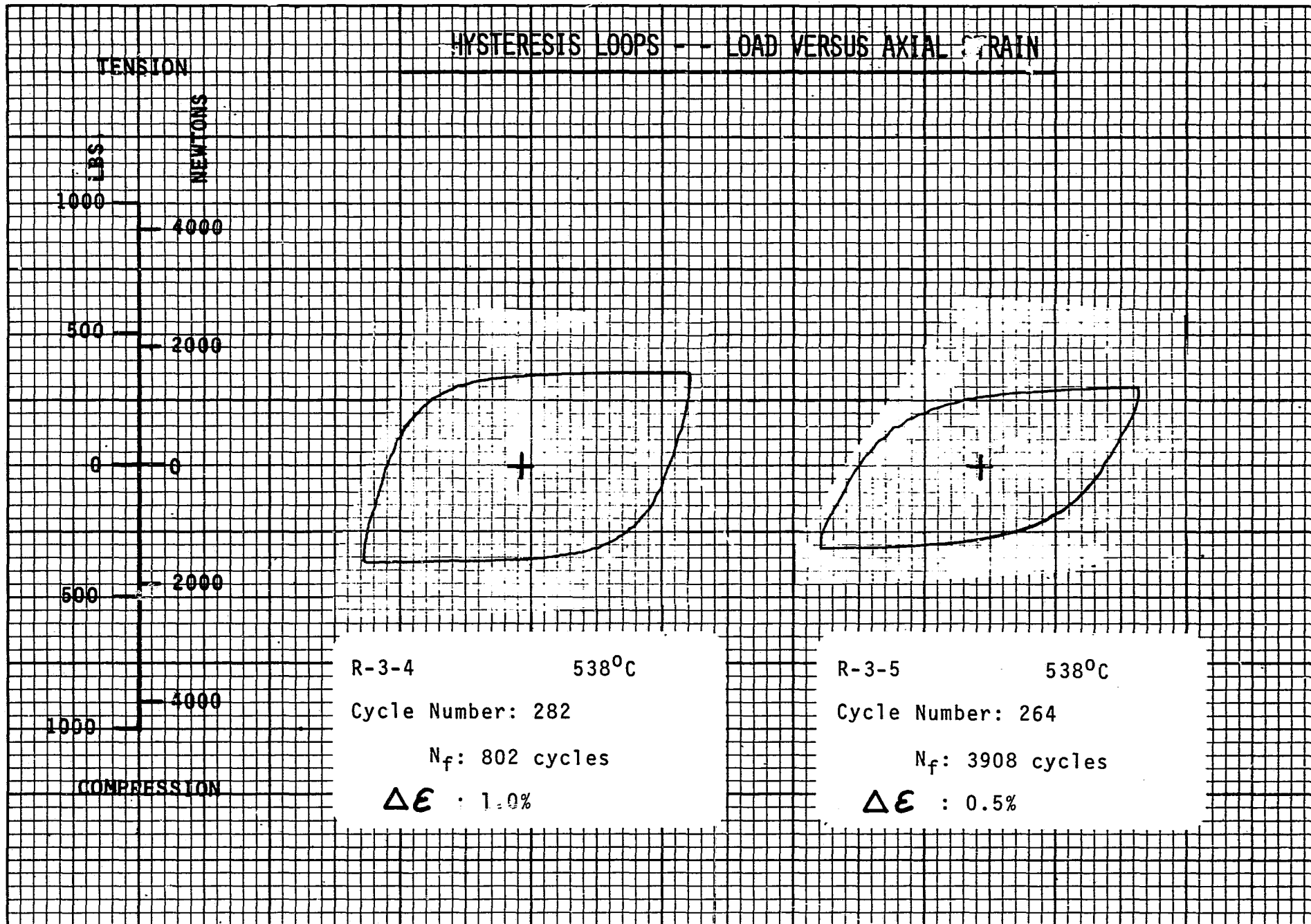


Figure 202



HYSTERESIS LOOPS -- LOAD VERSUS AXIAL STRAIN

TENSION
 2000
 1000
 0
 1000
 2000
 COMPRESSION

NEWTONS

8000

4000

0

4000

8000

R-4-1

538°C

Cycle Number: 32

N_f : 147 cycles

$\Delta \epsilon$: 2.0%

R-4-2

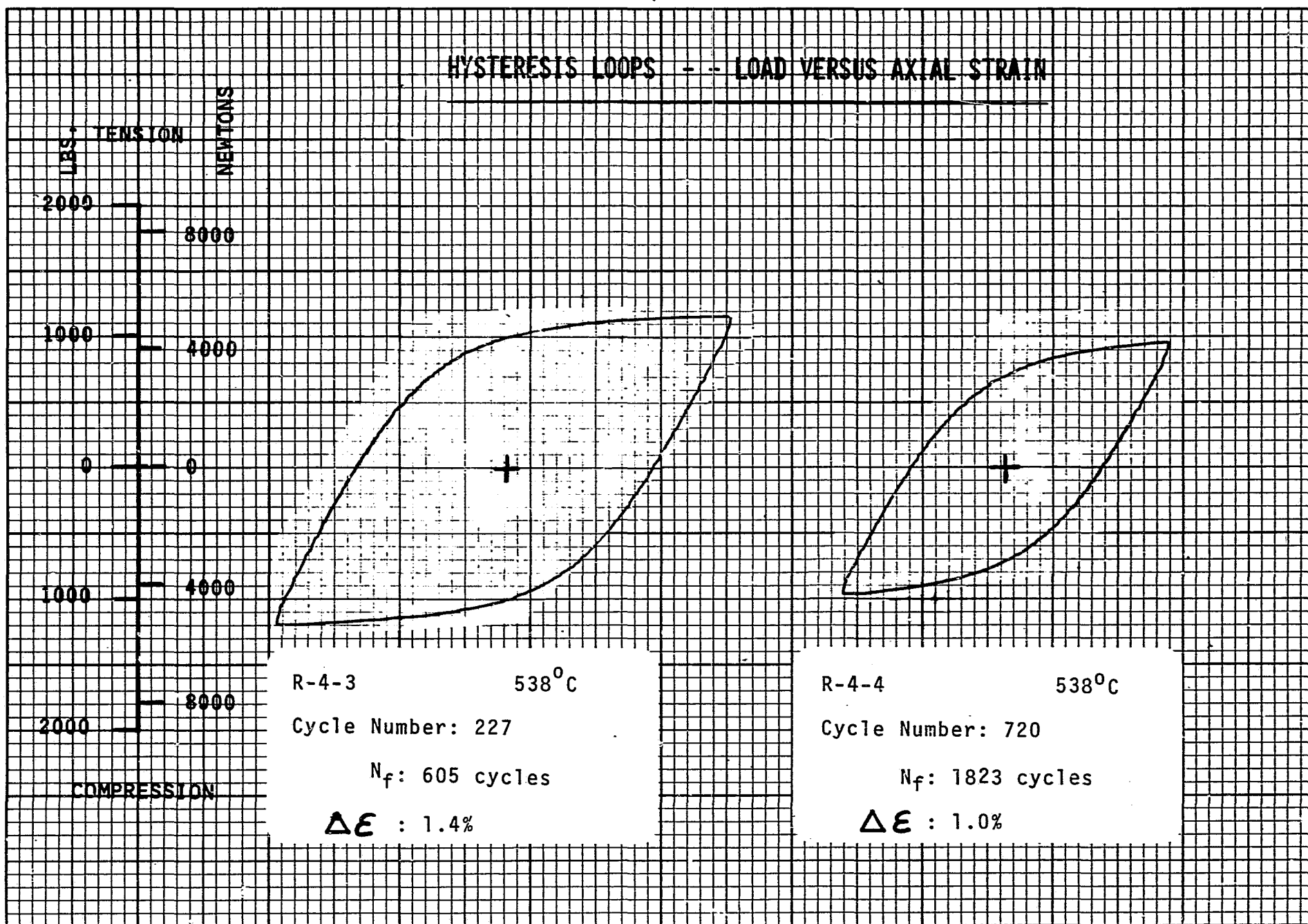
538°C

Cycle Number: 169

N_f : 354 cycles

$\Delta \epsilon$: 1.6%

HYSTERESIS LOOPS -- LOAD VERSUS AXIAL STRAIN



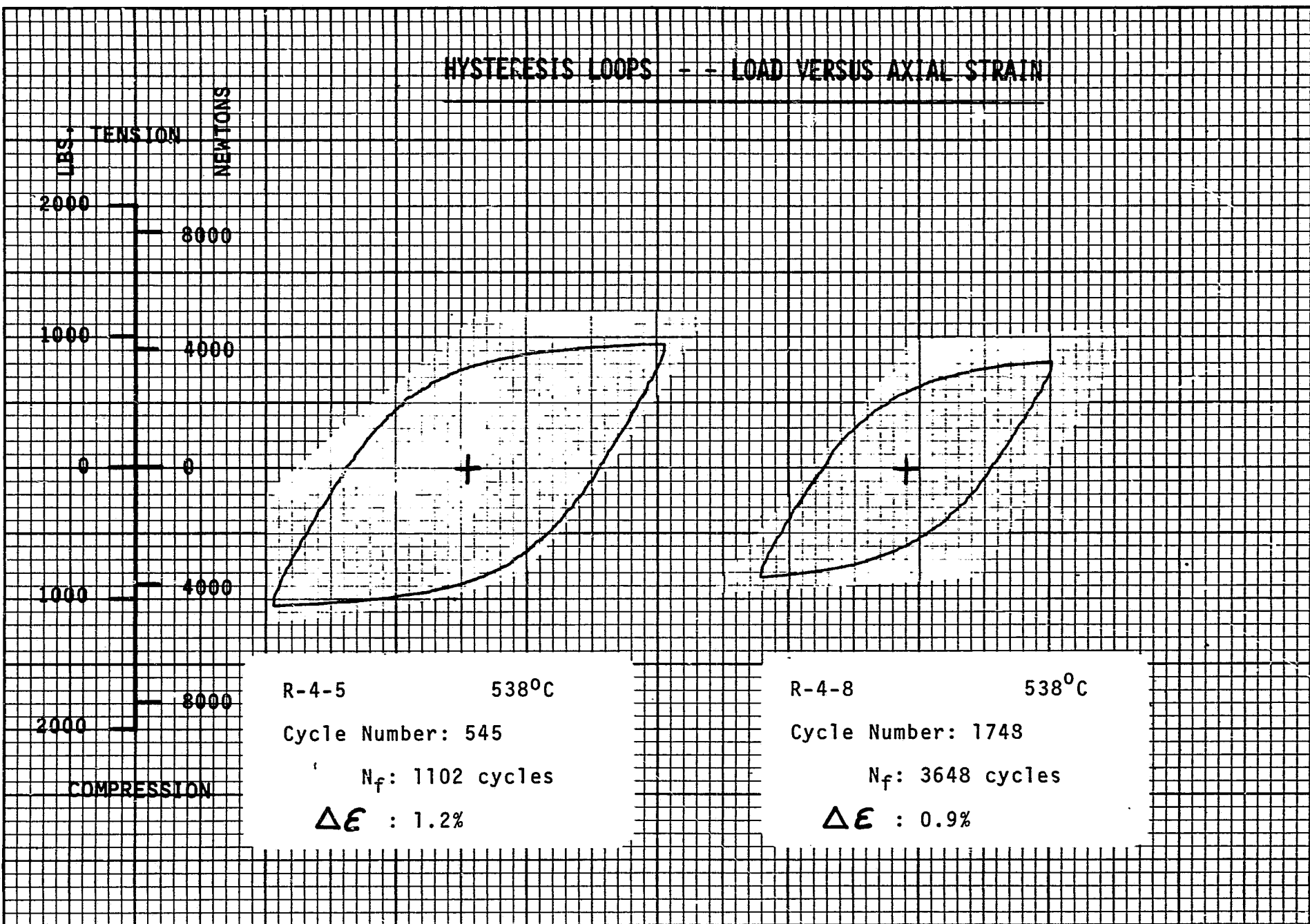
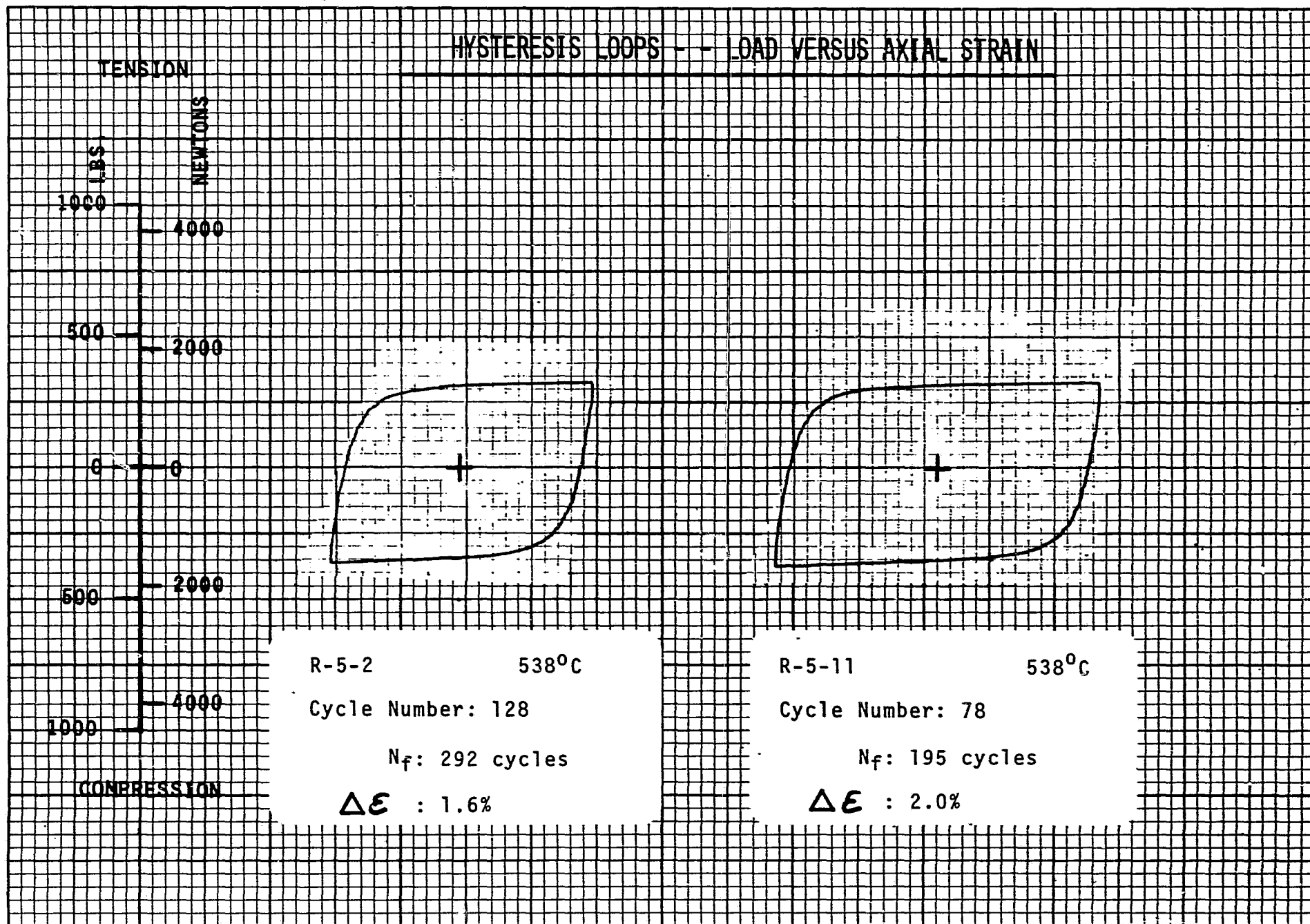


Figure 206

Figure 207



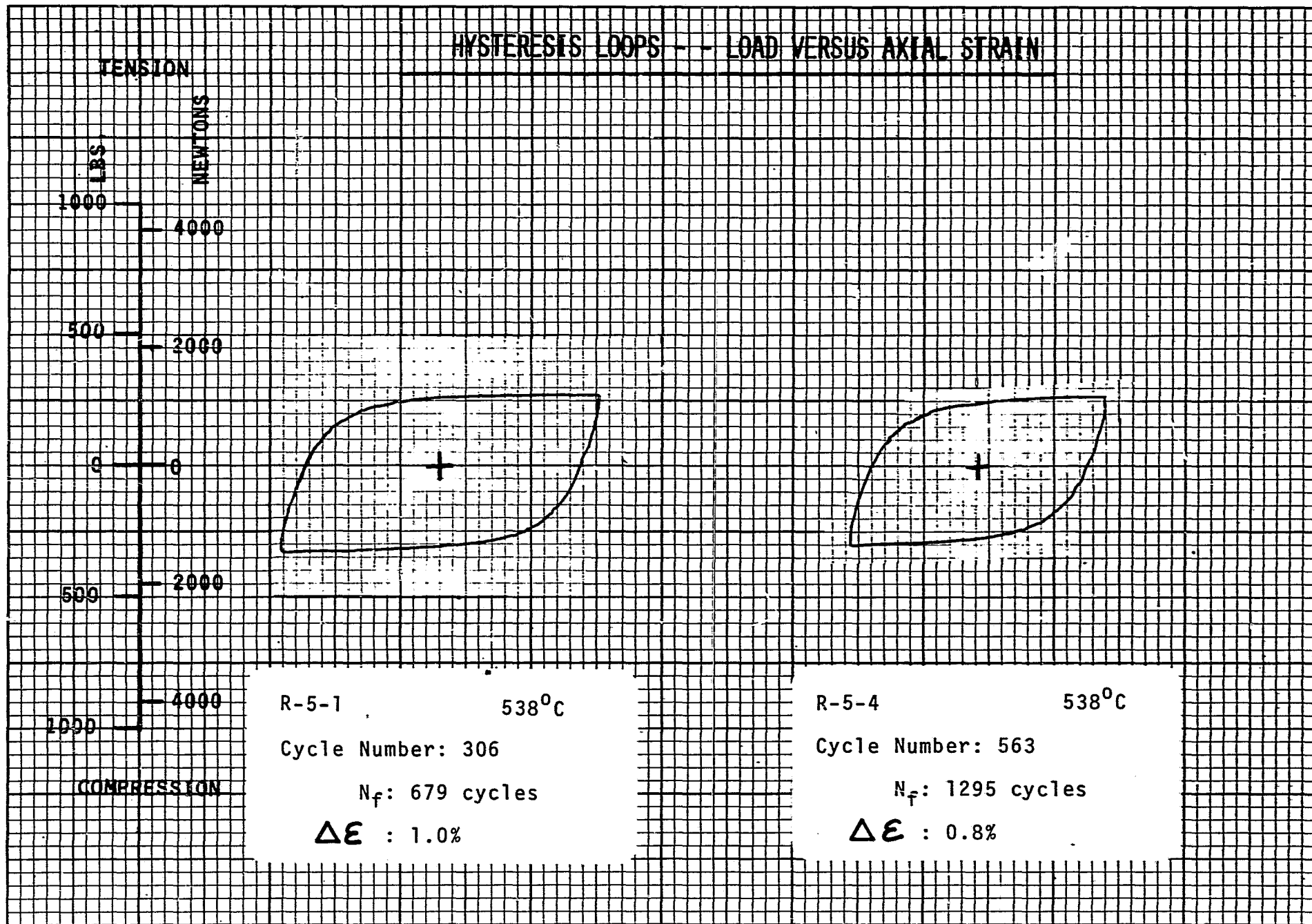
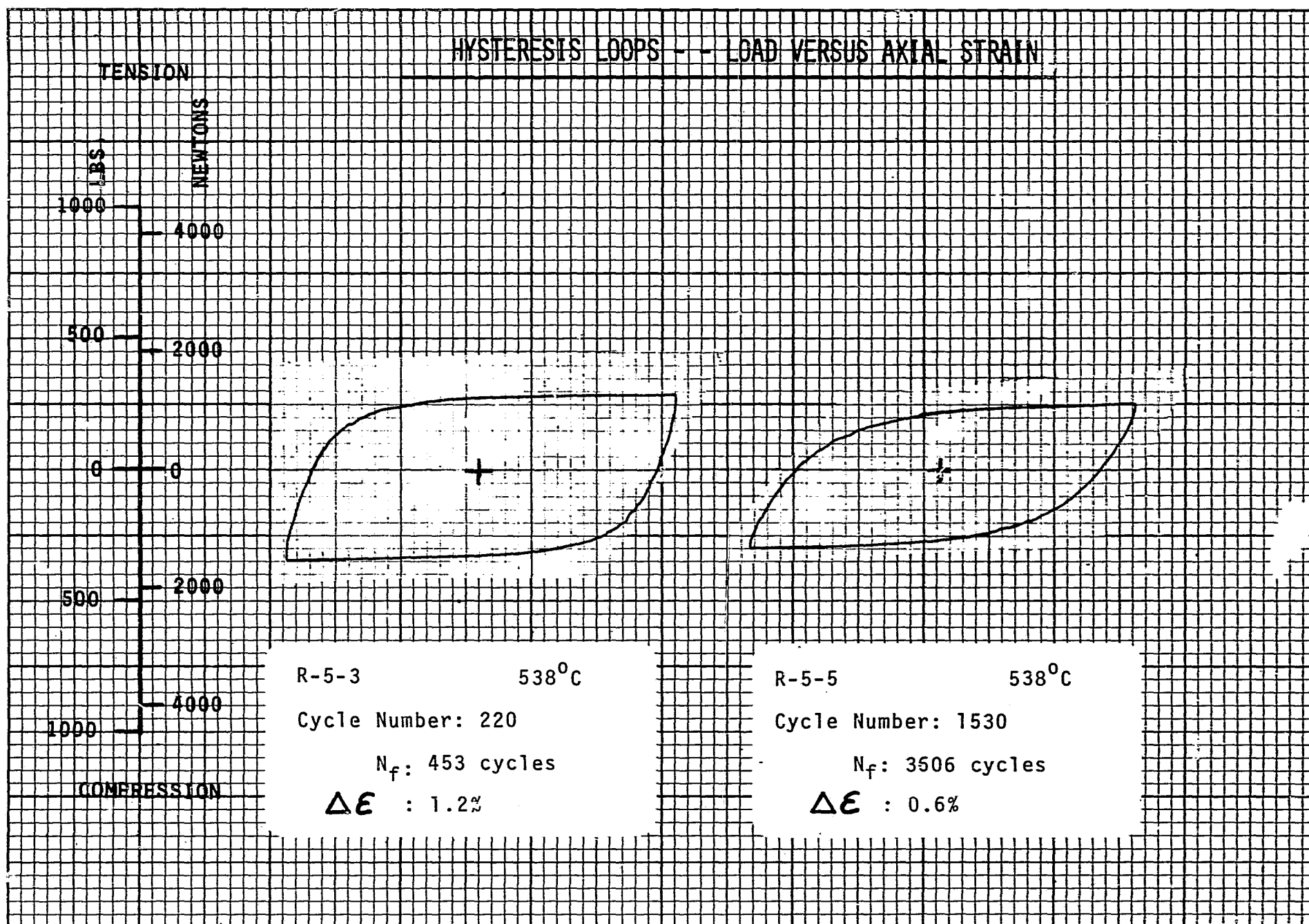


Figure 208



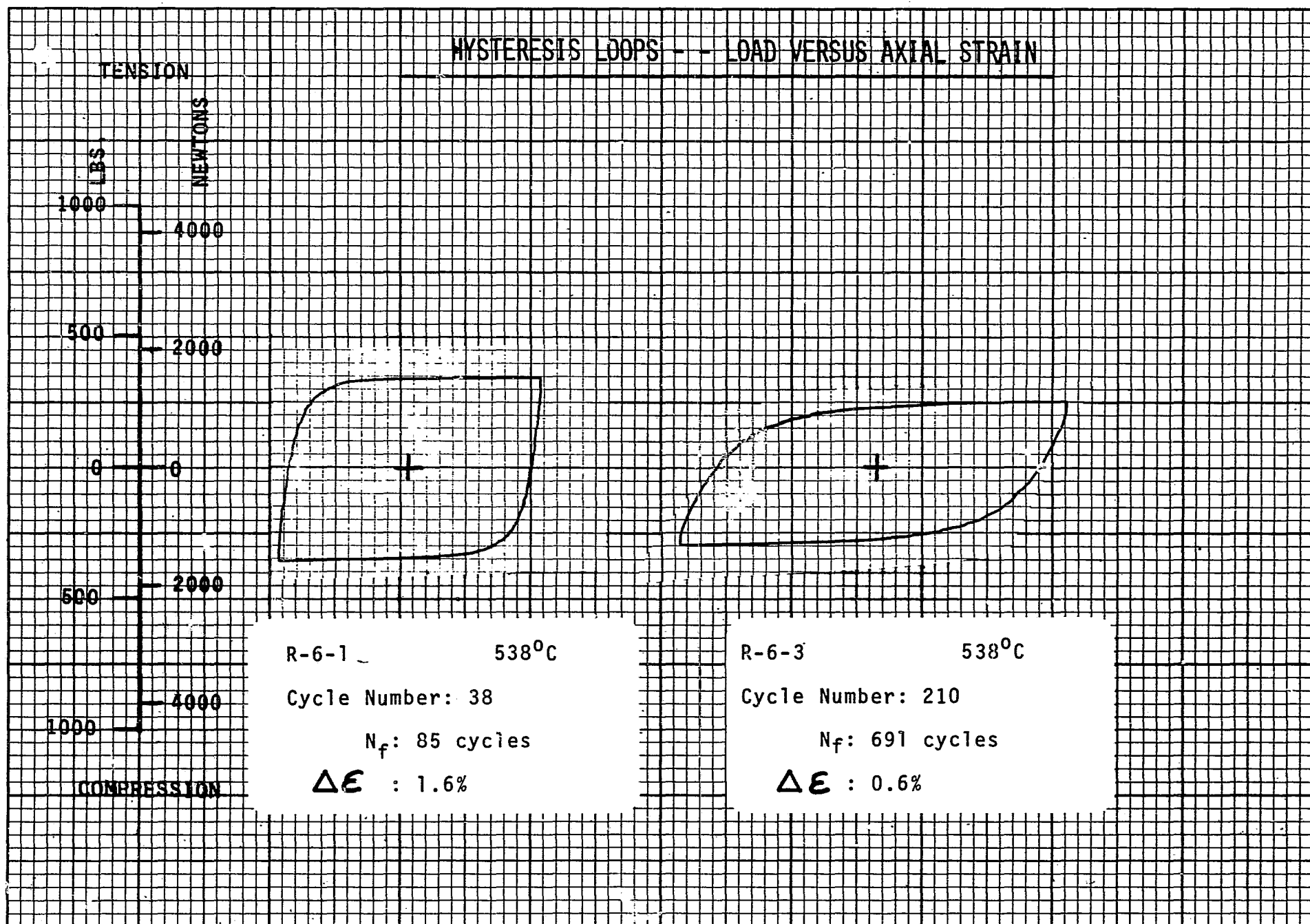
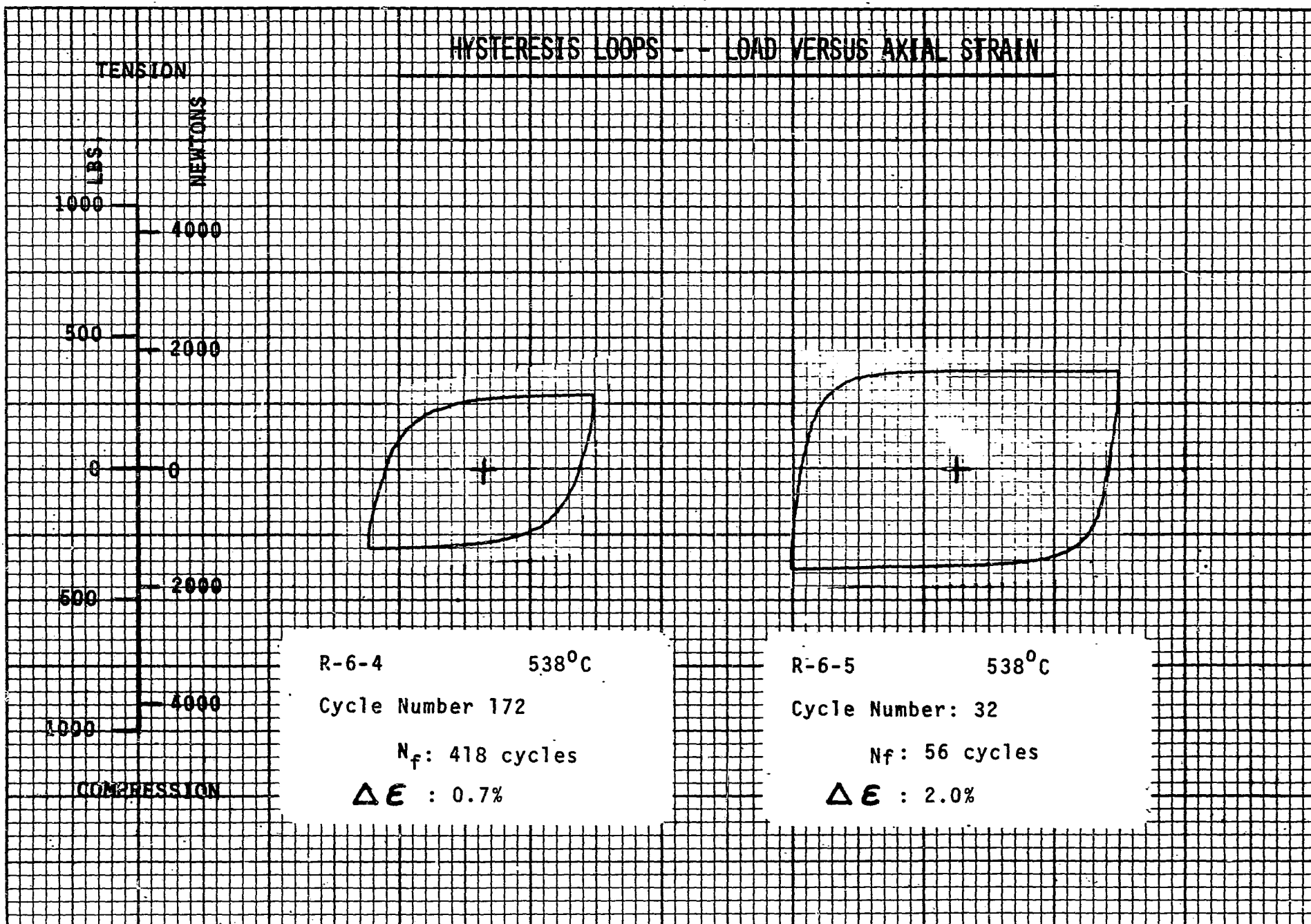


Figure 210



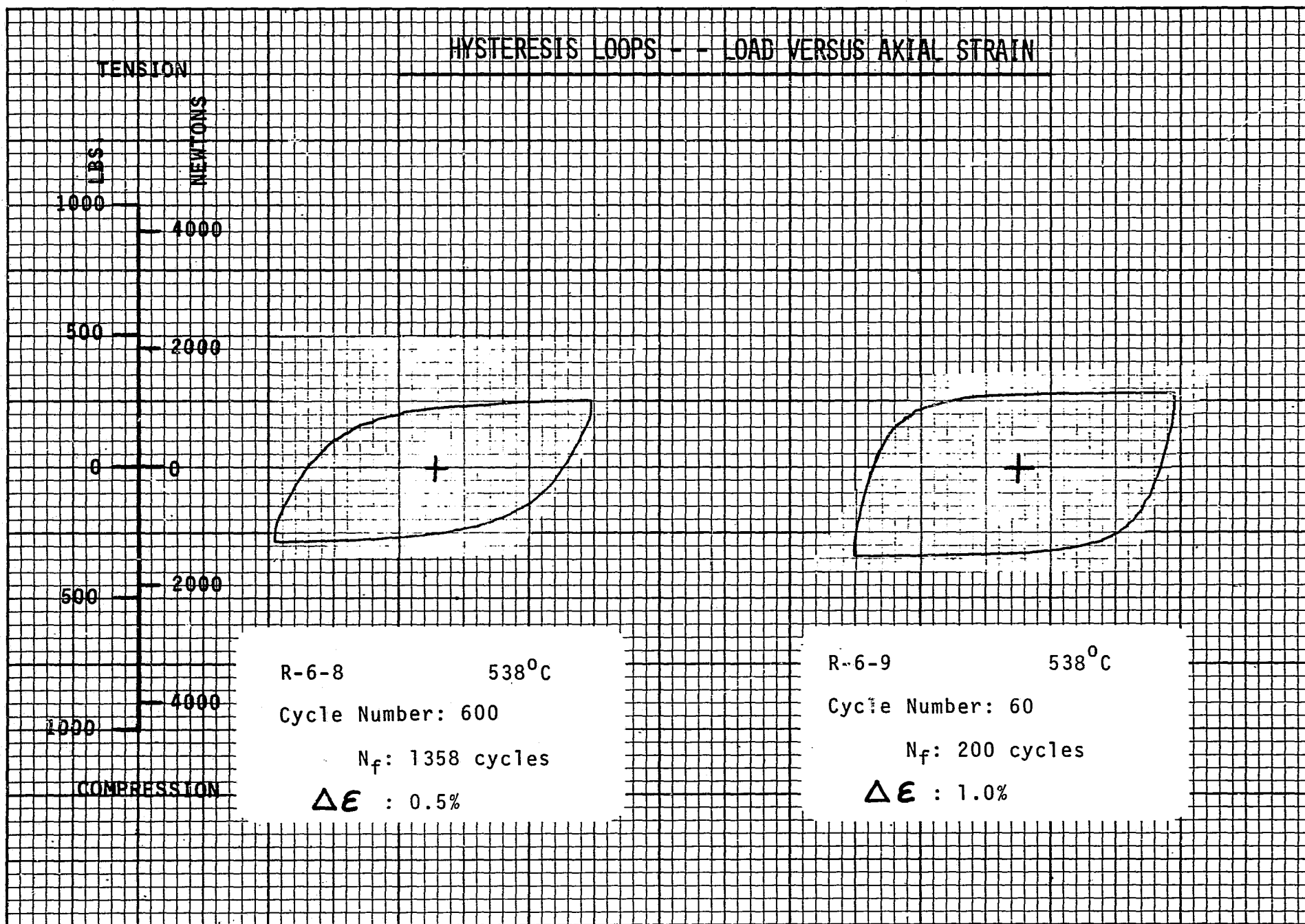
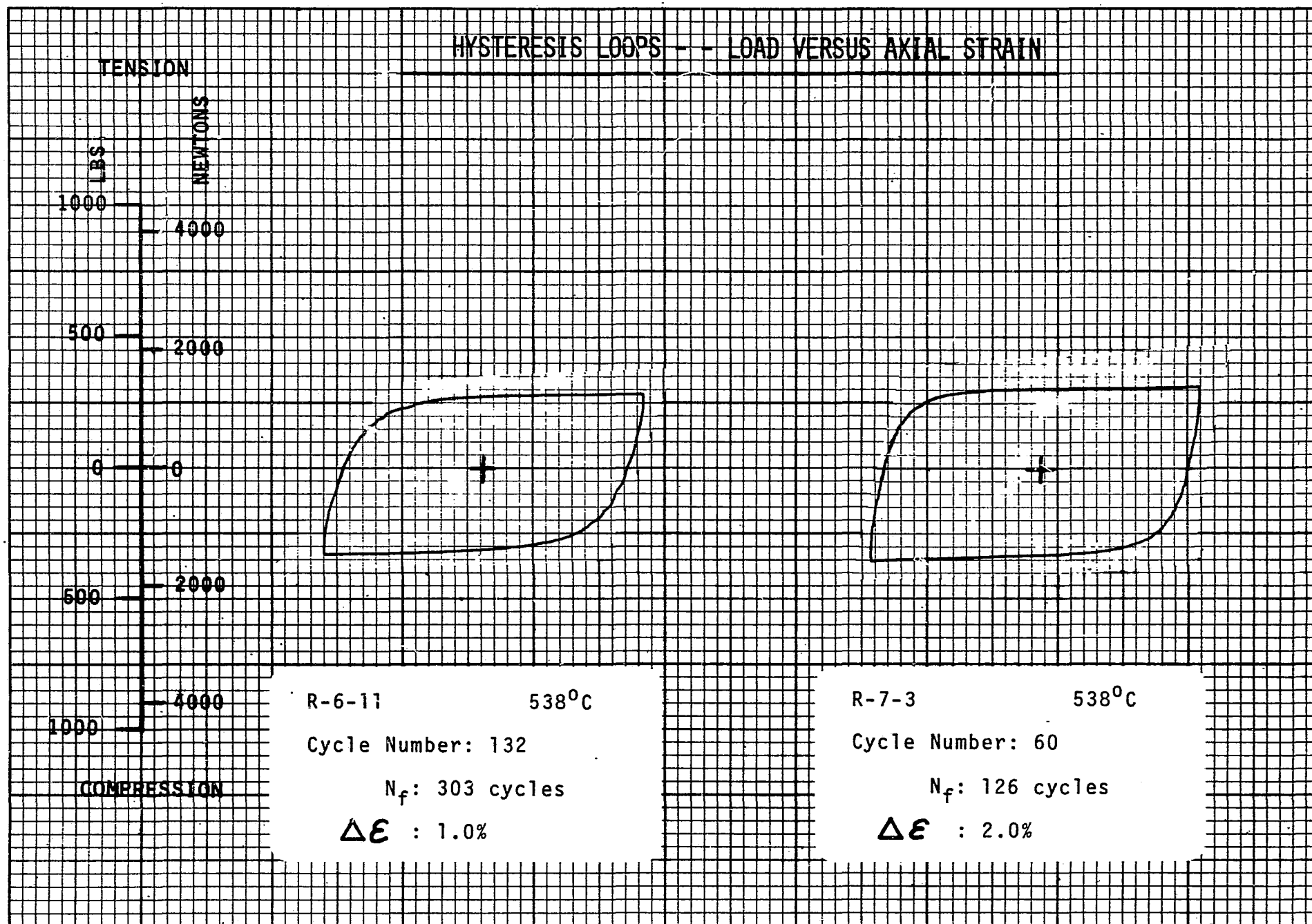


Figure 212



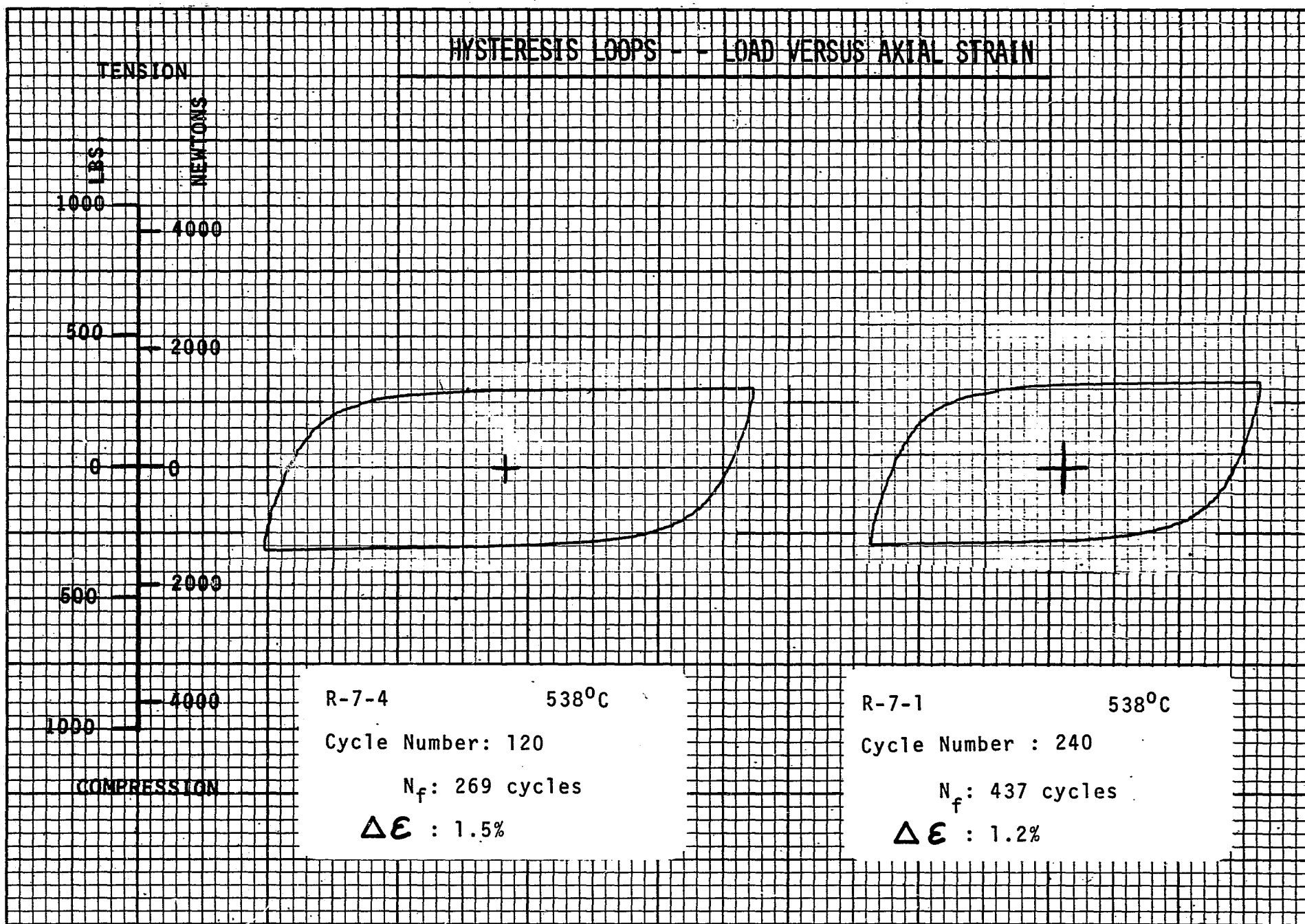
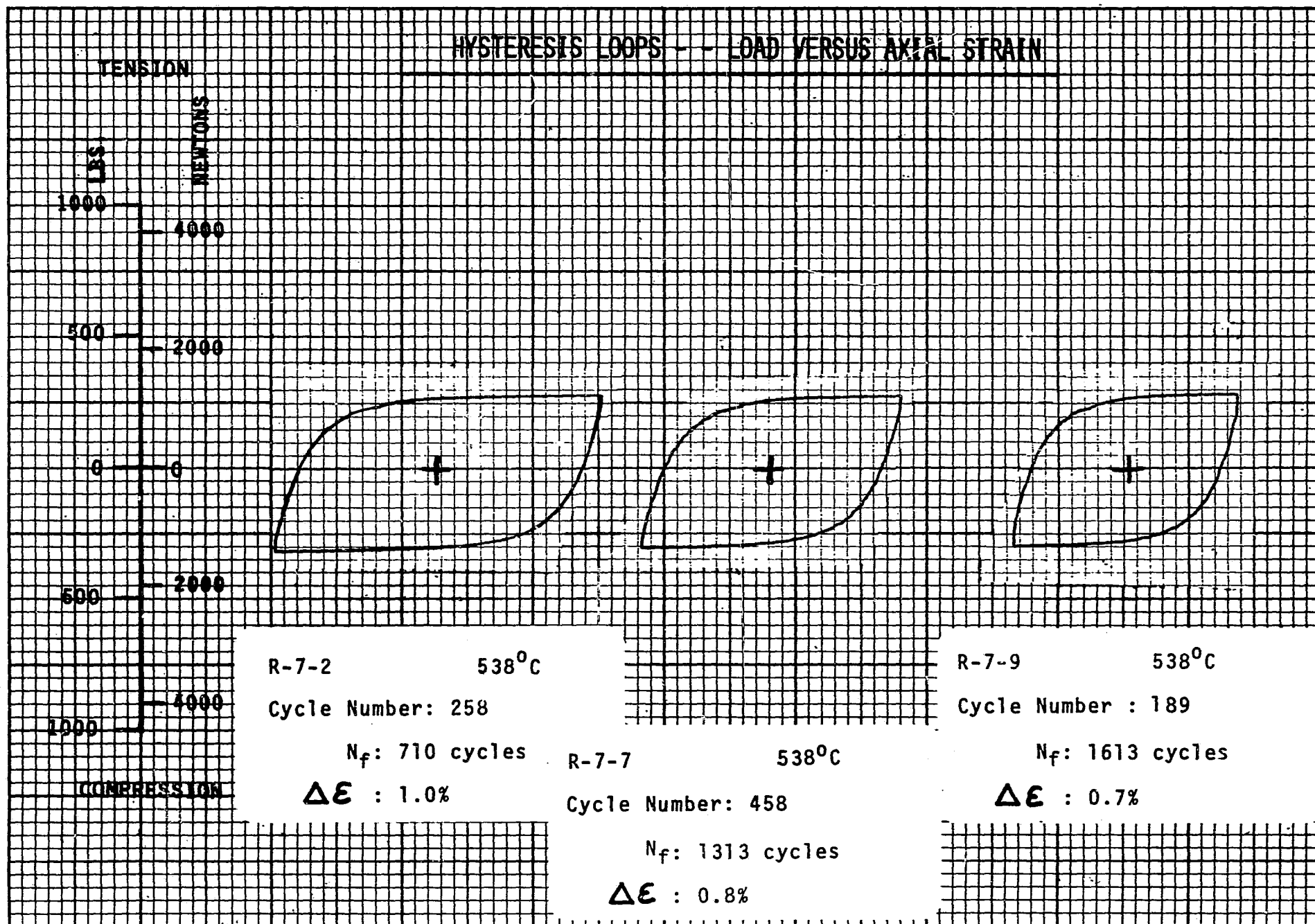


Figure 214



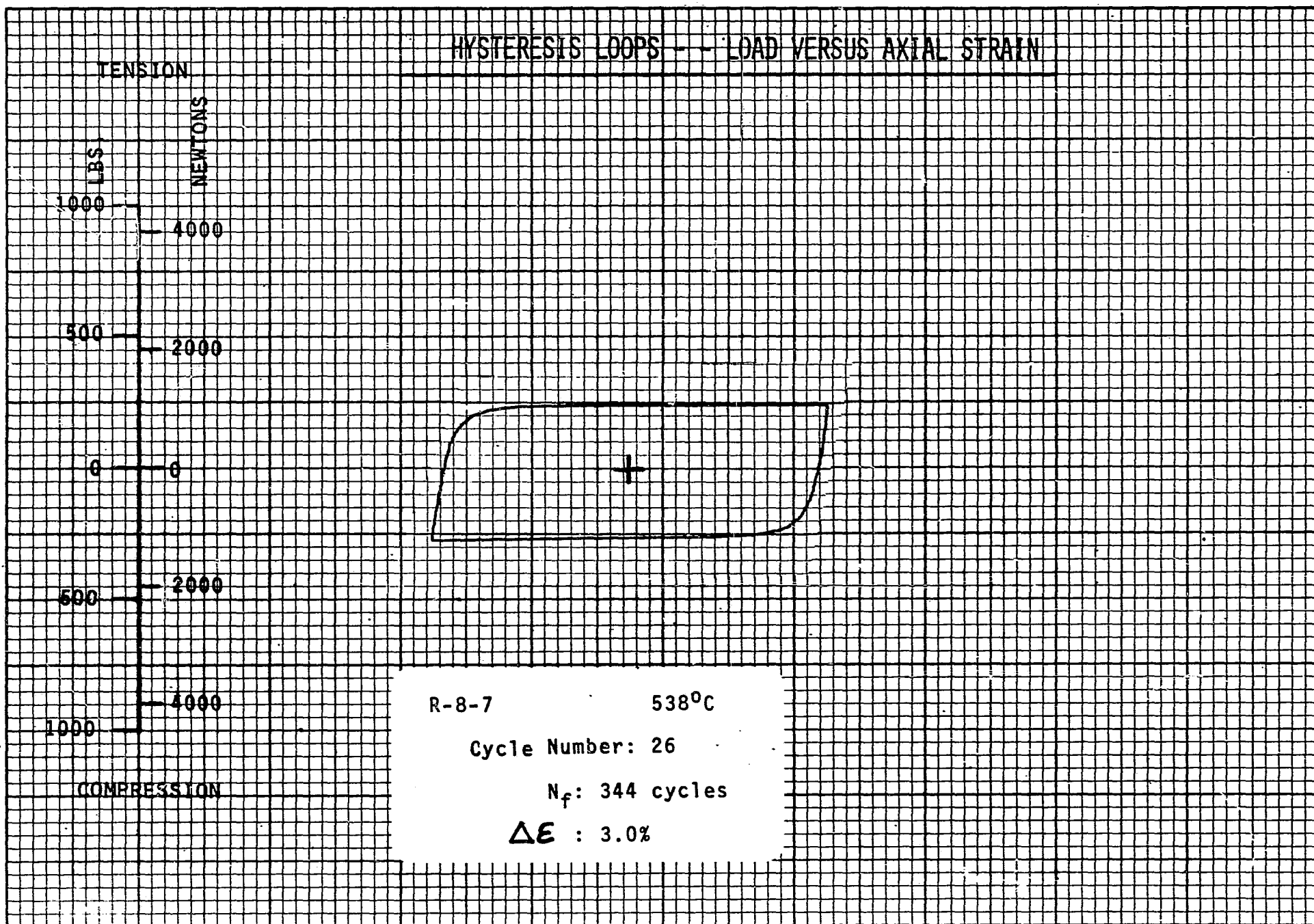


Figure 216

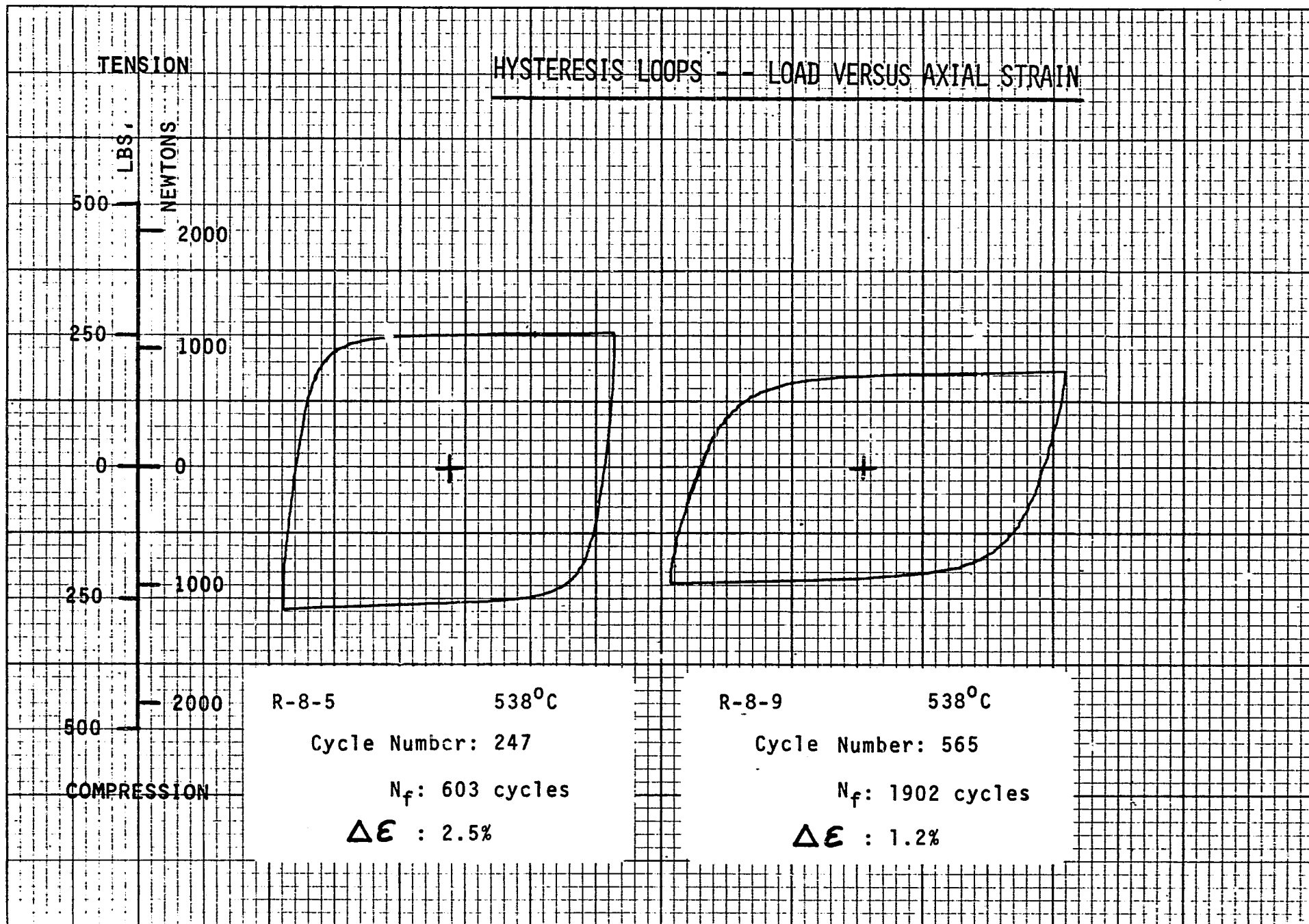


Figure 217

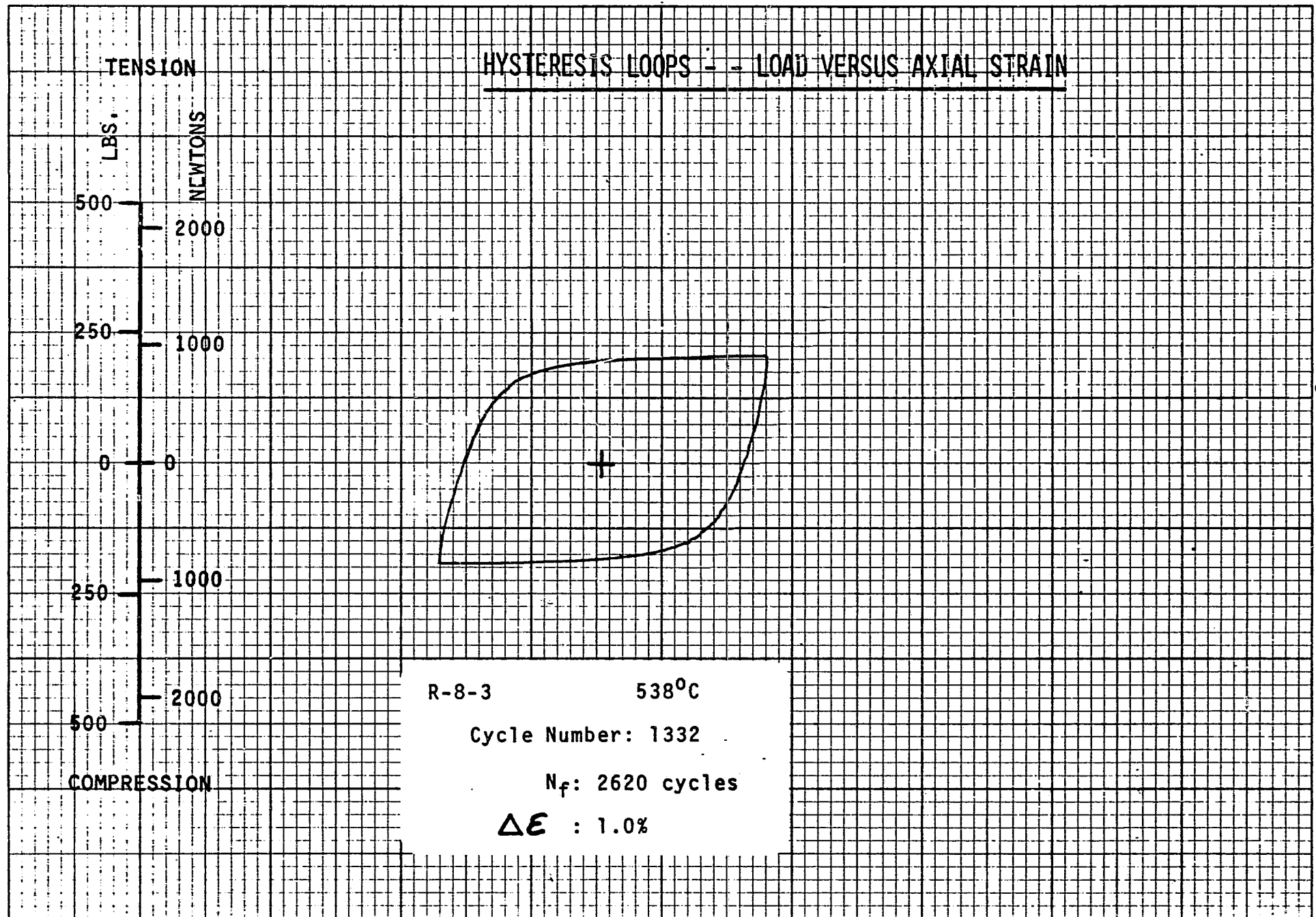


Figure 218

HYSTERESIS LOOPS -- LOAD VERSUS AXIAL STRAIN

TENSION

1000 LBS

500

0

500

1000

COMPRESSION

NEWTONS

4000

2000

0

2000

4000

R-8-2 538°C

Cycle Number: 369

N_f : 928 cycles

$\Delta \epsilon$: 2.0%

R-8-10 -538°C

Cycle Number: 680

N_f : 1381 cycles

$\Delta \epsilon$: 1.5%

HYSTERESIS LOOPS - - LOAD VERSUS AXIAL STRAIN

TENSION
2000
1000
0
1000
2000
NEWTONS
8000
4000
0
4000
8000

R-9-11 538°C

Cycle Number: 362

N_f : 843 cycles

$\Delta \epsilon$: 2.0%

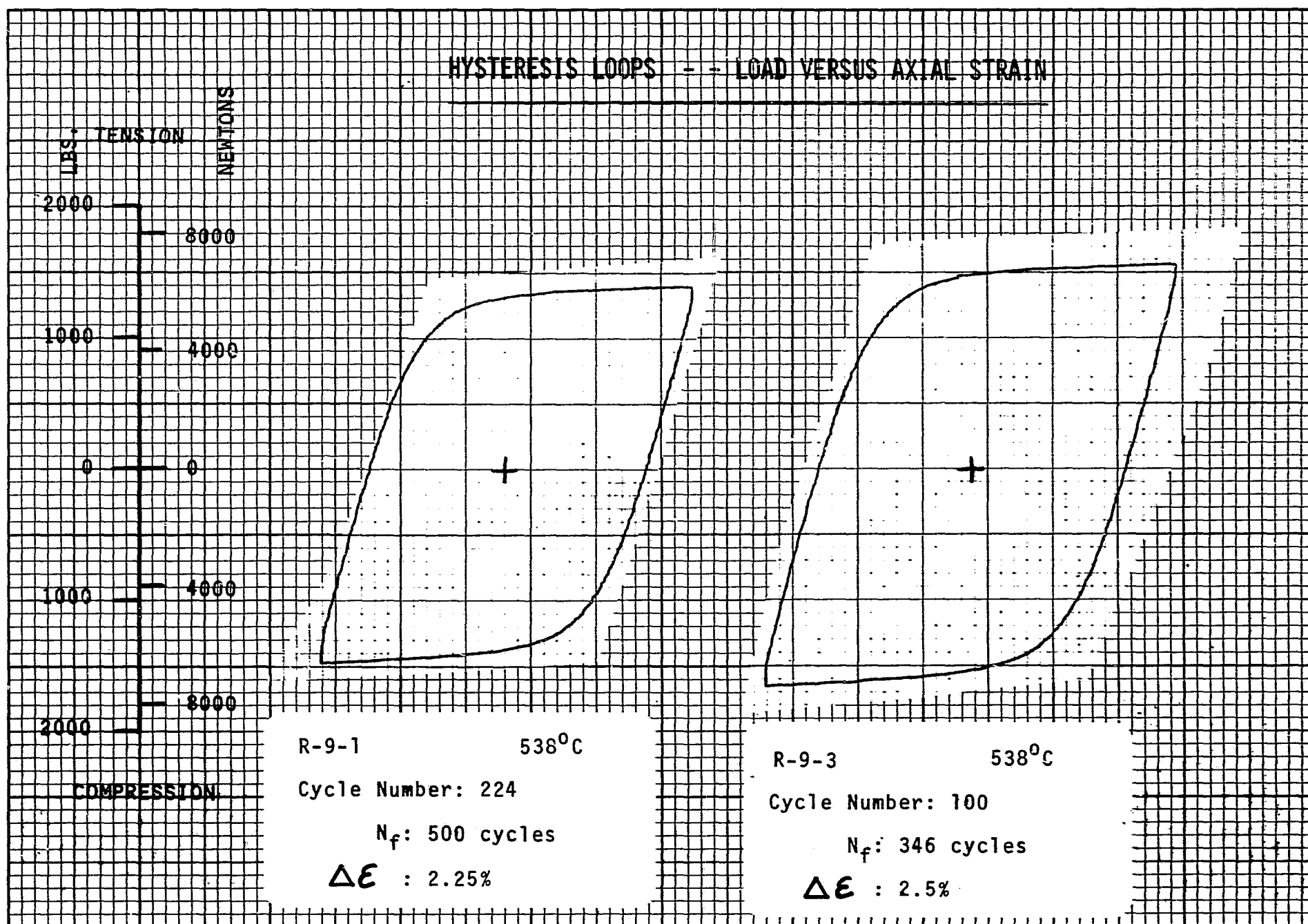
R-9-4 538°C

Cycle Number: 178

N_f : 357 cycles

$\Delta \epsilon$: 3.0%

HYSTERESIS LOOPS - - LOAD VERSUS AXIAL STRAIN



HYSTERESIS LOOPS - - LOAD VERSUS AXIAL STRAIN

LBS TENSION
NEWTONS

2000

8000

1000

4000

0

0

1000

4000

2000

8000

COMPRESSION

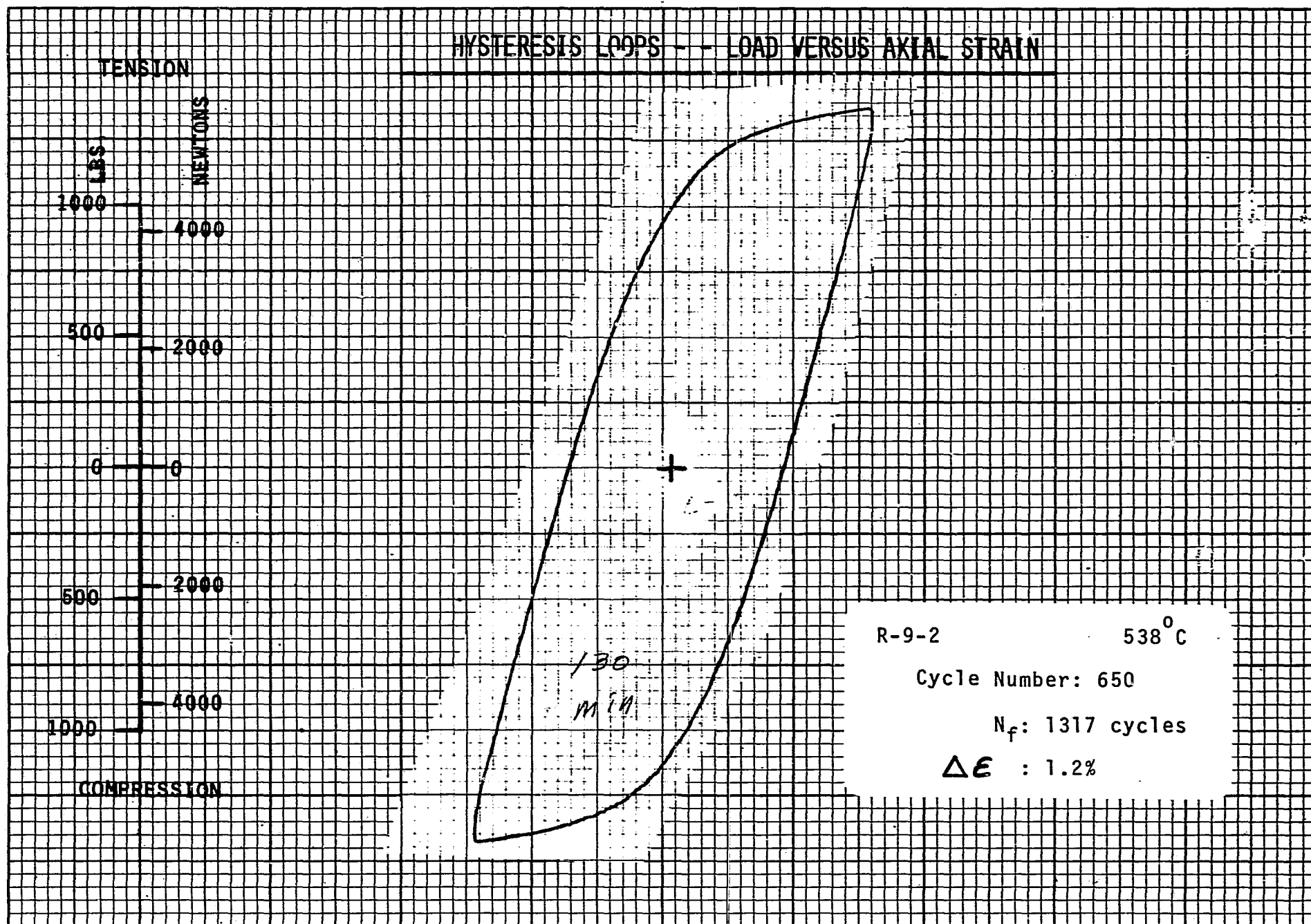
R-9-8

538°C

Cycle Number: 313

N_f : 2000 cycles

$\Delta\epsilon$: 1.4%



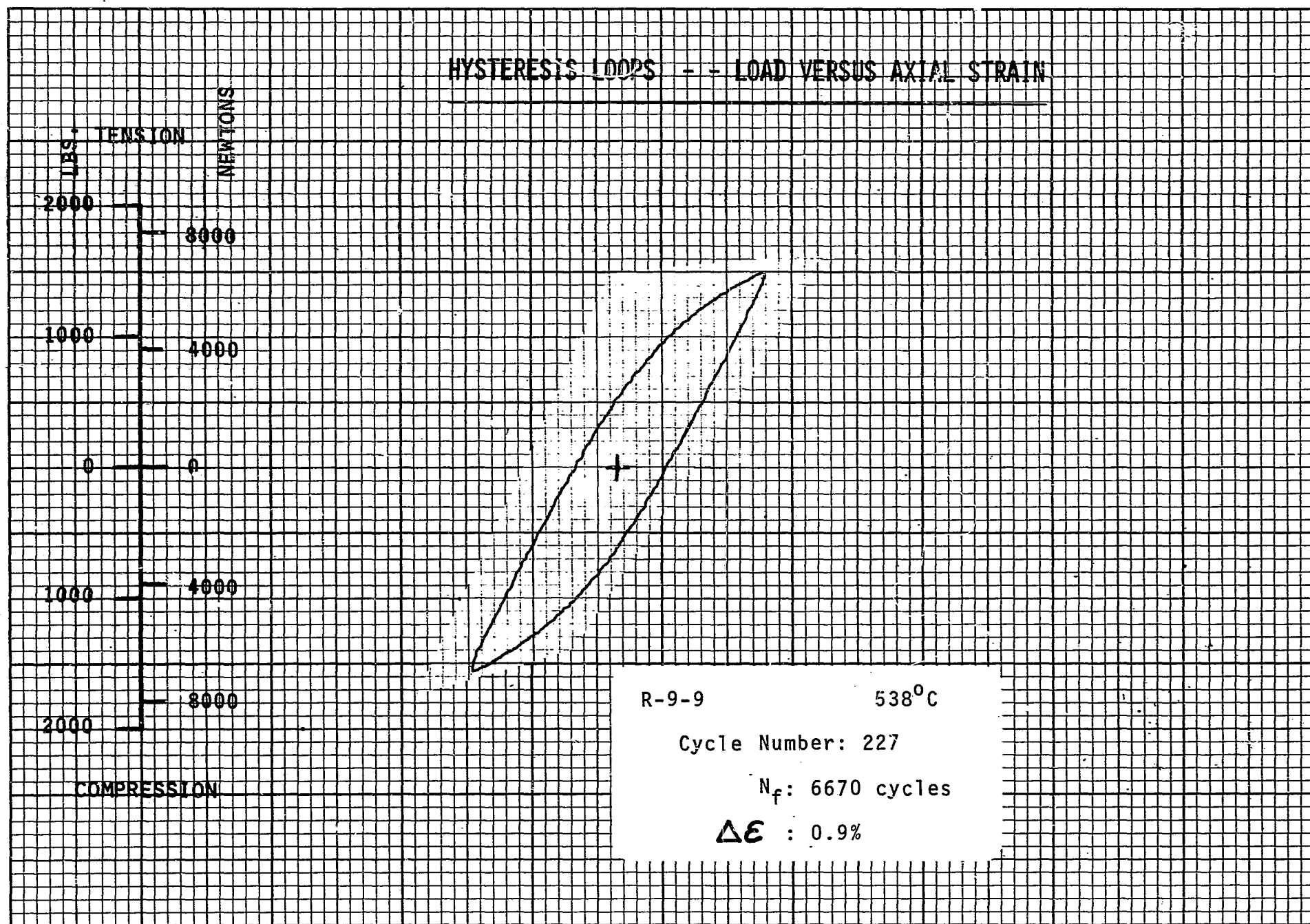
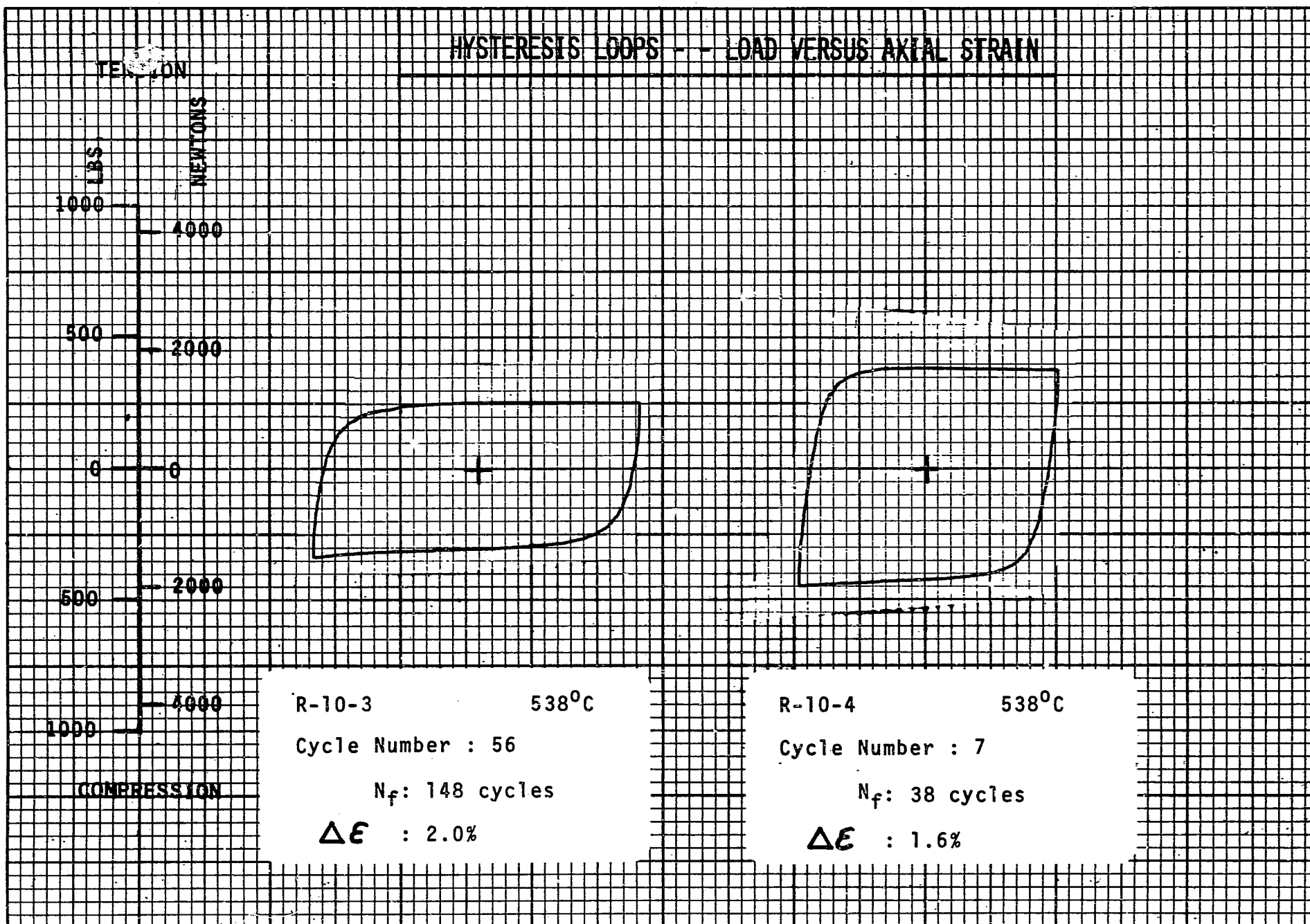


Figure 224



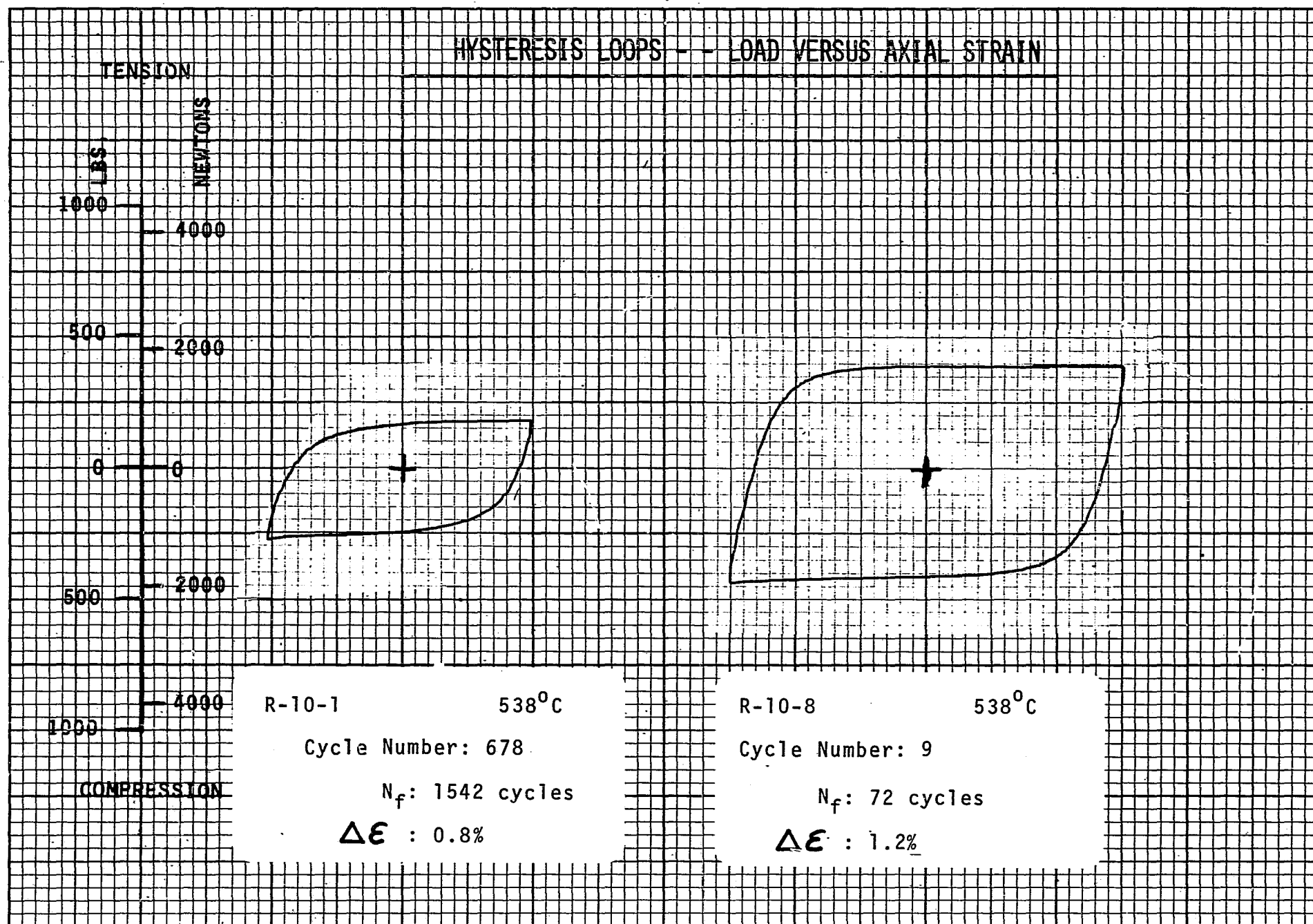


Figure 226

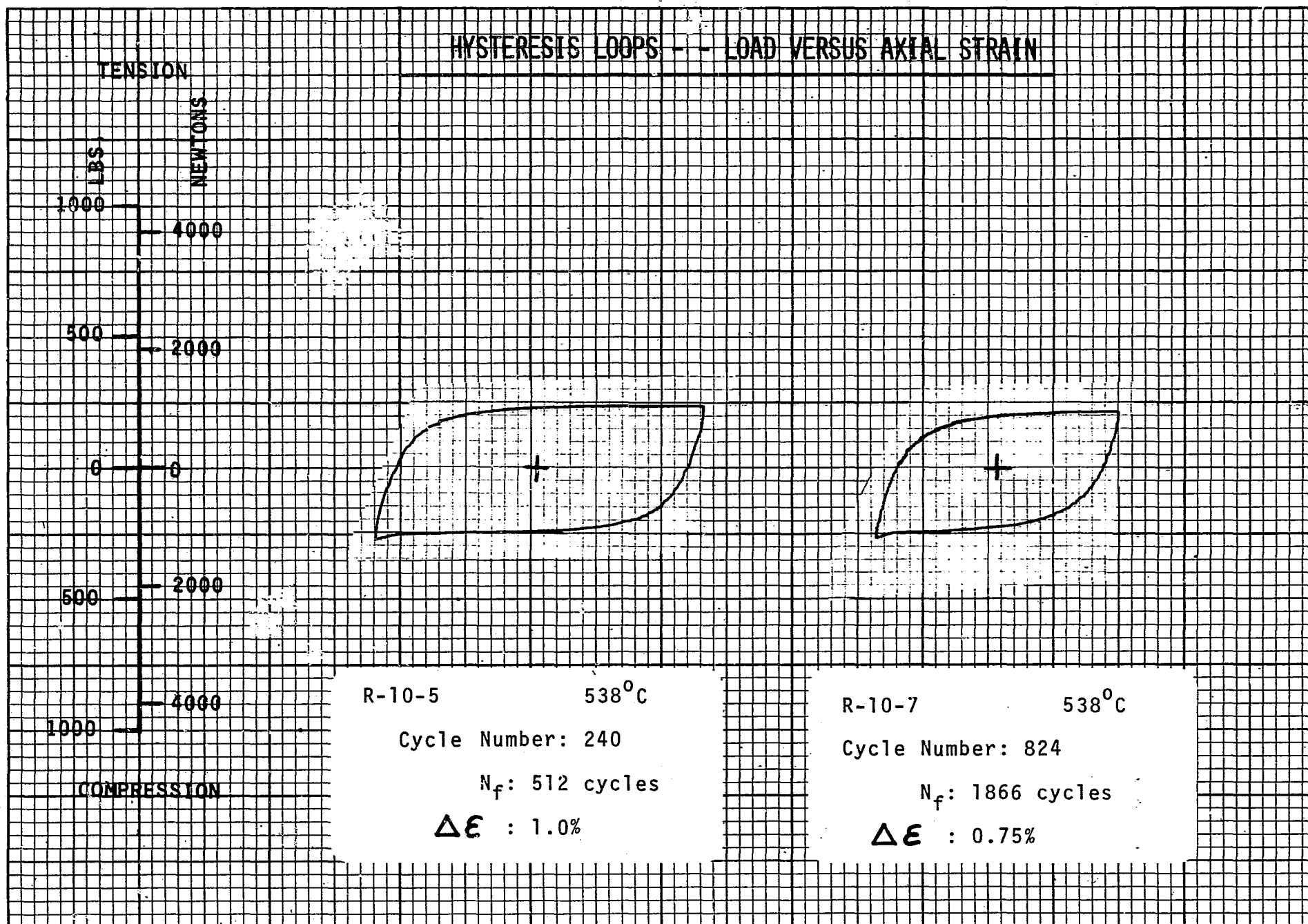


Figure 227

HYSTERESIS LOOPS - - LOAD VERSUS AXIAL STRAIN

LBS. TENSION
NEWTONS

2000

8000

1000

4000

0

0

1000

4000

2000

8000

COMPRESSION

R-13-5

538°C

Cycle Number: 45

N_f : 90 cycles

ΔE : 2.0%

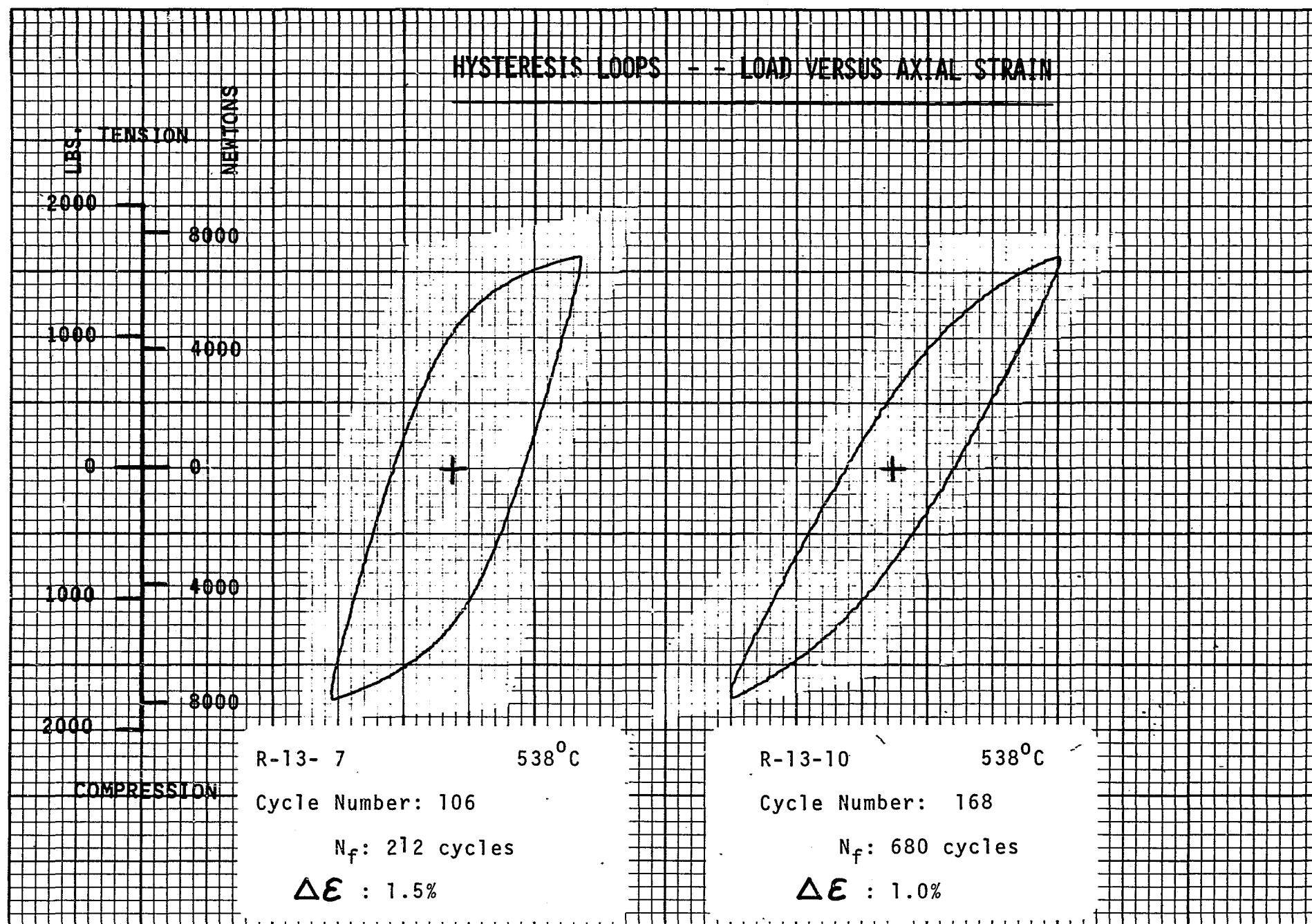
R-13-13

538°C

Cycle Number: 325

N_f : 644 cycles

ΔE : 1.2%



HYSTERESIS LOOPS - - LOAD VERSUS AXIAL STRAIN

LIB
TENSION
NEWTONS

2000

8000

1000

4000

0

0

1000

4000

2000

8000

COMPRESSION

R-13-6 538°C

Cycle Number: 480

N_f : 1615 cycles

$\Delta \epsilon$: 0.8%

R-13-8 538°C

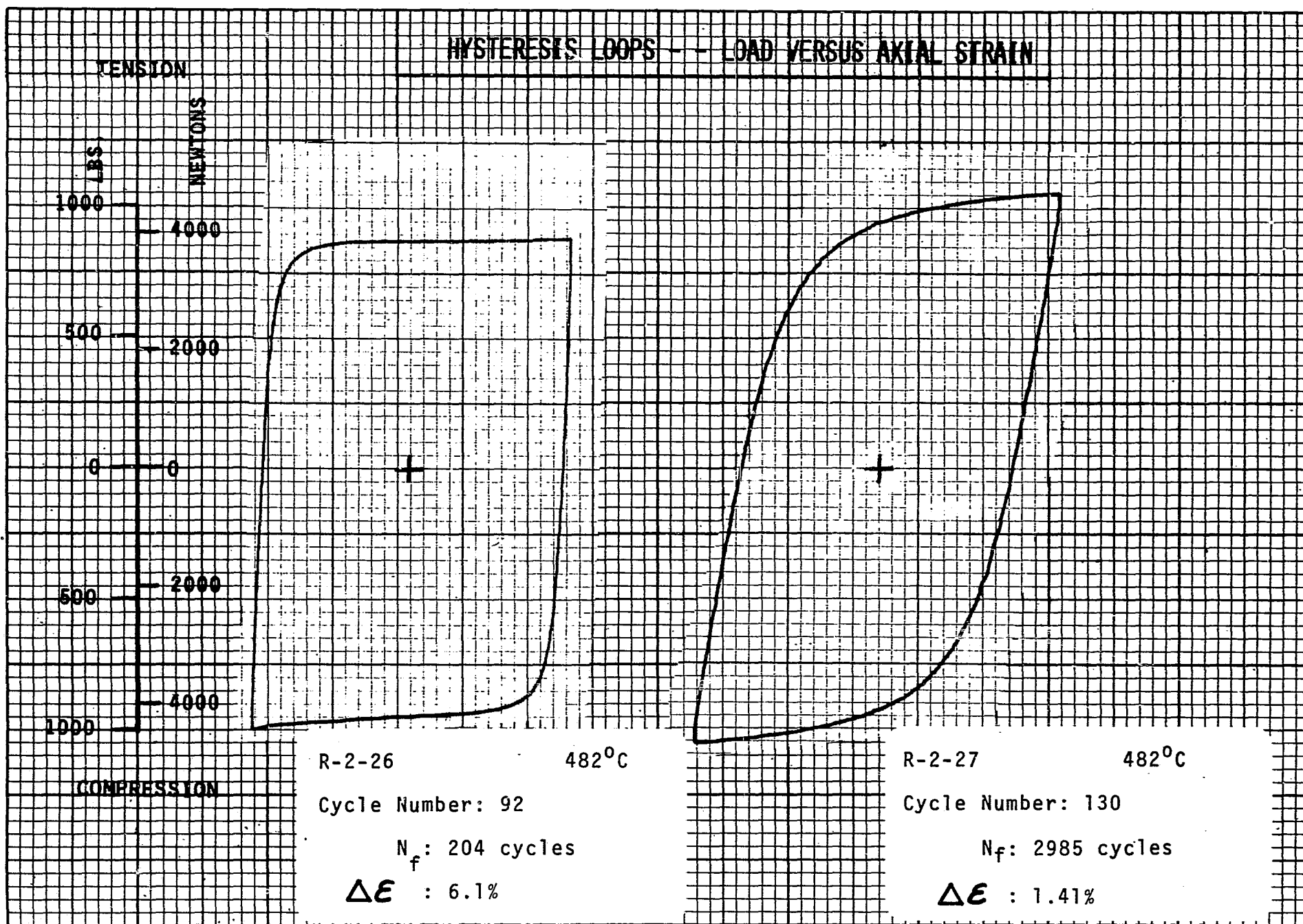
Cycle Number: 515

N_f : 3623 cycles

$\Delta \epsilon$: 0.7%

Figure 230

Figure 231



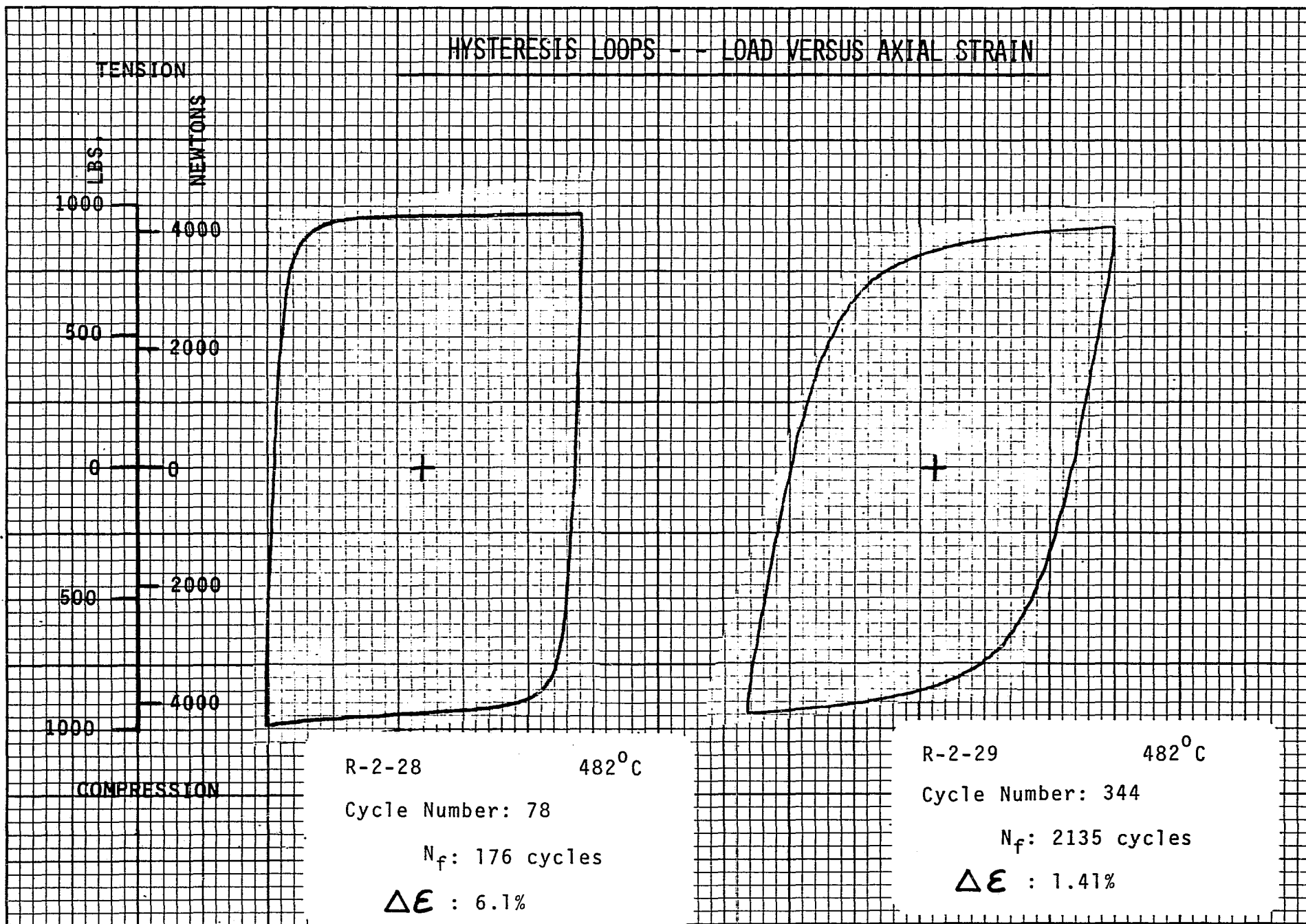
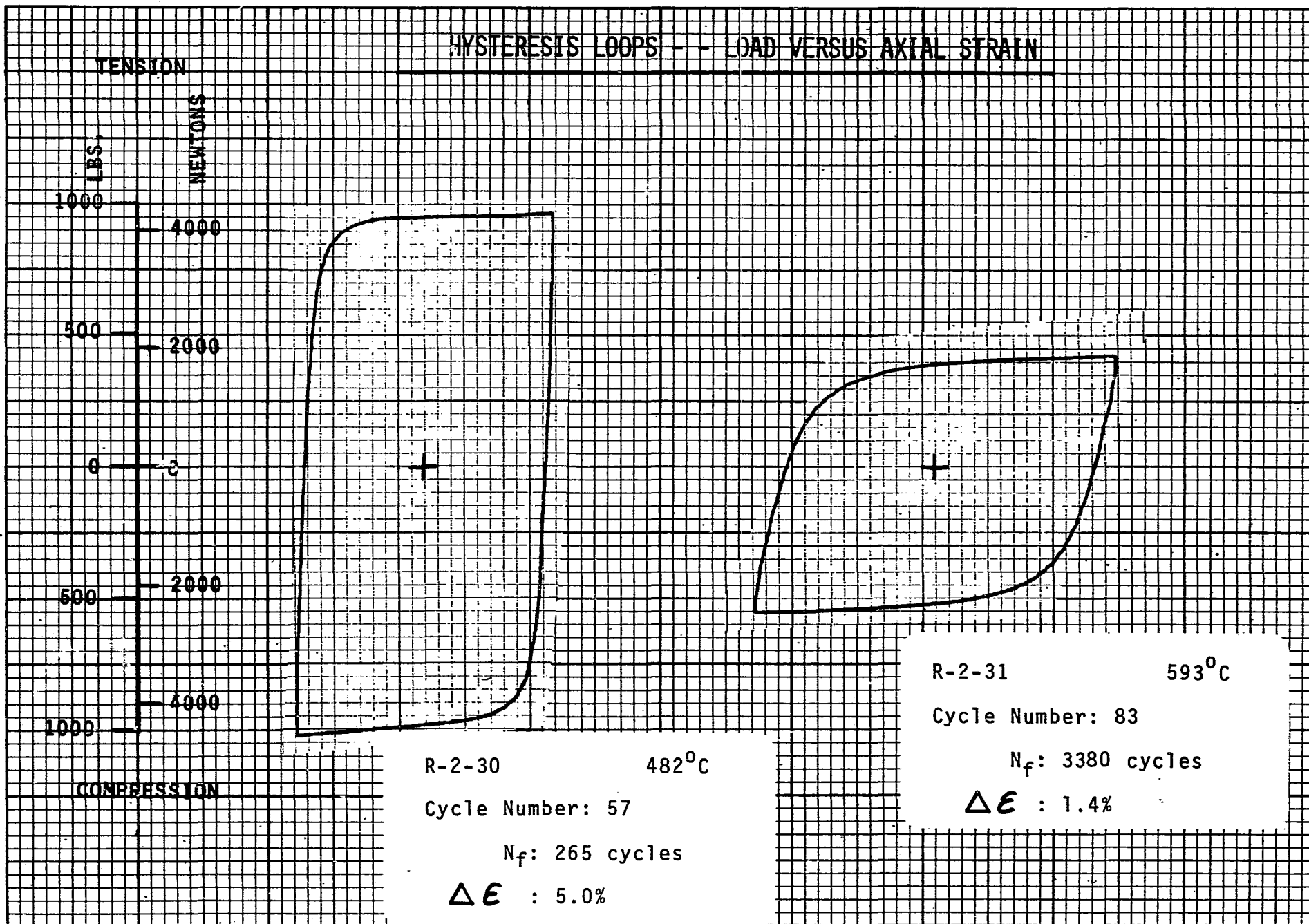


Figure 232



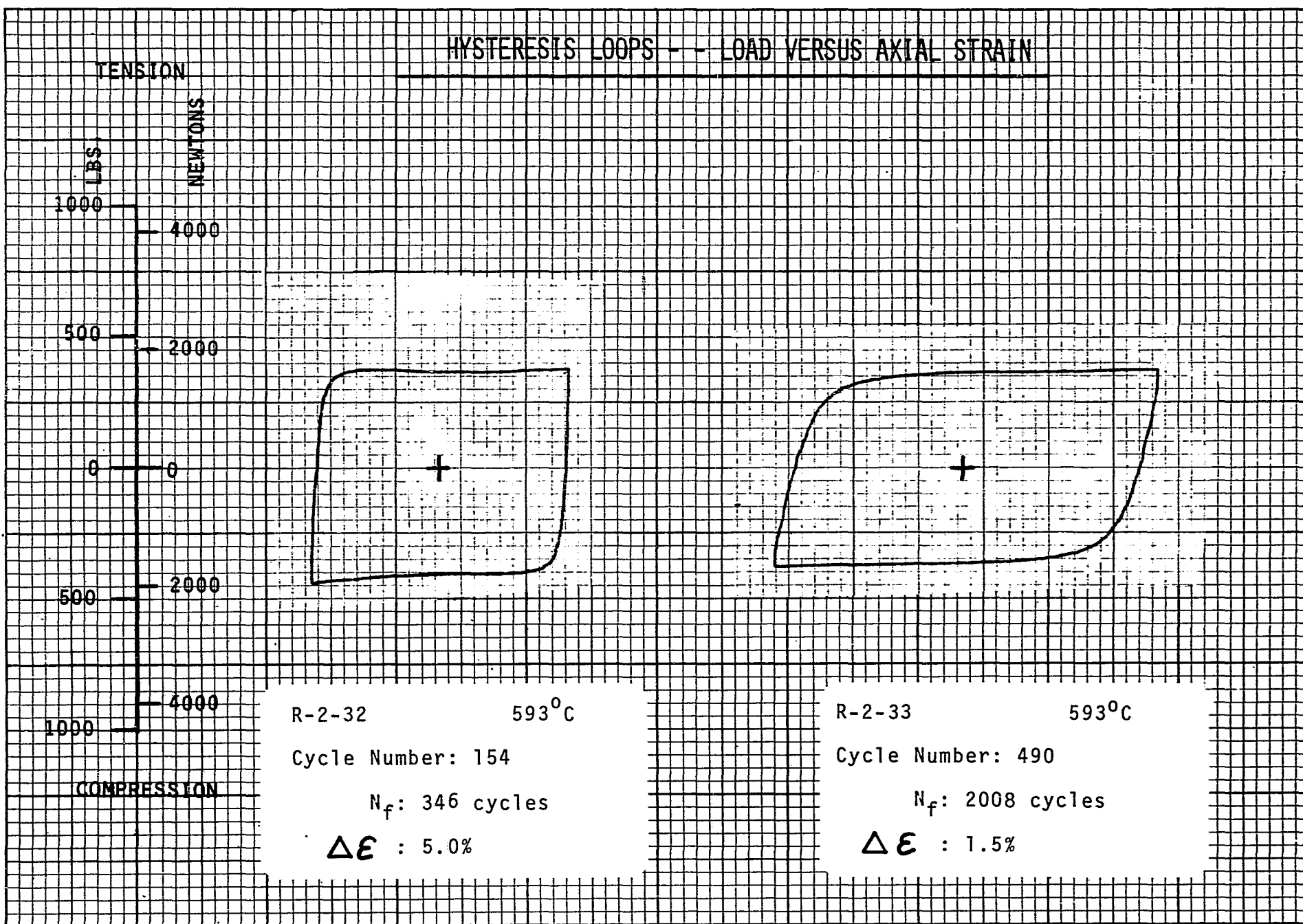
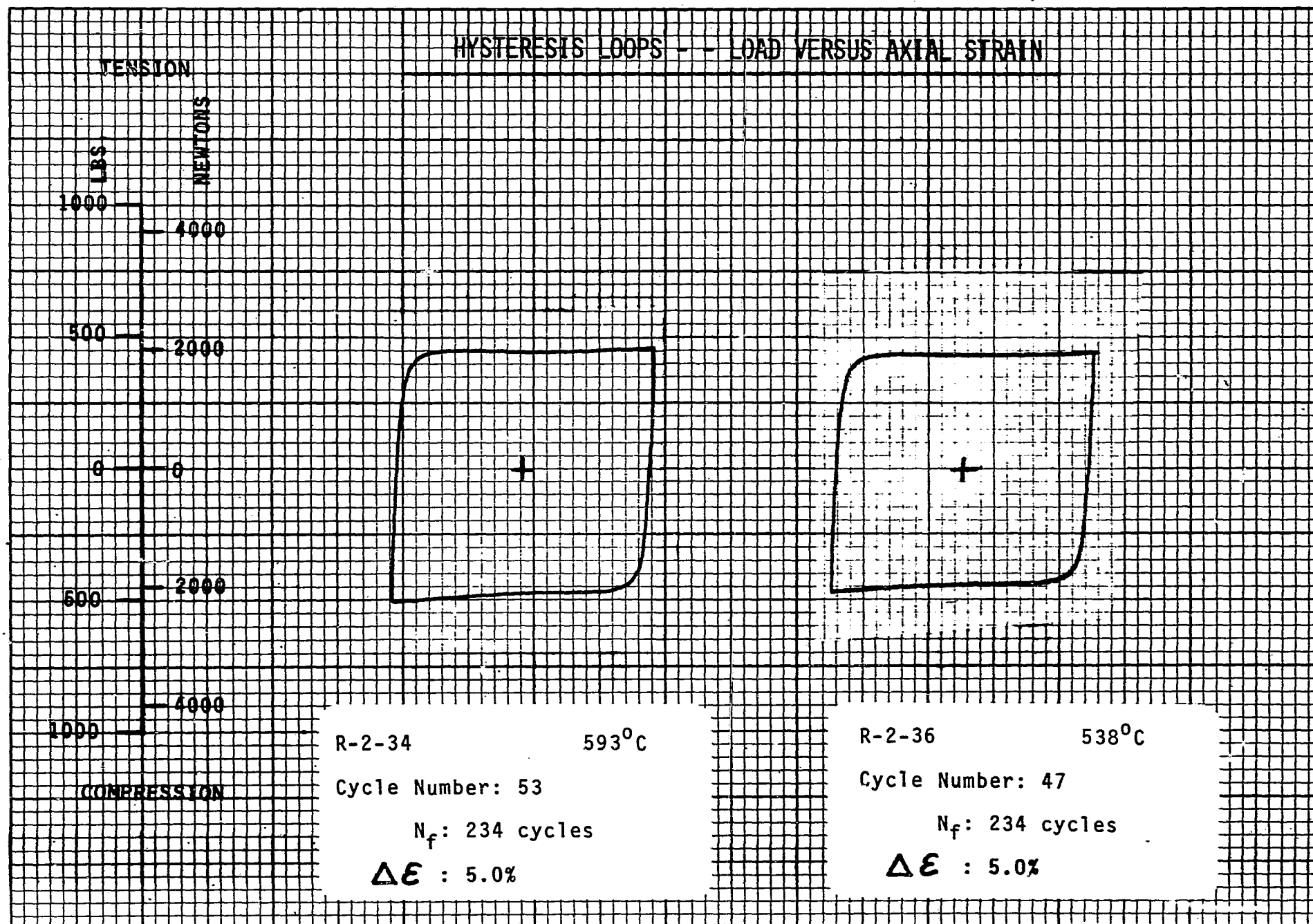


Figure 234

Figure 235



HYSTERESIS LOOPS -- LOAD VERSUS AXIAL STRAIN

TENSION

LBS

NEWTONS

1000

500

0

500

1000

4000

2000

2000

4000

R-2-37

538°C

Cycle Number: 60

N_f : 1613 cycles

$\Delta \epsilon$: 1.4%

R-2-39

538°C

Cycle Number: 10

N_f : 238 cycles

$\Delta \epsilon$: 5.0%

COMPRESSION

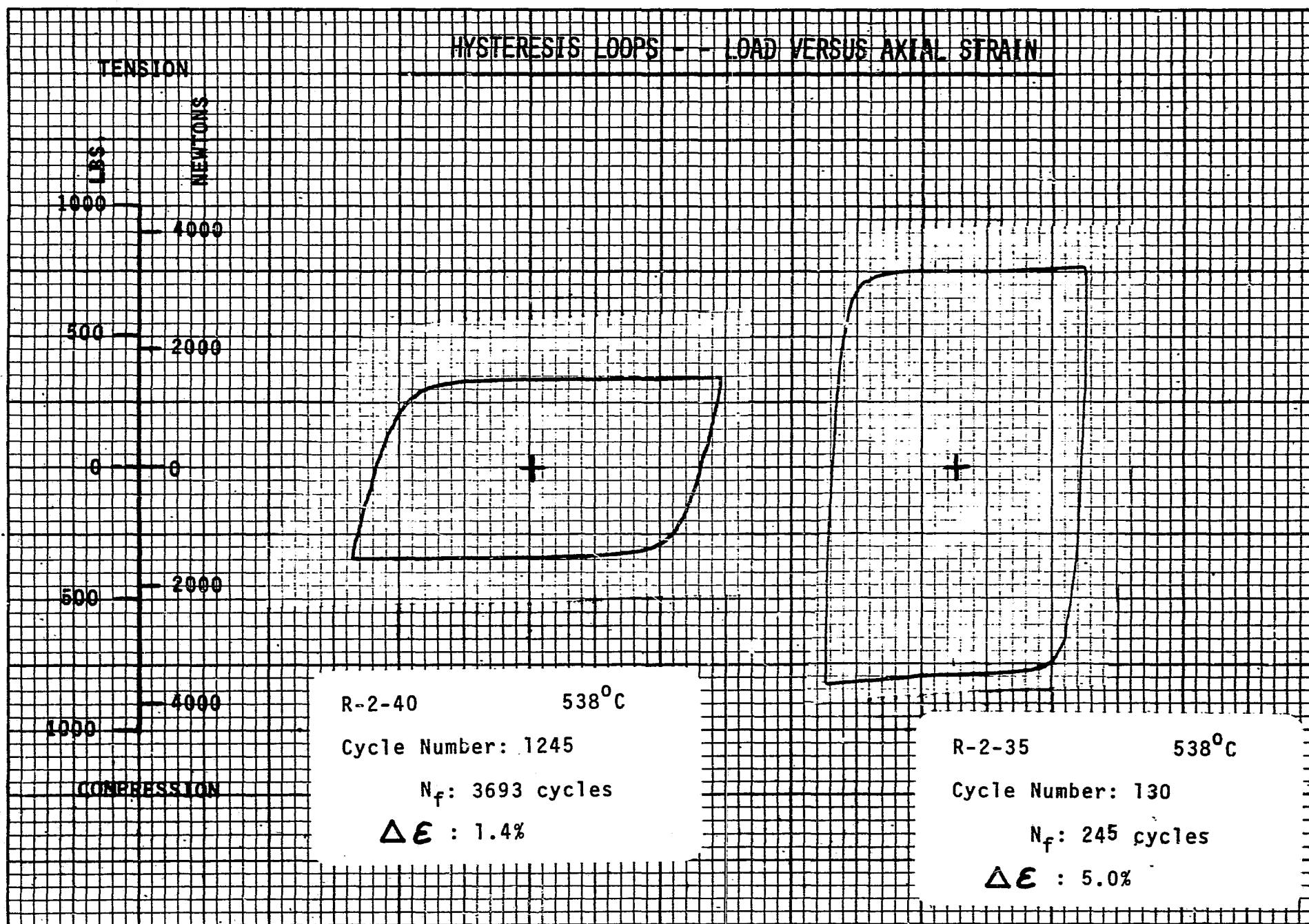


Figure 237

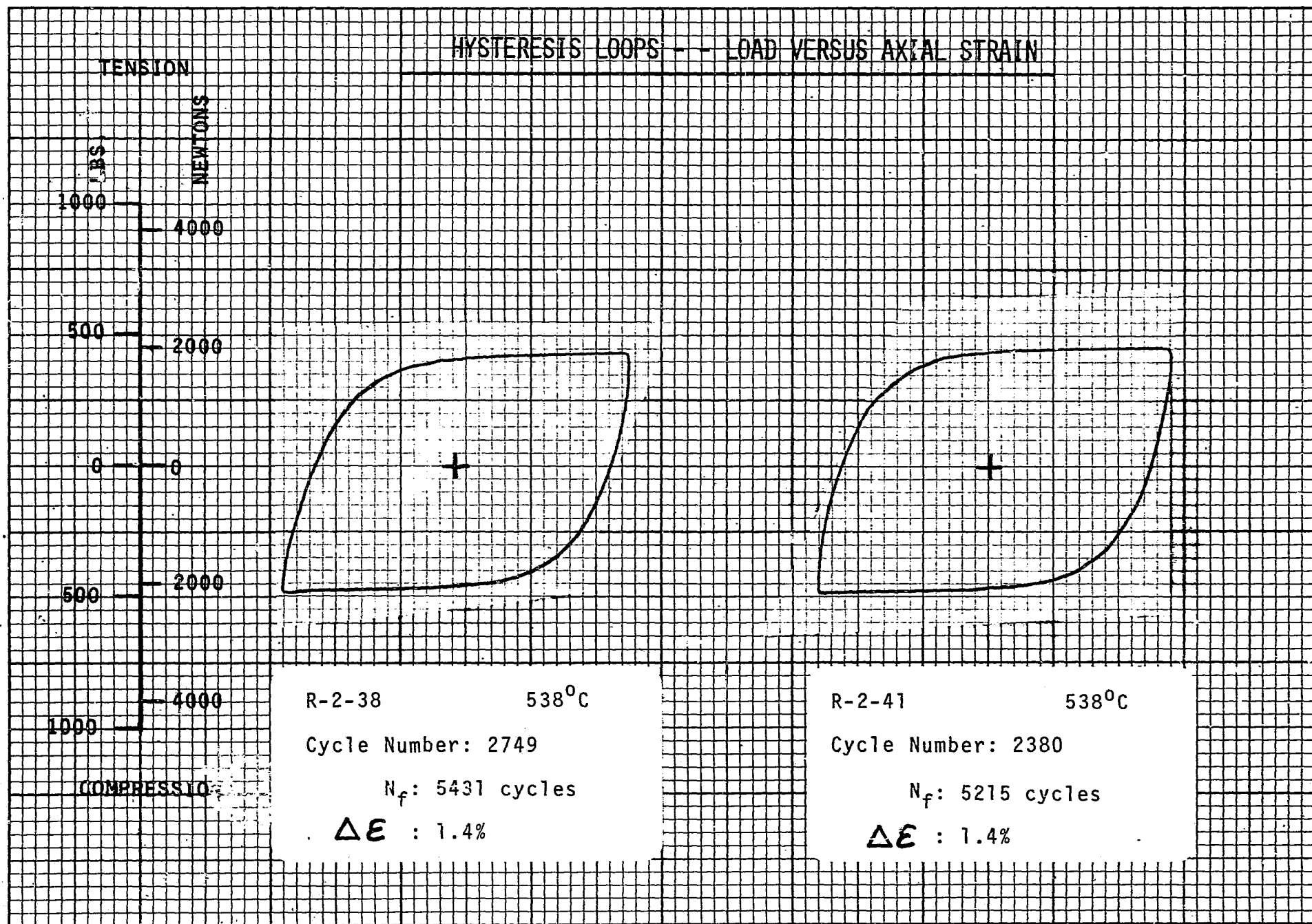
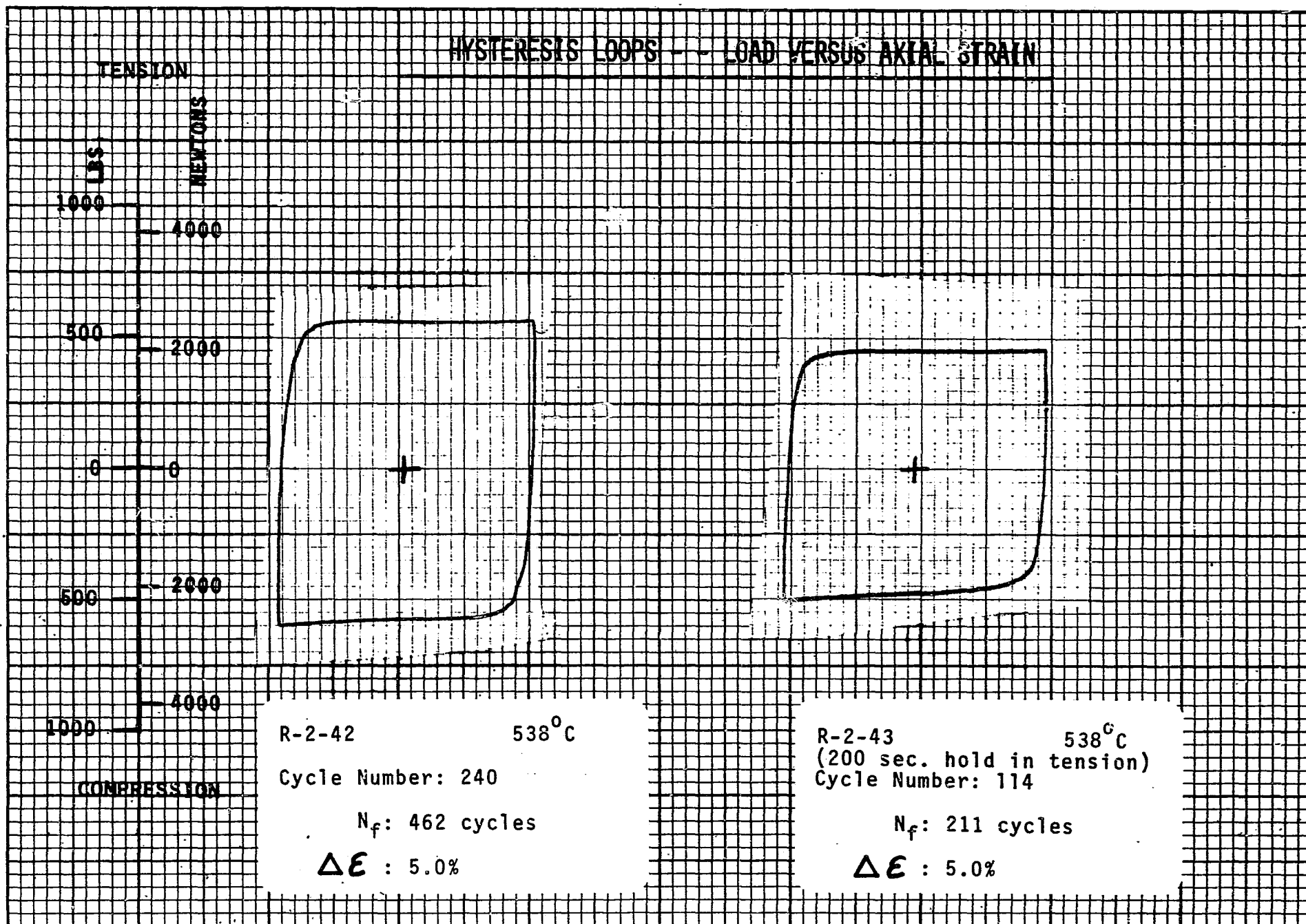


Figure 238



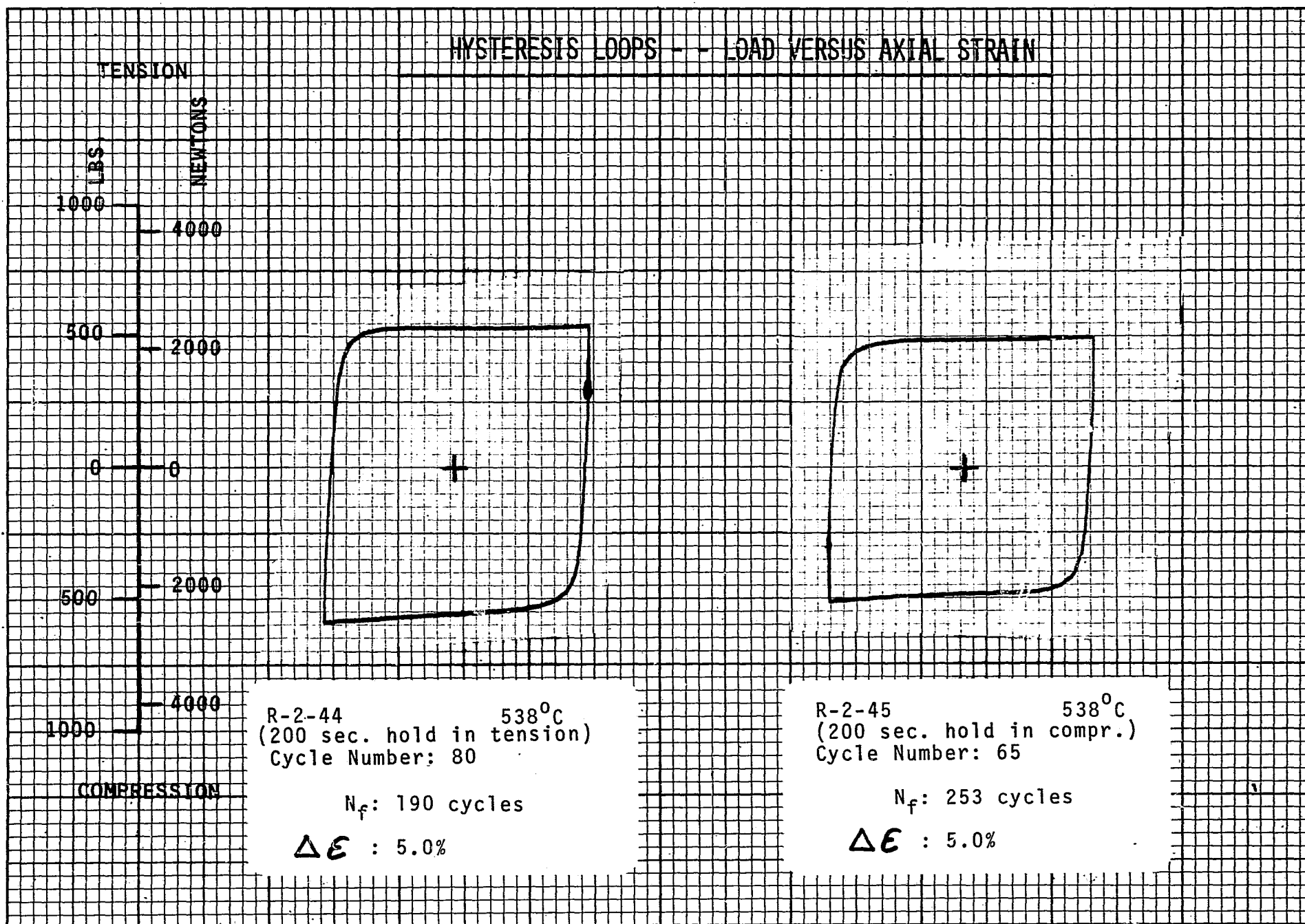


Figure 240

Figure 241

HYSTERESIS LOOPS - - LOAD VERSUS AXIAL STRAIN

TENSION

LBS

NEWTONS

1000

500

0

500

1000

4000

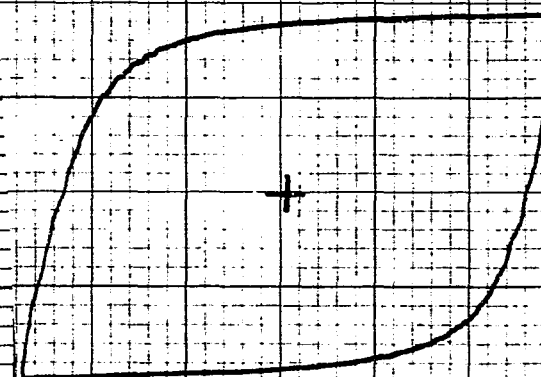
2000

2000

4000

R-2-46 538°C
(200 sec. hold in compr.)
Cycle Number: 89

N_f : 262 cycles
 ΔE : 5.0%



R-2-47 538°C
(56 sec. hold in tension)
Cycle Number: 163

N_f : 1152 cycles
 ΔE : 1.4%

COMPRESSION

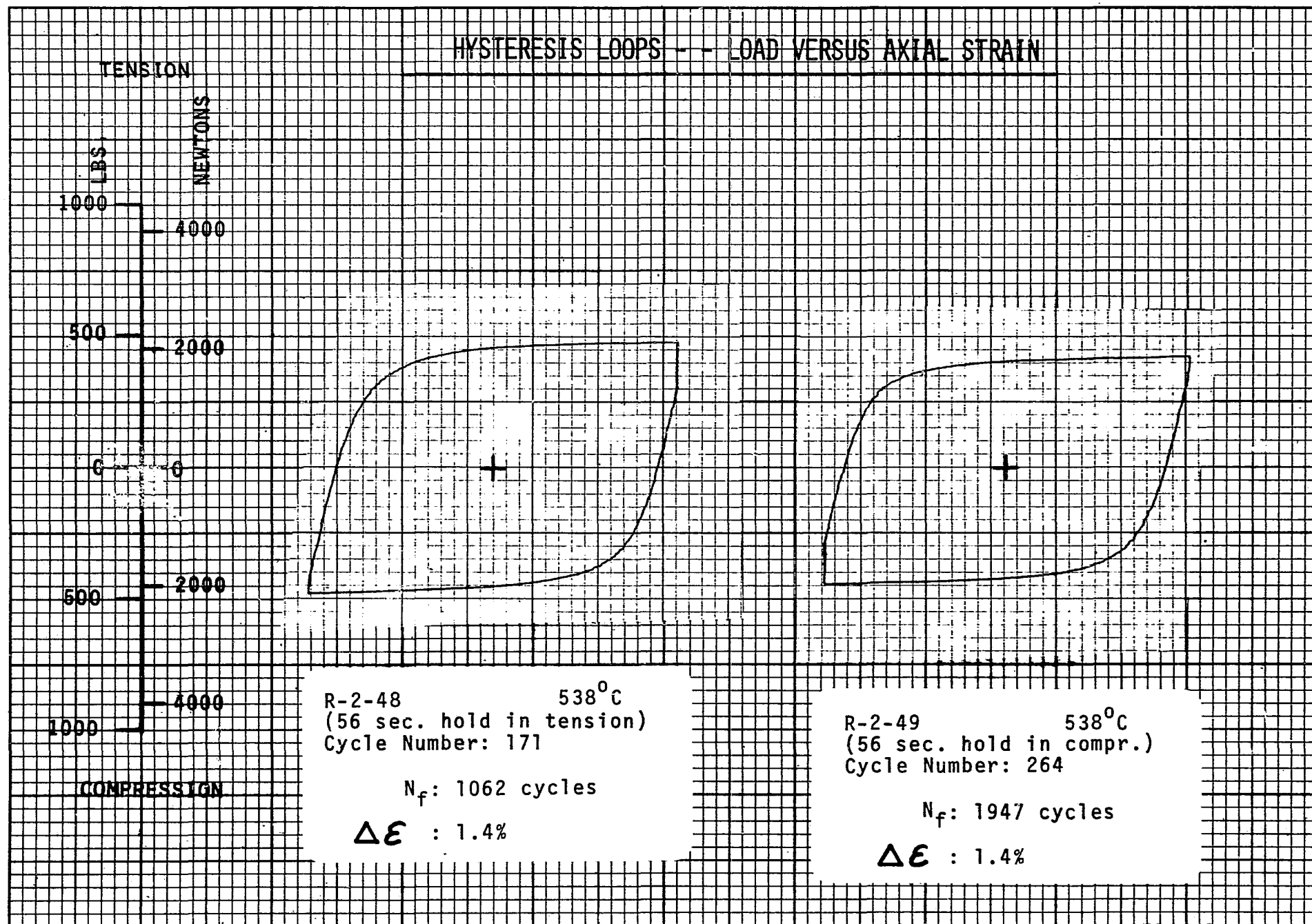
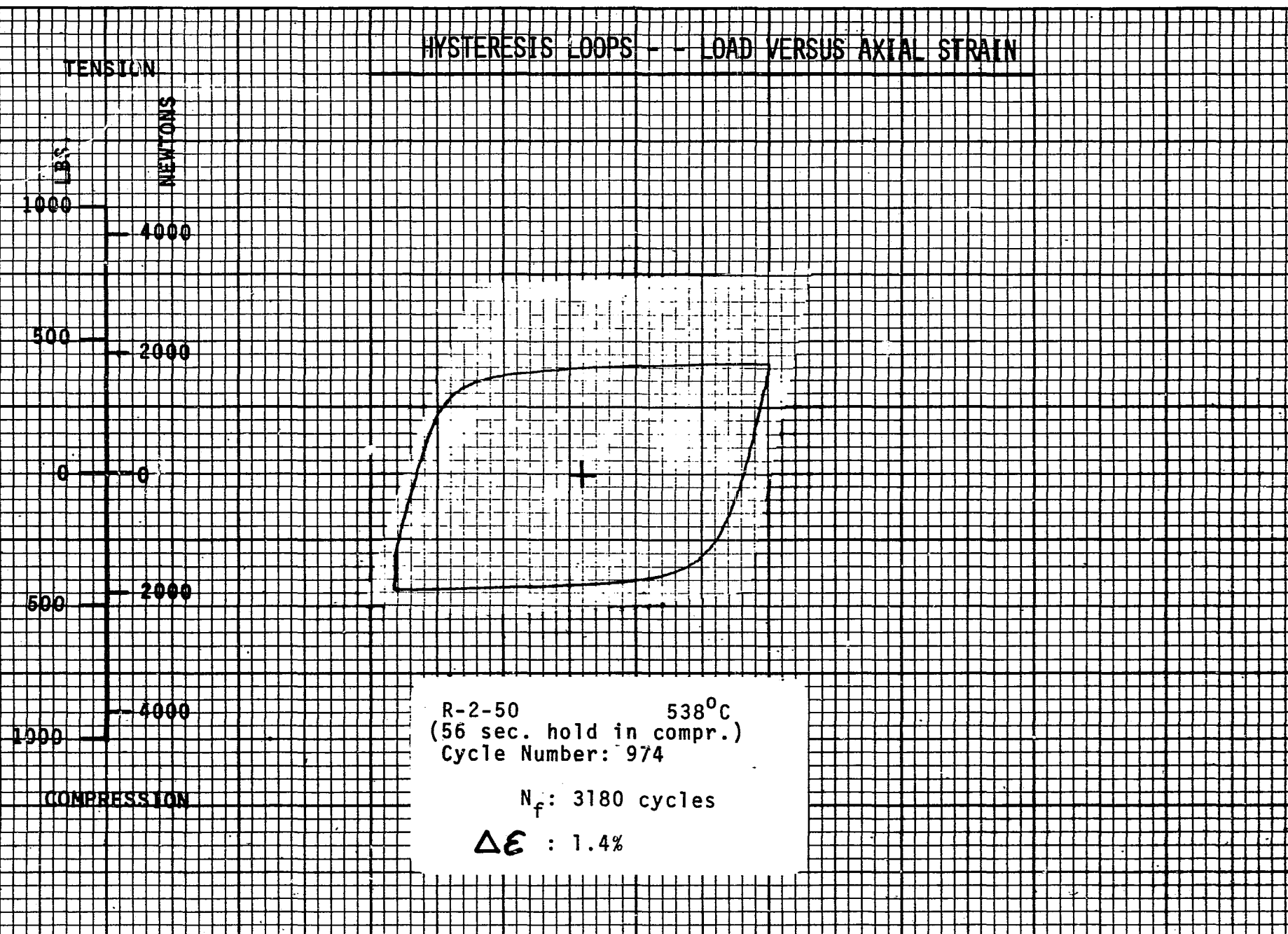


Figure 242



HYSTERESIS LOOPS - - LOAD VERSUS AXIAL STRAIN

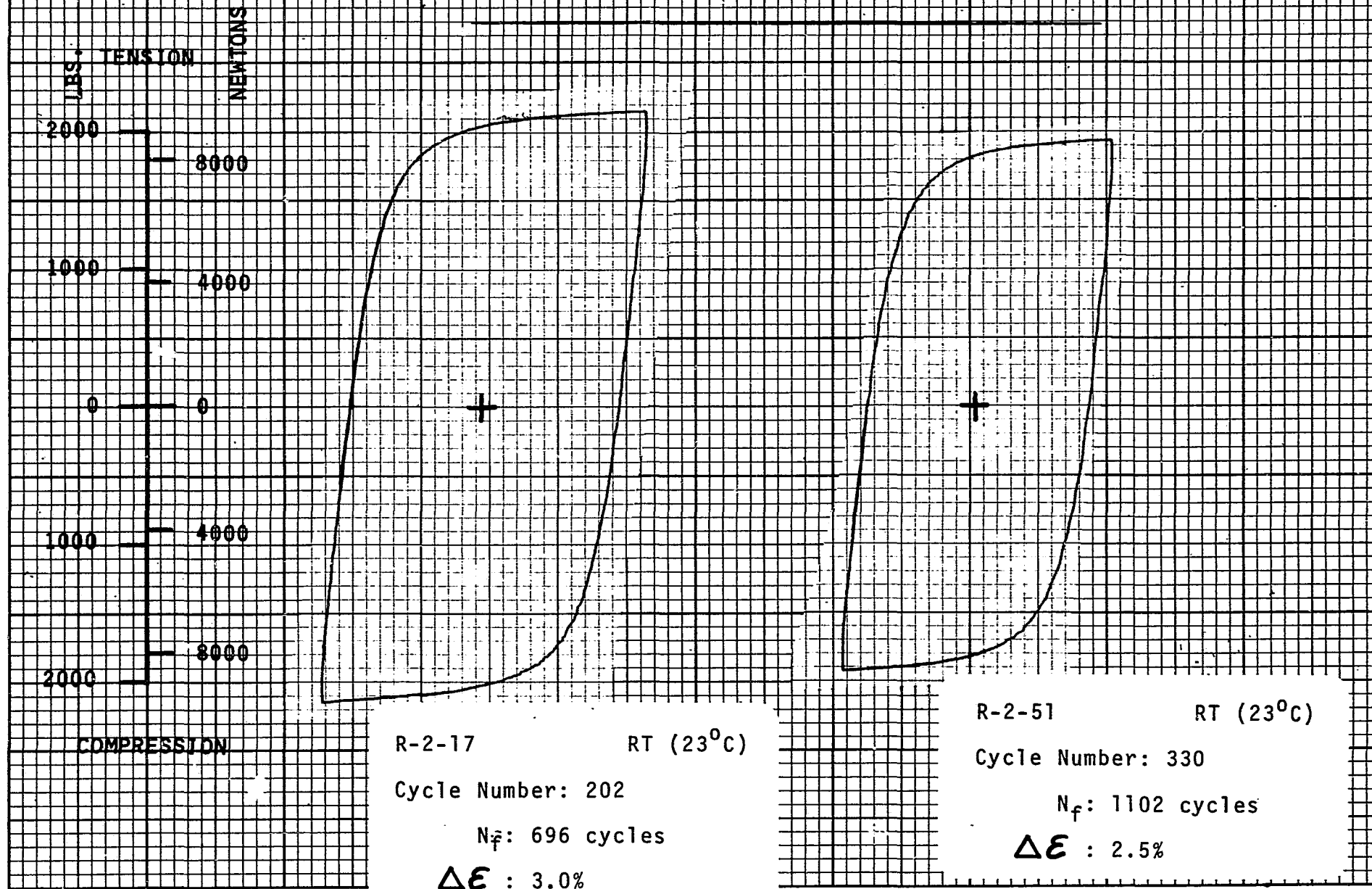
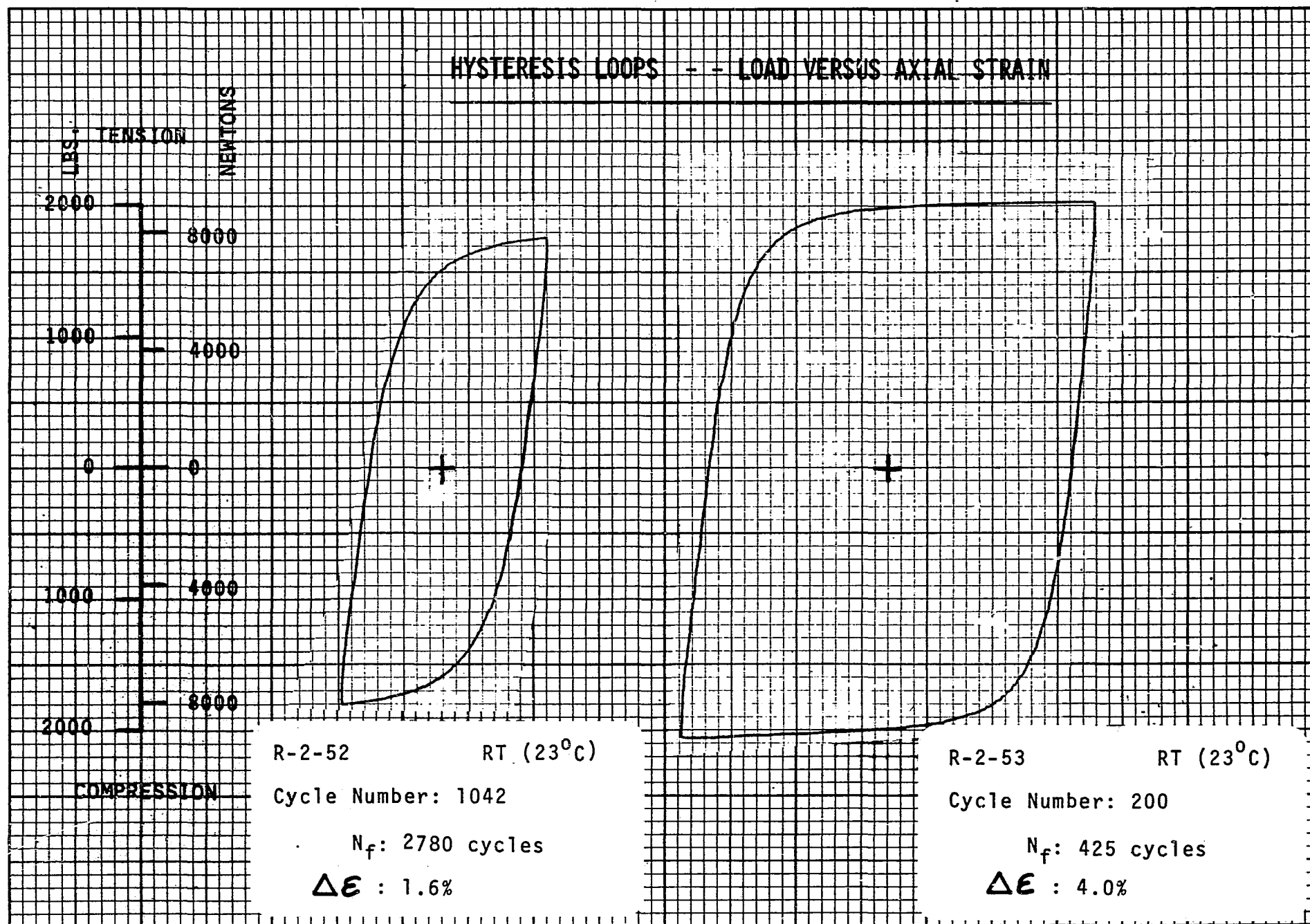


Figure 244



HYSTERESIS LOOPS - - LOAD VERSUS AXIAL STRAIN

TENSION
LBS.
NEWTONS

2000

8000

1000

4000

0

0

1000

4000

2000

8000

COMPRESSION

R-2-54 RT(23°C)

Cycle Number 1600

N_f : 3283 cycles

$\Delta \epsilon$: 1.4%

R-2-55

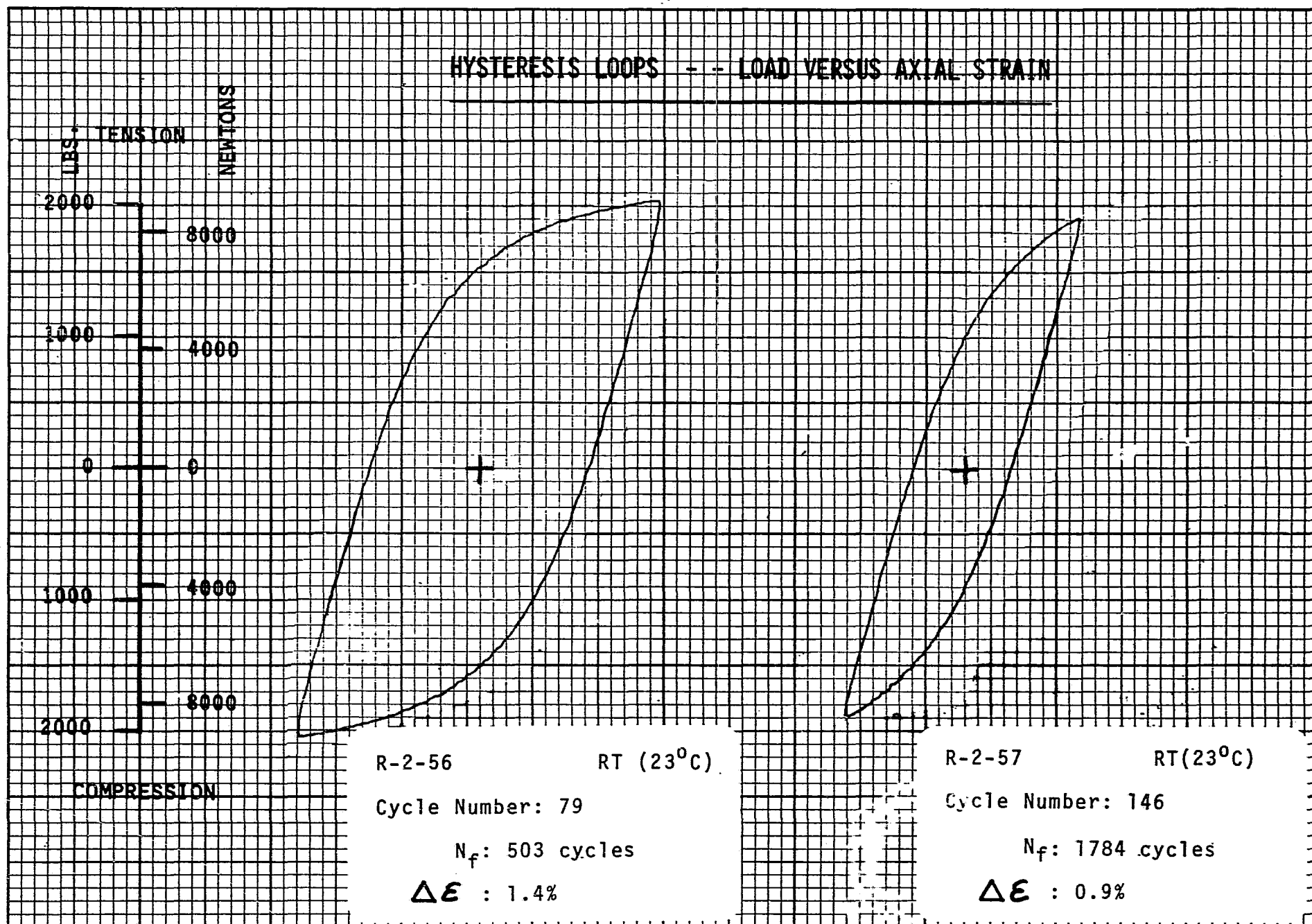
RT (23°C)

Cycle Number: 69

N_f : 141 cycles

$\Delta \epsilon$: 3.0%

Figure 246



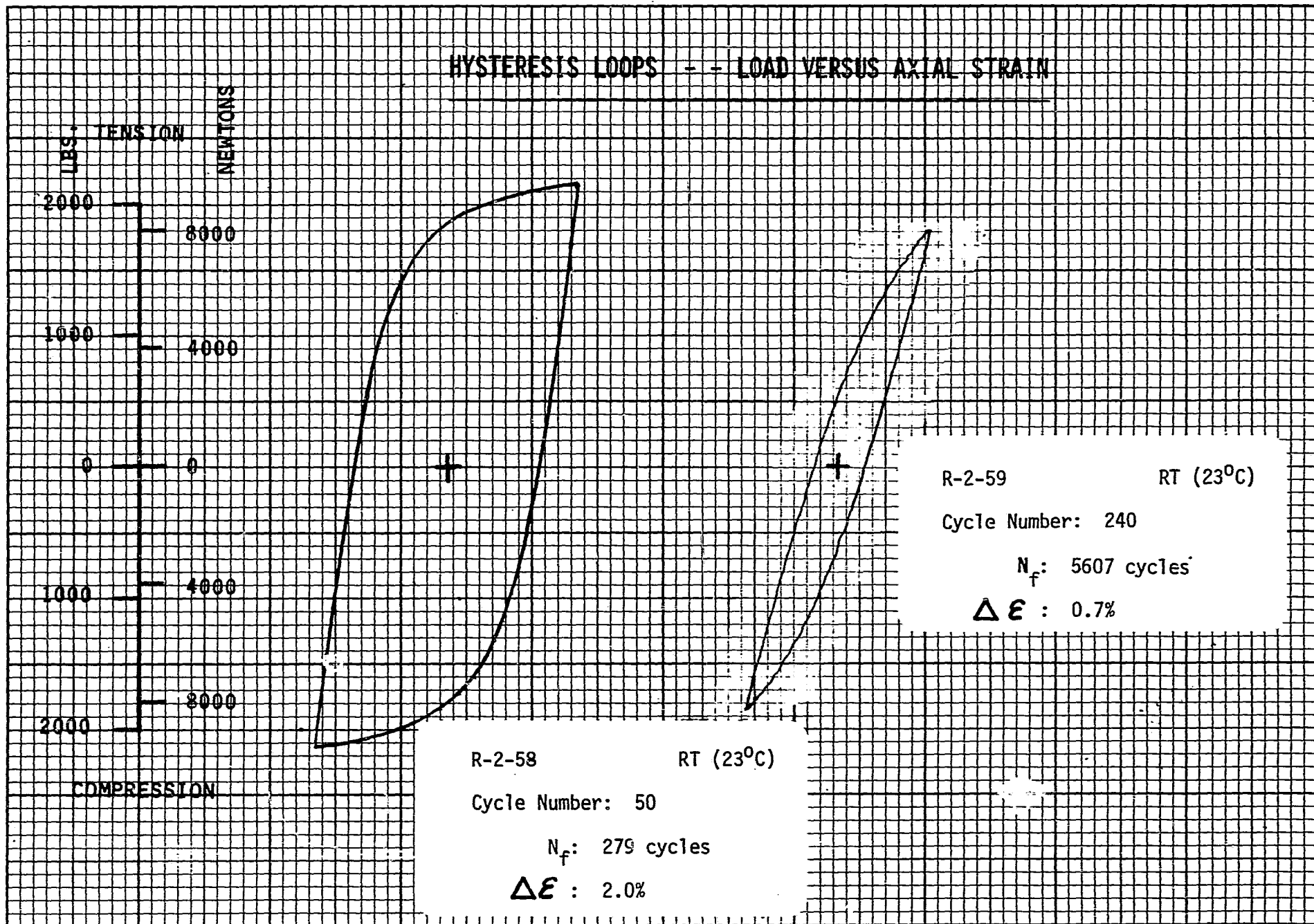
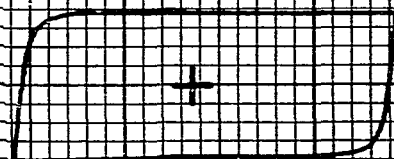


Figure 248

HYSTERESIS LOOPS - - LOAD VERSUS AXIAL STRAIN

TENSION
LBS
2000
1000
0
1000
2000
COMPRESSION

NEWTONS
8000
4000
0
4000
8000



R-20-1 538°C

Cycle Number: 113

 N_f : 390 cycles $\Delta \epsilon$: 5.0%

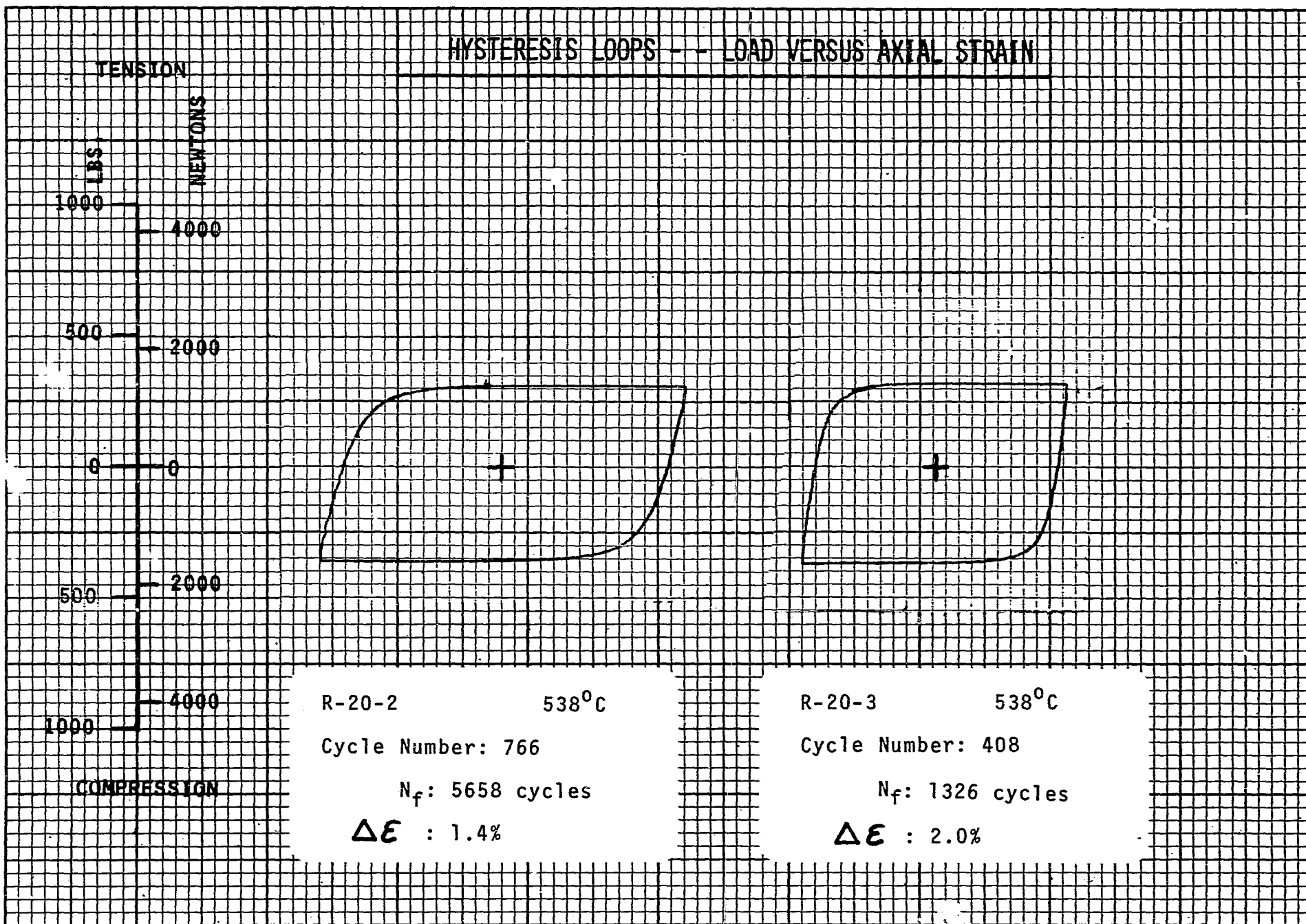
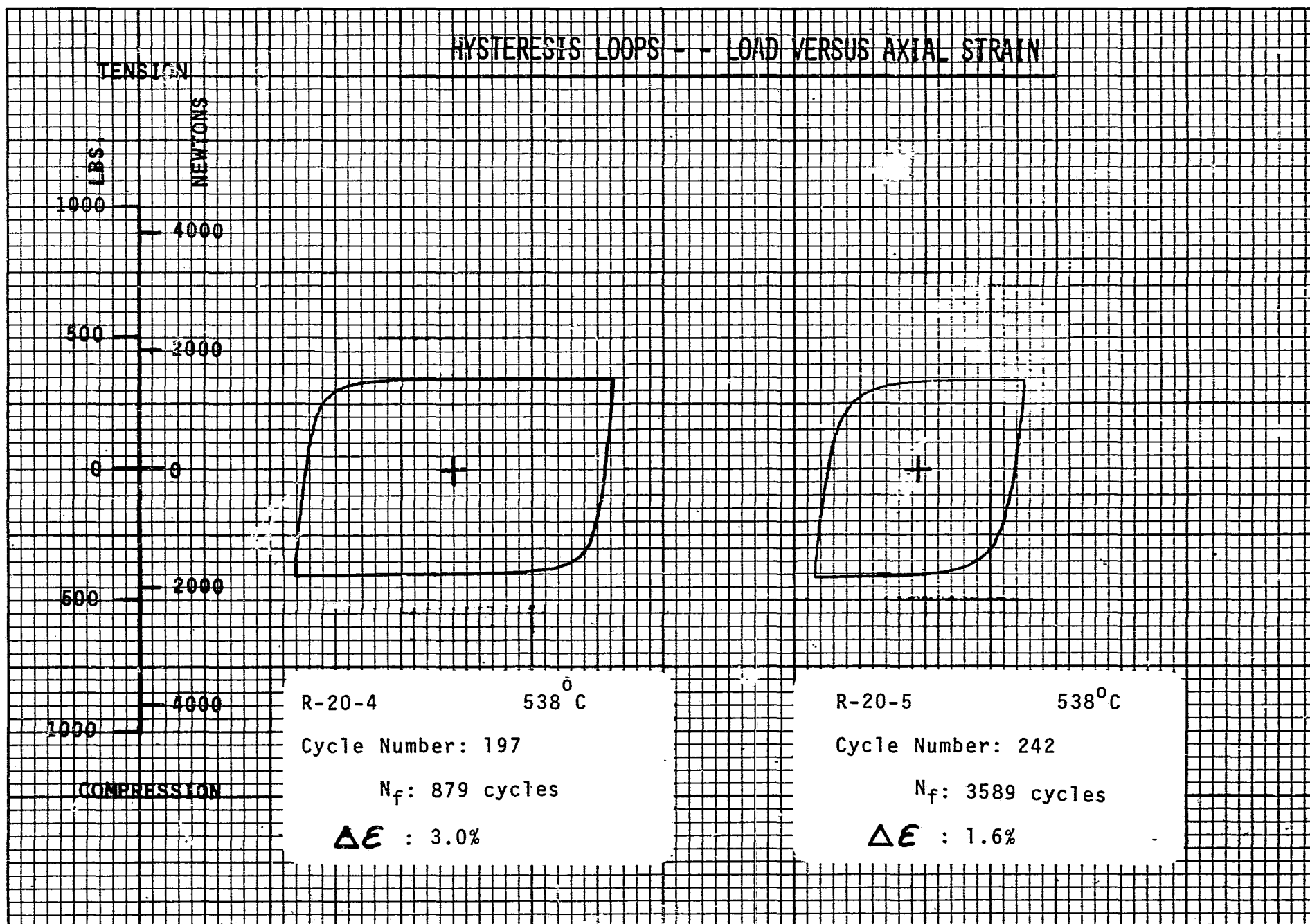


Figure 250



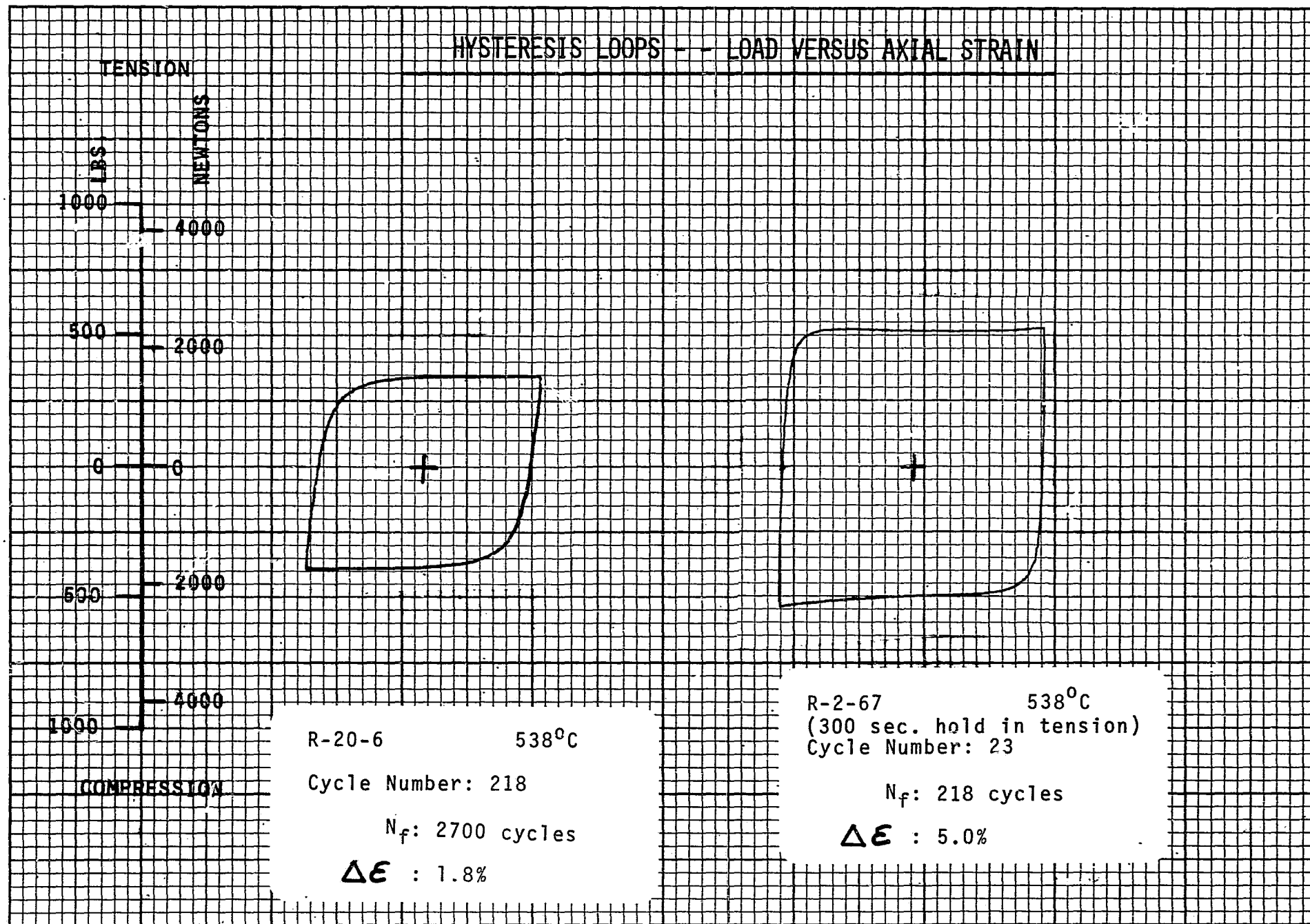
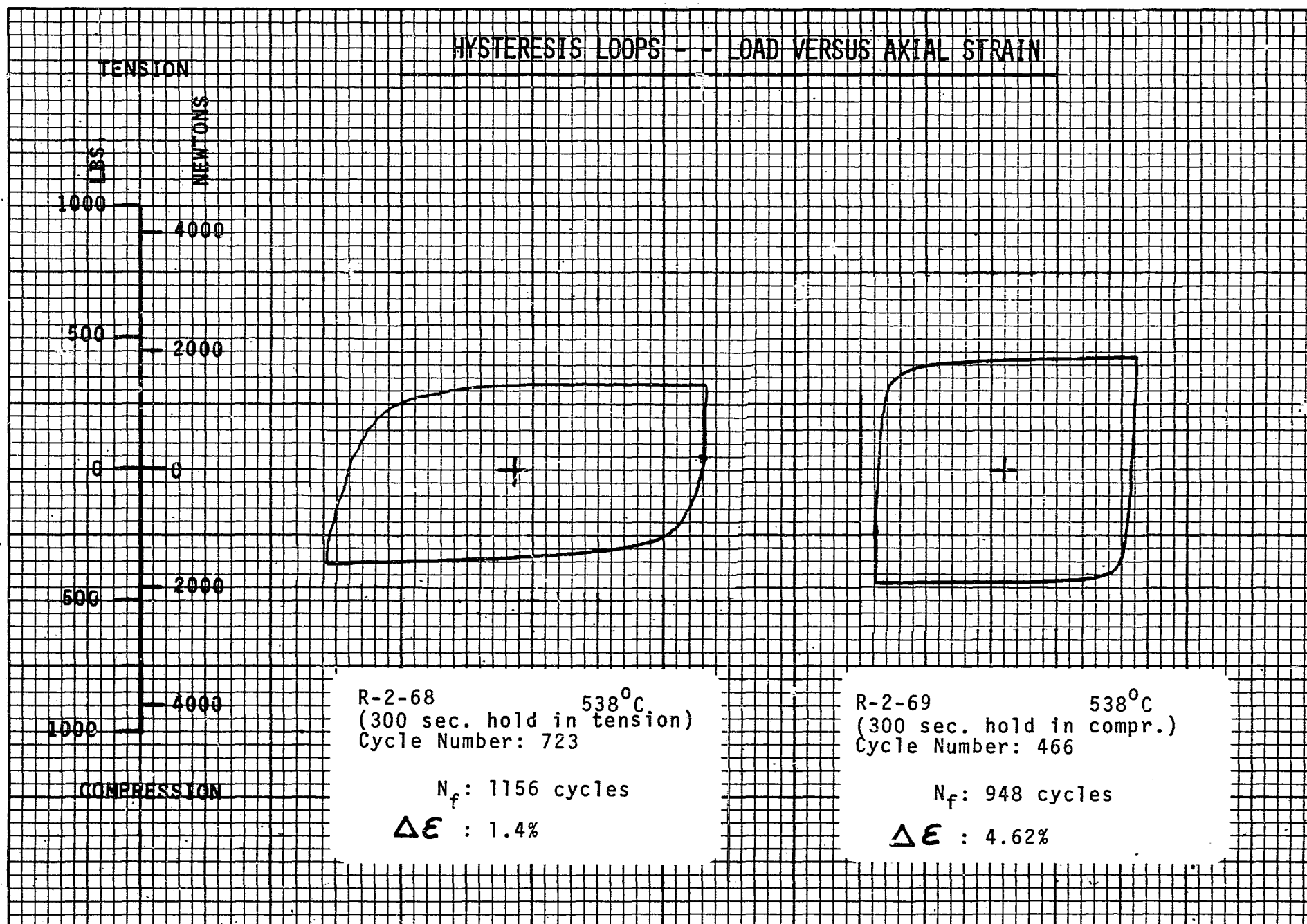


Figure 252



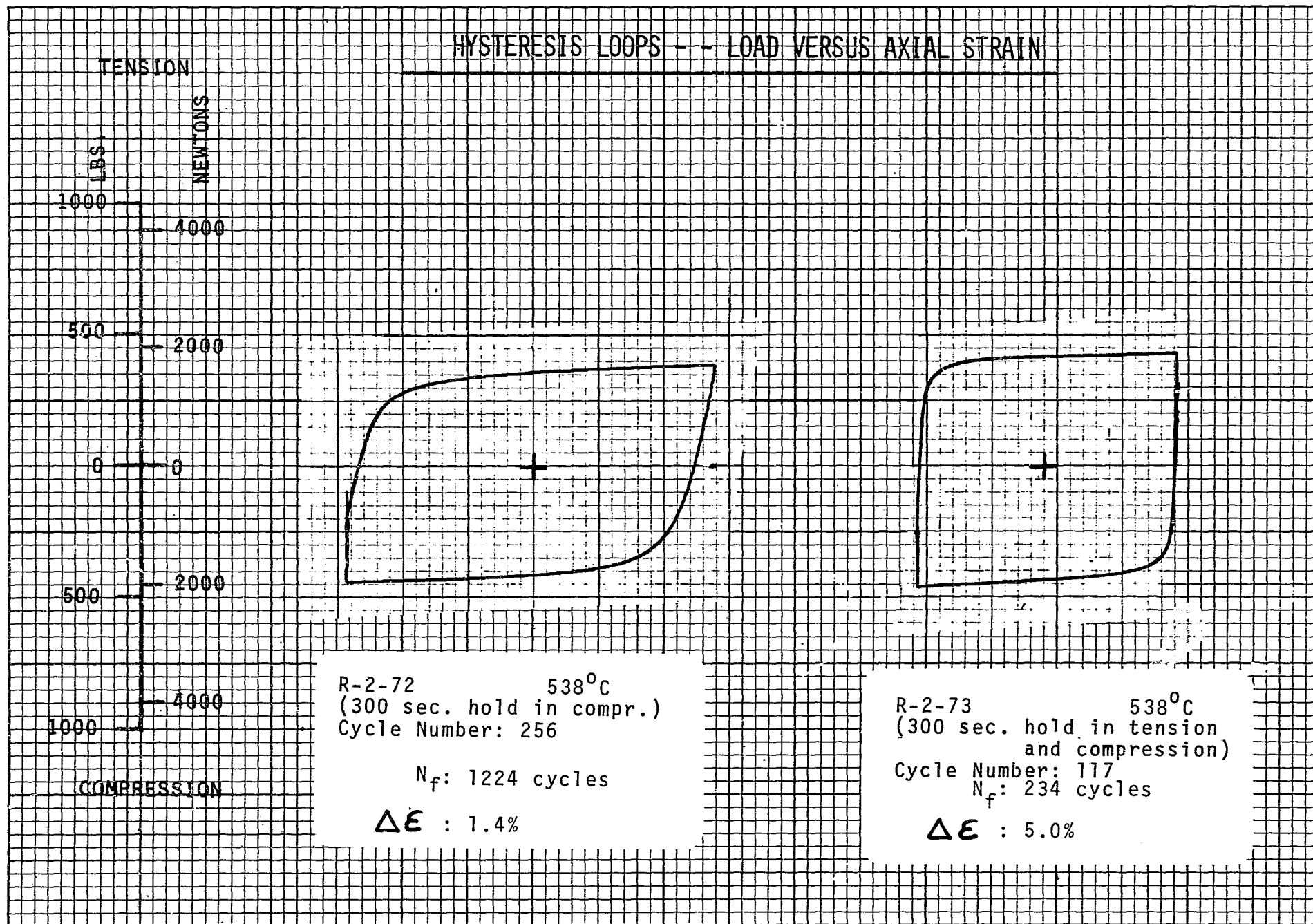
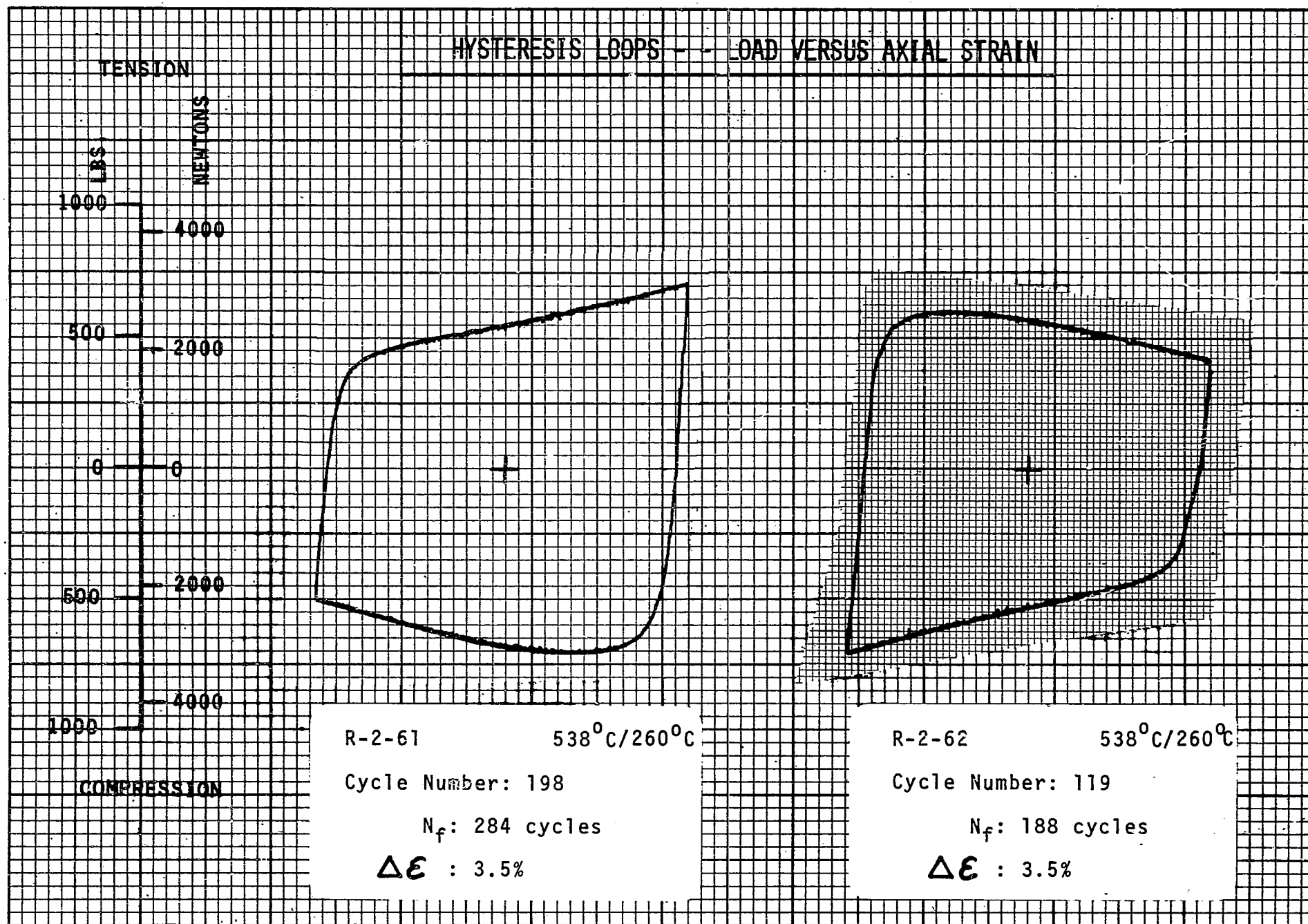


Figure 254



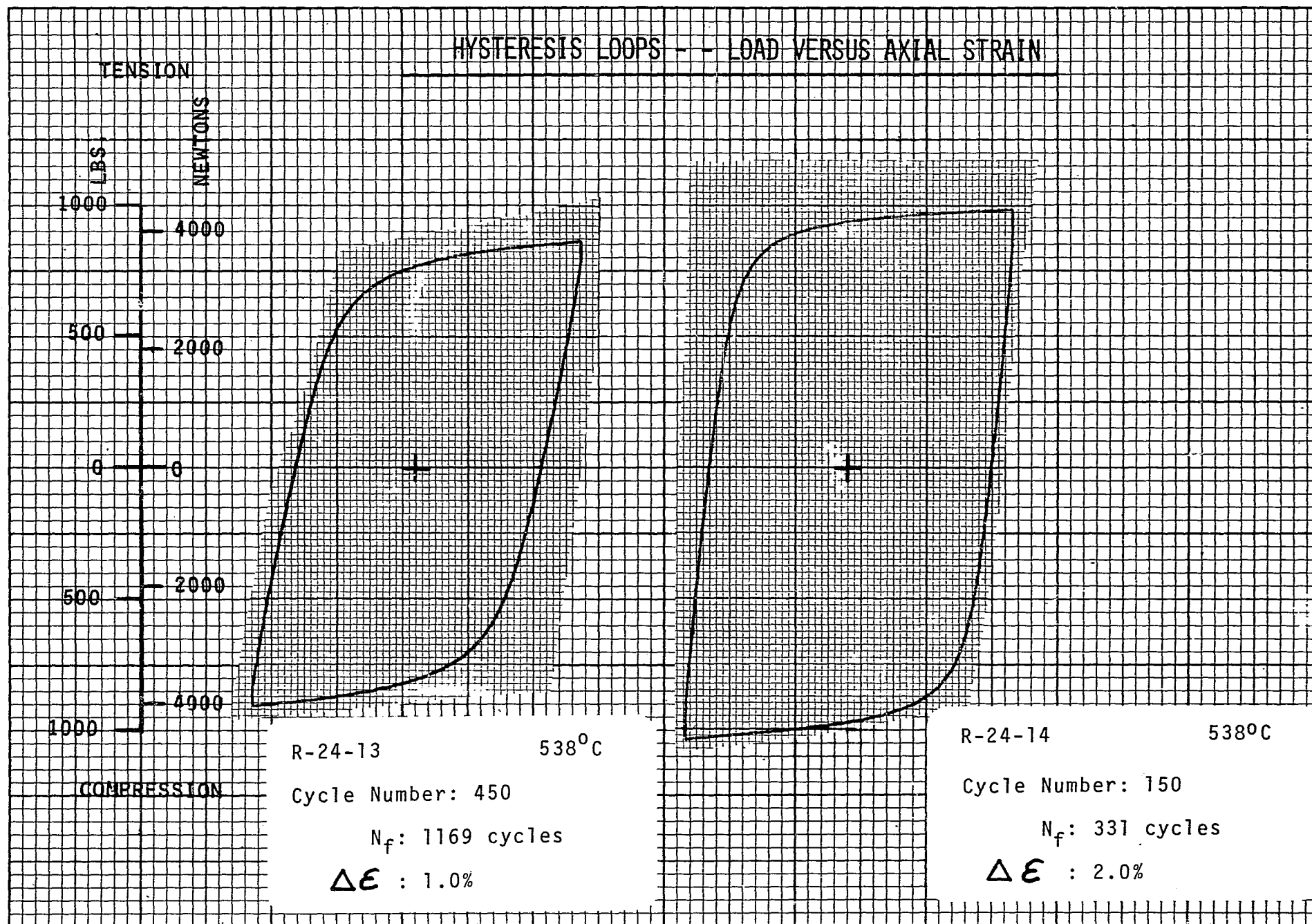
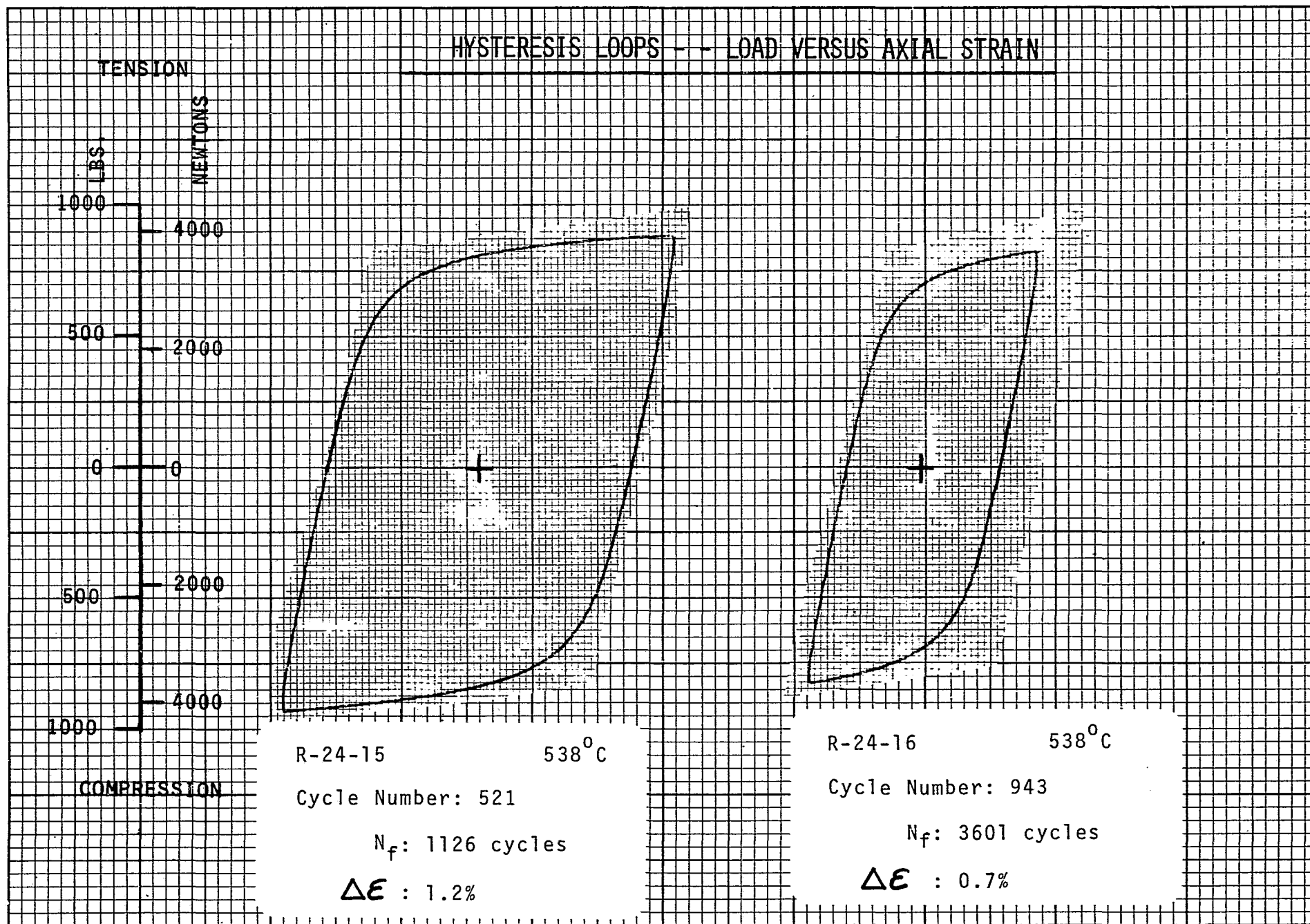


Figure 256



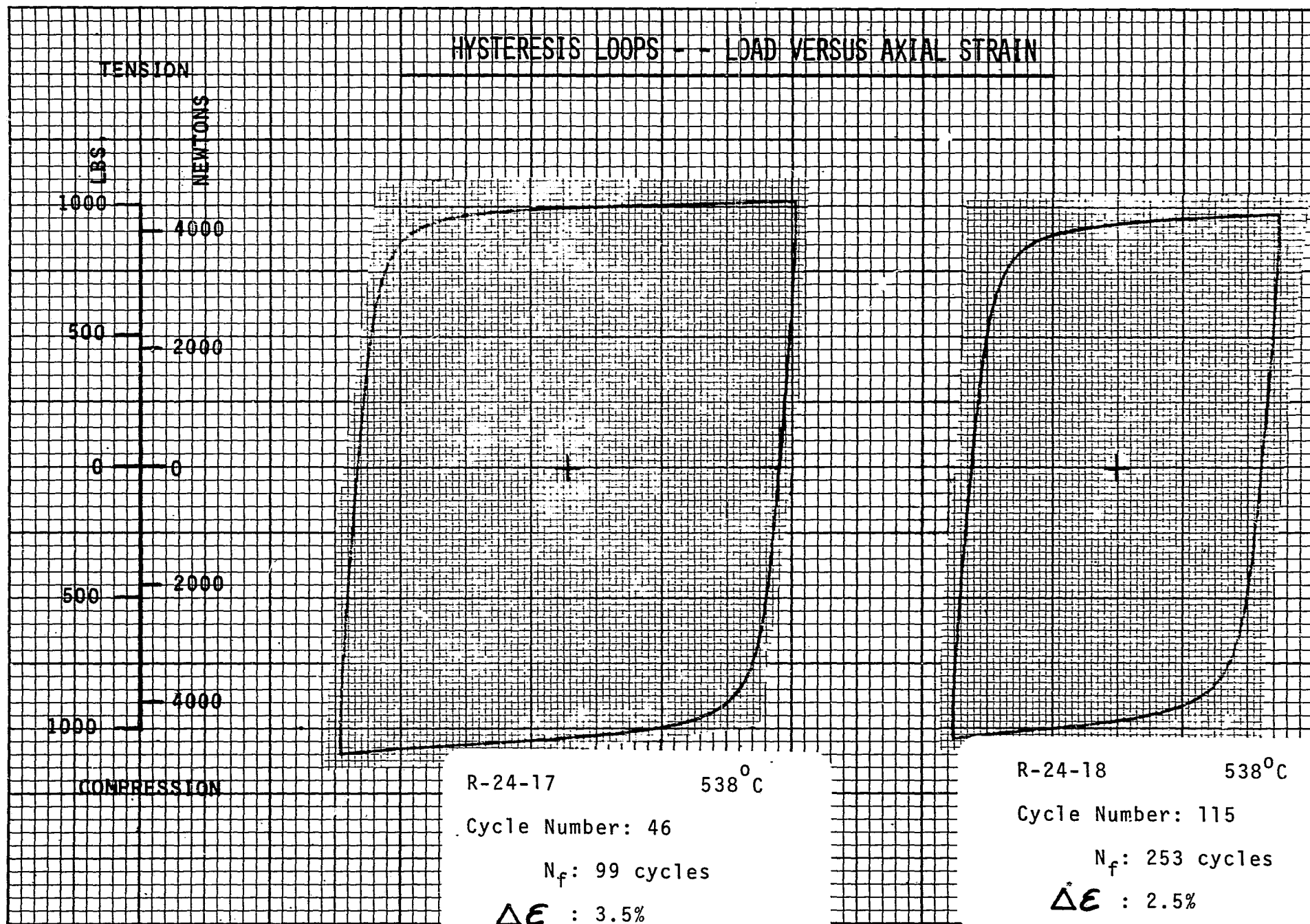
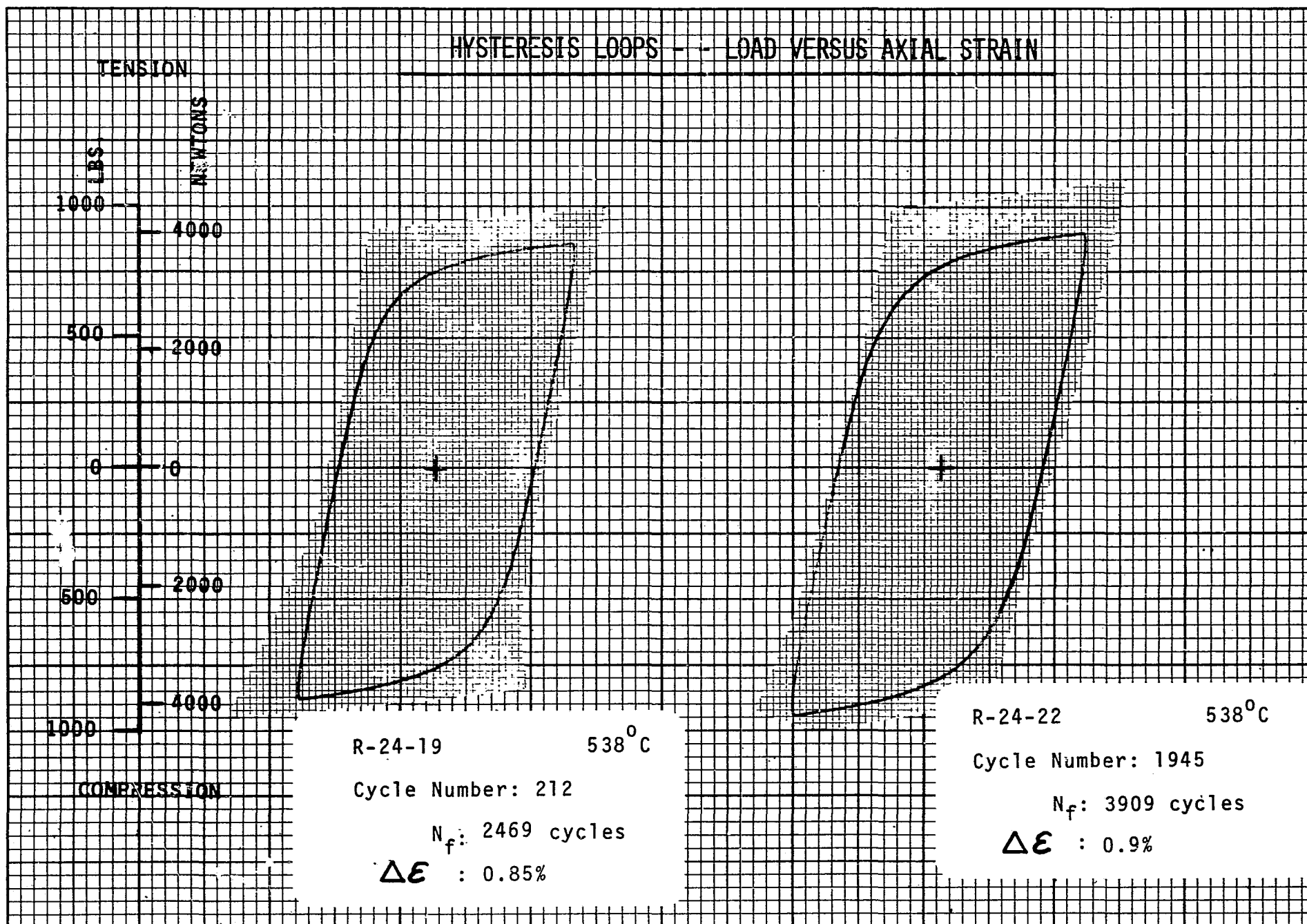


Figure 258



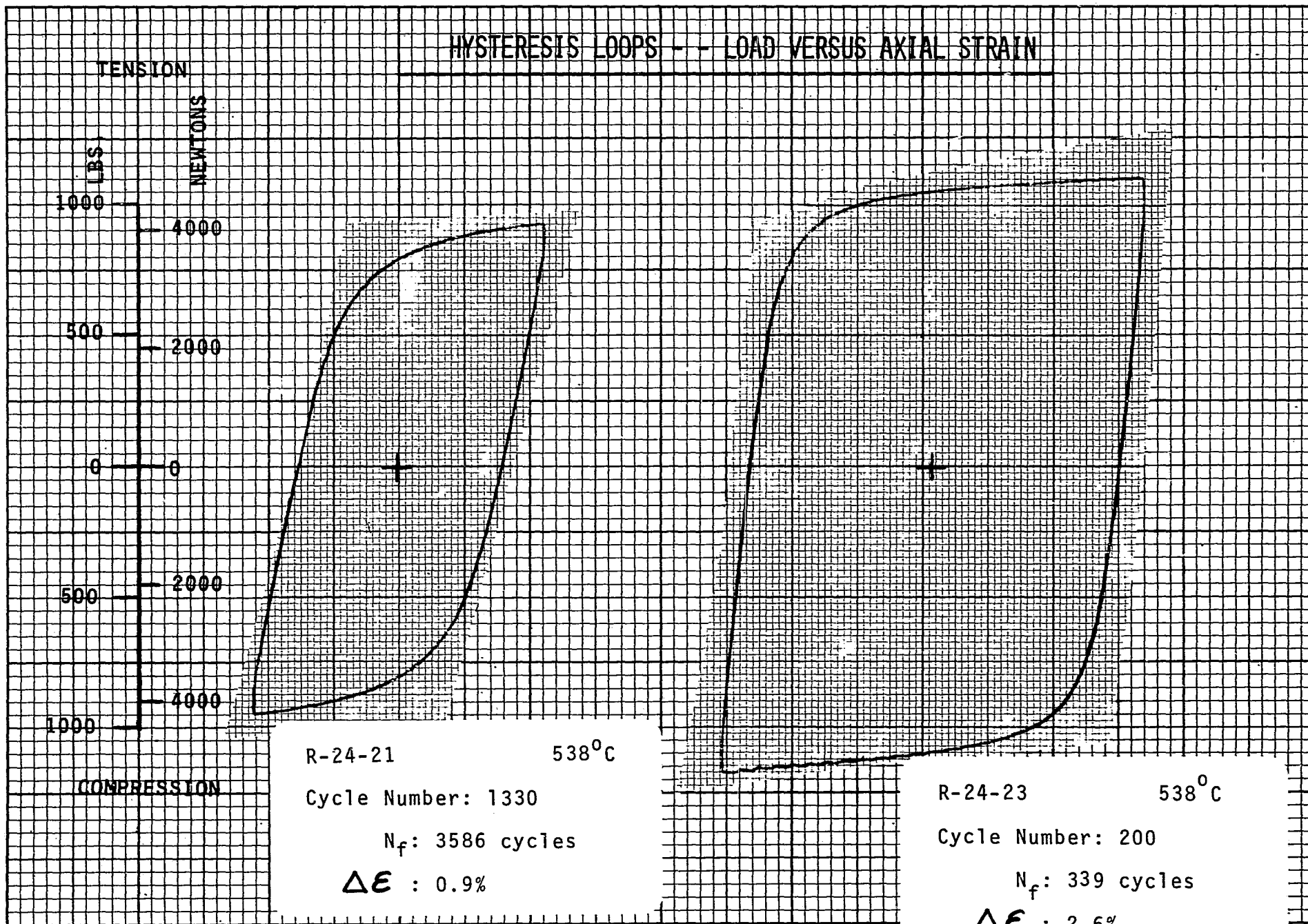
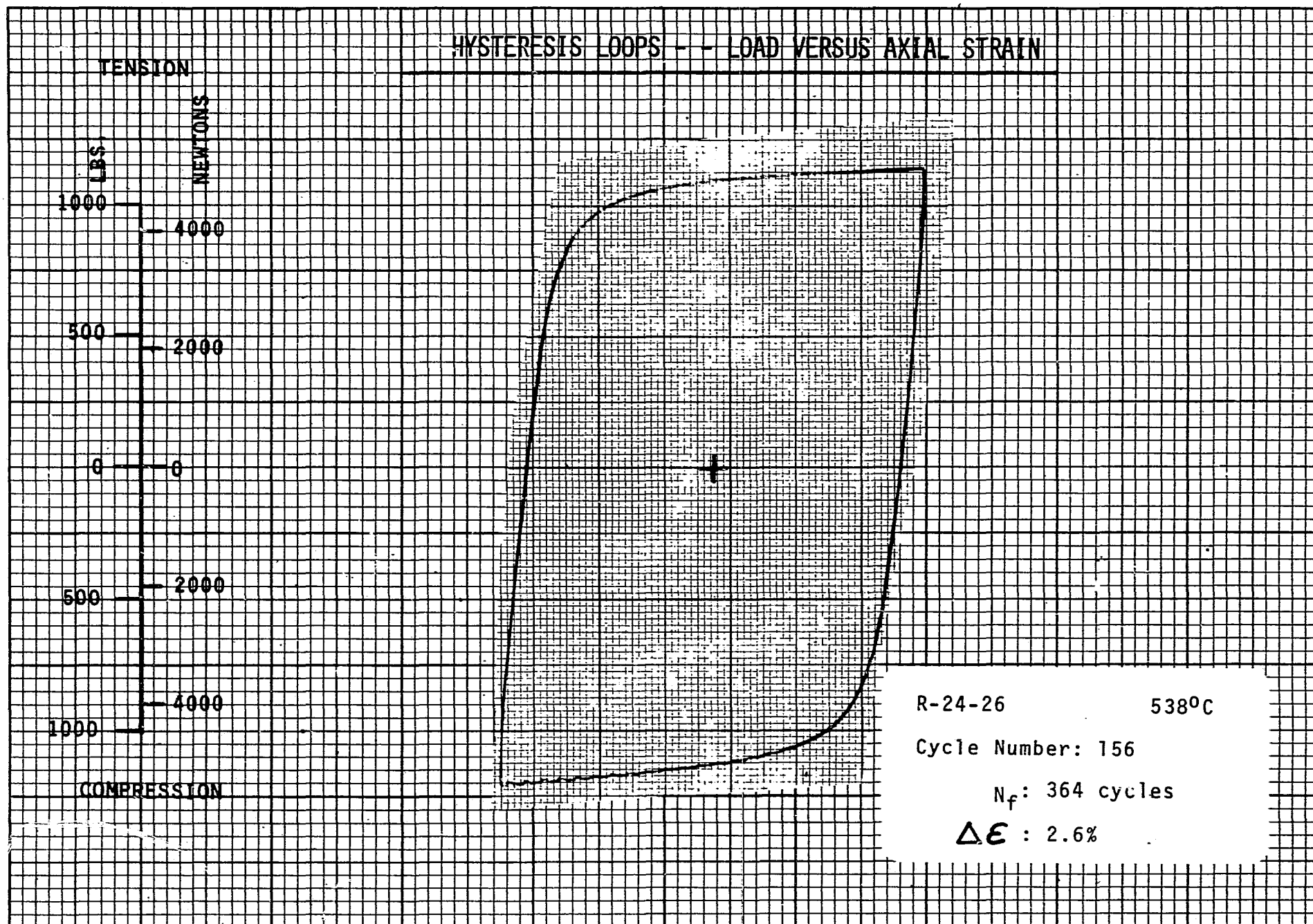


Figure 260



HYSTERESIS LOOPS -- LOAD VERSUS AXIAL STRAIN

TENSION

LBS

NEWTONS

1000

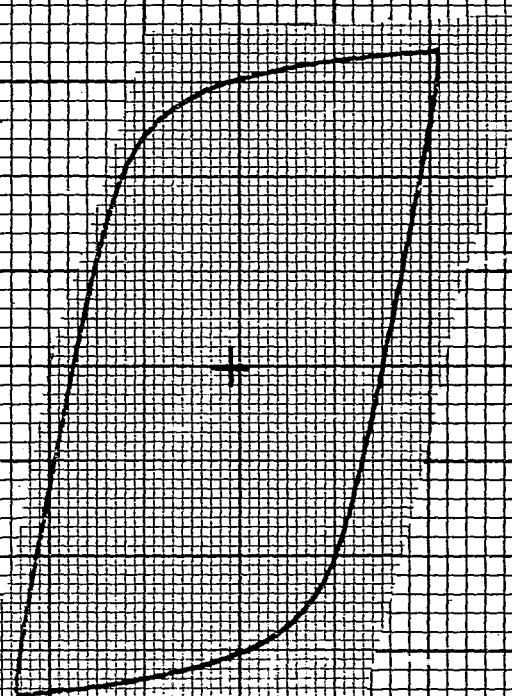
500

0

500

1000

COMPRESSION



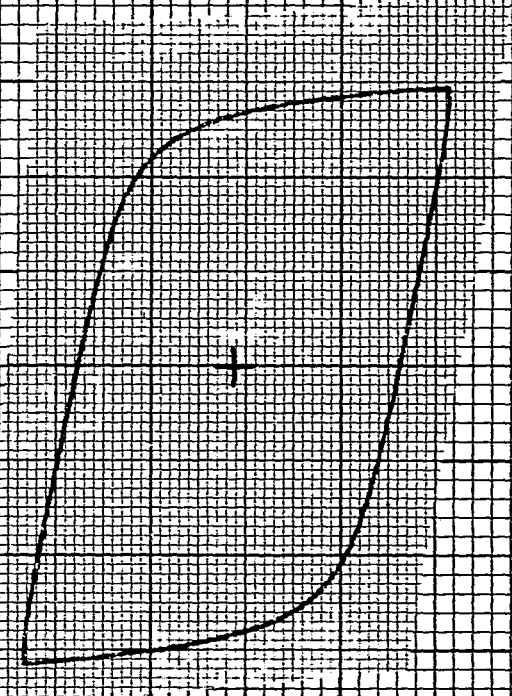
R-24-20

538°C

Cycle Number: 40

N_f : 1138 cycles

$\Delta \epsilon$: 0.9%



R-24-25

538°C

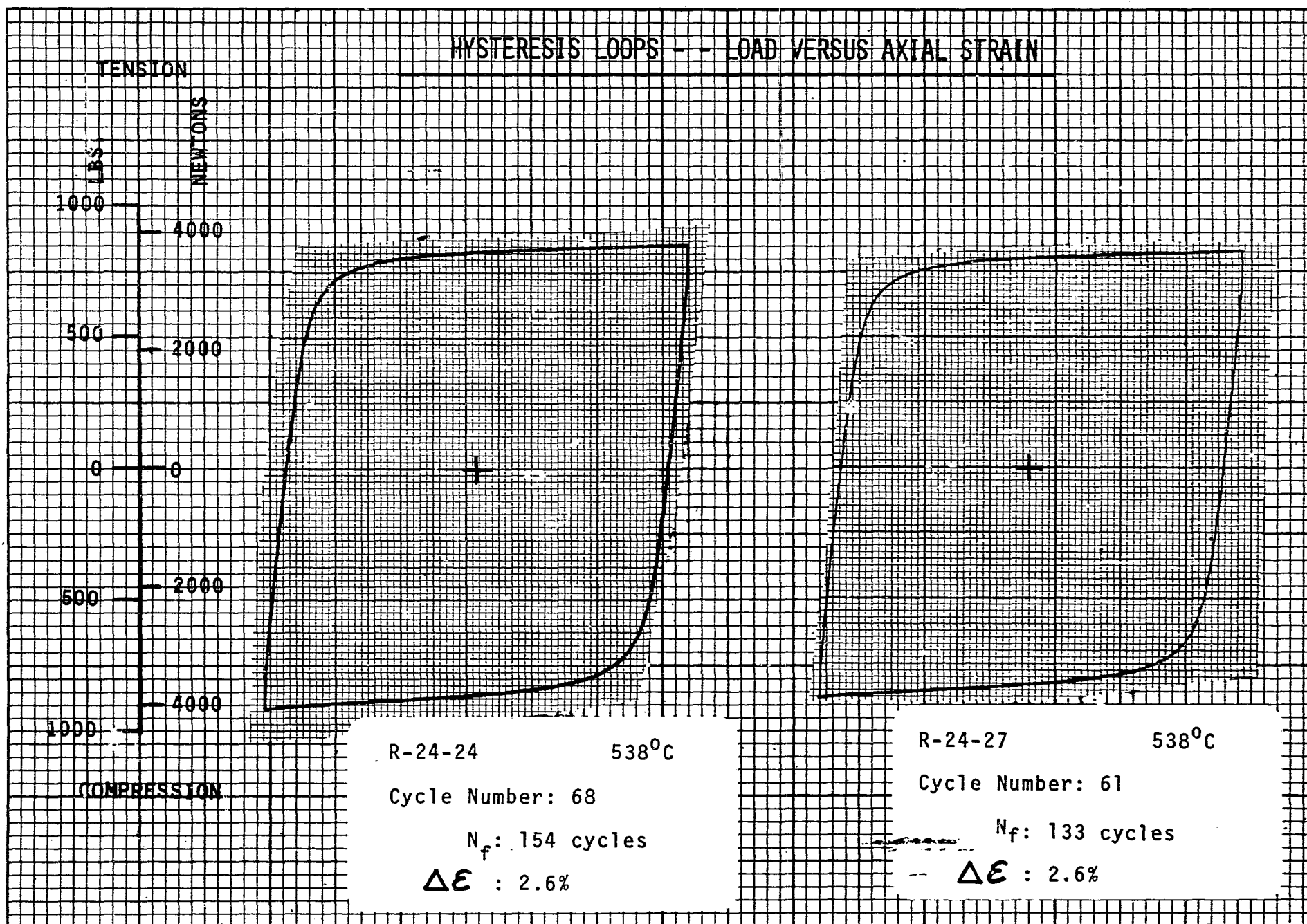
Cycle Number: 314

N_f : 1196 cycles

$\Delta \epsilon$: 0.9%

Figure 262

Figure 263



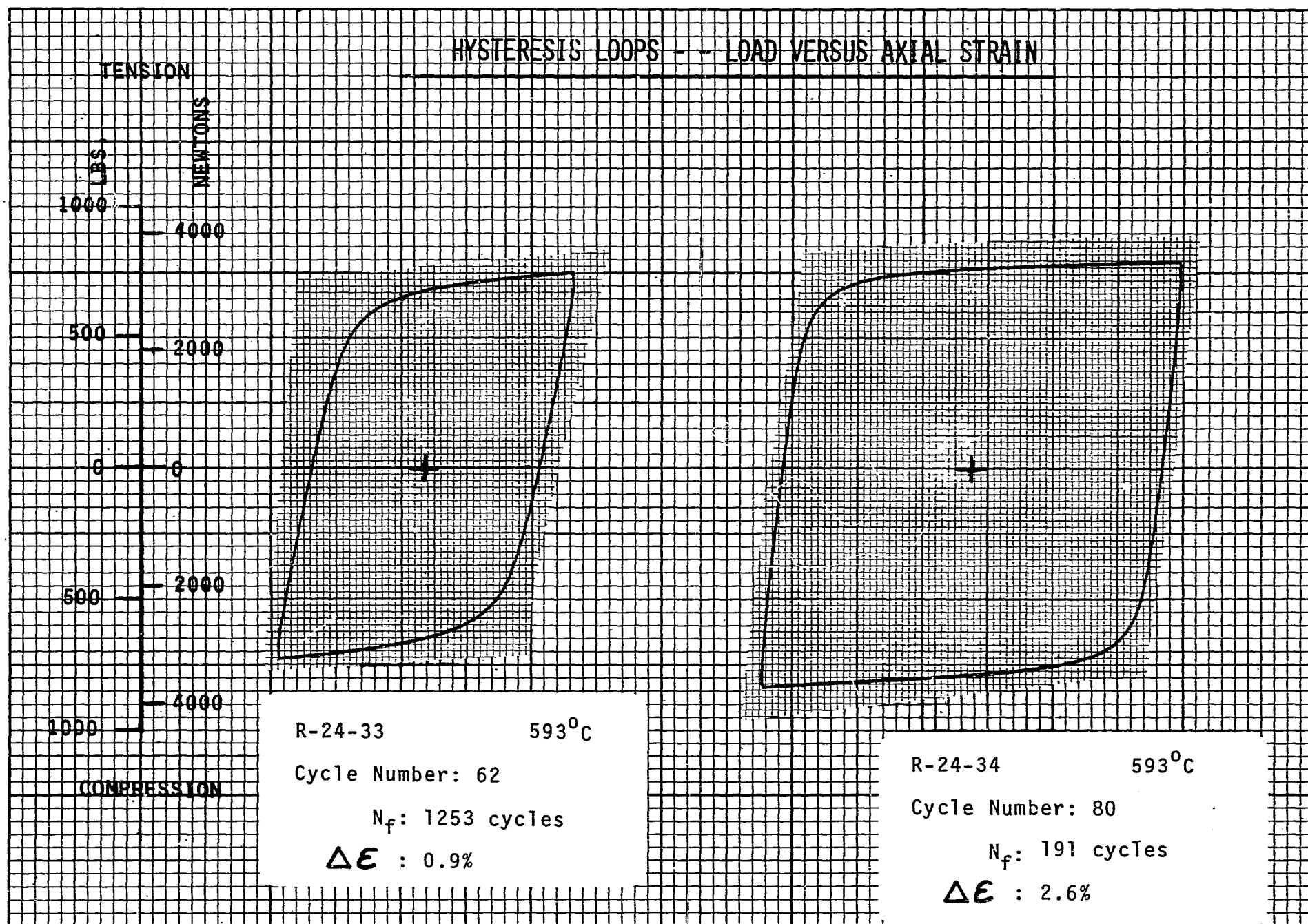
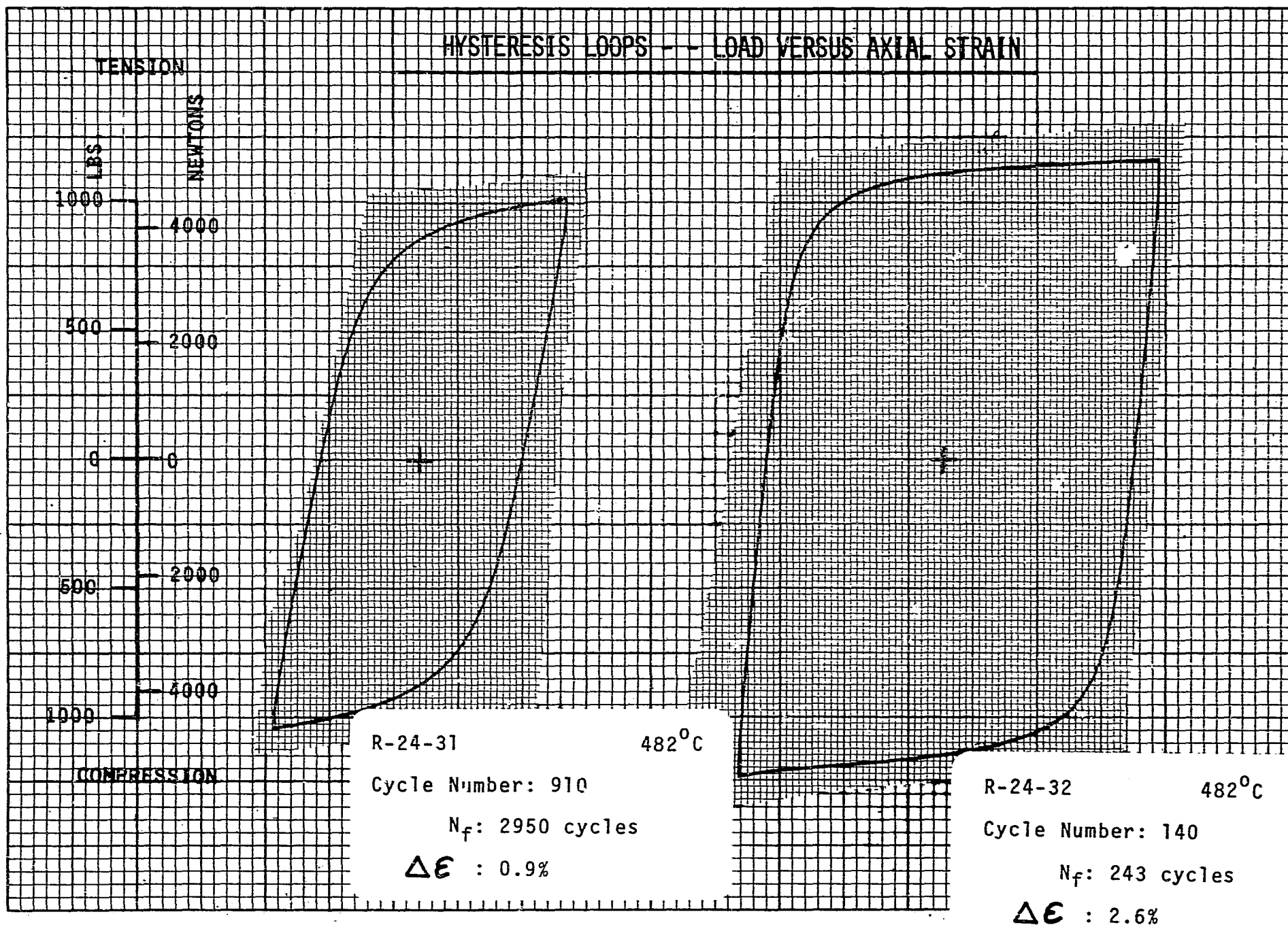


Figure 264



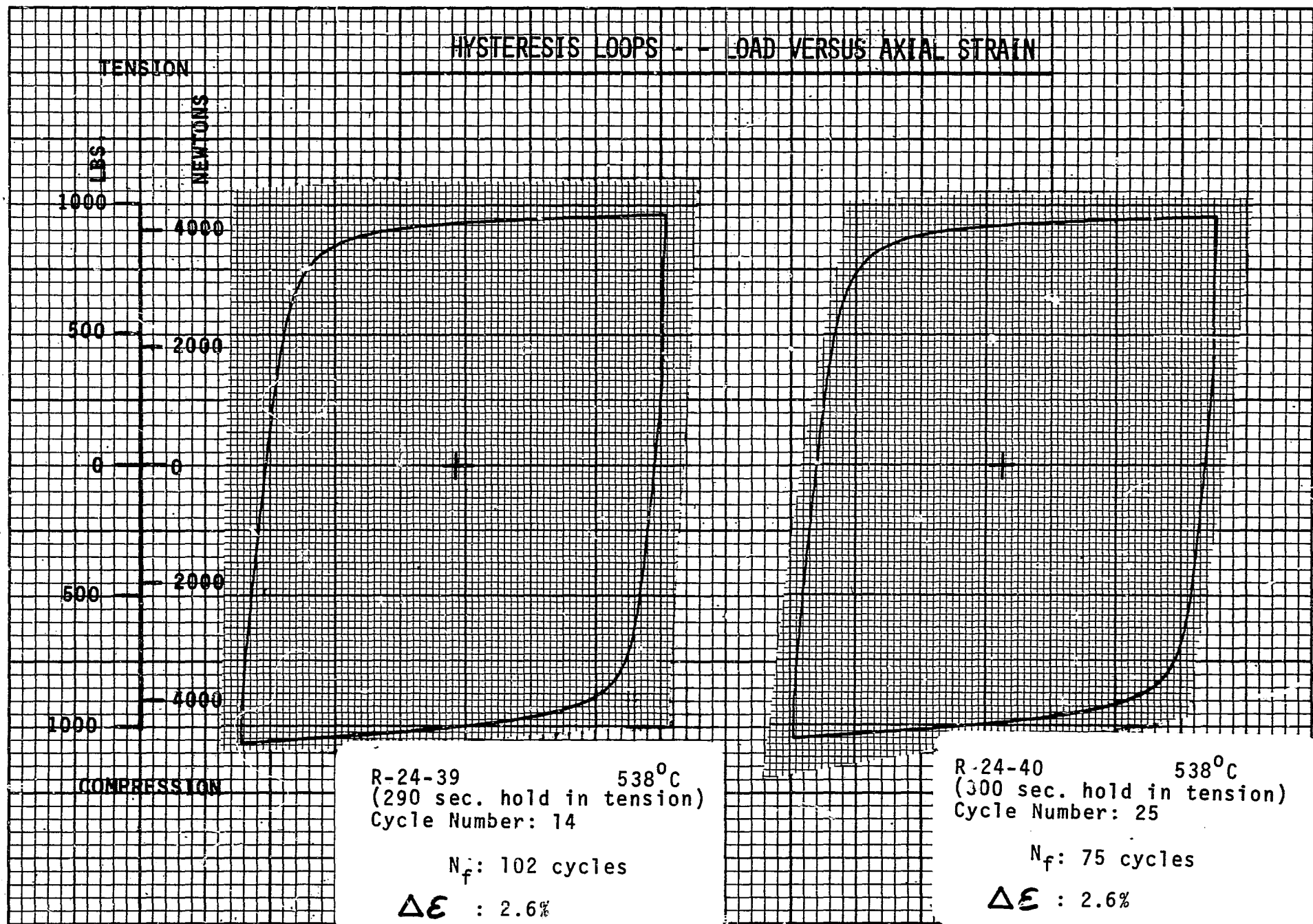
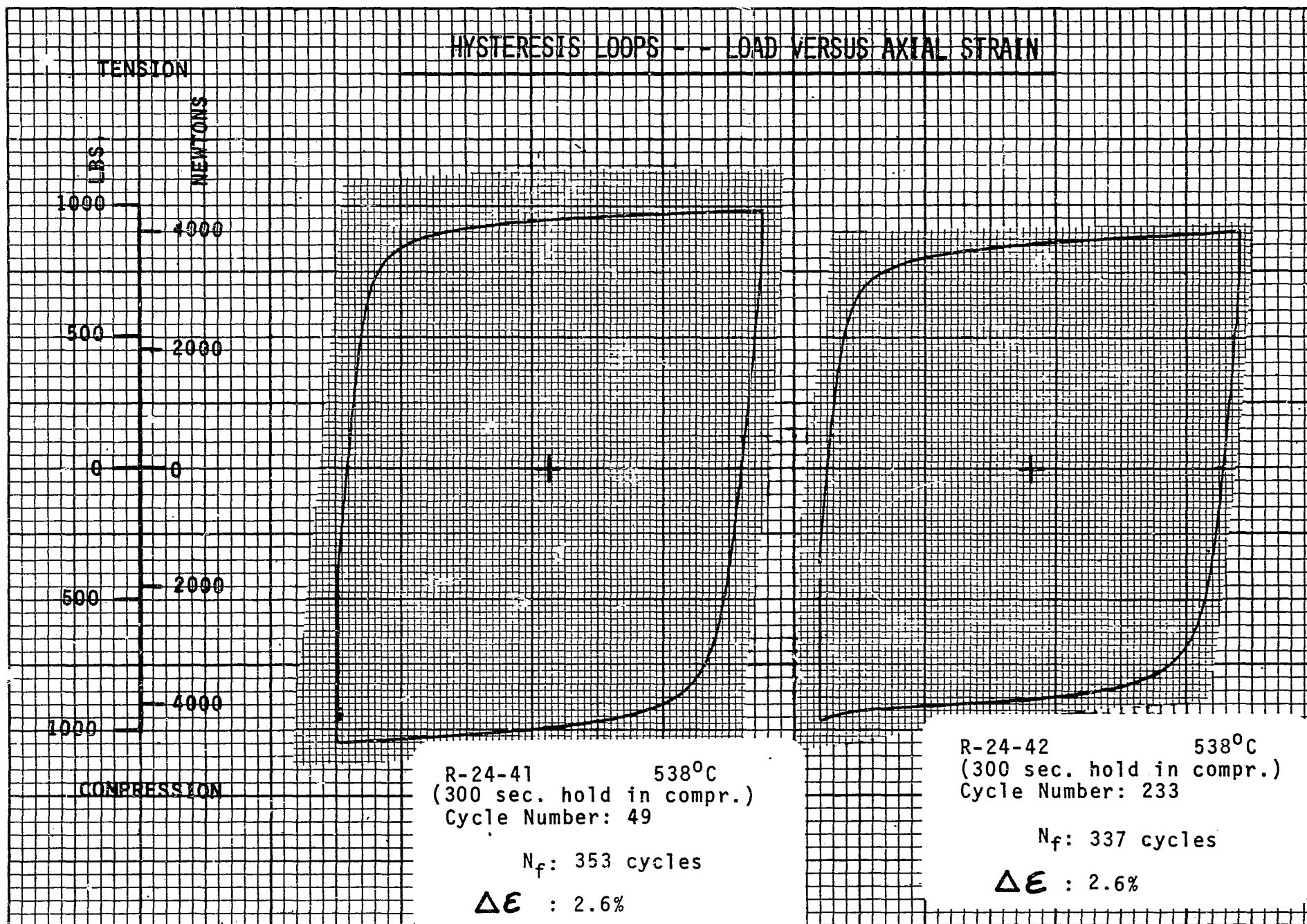


Figure 266



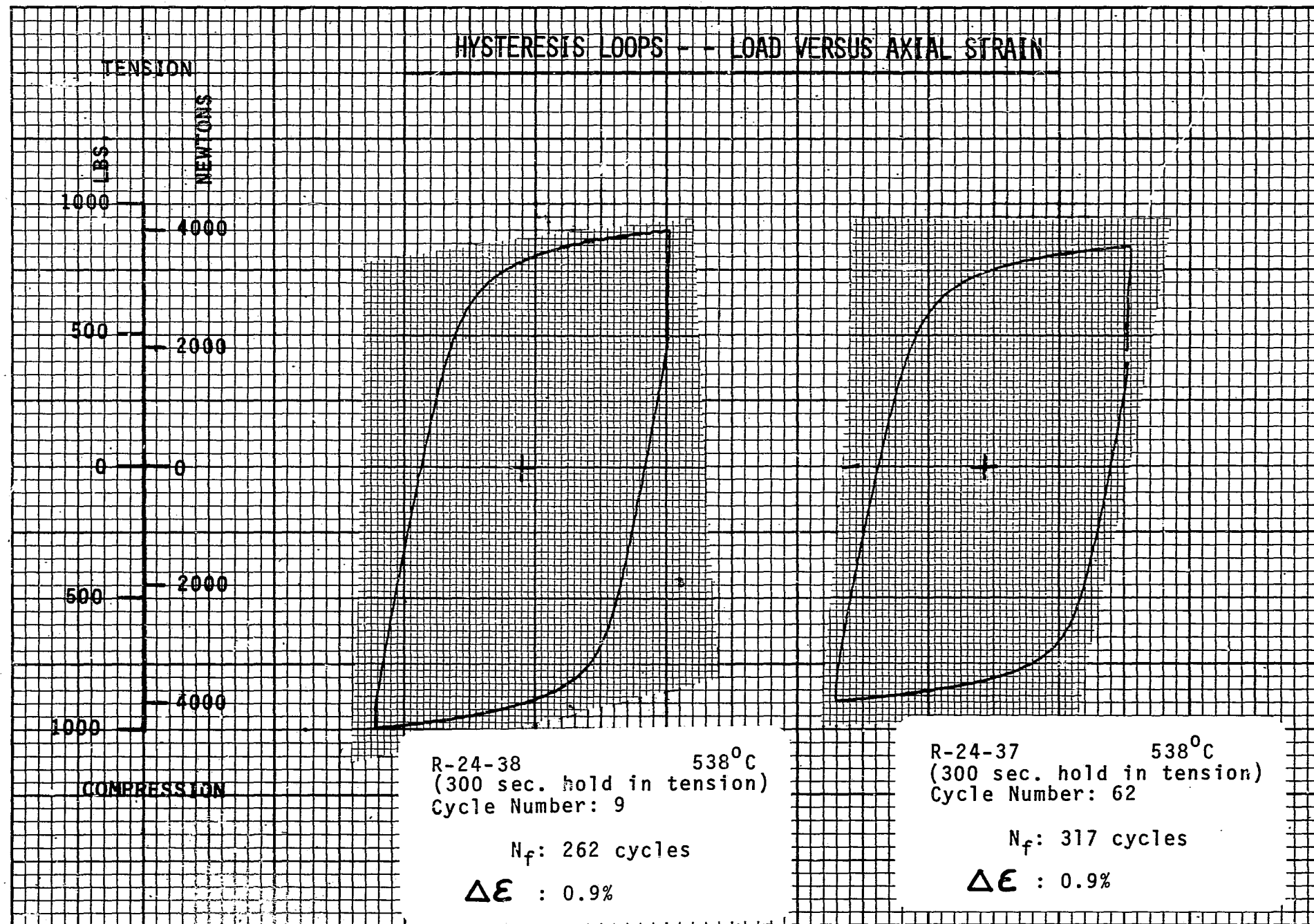


Figure 268

Figure 269

HYSTERESIS LOOPS -- LOAD VERSUS AXIAL STRAIN

TENSION

LBS

NEWTONS

1000

500

0

500

1000

4000

2000

0

2000

4000

COMPRESSION

R-24-43 538°C
 (300 sec. hold in compr.)
 Cycle Number: 1410

N_f : 2981 cycles

ΔE : 0.9%

R-24-45 538°C
 (300 sec. hold in compr.)
 Cycle Number: 1632

N_f : 3392 cycles

ΔE : 0.9%

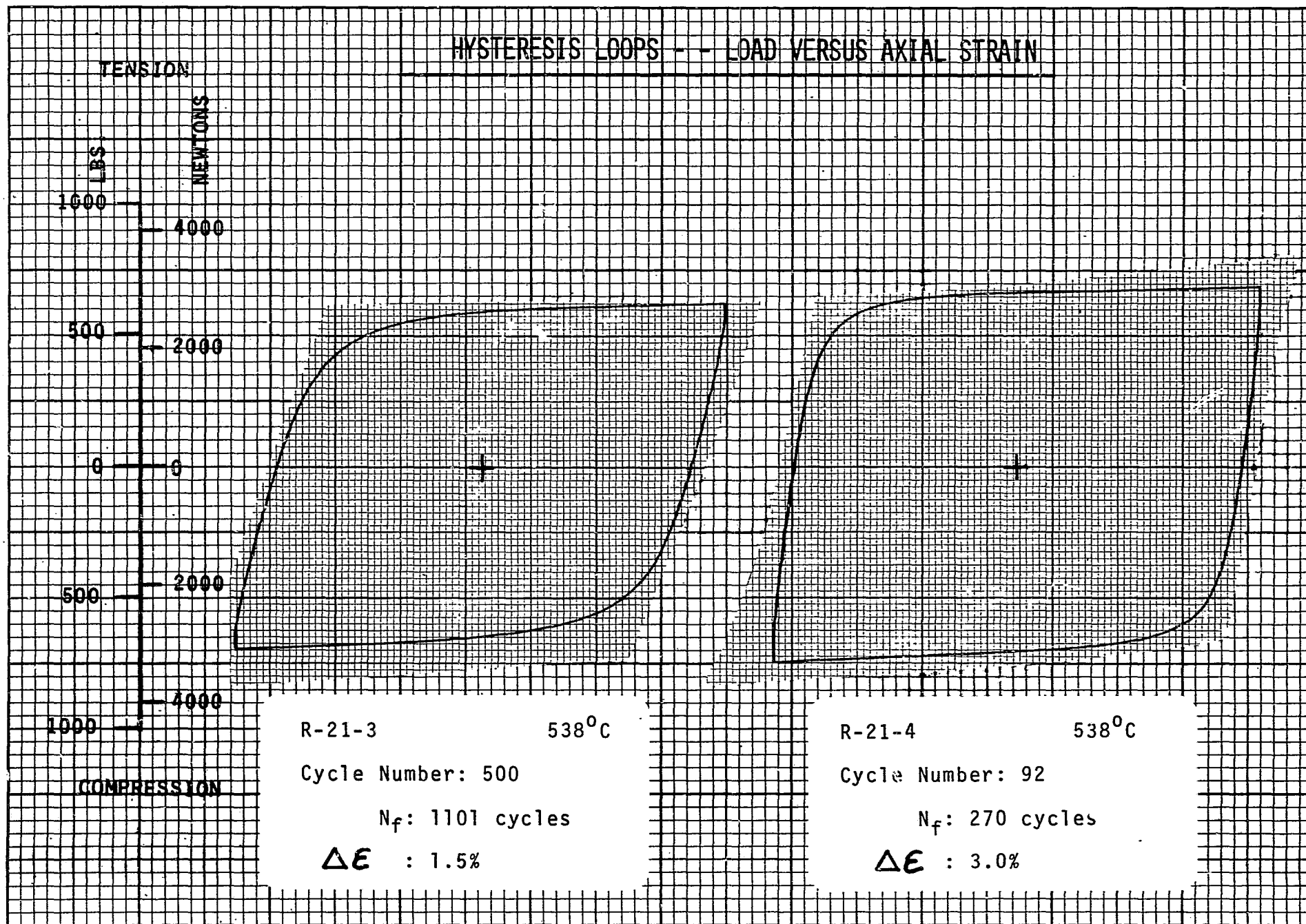
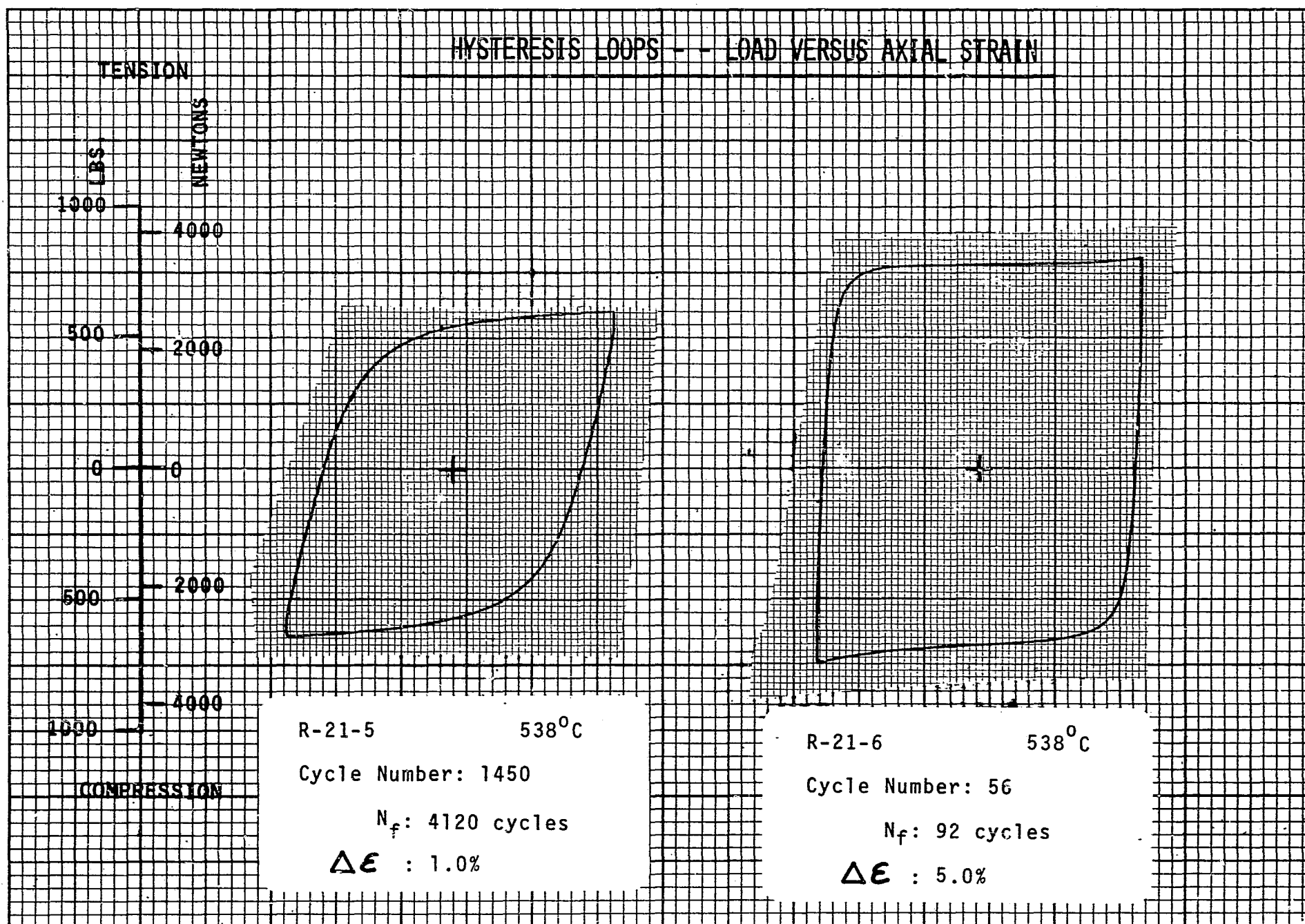


Figure 270

Figure 271



HYSTERESIS LOOPS - - LOAD VERSUS AXIAL STRAIN

TENSION

LBS. NEWTONS

1000 4000

500 2000

0 0

500 2000

1000 4000

R-22-4 538°C

Cycle Number: 300

N_f : 488 cycles

ΔE : 2.0%

COMPRESSION

R-22-5 538°C

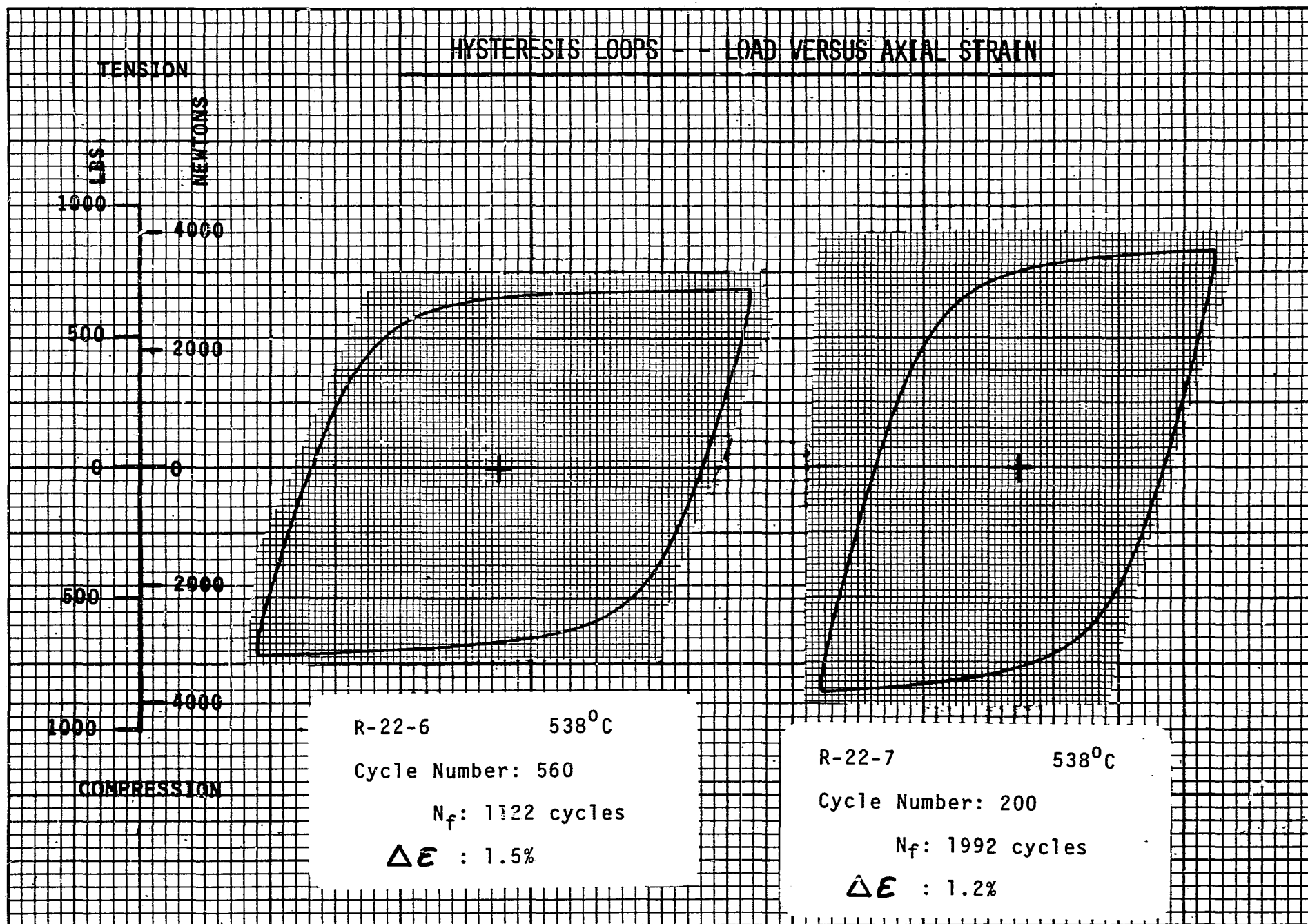
Cycle Number: 40

N_f : 106 cycles

ΔE : 4.0%

Figure 272

Figure 273



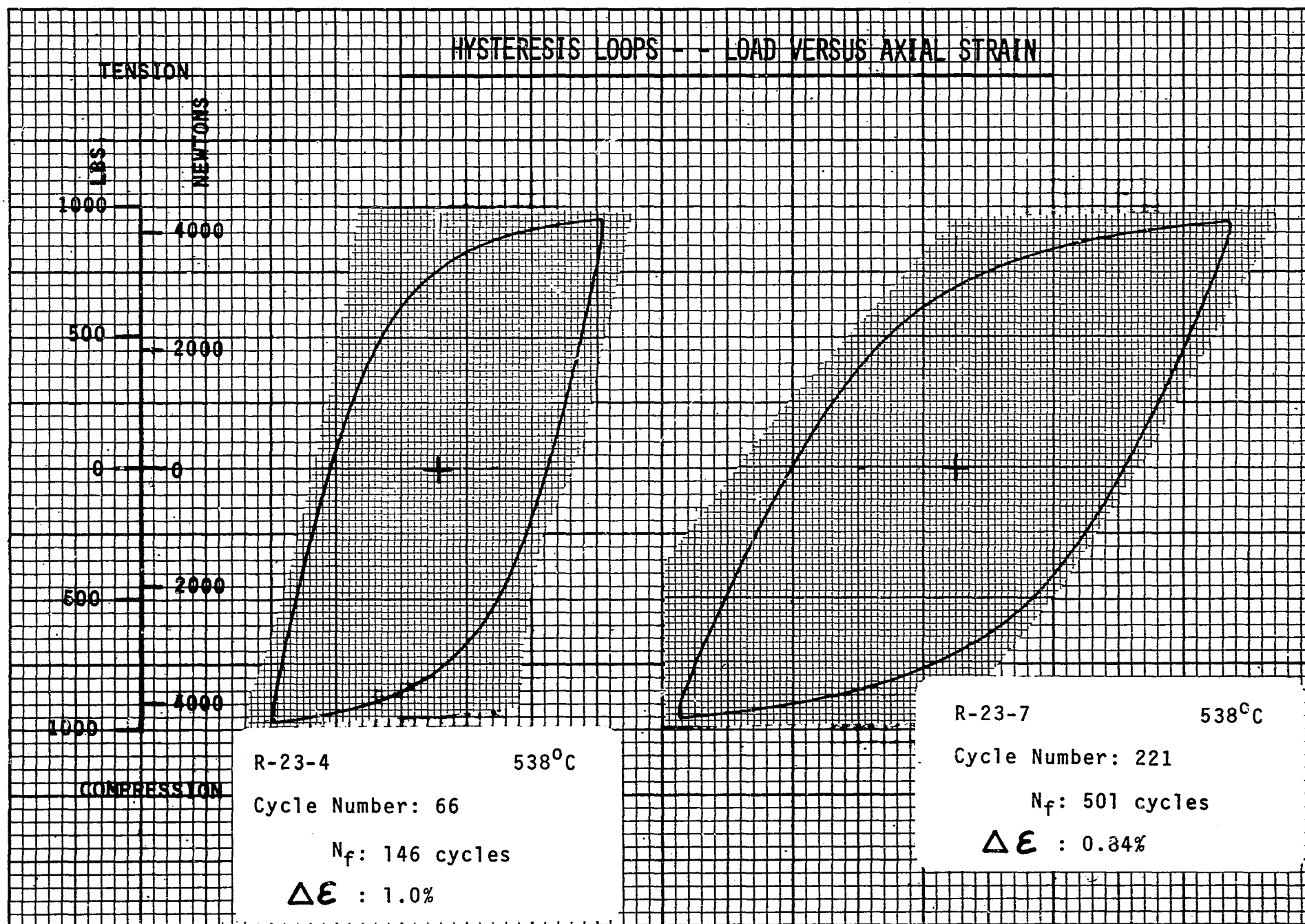
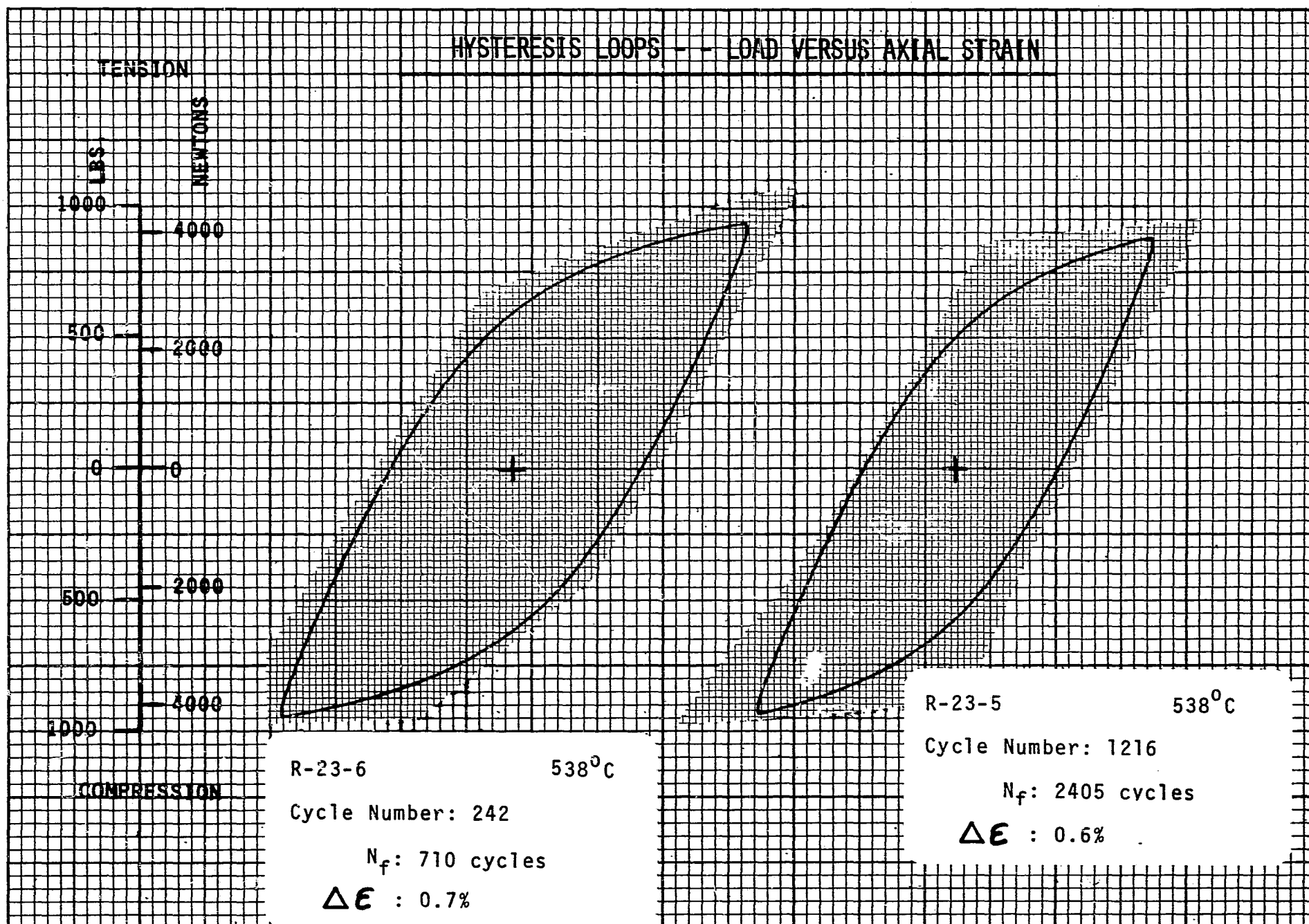


Figure 274

Figure 275



HYSTERESIS LOOPS -- LOAD VERSUS AXIAL STRAIN

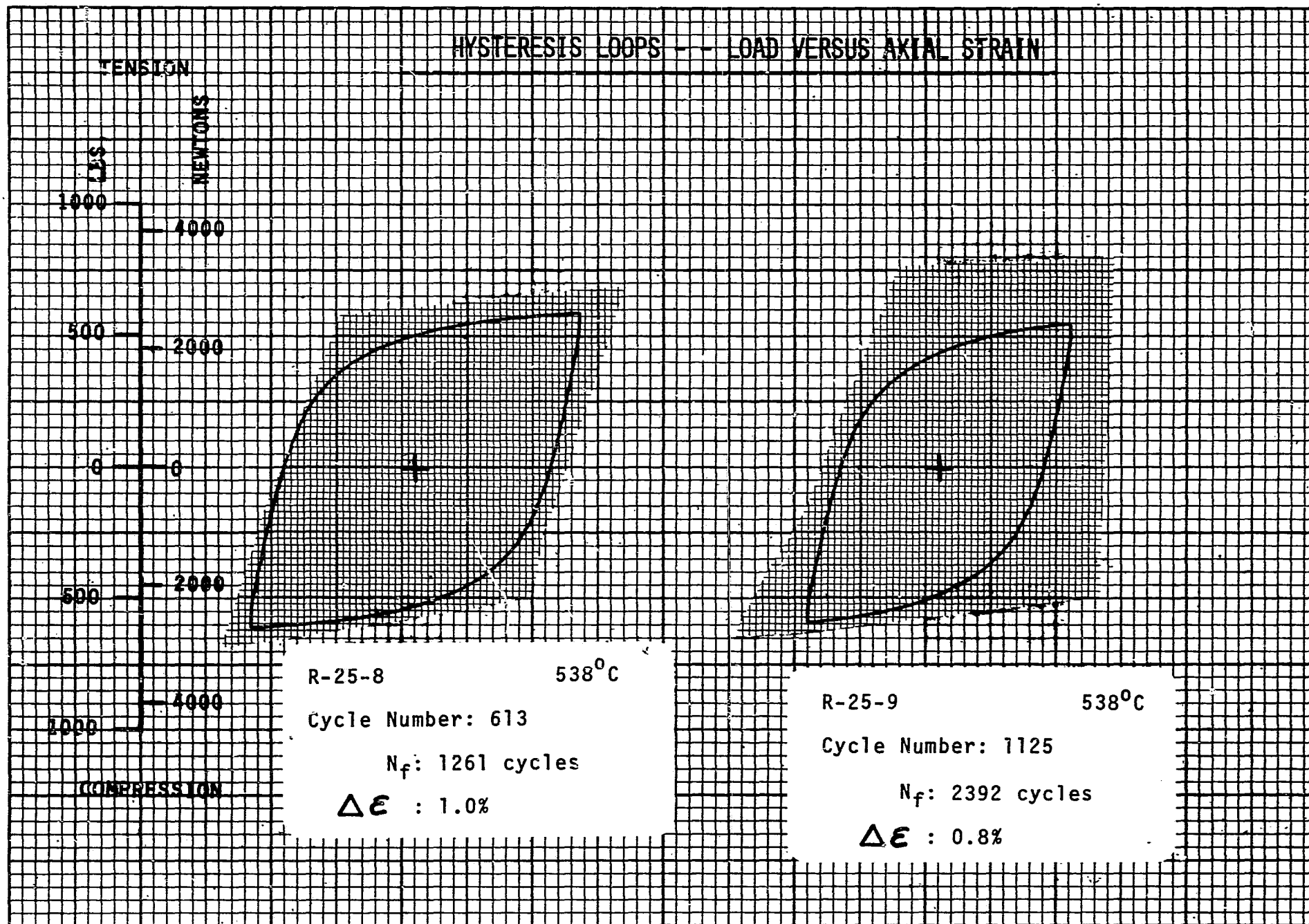
TENSION
LBS
NEWTONS
1000
500
0
500
1000
4000
2000
0
2000
4000
COMPRESSION

R-25-6 538°C
Cycle Number: 8
 N_f : 58 cycles
 $\Delta\epsilon$: 2.0%

R-25-7 538°C
Cycle Number: 57
 N_f : 109 cycles
 $\Delta\epsilon$: 2.0%

Figure 276

Figure 277



HYSTERESIS LOOPS -- LOAD VERSUS AXIAL STRAIN

TENSION

LBS. NEWTONS

1000 4000

500 2000

0 0

500 2000

1000 4000

R-26-2 538°C

Cycle Number: 45

N_f : 109 cycles

$\Delta \epsilon$: 2.0%

R-22-18 538°C

Cycle Number: 64

N_f : 120 cycles

$\Delta \epsilon$: 3.0%

Figure 278

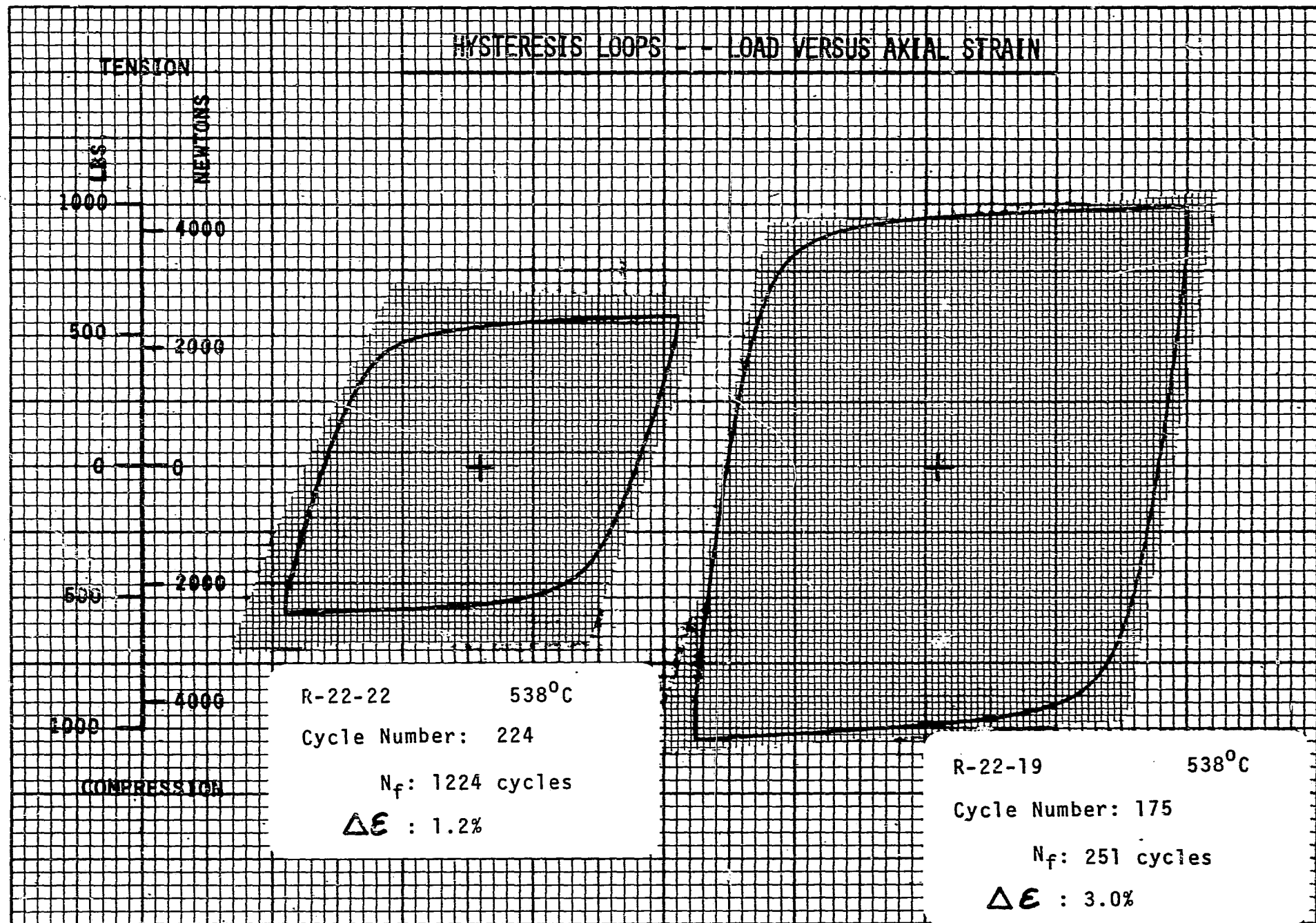


Figure 279

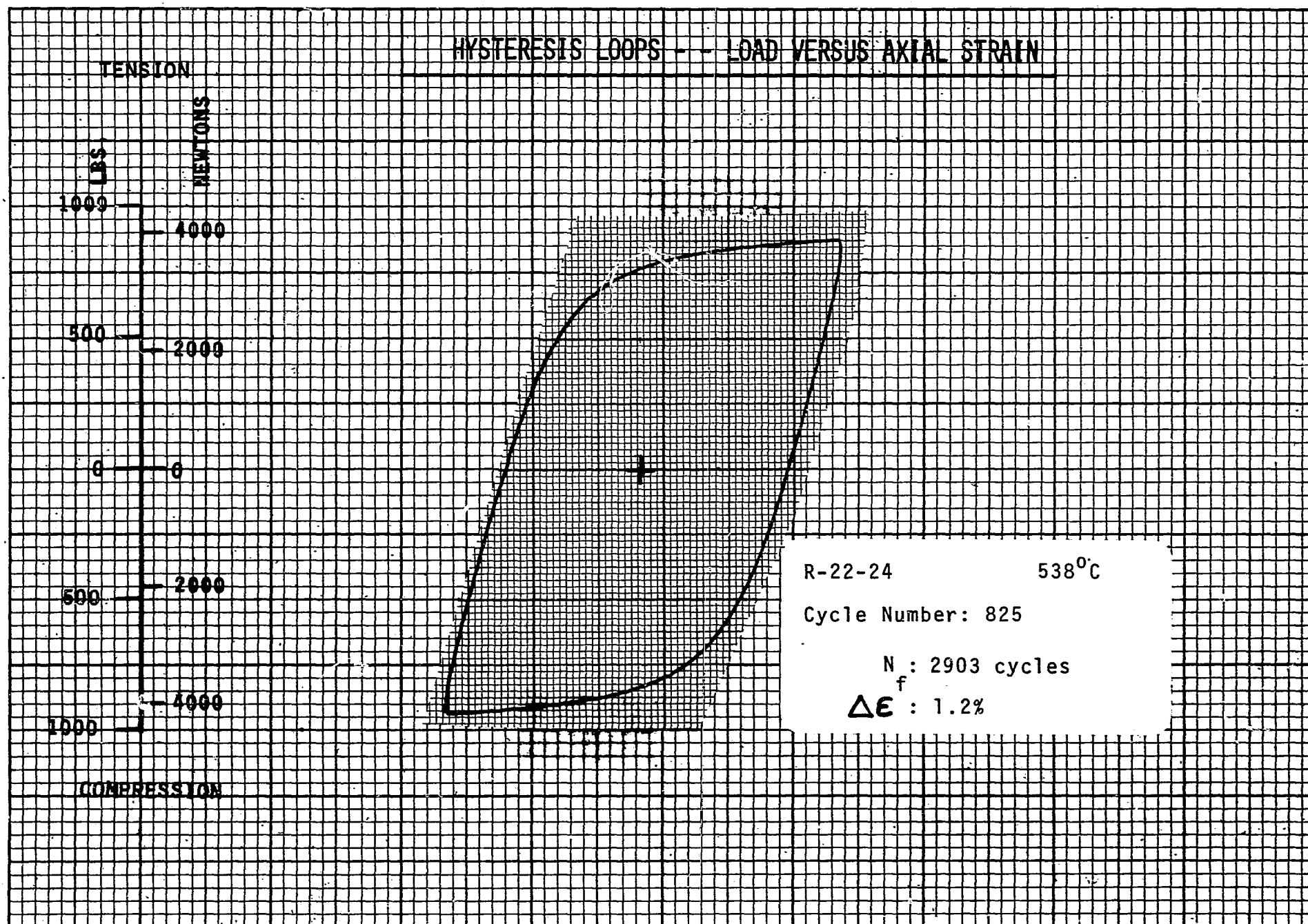
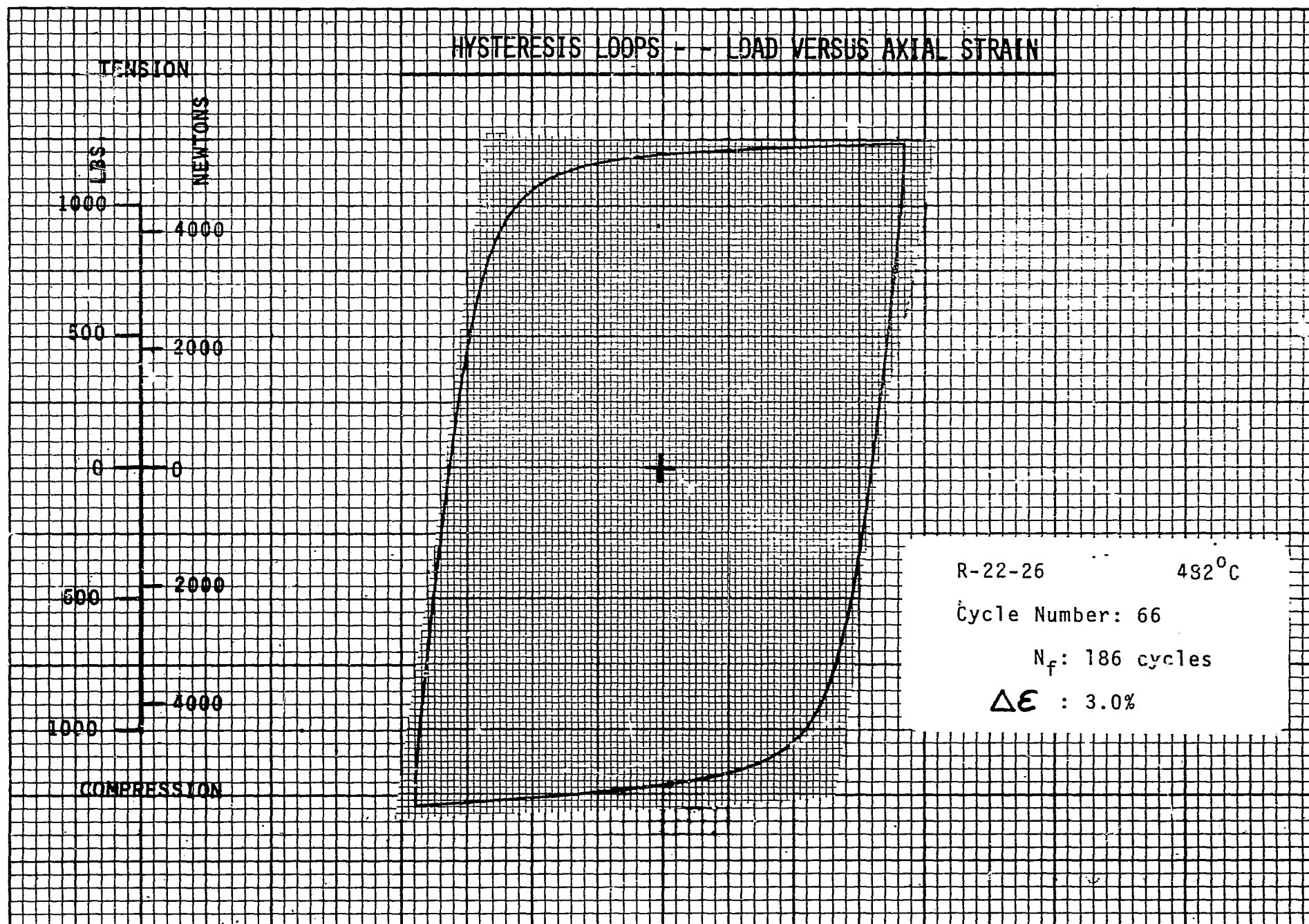


Figure 280



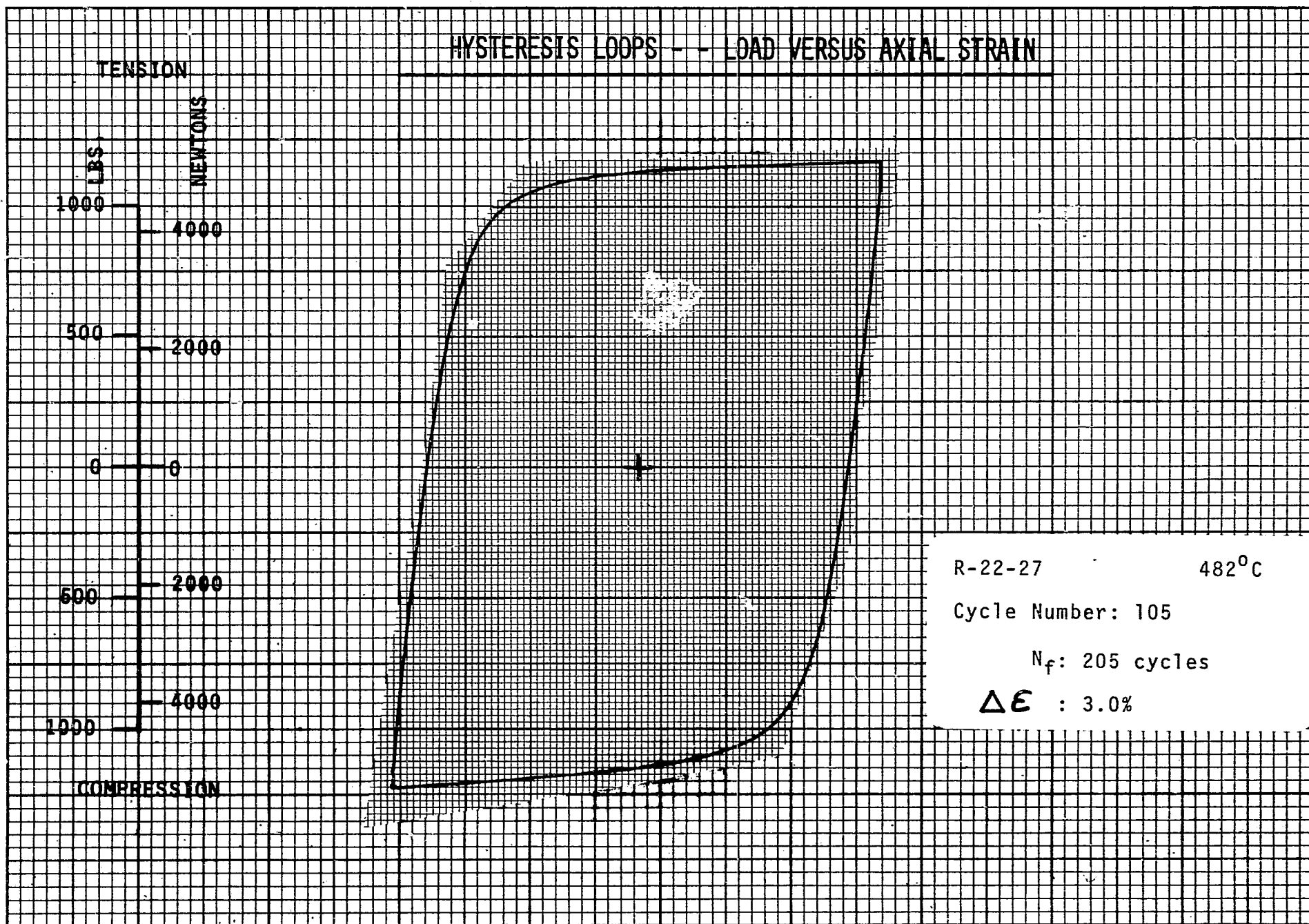
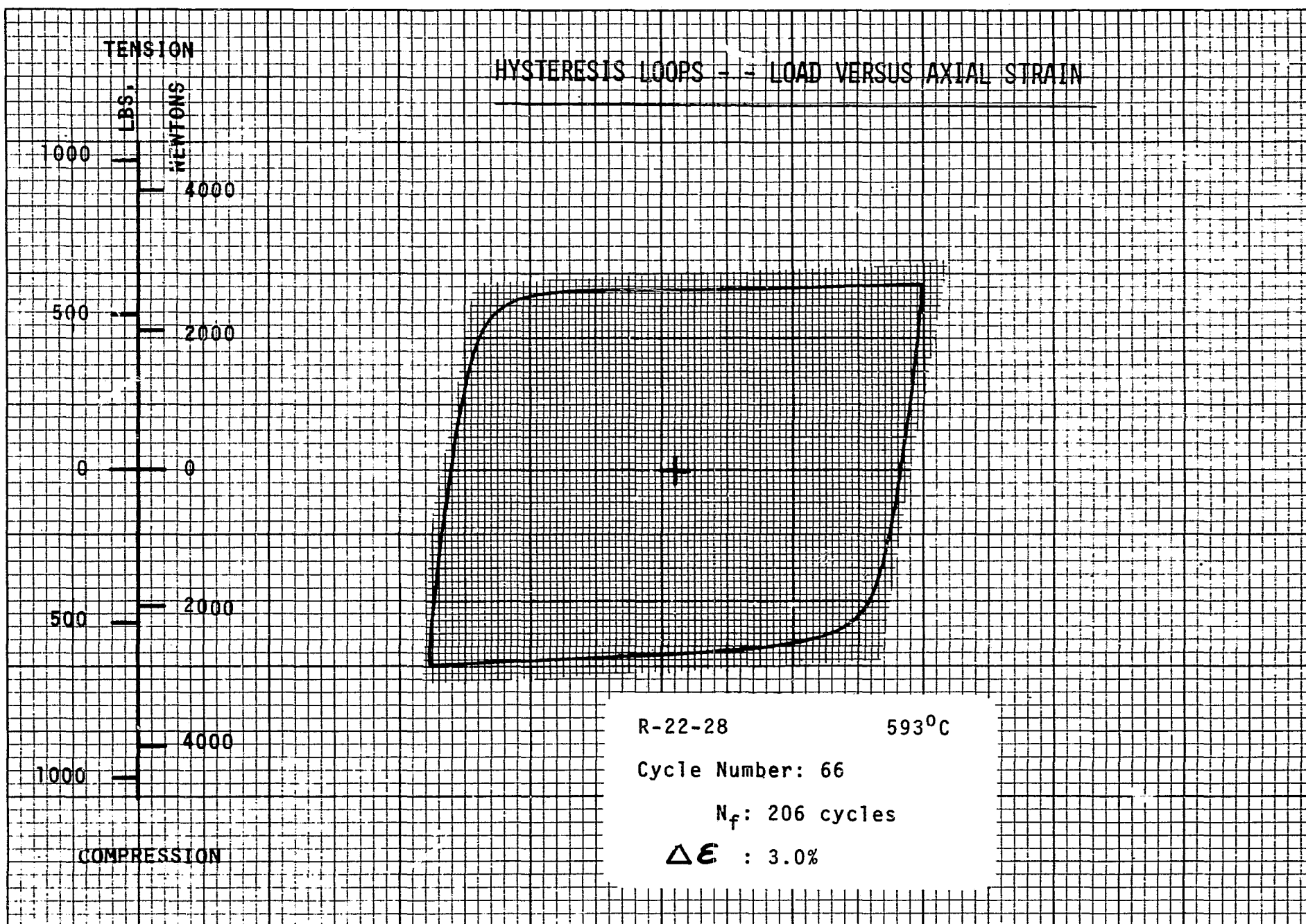


Figure 282



HYSTERESIS LOOPS - - LOAD VERSUS AXIAL STRAIN

TENSION

LB

NEWTONS

1000

500

0

500

1000

4000

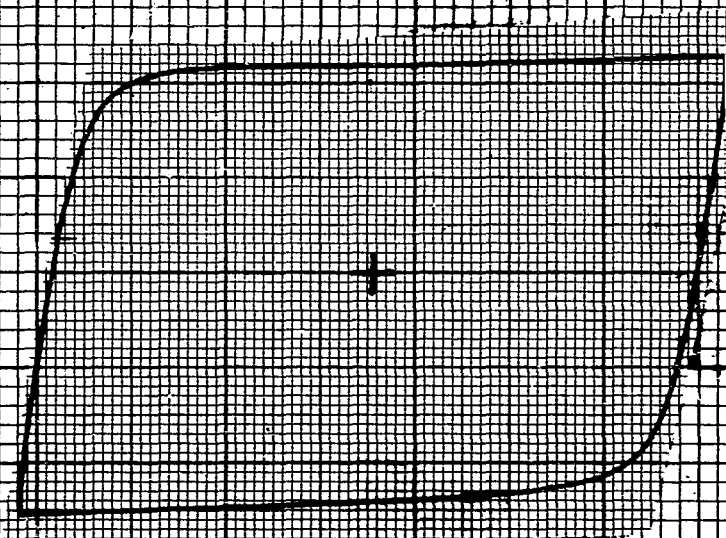
2000

0

2000

4000

COMPRESSION



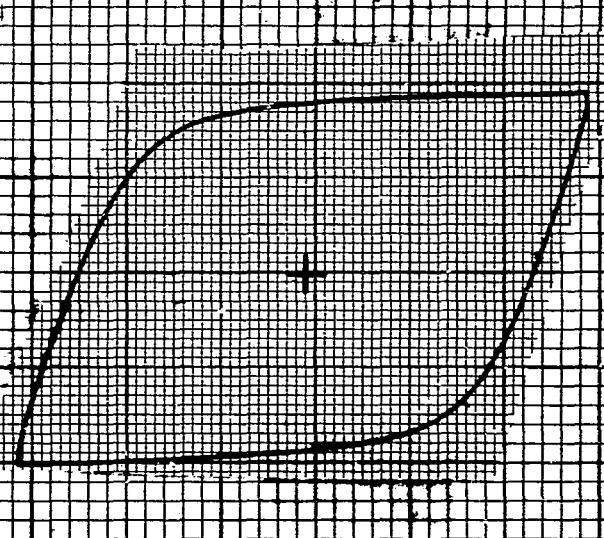
R-22-29

593°C

Cycle Number: 78

N_f : 192 cycles

ΔE : 3.0%



R-22-33

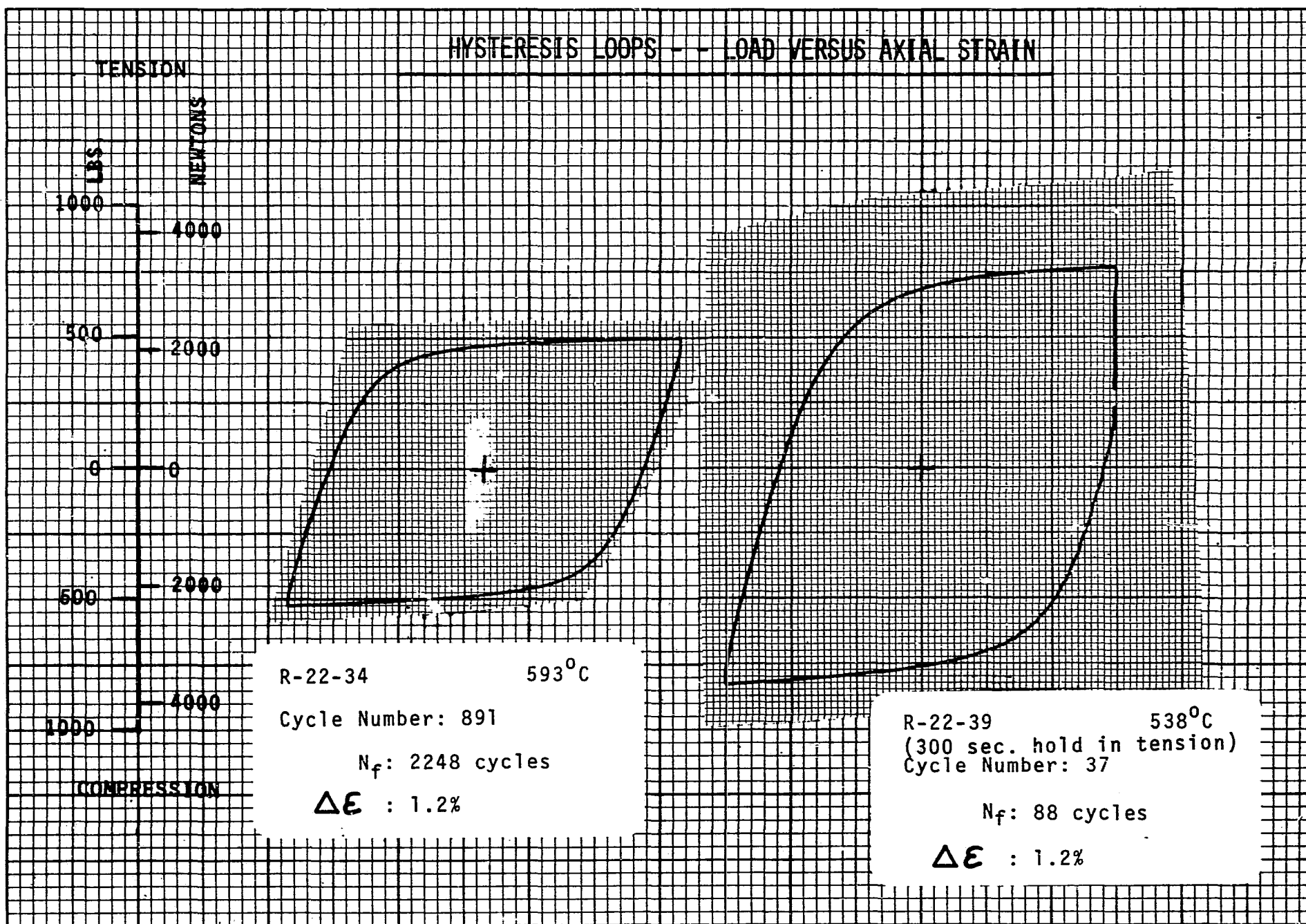
593°C

Cycle Number: 1450

N_f : 2893 cycles

ΔE : 1.2%

Figure 284



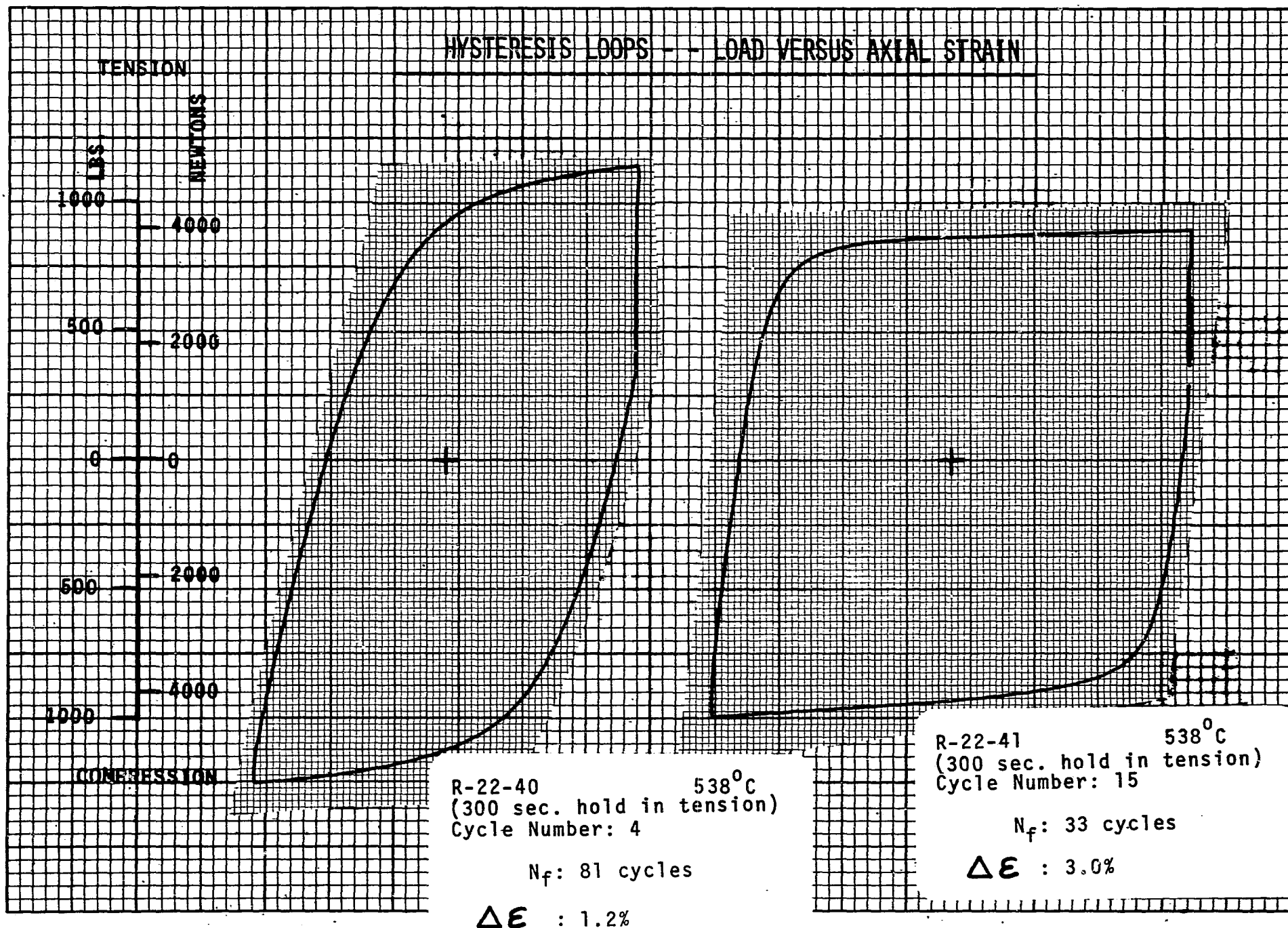


Figure 286

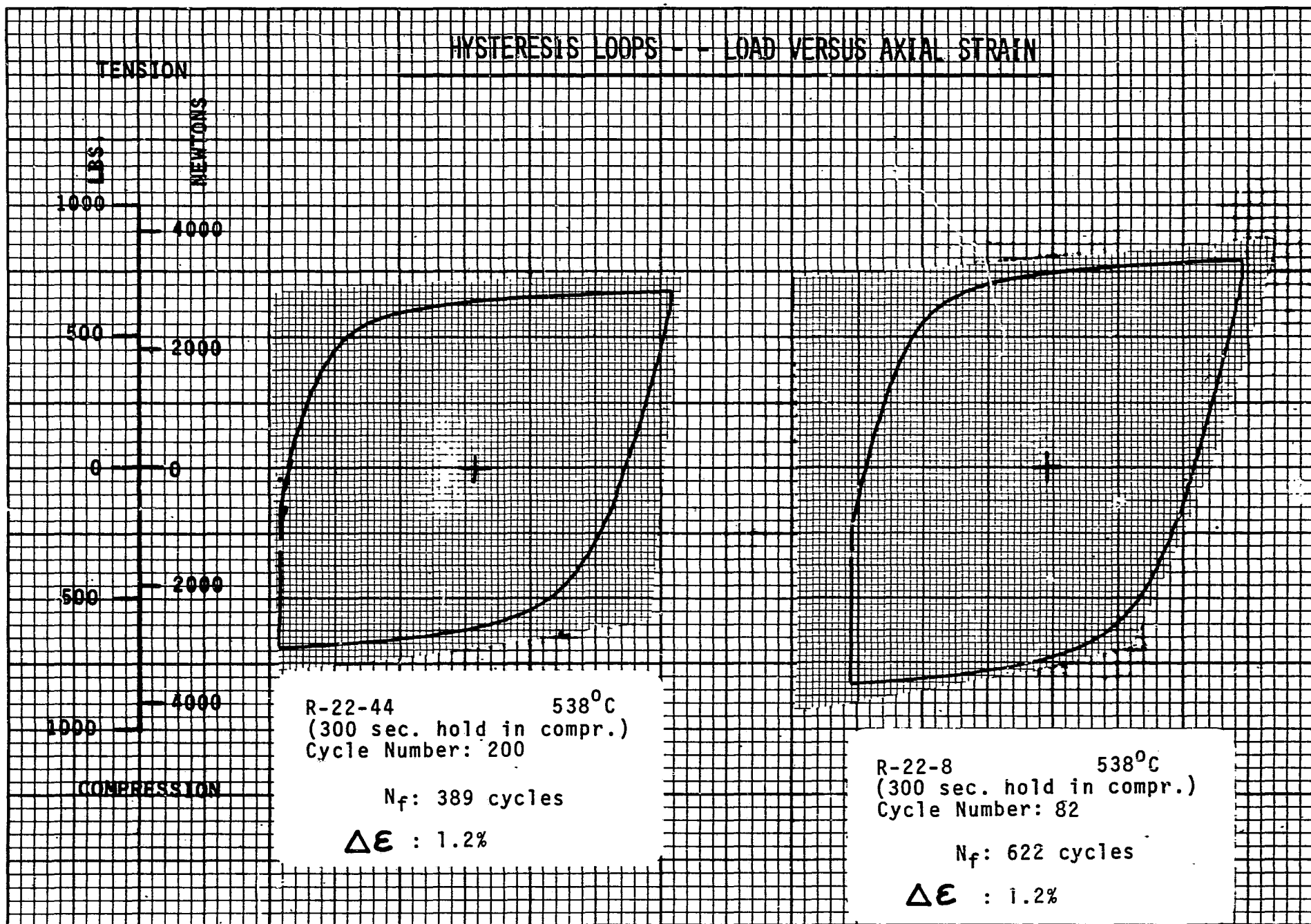


Figure 287

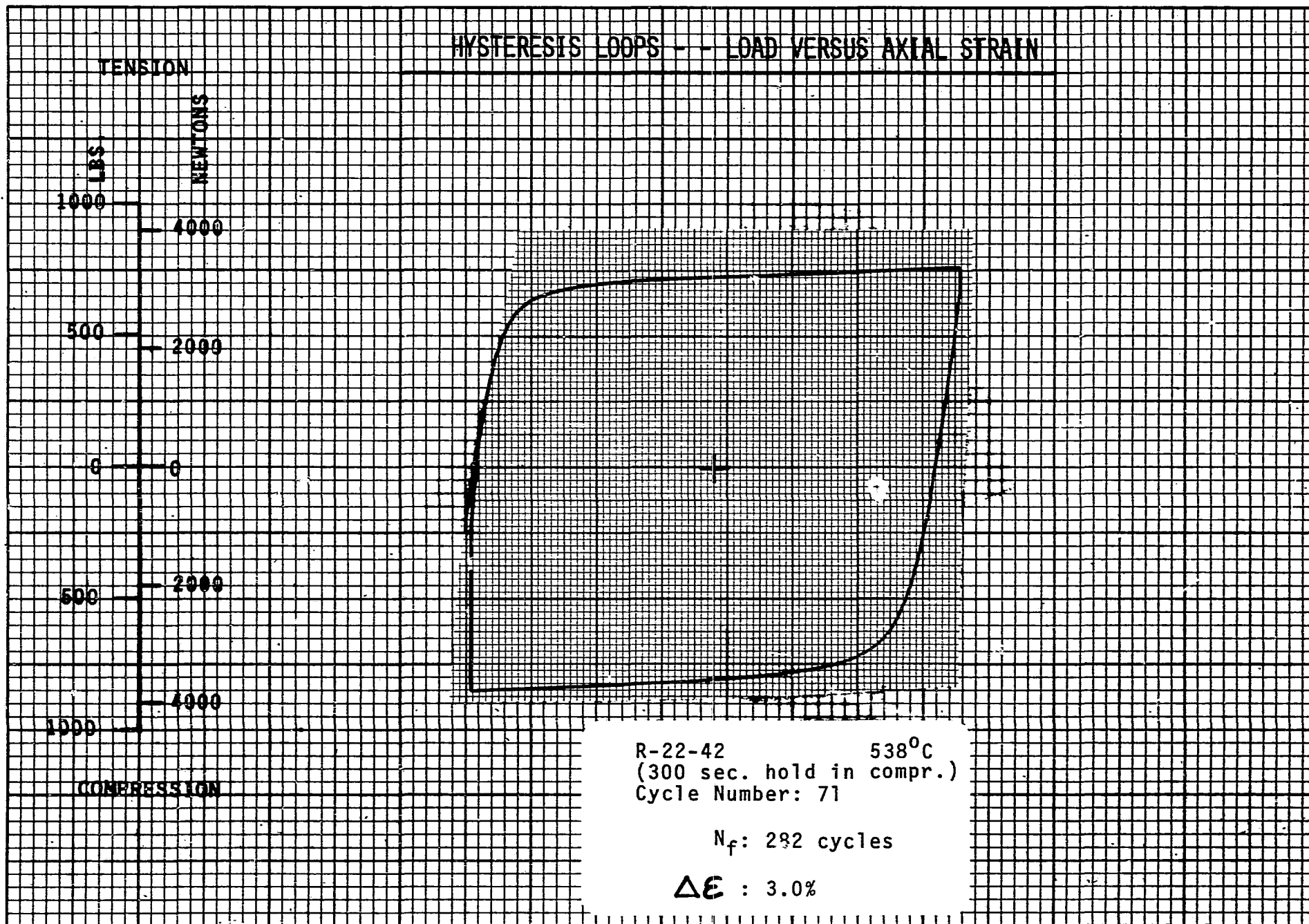


Figure 288

Table 2 - Values of N_5 and N_f for all Tests

(N_5 is the number of cycles to a five percent load range drop and N_f is the number of cycles to complete separation of the specimen)

Specimen Number	Temp., °C	Total Strain Range, %	Strain Rate, sec ⁻¹	N_5 , cycles	N_f , cycles
<u>Zirconium-copper, annealed</u>					
R-0-14	538	2.0	2×10^{-3}	680	1512
R-0-15	↓	1.5	↓	1200	4188
R-0-25	↓	3.5	↓	220	283
R-0-17	↓	3.0	↓	190	307
R-0-19	↓	1.7	↓	1000	2300
R-0-21	↓	2.5	↓	410	418
<u>Zirconium-copper, ½ Hard</u>					
R-1-3	538	2.5	2×10^{-3}	320	524
R-1-4	↓	1.6	↓	800	1088
R-1-9	↓	3.5	↓	170	562
R-1-10	↓	3.5	↓	180	447
R-1-11	↓	1.2	↓	4900	5590
R-1-12	↓	1.35	↓	2100	3660
<u>Zirconium-copper, ½ Hard</u>					
R-2-2	538	3.0	2×10^{-3}	1120	1615
R-2-4	↓	5.0	↓	280	366
R-2-5	↓	4.0	↓	480	552
R-2-10	↓	2.8	↓	780	1055
R-2-11	↓	2.0	↓	720	1239
R-2-6	↓	2.0	↓	1400	2051
R-2-12	↓	1.7	↓	1770	1770
R-2-13	↓	1.5	↓	2400	2453
<u>Tellurium-copper, ½ Hard</u>					
R-3-10	538	1.6	2×10^{-3}	385	390
R-3-13	↓	2.0	↓	116	117
R-3-1	↓	1.2	↓	390	462
R-3-2	↓	0.8	↓	970	1179
R-3-4	↓	1.0	↓	660	802
R-3-5	↓	0.5	↓	3900	3908
<u>Chromium-copper, SA and aged</u>					
R-4-1	538	2.0	2×10^{-3}	110	147
R-4-2	↓	1.6	↓	290	354
R-4-3	↓	1.4	↓	530	605

<u>Specimen Number</u>	<u>Temp., °C</u>	<u>Total Strain Range, %</u>	<u>Strain Rate, sec⁻¹</u>	<u>N₅, cycles</u>	<u>N_f, cycles</u>
<u>Chromium-copper, SA and aged</u>					
R-4-4	538	1.0	2×10^{-3}	1600	1823
R-4-5	↓	1.2	↓	970	1102
R-4-8		0.9		3500	3648
<u>OFHC Copper, Hard</u>					
R-5-2	538	1.6	2×10^{-3}	280	292
R-5-11	↓	2.0	↓	160	195
R-5-1		1.0		-	679
R-5-4		0.8		-	1295
R-5-3	↓	1.2	↓	450	453
R-5-5		0.6		2800	3606
<u>OFHC Copper, $\frac{1}{2}$ Hard</u>					
R-6-1	538	1.6	2×10^{-3}	82	85
R-6-3	↓	0.6	↓	630	691
R-6-4		0.7		300	418
R-6-5		2.0		40	56
R-6-8		0.5		900	1358
R-6-9	↓	1.0	↓	180	200
R-6-11		1.0		200	303
<u>OFHC Copper, Annealed</u>					
R-7-3	538	2.0	2×10^{-3}	75	126
R-7-4	↓	1.5	↓	230	269
R-7-1		1.2		400	437
R-7-2		1.0		660	710
R-7-7	↓	0.8	↓	1100	1313
R-7-9		0.7		1300	1613
<u>Silver, As-drawn</u>					
R-8-7	538	3.0	2×10^{-3}	-	344
R-8-5	↓	2.5	↓	590	603
R-8-9		1.2		1700	1902
R-8-3		1.0		2500	2620
R-8-2	↓	2.0	↓	850	928
R-8-10		1.5		1350	1381
<u>Zr-Cr-Mg Copper, SA, CW and Aged</u>					
R-9-11	538	2.0	2×10^{-3}	700	843
R-9-4	↓	3.0	↓	270	357
R-9-1		2.25		360	500
R-9-3	↓	2.5	↓	300	346

Table 2 continued

<u>Specimen Number</u>	<u>Temp., °C</u>	<u>Total Strain Range, %</u>	<u>Strain Rate, sec⁻¹</u>	<u>N₅, cycles</u>	<u>N_f, cycles</u>
<u>Zr-Cr-Mg Copper, SA, CW and aged</u>					
R-9-8	538	1.4	2×10^{-3}	1500	2000
R-9-2	↓	1.2	↓	1250	1317
R-9-9		0.9		4500	6670
<u>Electroformed Copper, 30-35 ksi</u>					
R-10-3	538	2.0	2×10^{-3}	30	148
R-10-4	↓	1.6	↓	25	38
R-10-1		0.8		450	1542
R-10-8		1.2		60	72
R-10-5	↓	1.0	↓	250	512
R-10-7		0.75		1000	1866
<u>Co-Be-Zr Copper, SA and aged</u>					
R-13-5	538	2.0	2×10^{-3}	70	90
R-13-13	↓	1.2	↓	350	644
R-13-7		1.5		130	212
R-13-10		1.0		450	680
R-13-6	↓	0.8	↓	700	1615
R-13-8		0.7		1800	3623
<u>Zirconium-Copper, ½ Hard</u>					
R-2-26	482	6.1	2×10^{-3}	100	204
R-2-27	↓	1.41	↓	2000	2985
R-2-28		6.1		98	176
R-2-29	↓	1.41	↓	1900	2135
R-2-30		5.0		145	265
R-2-31	593	1.4	2×10^{-3}	3100	3380
R-2-32	↓	5.0	↓	230	346
R-2-33	↓	1.5	↓	1750	2008
R-2-34		5.0		175	234
R-2-36	538	5.0	4×10^{-4}	170	234
R-2-37	↓	1.4	↓	-	1613
R-2-39		5.0		200	238
R-2-40		1.4		3000	3693
R-2-35		5.0	1×10^{-2}	-	245
R-2-38		1.4	↓	2500	5431
R-2-41		1.4	↓	2600	5215
R-2-42		5.0	↓	230	462
R-2-43		5.0	200 sec,T	170	211
R-2-44		5.0	200 sec,T	165	190
R-2-45	↓	5.0	200 sec,C	253	253
R-2-46		5.0	200 sec,C	210	262

Table 2 continued

Specimen Number	Temp., °C	Total Strain Range, %	Strain Rate, sec ⁻¹	N ₅ , cycles	N _f , cycles
<u>Zirconium-Copper, ½ Hard</u>					
R-2-47	538	1.4	56 sec, T	1100	1152
R-2-48	↓	1.4	56 sec, T	-	1062
R-2-49	↓	1.4	56 sec, C	-	1947
R-2-50	↓	1.4	56 sec, C	2800	3180
R-2-17	RT	3.0	5 x 10 ⁻³	-	696
R-2-51	↓	2.5	↓	1000	1102
R-2-52	↓	1.6	↓	-	2780
R-2-53	↓	4.0	↓	400	425
R-2-54	↓	1.4	↓	2900	3283
R-2-55	↓	3.0	↓	70	141
R-2-56	↓	1.4	↓	250	503
R-2-57	↓	0.9	3 x 10 ⁻³	750	1784
R-2-58	↓	2.0	5 x 10 ⁻³	130	279
R-2-59	↓	0.7	3 x 10 ⁻³	2000	5607
R-20-1	538	5.0	2 x 10 ⁻³	250	390
R-20-2	↓	1.4	↓	4000	5658
R-20-3	↓	2.0	↓	1200	1326
R-20-4	↓	3.0	↓	-	879
R-20-5	↓	1.6	↓	3400	3589
R-20-6	↓	1.8	↓	2600	2700
R-2-67	538	5.0	300 sec, T	-	218
R-2-68	↓	1.4	300 sec, T	-	1156
R-2-69	↓	4.62	300 sec, C	-	948
R-2-72	↓	1.4	300 sec, C	-	1224
R-2-73	↓	5.0	300 sec, T, C	-	234
R-2-61	538/260	3.5	8.75 x 10 ⁻⁵	240	284
R-2-62	538/260	3.5	8.75 x 10 ⁻⁵	-	188

Narloy Z, Cent. cast, hot-rolled,
solution annealed and aged

R-24-13	538	1.0	2 x 10 ⁻³	1150	1169
R-24-14	↓	2.0	↓	260	331
R-24-15	↓	1.2	↓	870	1126
R-24-16	↓	0.7	↓	3000	3601
R-24-17	↓	3.5	↓	92	99
R-24-18	↓	2.5	↓	200	253
R-24-19	↓	0.85	↓	1700	2469
R-24-22	↓	0.9	1 x 10 ⁻²	2800	3909
R-24-21	↓	0.9	↓	3230	3586
R-24-23	↓	2.6	↓	290	339
R-24-26	↓	2.6	↓	340	364
R-24-20	↓	0.9	4 x 10 ⁻⁴	860	1138
R-24-25	↓	0.9	4 x 10 ⁻⁴	880	1196

Table 2 continued

<u>Specimen Number</u>	<u>Temp., °C</u>	<u>Total Strain Range, %</u>	<u>Strain Rate, sec⁻¹</u>	<u>N₅, cycles</u>	<u>N_f, cycles</u>
<u>Narloy Z, Cent. cast, hot-rolled, solution annealed and aged</u>					
R-24-24	538	2.6	4×10^{-4}	120	154
R-24-27	538	2.6	4×10^{-4}	122	133
R-24-29	538	2.6	5.2×10^{-2}	474	474
R-24-30	538	2.6	5.2×10^{-2}	570	588
<u>Zirconium-Copper, $\frac{1}{2}$ Hard</u>					
R-2-74	538	2.6	5.2×10^{-2}	2500	3132
R-2-75	538	2.6	5.2×10^{-2}	-	3480
<u>Narloy Z, Cent. cast, hot-rolled, solution annealed and aged</u>					
R-24-33	593	0.9	2×10^{-3}	1180	1253
R-24-34	593	2.6	↓	160	191
R-24-31	482	0.9	↓	2800	2950
R-24-32	482	2.6	↓	180	243
R-24-39	538	2.6	290 sec,T	63	102
R-24-40	↓	2.6	300 sec,T	55	75
R-24-41	↓	2.6	300 sec,C	220	353
R-24-42	↓	2.6	300 sec,C	220	337
R-24-38	↓	0.9	300 sec,T	190	262
R-24-37	↓	0.9	300 sec,T	195	317
R-24-43	↓	0.9	300 sec,C	1500	2981
R-24-45	↓	0.9	300 sec,C	1500	3392
<u>NASA 1-1A Copper Alloy, Aged</u>					
R-21-3	538	1.5	2×10^{-3}	1090	1101
R-21-4	↓	3.0	↓	265	270
R-21-5	↓	1.0	↓	-	4120
R-21-6	↓	5.0	↓	85	92
<u>NASA 1-1B Copper Alloy, As-received</u>					
R-22-4	538	2.0	2×10^{-3}	-	488
R-22-5	↓	4.0	↓	90	106
R-22-6	↓	1.5	↓	-	1122
R-22-7	↓	1.2	↓	1950	1992
<u>Glidcop AL-10</u>					
R-23-4	538	1.0	2×10^{-3}	110	146
R-23-7	↓	0.84	↓	420	501
R-23-6	↓	0.7	↓	620	710
R-23-5	↓	0.6	↓	2000	2405

<u>Specimen Number</u>	<u>Temp., °C</u>	<u>Total Strain Range, %</u>	<u>Strain Rate, sec⁻¹</u>	<u>N₅, cycles</u>	<u>N_f, cycles</u>
<u>Sputtered Zirconium-Copper, Annealed</u>					
R-25-6	538	2.0	2×10^{-3}	50	58
R-25-7	↓	2.0	↓	80	109
R-25-8	↓	1.0	↓	900	1261
R-25-9	↓	0.8	↓	1700	2392
<u>Sputtered Zirconium-Copper, As-sputtered</u>					
R-26-2	538	2.0	2×10^{-3}	100	109
<u>NASA 1-1B Copper Alloy, As-received</u>					
R-22-18	538	3.0	4×10^{-4}	-	120
R-22-22	↓	1.2	4×10^{-4}	-	1224
R-22-19	↓	3.0	1×10^{-2}	190	251
R-22-24	↓	1.2	1×10^{-2}	2450	2903
R-22-26	482	3.0	2×10^{-3}	150	186
R-22-27	482	3.0	↓	155	205
R-22-28	593	3.0	↓	180	206
R-22-29	↓	3.0	↓	182	192
R-22-33	↓	1.2	↓	2880	2893
R-22-34	↓	1.2	↓	1900	2248
R-22-39	538	1.2	300 sec,T	77	88
R-22-40	↓	1.2	300 sec,T	67	81
R-22-41	↓	3.0	300 sec,T	29	33
R-22-44	↓	1.2	300 sec,C	389	389
R-22-8	↓	1.2	300 sec,C	-	622
R-22-42	↓	3.0	300 sec,C	240	282